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TRANSONIC WIND TUNNEL TESTS ON A SERIES OF TWO-DIMENSIONAL AERO--ETC(U)

JAN 79 B D FAIRLIE , N POLLOCK

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TRANSONIC WIND TUNNEL TESTS
ON A SERIES OF TWO-DIMENSIONAL AEROFOILS
IN A SOLID WALL TEST SECTION

by

B. D. FAIRLIE and N. POLLOCK

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10 B. D. FAIRLIE and N. POLLOCK

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SUMMARY

Transonic wind tunnel tests are reported on a series of two-dimensional models in a test section equipped with four solid walls. The tests were conducted on two geometrically similar models of each of two aerofoil sections—NACA-0012 and BGK-1—and covered a range of Mach numbers between 0.5 and 0.82. The main purpose of the tests was to provide a body of data suitable for testing the validity of linear interference theory without the complication of the unknown boundary conditions associated with non-solid wind tunnel walls.

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NOTATION

| | |
|----------------|---------------------------------------------------------------------|
| b | breadth of test-section |
| C_N | normal force coefficient = (normal force)/ qS |
| C_p | pressure coefficient = $(p - p_0)/q$ |
| C_m | pitching moment coefficient = (pitching moment about $c/4$)/ qSc |
| c | aerofoil chord |
| H | free stream total pressure |
| h | height of test section |
| K | torsional stiffness of model |
| M | free stream Mach number |
| p | local static pressure |
| p_0 | free stream static pressure |
| q | free stream kinetic pressure = $\gamma p_0 M^2/2$ |
| R | Reynolds number based on c |
| S | wing area = $b \times c$ |
| s | aerofoil arc length |
| t | aerofoil maximum thickness |
| x | chordwise ordinate: origin at leading edge |
| y | spanwise ordinate: origin at mid span |
| z | aerofoil thickness ordinate |
| α | angle of incidence |
| $\delta\alpha$ | aerofoil angle of twist due to aerodynamic loading |
| γ | ratio of specific heats: taken to be 1.4 |
| σ | angle of local tangent to aerofoil surface |

Subscripts

| | |
|------|-----------------------------------------|
| sc | denotes values at aerofoil shear centre |
|------|-----------------------------------------|

1. INTRODUCTION

For the past several years, A.R.L. has been conducting a theoretical and experimental investigation into transonic aerofoils. One of the major aims of this investigation has been to examine the problems of obtaining reliable wind tunnel data from aerofoil tests at high subsonic speeds. All such tests are subject to errors arising from wind tunnel wall interference, the magnitude of such errors increasing as the ratio of model chord to tunnel height is increased. For ease of model manufacture and to achieve the maximum possible Reynolds number, it is desirable to use the largest possible model chord without increasing the effects of wall interference beyond the point where current correction methods can be successfully applied.

In designing the experimental program for the investigation, one of the aims was therefore to produce a body of data which could be used to test the accuracy and range of applicability of current methods for accounting for wind-tunnel wall interference. To this end, two basic aerofoil shapes were selected for testing. The first, the NACA-0012 section, was chosen to provide a "standard" symmetrical section which, since it had been tested in a number of wind tunnels throughout the world, would provide reliable comparative data under both subcritical and supercritical conditions. The second, an aerofoil designed by the method of Bauer, Garabedian and Korn,¹ is an example of a modern "supercritical" section, with a reasonable extent of shock free supersonic flow on the upper surface and some degree of rear loading. This aerofoil (hereafter referred to as BGK-1) also provided a successful example of modern aerofoil design which presents new problems in wind tunnel testing due to the existence of shock free flow and high rear loading. At the same time, these two aerofoil sections were selected for a co-operative investigation into the problems of aerofoil testing sponsored by the Commonwealth Advisory Aeronautical Research Council*, and the A.R.L. experimental program was extended to include that suggested in the C.A.A.R.C. proposal².

When applying corrections to wind tunnel test results for the effects of wind-tunnel wall interference, one of the most popular approaches is to calculate the corrections theoretically from linearized potential flow theory³. A major problem to be faced when applying this theory to a practical wind tunnel situation is that of correctly representing the boundary conditions at the tunnel walls. In most transonic wind tunnels, porous or slotted (as is the case of the A.R.L. tunnel) walls are used. Such walls are usually employed primarily to allow the generation of low supersonic Mach numbers using diffuser suction, or to minimize shock reflections at high subsonic and low supersonic speeds. In such cases, the wall boundary conditions are complex, and it is usual to replace the mixed (part open, part closed) boundary condition with an equivalent homogeneous boundary condition which produces the same solution at the centre of the test section. However, the correct representation, especially in the case of slotted walls where the flow in the slots may be affected by viscosity, remains open to question⁴.

If data were available for a series of aerofoils tested in solid walls, in which case the boundary conditions are straightforward and amenable to mathematical analysis, it would be possible to examine the validity of the linearized approach of linear interference theory independently from the difficulties of the representation of non-solid wall boundary conditions. It was therefore decided to supplement the tests on the series of aerofoil sections already conducted in a slotted wall test section^{5,6} by a similar series of tests in a test section equipped with solid walls. These tests were conducted during May 1976.

This note contains results from tests on two models of differing chord lengths of the BGK-1 section and a similar pair of models of the NACA-0012 conducted in a test section with all solid walls. The tests cover a range of Mach numbers from 0.5 to 0.82 and of angle of incidence from -2.5° to 5° for the BGK-1 section and from 0° to 4° for the NACA-0012. The test Mach

* Australia, Canada, India, New Zealand and the United Kingdom are active members of C.A.A.R.C.

number range was deliberately extended closer than normal to the onset of choking (see Section 3) to provide data on the limits of applicability of linear interference theory.

2. TEST DETAILS

2.1 Section Definition

Aerofoil section BGK-1 was one of the earliest designed by Garabedian and his colleagues using their numerical hodograph method¹. The section was designed to produce shock-free flow on the upper surface at $M = 0.75$ and $\alpha = 0^\circ$, giving a C_L of 0.63 (inviscid) (Fig. 1). The ordinates of the aerofoil are presented in Table 1: these were derived from those published in Ref. 7, with the exception of those close to the trailing edge, where the presence of a cusp required some modification to obtain a practical shape for manufacture. These modifications involved thickening the section in the region $0.95 \leq (x/c) \leq 1.00$ and are noted in Table 1.

The NACA-0012 is a symmetrical section, 12% thick, with ordinates given by:

$$\frac{z}{c} = 0.6 \left\{ 0.29690 \sqrt{\frac{x}{c}} - 0.12600 \frac{x}{c} - 0.35160 \left(\frac{x}{c}\right)^2 + 0.28430 \left(\frac{x}{c}\right)^3 - 0.10150 \left(\frac{x}{c}\right)^4 \right\}$$

with a leading edge radius of $0.0158 c$. The basic section has a blunt base of thickness $0.00252 c$. However, to avoid confusing comparisons between models arising from base pressure variations with Reynolds number, the upper and lower surfaces were linearly extended to a sharp (0.1 mm thick) trailing edge. Due to this extension the actual physical chord length of the models were slightly greater than the nominal chord. The calculated force coefficients are however based on the nominal chord (c).

2.2 Model Details and Accuracy

The tests in the slotted wall test section^{5,6} were conducted on two models of the BGK-1 section (with chords of 203.2 mm and 101.6 mm) and three models of the NACA-0012 section (203.2 mm, 101.6 mm and 50.8 mm chords). The present tests covered the same group of aerofoils with the exception of the smallest (50.8 mm chord) model of the NACA-0012 section which was excluded since previous tests had shown results from this aerofoil to be inaccurate. The two BGK-1 models were manufactured in stainless steel from a number of pieces, rigidly joined prior to final machining, enabling surface pressure hole connections to be made without spoiling the windswept surfaces. The NACA-0012 models were cast in an epoxy resin using a development of the wax mould technique described in Reference 8.

Measured profile errors for the completed models are plotted in Figures 2 to 5. For the BGK-1 models, the accuracy in general falls within the tolerances suggested in Ref. 2 (local waviness less than $0.0002 c$ and overall dimensional errors less than $0.0005 c$). The 203.2 mm model of NACA-0012 also falls within these tolerances. However the absolute accuracy achieved for all the NACA-0012 models was similar, leading to a deterioration of non-dimensional accuracy as the model chord became smaller. Although the 101.6 mm chord NACA-0012 does not fall within the tolerances of Ref. 2 the accuracy was considered to be sufficient for the present tests.

All models were extensively pressure tapped to enable surface pressure distributions to be obtained. Pressure hole diameters for all models were 0.4 mm, and chordwise locations of the pressure holes are given in Table 2. For the 203.2 mm chord BGK-6 model, the pressure holes were distributed over a spanwise region $0.19 c$ wide and centred at midspan, with 37 pressure holes on the upper and 17 on the lower surface. The 101.6 mm chord BGK-1 model had 34 pressure holes on the upper surface and 14 on the lower surface, distributed over $0.38 c$. The 203.2 mm chord NACA-0012 model had 38 pressure holes in one surface only, distributed over a spanwise region of $0.32 c$, and the 101.6 mm chord NACA-0012 model had 18 holes in the upper surface and 14 in the lower surface distributed over $0.29 c$.

2.3 Wind Tunnel

All tests were carried out in the A.R.L. variable pressure transonic wind tunnel. The test section fitted for these tests had four solid walls, with dimensions at the model location of

$b = 533$ mm, $h = 813$ mm. Test section total pressure was derived from measurements of the static pressure in the entry to the contraction. Test section static pressure was measured at a side-wall pressure tapping located on the tunnel central plane 775 mm upstream of the model centreline. Preliminary empty tunnel tests had shown that the Mach number derived from this static pressure remained within ± 0.001 of the Mach number at the model centreline throughout the Mach number range of interest.

All models were supported by means of integral end tongues clamped in sidewall-mounted, steel discs replacing the usual glass windows. The models completely spanned the width of the test section, giving aspect ratios of 2.62 and 5.25 for the 203.2 mm and 101.6 mm chord models respectively. These aspect ratios should be large enough to ensure sensibly two-dimensional flow conditions, at least over the central measuring sections. Small gaps, approximately 0.6 mm wide existed between the ends of the models and the steel "windows". For all tests these gaps were sealed with a silicone rubber compound.

2.4 Pressure Measuring Equipment

Model surface pressure distributions were measured using two 48 port Scanivalves fitted with ± 33.5 kPa (± 5 p.s.i.) Statham differential pressure transducers referenced to p_0 . Stagger scanning was used to ensure the maximum settling time between readings. Scanning speed was normally 14 ports per second, with free stream conditions (p_0 , H and T) being recorded after each group of 12 ports. This allowed the free stream conditions to be monitored throughout the four-second period of measurement; measurements were repeated if the free stream conditions showed significant variation. Values of p_0 and H used to calculate p/H and C_p were the average of the p_0 and H readings bounding a particular pressure measurement.

All pressure measurements were processed by the tunnel PDP8/I computer data processing system, the data being reduced to pressure coefficient and pressure ratio forms for on-line printing and display. The data were also recorded on magnetic tape for further processing by the central site computer.

2.5 Transition Fixing

The majority of the slotted wall tests^{5,6} were conducted with transition fixed on both surfaces. For consistency it was decided that identical transition fixing should again be used for the majority of tests in the solid wall section. Transition was therefore fixed on both upper and lower surfaces of all models by spanwise roughness bands consisting of carborundum particles attached to the model by a thin (0.03 mm) layer of lacquer. All bands were located at $x/c = 0.05$ and were $0.015c$ wide with a particle coverage of 10–20%. Particle size was 0.15 mm for the 203.2 mm chord models and 0.08 mm for the 101.6 mm chord models. Surface oil flow observations indicated that at the test Reynolds number for each model, the roughness bands effectively produced boundary layer transition.

2.6 Test Program

The test program for the present tests was a subset of that suggested in the C.A.A.R.C. proposal² and utilized for the tests in slotted walls.^{5,6} The same Mach number range was covered, but the incidence range was roughly halved, always ensuring sufficient points were included to adequately define a lift curve slope. The complete program for the tests in solid walls is presented in Table 3.

The Reynolds numbers employed were the same as those for the slotted wall tests, being chosen such that the maximum available Reynolds number was utilized at the highest Mach number ($M = 0.82$) and the Reynolds number maintained constant throughout the Mach number range by adjusting tunnel starting pressure. Due to small tunnel temperature variations during a run, the Reynolds number would vary slightly, the values being in the bands $1.63 \pm 0.03 \times 10^6$ and $0.80 \pm 0.02 \times 10^6$ for the 203.2 mm and 101.6 mm chord BGK-1 models and $1.54 \pm 0.13 \times 10^6$ and $0.81 \pm 0.02 \times 10^6$ for the respective NACA-0012 models. Since tests in slotted walls^{5,6} on the larger models at the Reynolds numbers of the smaller had shown very little difference, such tests were not repeated in the present tests.

Since the slotted wall tests¹⁰ had shown that with the addition of transition bands it was no longer possible to obtain a "design" pressure distribution on the BGK-1 sections, a brief investigation was undertaken on these models without transition fixing to investigate whether a "design" distribution could be obtained in solid walls.

2.7 Data Reduction and Presentation

2.7.1 Surface Pressure Measurements

The 203·2 mm chord model of the NACA-0012 section was pressure tapped on one surface only. To obtain "upper" and "lower" surface pressures, this model was tested twice, at equal positive and negative angles of incidence.

All surface pressure data were reduced to pressure ratio (p/H) and pressure coefficient (C_p) form are tabulated in Appendix A.* Appendix B* contains plots of p/H versus x/c for all test conditions. The data were plotted in this form rather than C_p versus x/c to allow direct comparisons with other tunnel tests or theory to be made: adjustments to the values of Mach number or angle of incidence to take account of interference or other effects do not affect the p/H values.

Surface pressure data were integrated to obtain force coefficients as follows:

$$\text{normal force coefficient: } C_N = \frac{1}{c} \oint C_p \cos \sigma \, ds$$

$$\text{pitching moment coefficient: } C_m = \frac{1}{c^2} \oint C_p (x - 0.25) \cos \sigma \, ds$$

where s is arc length, measured clockwise from the leading edge, and σ is the angle between the local tangent to the surface and the chord line. The full expression for pitching moment coefficient contains a term:

$$\frac{1}{c^2} \oint C_p z \sin \sigma \, ds.$$

However due to the absence of leading edge pressure holes on the NACA-0012 models, this integral was poorly defined, and since rough check calculations indicated the term to be small, it was not included in the present calculations of C_m . For consistency, the BGK-1 section results were treated similarly. For the same reasons, integrated chordwise force coefficients were not calculated.

The surface pressure integrations were carried out numerically using a method derived from that of Woodward.⁹ Due to the scarcity of pressure tappings near the trailing edge on all models, and near the leading edge on the NACA-0012 models, some extra information was required to properly define these integrals. After investigation of a number of approaches, the following method was adopted. For all models a fictitious trailing edge pressure was generated by linear extrapolation of the upper surface pressure distribution. For the NACA-0012 models, a leading edge pressure was found by parabolic extrapolation of the upper and lower surface pressure distributions with the extrapolated curves being tangent to the C_p axis. It was found that this procedure gave values of C_N and C_m in good agreement with those obtained from curves manually fitted to the experimental data points.

2.7.2 Correction to Angles of Incidence

The model angle of incidence was measured with a transducer attached to the starboard tunnel "window". Since the port "window" was mechanically slaved to the starboard one, it was assumed that the model was not twisted by differential window rotations. The zero incidence position for each model was determined from tests with the model upright and inverted at one Mach number ($M = 0.6$).

Conventional metal aerofoil models tested in the A.R.L. transonic tunnel normally have sufficient torsional stiffness to ensure that the central pressure tapped section of the model can be regarded as having the same angle of incidence as that measured at the ends. This was the case for the two models of the BGK-1 section. However, composite plastic models such as those of

* Appendix A and Appendix B have been reproduced in microfiche form.

the NACA-0012 section used for these tests show significant spanwise twist due to aerodynamic loading. The angles of incidence measured for these models were therefore corrected for twist under aerodynamic load as follows. By appropriately loading the model with weights, the torsional stiffness per unit length (K) and the location of the shear centre $[(x/c)_{sc}]$, i.e. the chordwise position at which an applied normal force produces no rotation of the section, were determined. The torsional characteristics of the two NACA-0012 models included in the solid wall tests are summarized in the following table:

| Model chord (mm) | K (N rad ⁻¹) | $(x/c)_{sc}$ |
|---------------------|-------------------------------|--------------|
| 203.2 | 0.956×10^7 | 0.30 |
| 101.6 | 0.291×10^6 | 0.35 |

The angle of twist of the model with respect to the midspan section is given by:*

$$\delta\alpha|_{y=y'} = \frac{qc^2}{K} \int_0^{y'} \int_0^{y'} \left\{ C_N \left[\left(\frac{x}{c} \right)_{sc} - 0.25 \right] + C_m \right\} dy dy'$$

The angle of twist of the midspan section with respect to the sidewalls, $\delta\alpha|_{y=b/2}$, is calculated from the above equation by successive approximation. For the first iteration C_N and C_m are assumed constant at their midspan values (obtained from integration of the pressure distribution). The expression for $\delta\alpha$ is evaluated numerically for 20 stations equally spaced over the semispan. For all successive iterations C_N and C_m are assumed to vary linearly with angle of incidence so that:

$$C_N|_{y=y'} = C_N \text{ midspan} \times \frac{\alpha_{\text{sidewall}} + \alpha|_{y=b/2} - \alpha|_{y=y'}}{\alpha_{\text{sidewall}} + \alpha|_{y=b/2}}$$

with a similar expression for C_m . The calculation was continued until the angle of twist converged to within $\pm 0.01^\circ$ of its final value. For the present results, a maximum of three iterations was required.

Incidence angles corrected for model twist are used for all plotted results and for all interference correction calculations.

In Table 3 the measured incidence angles for the NACA-0012 are listed as "Nominal α " and the incidence angles corrected for model twist are listed as "Corrected α ".

3. RESULTS AND DISCUSSION

Tabulations and plots of all surface pressure data are contained in Appendices A and B. None of the pressure distributions show conclusive evidence of the occurrence of boundary layer separation, although separation may have been present on the upper surface of the BGK-1 models at the higher values of Mach number and angle of incidence.

Integrated force coefficients (normal force C_N , and pitching moment C_m) for all models are included in Table 3 and are plotted against angle of incidence in Figures 6 to 13. The erroneous zero incidence values for C_N and C_m obtained from integration of the 101.6 mm chord NACA-0012 surface pressures (Figures 9 and 13) are caused by a combination of the sparse and uneven distribution of pressure tapings near the leading edge of that model with a peculiarity of the integration routine used (see Section 2.7.1). Manual curve fitting and integration has confirmed that this error affects only the zero incidence results.

The 203.2 mm chord BGK-1 model shows marked non-linearities in its normal force and pitching moment characteristics (Figures 6 and 10) for Mach numbers above 0.76. Similar non-linear behaviour is shown by the 203.2 mm chord NACA-0012 model at $M = 0.80$ and $M = 0.82$ (Figures 8 and 12) and to a lesser extent by the 101.6 mm chord models of both sections. These non-linearities are believed to be due to the effects of incipient choking changing

* By working with the y origin at midspan the problem is simplified since from considerations of symmetry it is evident that the torsional moment acting on the midspan section is zero.

the basic characteristics of the model flow field. Tests with the models at zero incidence indicated choking Mach numbers of 0.824 ± 0.005 for the 203.2 mm chord models and 0.865 ± 0.005 for the 101.6 mm chord models. These values are in good agreement with those given by one-dimensional inviscid theory of 0.82 and 0.87 respectively. Examination of pressure distributions corresponding to the appearance of these non-linearities reveals apparently shock-free recompressions from supersonic velocities, particularly for the lower surface of the BGK-1 models at negative incidence. The decrease in C_N with increasing incidence evident for the 203.2 mm chord NACA-0012 model at $M = 0.82$ (Figure 8) is caused by the upper and lower surface shock waves reversing their normal direction of movement as incidence is varied.

The normal force curve slope $[(\partial C_N / \partial \alpha)_{C_N=0}]$ is plotted against Mach number for the two BGK-1 models in Figure 14 and for the two NACA-0012 models in Figure 15. As would be expected for tests in solid walls, the larger model in both cases shows the higher normal force curve slope.

The tests on the BGK-1 models without the transition fixing roughness bands showed that it was indeed possible to find a "design" pressure distribution in solid walls. Pressure distributions closely approximating the design distribution were obtained at $M = 0.748$, $\alpha = 0.55^\circ$ for the 203.2 mm chord model and at $M = 0.753$, $\alpha = 0.60^\circ$ for the 101.6 mm chord model (see Figures 16 and 17). The sensitivity of the experimental design point pressure distributions noted in the slotted wall tests¹⁰ was even more marked in the present tests. Quite significantly different pressure distributions could be obtained at the same nominal values of Mach number and angle of incidence which had setting accuracies of ± 0.001 and $\pm 0.01^\circ$ respectively.

4. CONCLUSIONS

Measurements of surface pressure distributions have been conducted on two aerofoil sections—NACA-0012 and BGK-1—in a test section with all solid walls. The tests were conducted on two geometrically similar models of each section over a range of Mach numbers between 0.5 and 0.82. The results generally show the characteristics of each section to be similar to those obtained in a similar series of tests conducted in a slotted wall test section. However, at the highest Mach numbers, distinct non-linearities are apparent which are believed to be due to the effects of incipient test section choking.

The main purpose of the tests was to provide a body of data which could be used to test the validity of linear interference theory without the complication of the unknown boundary conditions involved with non-solid walls.

Tests on the BGK-1 section without artificial transition fixing show that pressure distributions closely approximate the "design" distribution could be obtained in the solid wall test section. The sensitivity of these distributions to small changes in Mach number or angle of incidence was found to be greater than that for similar results from the slotted wall tests. As for the slotted wall tests, it was again found that with artificial transition fixing present, the design pressure distributions could no longer be obtained.

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TABLE 1 (a)
Profile of Aerofoil—BGK-1

Upper Surface
 $0.00 \leq x/c \leq 0.95$

| x/c | z/c | x/c | z/c | x/c | z/c |
|---------|---------|---------|---------|---------|---------|
| 0.00000 | 0.00219 | 0.07532 | 0.04352 | 0.27237 | 0.06271 |
| 0.00001 | 0.00298 | 0.07597 | 0.04366 | 0.28741 | 0.06336 |
| 0.00006 | 0.00403 | 0.07660 | 0.04379 | 0.30299 | 0.06396 |
| 0.00017 | 0.00528 | 0.07724 | 0.04392 | 0.31906 | 0.06450 |
| 0.00027 | 0.00603 | 0.07786 | 0.04405 | 0.33556 | 0.06496 |
| 0.00039 | 0.00673 | 0.07849 | 0.04418 | 0.35242 | 0.06535 |
| 0.00058 | 0.00758 | 0.07911 | 0.04430 | 0.36959 | 0.06566 |
| 0.00080 | 0.00832 | 0.08045 | 0.04457 | 0.38700 | 0.06588 |
| 0.00109 | 0.00913 | 0.07978 | 0.04444 | 0.40413 | 0.06602 |
| 0.00141 | 0.00987 | 0.08253 | 0.04496 | 0.42139 | 0.06608 |
| 0.00154 | 0.01015 | 0.08113 | 0.04470 | 0.43871 | 0.06604 |
| 0.00215 | 0.01124 | 0.08182 | 0.04483 | 0.45601 | 0.06592 |
| 0.00269 | 0.01203 | 0.08327 | 0.04510 | 0.47325 | 0.06572 |
| 0.00333 | 0.01286 | 0.08404 | 0.04524 | 0.49034 | 0.06542 |
| 0.00428 | 0.01392 | 0.08484 | 0.04539 | 0.50724 | 0.06504 |
| 0.00534 | 0.01498 | 0.08574 | 0.04555 | 0.52388 | 0.06457 |
| 0.00689 | 0.01637 | 0.08670 | 0.04572 | 0.53954 | 0.06405 |
| 0.00865 | 0.01778 | 0.08773 | 0.04589 | 0.55487 | 0.06345 |
| 0.01107 | 0.01953 | 0.08885 | 0.04608 | 0.56982 | 0.06278 |
| 0.01458 | 0.02178 | 0.09006 | 0.04628 | 0.58435 | 0.06203 |
| 0.01964 | 0.02464 | 0.09138 | 0.04650 | 0.59841 | 0.06122 |
| 0.02348 | 0.02658 | 0.09282 | 0.04674 | 0.61194 | 0.06035 |
| 0.02831 | 0.02881 | 0.09441 | 0.04699 | 0.62490 | 0.05943 |
| 0.03389 | 0.03116 | 0.09626 | 0.04728 | 0.63727 | 0.05847 |
| 0.03779 | 0.03267 | 0.09831 | 0.04759 | 0.64844 | 0.05754 |
| 0.04089 | 0.03380 | 0.10058 | 0.04793 | 0.65902 | 0.05659 |
| 0.05268 | 0.03766 | 0.10310 | 0.04829 | 0.66899 | 0.05564 |
| 0.05397 | 0.03805 | 0.10591 | 0.04869 | 0.67832 | 0.05469 |
| 0.05543 | 0.03848 | 0.10901 | 0.04912 | 0.68701 | 0.05377 |
| 0.05692 | 0.03891 | 0.11245 | 0.04959 | 0.69509 | 0.05288 |
| 0.05769 | 0.03913 | 0.11625 | 0.05009 | 0.70694 | 0.05149 |
| 0.05846 | 0.03935 | 0.12062 | 0.05064 | 0.72841 | 0.04877 |
| 0.05922 | 0.03956 | 0.12545 | 0.05124 | 0.73865 | 0.04738 |
| 0.05997 | 0.03977 | 0.13078 | 0.05187 | 0.76139 | 0.04410 |
| 0.06071 | 0.03997 | 0.13664 | 0.05254 | 0.77701 | 0.04168 |
| 0.06145 | 0.04017 | 0.14305 | 0.05324 | 0.79224 | 0.03920 |
| 0.06217 | 0.04037 | 0.15004 | 0.05398 | 0.80907 | 0.03633 |
| 0.06291 | 0.04056 | 0.15763 | 0.05474 | 0.82499 | 0.03349 |
| 0.06444 | 0.04096 | 0.16585 | 0.05552 | 0.83695 | 0.03128 |
| 0.06593 | 0.04133 | 0.17493 | 0.05634 | 0.85719 | 0.02742 |
| 0.06738 | 0.04169 | 0.18470 | 0.05717 | 0.87562 | 0.02379 |
| 0.06879 | 0.04204 | 0.19518 | 0.05801 | 0.89827 | 0.01923 |
| 0.07016 | 0.04236 | 0.20636 | 0.05884 | 0.91369 | 0.01610 |
| 0.07148 | 0.04267 | 0.21825 | 0.05967 | 0.92769 | 0.01328 |
| 0.07277 | 0.04296 | 0.23081 | 0.06048 | 0.93981 | 0.01087 |
| 0.07401 | 0.04324 | 0.24405 | 0.06126 | | |
| 0.07467 | 0.04338 | 0.25792 | 0.06201 | | |

TABLE 1 (b)
Profile of Aerofoil—BGK-1

Lower Surface
 $0.00 \leq x/c \leq 0.95$

| x/c | z/c | x/c | z/c |
|---------|----------|---------|----------|
| 0.00001 | 0.00149 | 0.26400 | -0.05141 |
| 0.00004 | 0.00080 | 0.28205 | -0.05150 |
| 0.00018 | -0.00068 | 0.29705 | -0.05144 |
| 0.00025 | -0.00114 | 0.31015 | -0.05128 |
| 0.00035 | -0.00169 | 0.32408 | -0.05103 |
| 0.00046 | -0.00219 | 0.34487 | -0.05046 |
| 0.00068 | -0.00301 | 0.36350 | -0.04976 |
| 0.00101 | -0.00395 | 0.38765 | -0.04860 |
| 0.00148 | -0.00500 | 0.40449 | -0.04763 |
| 0.00188 | -0.00575 | 0.42631 | -0.04616 |
| 0.00232 | -0.00649 | 0.45002 | -0.04432 |
| 0.00306 | -0.00760 | 0.47538 | -0.04206 |
| 0.00420 | -0.00903 | 0.52184 | -0.03720 |
| 0.00475 | -0.00964 | 0.54984 | -0.03383 |
| 0.00578 | -0.01069 | 0.57861 | -0.03008 |
| 0.00704 | -0.01181 | 0.60531 | -0.02636 |
| 0.00827 | -0.01279 | 0.62348 | -0.02373 |
| 0.00997 | -0.01401 | 0.64215 | -0.02096 |
| 0.01229 | -0.01547 | 0.65754 | -0.01865 |
| 0.01535 | -0.01718 | 0.67121 | -0.01659 |
| 0.01928 | -0.01911 | 0.69034 | -0.01373 |
| 0.02534 | -0.02170 | 0.70889 | -0.01101 |
| 0.02938 | -0.02324 | 0.72669 | -0.00850 |
| 0.03910 | -0.02653 | 0.74668 | -0.00585 |
| 0.05252 | -0.03034 | 0.76850 | -0.00322 |
| 0.06247 | -0.03278 | 0.78489 | -0.00148 |
| 0.07333 | -0.03514 | 0.80115 | 0.00006 |
| 0.08529 | -0.03744 | 0.81919 | 0.00149 |
| 0.09982 | -0.03988 | 0.83855 | 0.00273 |
| 0.11856 | -0.04253 | 0.85882 | 0.00369 |
| 0.14099 | -0.04511 | 0.87943 | 0.00431 |
| 0.16719 | -0.04745 | 0.90372 | 0.00459 |
| 0.20293 | -0.04966 | 0.92651 | 0.00439 |
| 0.22851 | -0.05068 | 0.94365 | 0.00395 |
| 0.24790 | -0.05117 | | |

TABLE 1 (c)
Profile of Aerofoil—BGK-1

$0.95 \leq x/c \leq 1.00$
Upper Surface

| x/c | z/c Ref. 5 | z/c 101.6 mm Model | z/c 203.2 mm Model |
|---------|-----------------|-------------------------|-------------------------|
| 0.95200 | 0.00849 | 0.00855 | 0.00851 |
| 0.97179 | 0.00480 | 0.00503 | 0.00488 |
| 0.98910 | 0.00182 | 0.00218 | 0.00200 |
| 0.99786 | 0.00048 | 0.00095 | 0.00071 |
| 1.00000 | 0.00018 | 0.00067 | 0.00043 |

Lower Surface

| x/c | z/c Ref. 5 | z/c 101.6 mm Model | z/c 203.2 mm Model |
|---------|-----------------|-------------------------|-------------------------|
| 0.95867 | 0.00333 | 0.00323 | 0.00327 |
| 0.97362 | 0.00247 | 0.00228 | 0.00238 |
| 0.98619 | 0.00154 | 0.00120 | 0.00138 |
| 0.99658 | 0.00056 | 0.00013 | 0.00034 |
| 1.00000 | 0.00018 | -0.00033 | -0.00001 |

TABLE 2 (a)
Pressure Hole Locations—BGK-1
 203·2 mm Chord Model

| Upper Surface | | Lower Surface | |
|---------------|--------|---------------|---------|
| x/c | z/c | x/c | z/c |
| 0·0000 | 0·0047 | 0·0051 | —0·0103 |
| 0·0035 | 0·0129 | 0·0098 | —0·0141 |
| 0·0079 | 0·0179 | 0·0201 | —0·0197 |
| 0·0196 | 0·0247 | 0·0300 | —0·0235 |
| 0·0294 | 0·0293 | 0·0500 | —0·0284 |
| 0·0495 | 0·0367 | 0·0749 | —0·0353 |
| 0·0741 | 0·0433 | 0·1000 | —0·0398 |
| 0·0991 | 0·0480 | 0·1498 | —0·0459 |
| *0·1241 | 0·0513 | 0·2000 | —0·0495 |
| 0·1591 | 0·0549 | 0·2499 | —0·0512 |
| 0·1994 | 0·0583 | 0·3502 | —0·0503 |
| 0·2382 | 0·0609 | 0·4500 | —0·0443 |
| 0·2794 | 0·0631 | 0·5490 | —0·0338 |
| 0·3188 | 0·0645 | 0·6492 | —0·0199 |
| 0·3594 | 0·0655 | 0·7498 | —0·0037 |
| 0·3993 | 0·0660 | 0·8499 | 0·0032 |
| 0·4194 | 0·0660 | 0·9000 | 0·0045 |
| 0·4395 | 0·0660 | | |
| 0·4585 | 0·0659 | | |
| 0·4787 | 0·0656 | | |
| 0·4983 | 0·0653 | | |
| 0·5184 | 0·0648 | | |
| 0·5396 | 0·0641 | | |
| 0·5592 | 0·0632 | | |
| 0·5787 | 0·0624 | | |
| 0·5989 | 0·0612 | | |
| 0·6180 | 0·0599 | | |
| 0·6484 | 0·0576 | | |
| 0·6747 | 0·0551 | | |
| 0·6989 | 0·0525 | | |
| 0·7237 | 0·0494 | | |
| 0·7488 | 0·0459 | | |
| 0·7738 | 0·0422 | | |
| 0·7990 | 0·0381 | | |
| 0·8240 | 0·0339 | | |
| 0·8489 | 0·0289 | | |
| 0·8989 | 0·0191 | | |

* Damaged orifice.

TABLE 2 (b)
Pressure Hole Locations—BGK-1

101.6 mm Chord Model

| Upper Surface | | Lower Surface | |
|---------------|--------|---------------|---------|
| x/c | z/c | x/c | z/c |
| 0.0000 | 0.0047 | 0.0096 | -0.0142 |
| 0.0089 | 0.0178 | 0.0205 | -0.0198 |
| 0.0206 | 0.0248 | 0.0306 | -0.0236 |
| 0.0306 | 0.0300 | 0.0506 | -0.0285 |
| *0.0504 | 0.0367 | 0.0754 | -0.0354 |
| 0.0752 | 0.0434 | 0.1001 | -0.0398 |
| 0.0999 | 0.0481 | | |
| 0.1252 | 0.0514 | 0.2000 | -0.0495 |
| 0.1599 | 0.0550 | 0.2496 | -0.0512 |
| 0.2000 | 0.0584 | 0.3495 | -0.0503 |
| 0.2407 | 0.0610 | 0.4486 | -0.0444 |
| 0.2798 | 0.0631 | 0.5493 | -0.0338 |
| 0.3202 | 0.0645 | 0.6499 | -0.0199 |
| 0.3603 | 0.0655 | 0.7499 | -0.0037 |
| 0.4001 | 0.0660 | 0.8500 | 0.0032 |
| 0.4198 | 0.0660 | | |
| 0.4402 | 0.0660 | | |
| 0.4599 | 0.0659 | | |
| 0.4796 | 0.0655 | | |
| 0.4996 | 0.0652 | | |
| 0.5405 | 0.0640 | | |
| 0.5602 | 0.0630 | | |
| 0.5803 | 0.0623 | | |
| 0.6004 | 0.0611 | | |
| 0.6203 | 0.0598 | | |
| 0.6505 | 0.0575 | | |
| 0.6754 | 0.0551 | | |
| 0.7005 | 0.0524 | | |
| 0.7255 | 0.0493 | | |
| 0.7504 | 0.0458 | | |
| 0.7753 | 0.0421 | | |
| 0.8005 | 0.0380 | | |
| 0.8255 | 0.0338 | | |
| 0.8503 | 0.0288 | | |

* Damaged orifice.

TABLE 2 (c)
Pressure Hole Locations—NACA-0012

203·2 mm Chord Model

| Hole No. | x/c | Hole No. | x/c |
|----------|--------|----------|--------|
| 1 | 0·0119 | 20 | 0·4871 |
| 2 | 0·0371 | 21 | 0·5120 |
| 3 | 0·0623 | 22 | 0·5371 |
| 4 | 0·0872 | 23 | 0·5621 |
| 5 | 0·1123 | 24 | 0·5870 |
| 6 | 0·1372 | 25 | 0·6122 |
| 7 | 0·1623 | 26 | 0·6371 |
| 8 | 0·1872 | 27 | 0·6661 |
| 9 | 0·2122 | 28 | 0·6869 |
| 10 | 0·2372 | 29 | 0·7121 |
| 11 | 0·2621 | 30 | 0·7369 |
| 12 | 0·2872 | 31 | 0·7620 |
| 13 | 0·3122 | 32 | 0·7870 |
| 14 | 0·3376 | 33 | 0·8117 |
| 15 | 0·3618 | 34 | 0·8369 |
| 16 | 0·3873 | 35 | 0·8619 |
| 17 | 0·4124 | 36 | 0·8869 |
| 18 | 0·4371 | 37 | 0·9120 |
| 19 | 0·4621 | 38 | 0·9416 |

TABLE 2 (d)
Pressure Hole Locations—NACA-0012

101·6 mm Chord Model

| Upper Surface | | Lower Surface | |
|---------------|--------|---------------|--------|
| Hole No. | x/c | Hole No. | x/c |
| 1 | 0·0207 | 1 | 0·0757 |
| 2 | 0·0787 | 2 | 0·1294 |
| 3 | 0·1299 | 3 | 0·1830 |
| 4 | 0·1813 | 4 | 0·2368 |
| 5 | 0·2307 | 5 | 0·2864 |
| 6 | 0·2836 | 6 | 0·3383 |
| 7 | 0·3337 | 7 | 0·3904 |
| 8 | 0·3858 | 8 | 0·4376 |
| 9 | 0·4358 | 9 | 0·4870 |
| 10 | 0·4860 | 10 | 0·5371 |
| 11 | 0·5380 | 11 | 0·5879 |
| 12 | 0·5882 | 12 | 0·6371 |
| 13 | 0·6395 | 13 | 0·6874 |
| 14 | 0·6894 | 14 | 0·7359 |
| 15 | 0·7443 | | |
| 16 | 0·7954 | | |
| 17 | 0·8414 | | |
| 18 | 0·8891 | | |

TABLE 3 (a)
Test Program — BGK-1
203·2 mm Chord Model

| M | α | C_N | C_m |
|--------|----------|---------|---------|
| 0·750* | 0·55 | 0·5849 | -0·1301 |
| 0·749* | 0·55 | 0·5656 | -0·1263 |
| 0·749* | 0·55 | 0·5858 | -0·1301 |
| 0·50 | -2·5 | -0·0313 | -0·0830 |
| 0·50 | 0 | 0·2760 | -0·0835 |
| 0·50 | 1 | 0·3945 | -0·0830 |
| 0·50 | 2 | 0·5313 | -0·0829 |
| 0·50 | 3 | — | — |
| 0·50 | 4 | 0·7530 | -0·0775 |
| 0·60 | -2·5 | -0·0529 | -0·0881 |
| 0·60 | 0 | 0·2893 | -0·0886 |
| 0·60 | 1 | 0·4196 | -0·0875 |
| 0·60 | 2 | 0·5486 | -0·0855 |
| 0·60 | 3 | 0·6714 | -0·0811 |
| 0·60 | 4 | 0·8034 | -0·0747 |
| 0·65 | -2·5 | -0·0634 | -0·0910 |
| 0·65 | 0 | 0·2960 | -0·0917 |
| 0·65 | 1 | 0·4358 | -0·0901 |
| 0·65 | 2 | 0·5686 | -0·0857 |
| 0·65 | 3 | 0·7185 | -0·0813 |
| 0·70 | -2·5 | -0·0915 | -0·0954 |
| 0·70 | 0 | 0·3016 | -0·0950 |
| 0·70 | 1 | 0·4519 | -0·0917 |
| 0·70 | 2 | 0·6194 | -0·0869 |
| 0·70 | 3 | 0·7968 | -0·0854 |
| 0·72 | -2·5 | -0·1116 | -0·0979 |
| 0·72 | 0 | 0·3060 | -0·0966 |
| 0·72 | 1 | 0·4674 | -0·0927 |
| 0·72 | 2 | 0·6516 | -0·0896 |
| 0·74 | -2·5 | -0·1132 | -0·1008 |
| 0·74 | -1 | 0·1448 | -0·0994 |
| 0·74 | 0 | 0·3062 | -0·0978 |
| 0·74 | 1 | 0·4909 | -0·0944 |
| 0·74 | 2 | 0·6665 | -0·0982 |
| 0·76 | -2·5 | -0·1525 | -0·0735 |
| 0·76 | -1 | 0·1335 | -0·1030 |
| 0·76 | 0 | 0·3056 | -0·1000 |
| 0·76 | 1 | 0·4987 | -0·0972 |
| 0·76 | 2 | 0·5759 | -0·0973 |
| 0·78 | -2·5 | -0·0130 | -0·0823 |
| 0·78 | -1 | 0·0995 | -0·1009 |
| 0·78 | 0 | 0·3008 | -0·1059 |
| 0·78 | 1 | 0·4383 | -0·1013 |
| 0·78 | 2 | 0·4756 | -0·0949 |
| 0·80 | -2·5 | -0·1015 | -0·0552 |
| 0·80 | -1 | -0·0039 | -0·0534 |
| 0·80 | 0 | 0·1887 | -0·0914 |
| 0·80 | 1 | 0·3249 | -0·0985 |
| 0·80 | 2 | 0·3996 | -0·0982 |

* Free transition.

TABLE 3 (b)
TEST PROGRAM — BGK-1

101.6 mm Chord

| M | α | C_N | C_m |
|--------|----------|---------|---------|
| 0.753* | 0.60 | 0.5444 | -0.1146 |
| 0.752* | 0.60 | 0.5464 | -0.1145 |
| 0.755* | 0.60 | 0.5632 | -0.1182 |
| 0.50 | -2.5 | -0.0336 | -0.0753 |
| 0.50 | 0 | 0.2752 | -0.0770 |
| 0.50 | 1 | 0.3910 | -0.0764 |
| 0.50 | 2 | 0.5037 | -0.0753 |
| 0.50 | 3 | 0.6148 | -0.0731 |
| 0.50 | 4 | 0.7239 | -0.0705 |
| 0.60 | -2.5 | -0.0512 | -0.0818 |
| 0.60 | 0 | 0.2846 | -0.0814 |
| 0.60 | 1 | 0.4110 | -0.0804 |
| 0.60 | 2 | 0.5342 | -0.0781 |
| 0.60 | 3 | 0.6456 | -0.0741 |
| 0.60 | 4 | 0.7653 | -0.0686 |
| 0.65 | -2.5 | 0.0663 | -0.0846 |
| 0.65 | 0 | 0.2902 | -0.0842 |
| 0.65 | 1 | 0.4192 | -0.0819 |
| 0.65 | 2 | 0.5474 | -0.0783 |
| 0.65 | 3 | 0.6736 | -0.0733 |
| 0.70 | -2.5 | -0.0887 | -0.0907 |
| 0.70 | 0 | 0.2941 | -0.0877 |
| 0.70 | 1 | 0.4415 | -0.0838 |
| 0.70 | 2 | 0.5933 | -0.0789 |
| 0.70 | 3 | 0.7559 | -0.0743 |
| 0.72 | -2.5 | -0.1086 | -0.0931 |
| 0.72 | 0 | 0.2943 | -0.0886 |
| 0.72 | 1 | 0.4489 | -0.0845 |
| 0.72 | 2 | 0.6172 | -0.0795 |
| 0.74 | -2.5 | -0.1294 | -0.0954 |
| 0.74 | -1 | 0.1397 | -0.0923 |
| 0.74 | 0 | 0.2998 | -0.0897 |
| 0.74 | 1 | 0.4627 | -0.0854 |
| 0.74 | 2 | 0.6405 | -0.0827 |
| 0.76 | -2.5 | -0.1624 | -0.0841 |
| 0.76 | -1 | 0.1241 | -0.0950 |
| 0.76 | 0 | 0.3001 | -0.0912 |
| 0.76 | 1 | 0.4864 | -0.0880 |
| 0.76 | 2 | 0.6132 | -0.0867 |
| 0.78 | -2.5 | -0.1500 | -0.0680 |
| 0.78 | -1 | 0.1046 | -0.0952 |
| 0.78 | 0 | 0.2934 | -0.0945 |
| 0.78 | 1 | 0.4684 | -0.0915 |
| 0.78 | 2 | 0.5375 | -0.0862 |
| 0.80 | -2.5 | -0.1000 | -0.0687 |
| 0.80 | -1 | 0.0723 | -0.0837 |
| 0.80 | 0 | 0.2447 | -0.0951 |
| 0.80 | 1 | 0.3734 | -0.0894 |
| 0.80 | 2 | 0.4498 | -0.0839 |

* Free transition.

TABLE 3 (c)
TEST PROGRAM — NACA-0012

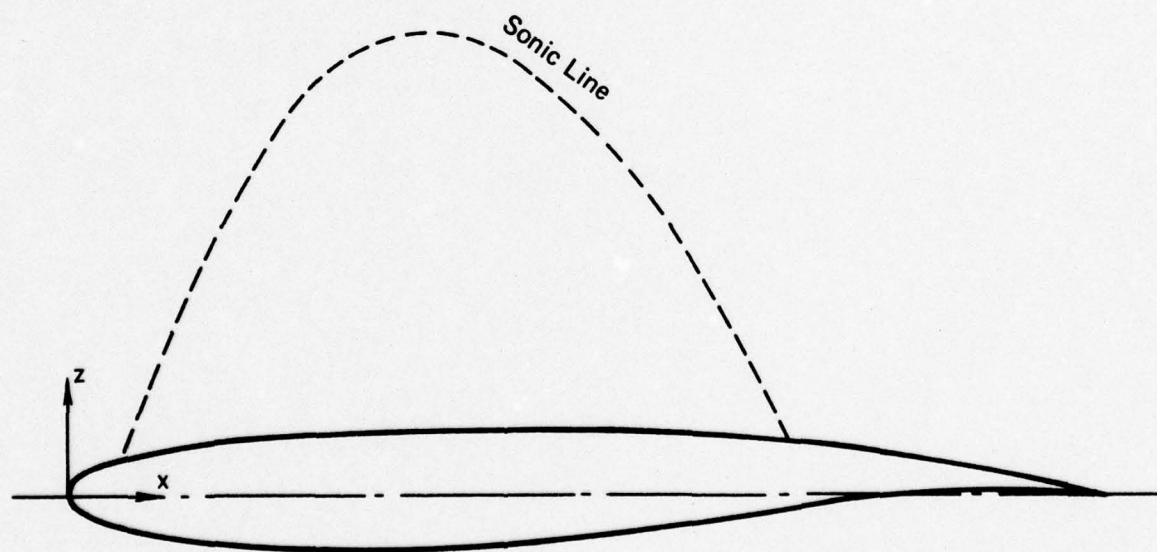
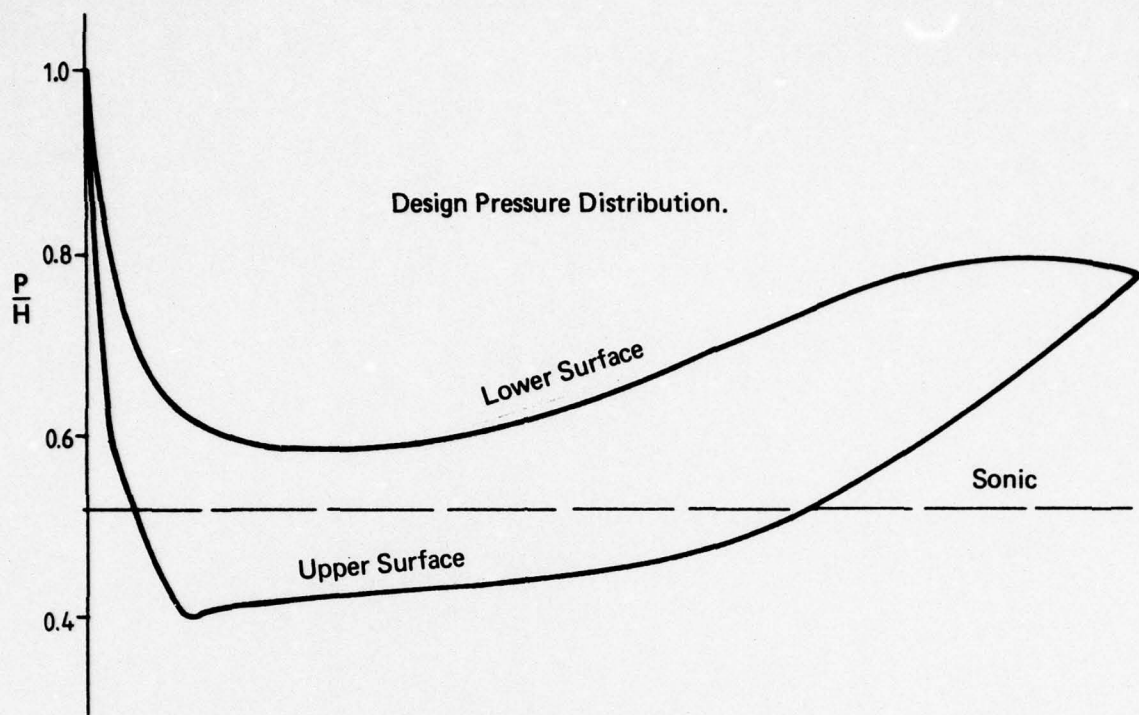
203·2 mm Chord Model

| <i>M</i> | Nominal α | Corrected α | C_N | C_m |
|----------|---------------------|-----------------------|---------|---------|
| 0·50 | 0 | 0 | —0·0008 | 0·0004 |
| 0·50 | 1 | 1 | 0·1154 | 0·0017 |
| 0·50 | 2 | 2 | 0·2304 | 0·0032 |
| 0·50 | 3 | 3 | 0·3423 | 0·0052 |
| 0·50 | 4 | 4 | 0·4525 | 0·0075 |
| 0·60 | 0 | 0 | 0·0002 | 0·0003 |
| 0·60 | 1 | 1 | 0·1211 | 0·0029 |
| 0·60 | 2 | 2 | 0·2471 | 0·0049 |
| 0·60 | 3 | 3 | 0·3713 | 0·0084 |
| 0·65 | 0 | 0 | 0·0009 | 0·0002 |
| 0·65 | 1 | 1 | 0·1324 | 0·0028 |
| 0·65 | 2 | 2 | 0·2596 | 0·0070 |
| 0·65 | 3 | 3 | 0·3968 | 0·0121 |
| 0·675 | 0 | 0 | 0·0011 | 0·0004 |
| 0·675 | 1 | 1 | 0·1344 | 0·0039 |
| 0·675 | 2 | 2 | 0·2747 | 0·0081 |
| 0·70 | 0 | 0 | —0·0012 | —0·0000 |
| 0·70 | 1 | 1 | 0·1424 | 0·0048 |
| 0·70 | 2 | 2 | 0·2917 | 0·0101 |
| 0·72 | 0 | 0 | —0·0047 | 0·0001 |
| 0·72 | 1 | 1 | 0·1492 | 0·0053 |
| 0·72 | 2 | 2 | 0·3208 | 0·0111 |
| 0·74 | 0 | 0 | —0·0071 | 0·0001 |
| 0·74 | 1 | 1 | 0·1663 | 0·0057 |
| 0·74 | 2 | 2 | 0·3337 | 0·0087 |
| 0·76 | 0 | 0 | —0·0045 | —0·0000 |
| 0·76 | 1 | 1 | 0·1804 | 0·0027 |
| 0·76 | 2 | 2 | 0·3456 | —0·0005 |
| 0·78 | 0 | 0 | 0·0016 | —0·0003 |
| 0·78 | 0·5 | 0·5 | 0·1018 | —0·0016 |
| 0·78 | 1 | 1 | 0·1844 | —0·0049 |
| 0·80 | 0 | 0 | —0·0061 | 0·0015 |
| 0·80 | 0·5 | 0·5 | 0·0644 | —0·0021 |
| 0·80 | 1 | 1 | 0·0933 | —0·0002 |
| 0·82 | 0 | 0 | 0·0374 | —0·0151 |
| 0·82 | 0·5 | 0·5 | —0·0363 | 0·0287 |
| 0·82 | 1 | 1 | —0·0605 | 0·0493 |

TABLE 3 (d)
TEST PROGRAM — NACA-0012

101.6 mm Chord Model

| M | Nominal | Corrected | C_N | C_m |
|-------|---------|-----------|--------|---------|
| 0.50 | 0 | 0.01 | 0.0252 | 0.0034 |
| 0.50 | 1 | 1.02 | 0.1091 | 0.0012 |
| 0.50 | 2 | 2.03 | 0.2247 | 0.0024 |
| 0.50 | 3 | 3.05 | 0.3361 | 0.0048 |
| 0.50 | 4 | 4.07 | 0.4451 | 0.0073 |
| 0.60 | 0 | 0.01 | 0.0211 | 0.0040 |
| 0.60 | 1 | 1.02 | 0.1194 | 0.0023 |
| 0.60 | 2 | 2.05 | 0.2387 | 0.0047 |
| 0.60 | 3 | 3.07 | 0.3591 | 0.0087 |
| 0.65 | 0 | 0.01 | 0.0271 | 0.0041 |
| 0.65 | 1 | 1.03 | 0.1271 | 0.0030 |
| 0.65 | 2 | 2.06 | 0.2521 | 0.0073 |
| 0.65 | 3 | 3.09 | 0.3920 | 0.0148 |
| 0.675 | 0 | 0.01 | 0.0263 | 0.0045 |
| 0.675 | 1 | 1.03 | 0.1319 | 0.0044 |
| 0.675 | 2 | 2.07 | 0.2770 | 0.0121 |
| 0.70 | 0 | 0.01 | 0.0280 | 0.0052 |
| 0.70 | 1 | 1.04 | 0.1428 | 0.0067 |
| 0.70 | 2 | 2.10 | 0.3232 | 0.0203 |
| 0.72 | 0 | 0.02 | 0.0297 | 0.0054 |
| 0.72 | 1 | 1.05 | 0.1719 | 0.0113 |
| 0.72 | 2 | 2.09 | 0.3110 | 0.0166 |
| 0.74 | 0 | 0.02 | 0.0318 | 0.0059 |
| 0.74 | 1 | 1.06 | 0.1909 | 0.0143 |
| 0.74 | 2 | 2.09 | 0.3346 | 0.0134 |
| 0.76 | 0 | 0.01 | 0.0208 | 0.0042 |
| 0.76 | 1 | 1.06 | 0.2025 | 0.0135 |
| 0.76 | 2 | 2.08 | 0.3417 | 0.0086 |
| 0.78 | 0 | 0.01 | 0.0139 | 0.0026 |
| 0.78 | 0.5 | 0.52 | 0.0900 | 0.0009 |
| 0.78 | 1 | 1.05 | 0.1918 | 0.0049 |
| 0.80 | 0 | 0.01 | 0.0076 | 0.0019 |
| 0.80 | 0.5 | 0.53 | 0.1172 | 0.0041 |
| 0.80 | 1 | 1.03 | 0.1605 | -0.0028 |
| 0.82 | 0 | 0 | 0.0027 | 0.0020 |
| 0.82 | 0.5 | 0.53 | 0.0570 | 0.0076 |
| 0.82 | 1 | 1.05 | 0.0699 | 0.0160 |



Design conditions $M = 0,750$ $\alpha = 0^\circ$ $C_L = 0.63$

FIG. 1 SUPERCRITICAL LIFTING AEROFOIL BGK 1

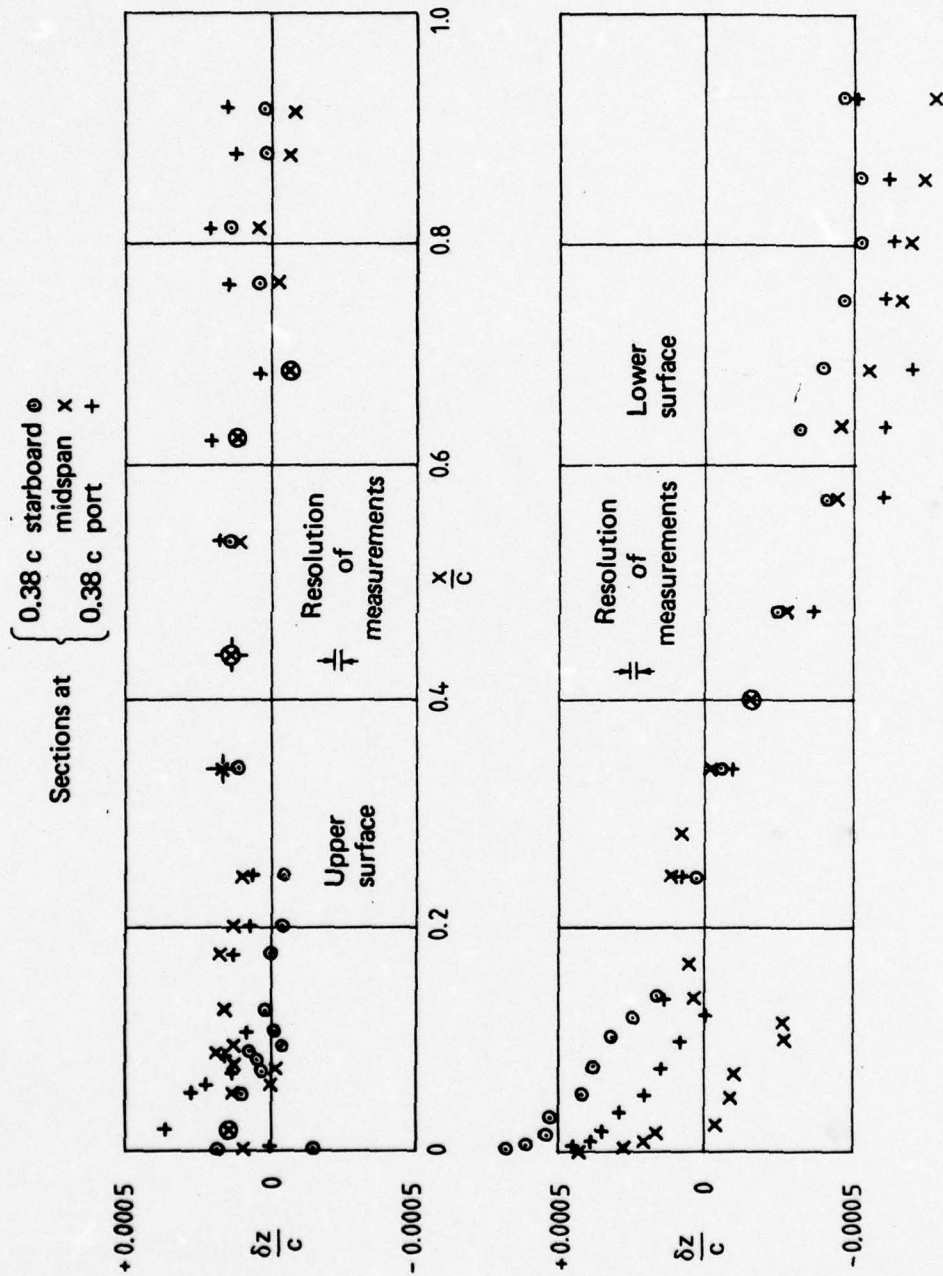
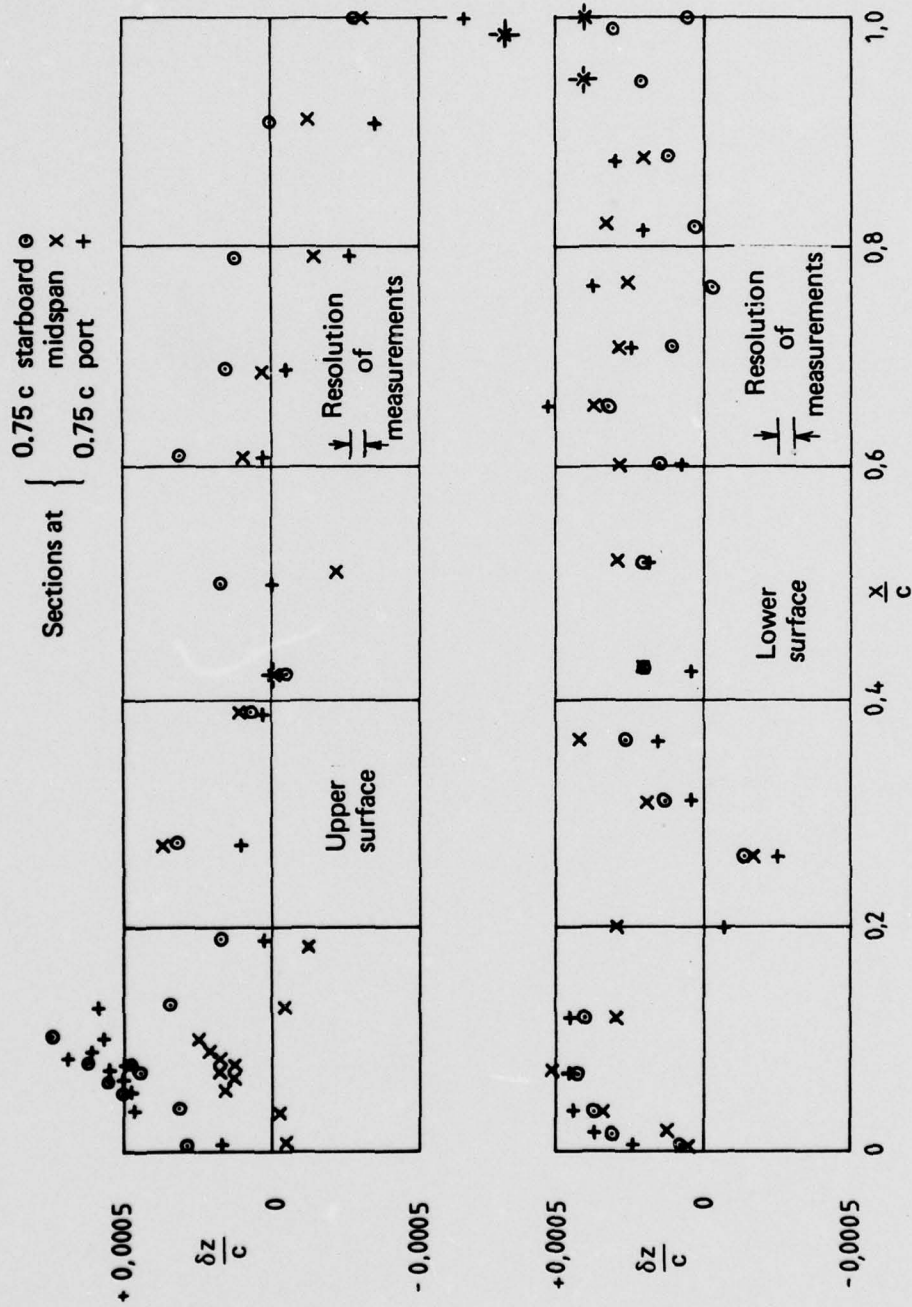


FIG. 2. PROFILE ERRORS — BGK-1; 203.2 mm CHORD



δz = Measured ordinates -- manufacturing ordinates
(positive for excess metal)

FIG. 3. PROFILE ERRORS -- BGK-1; 101.6 mm CHORD

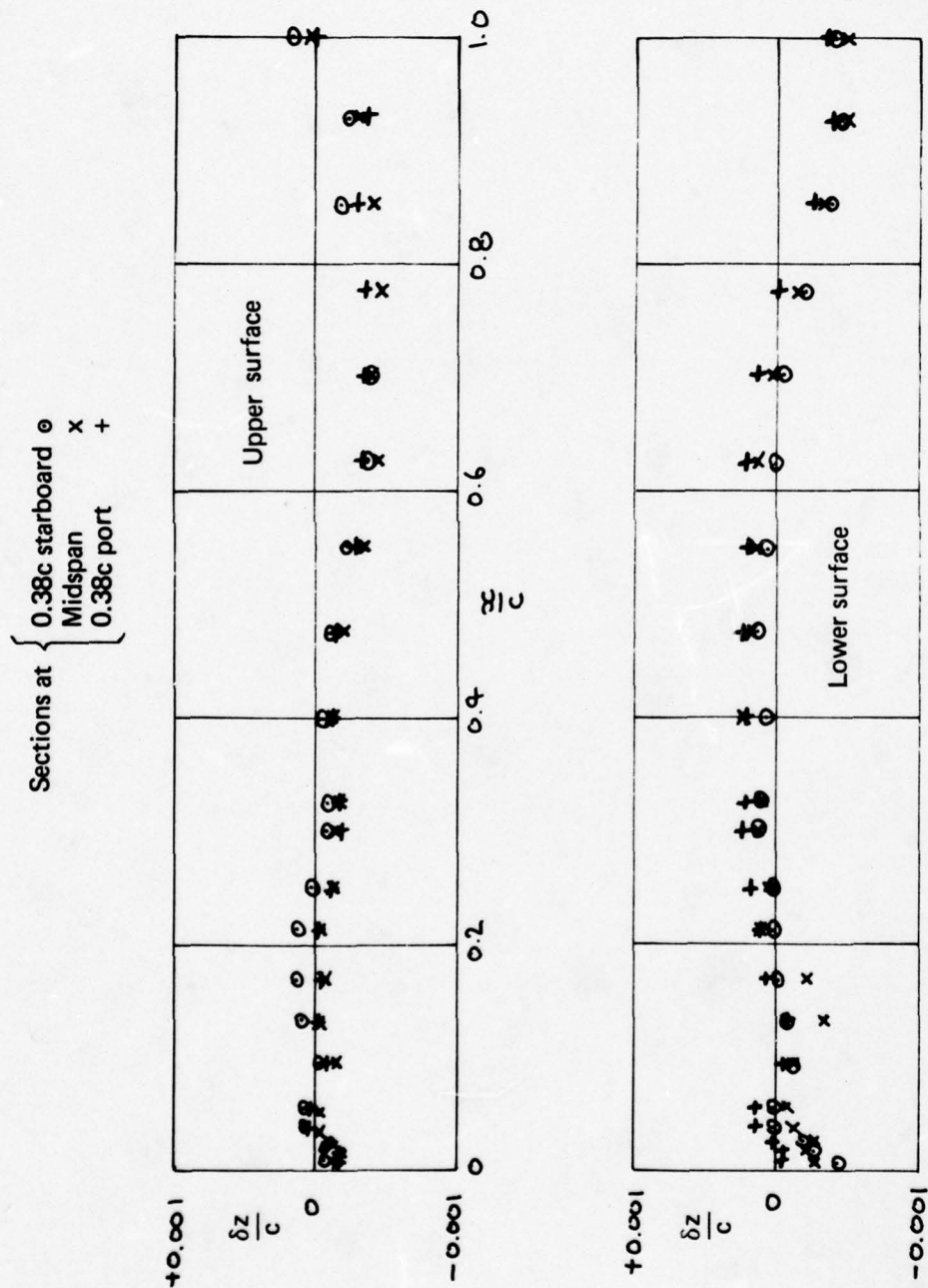
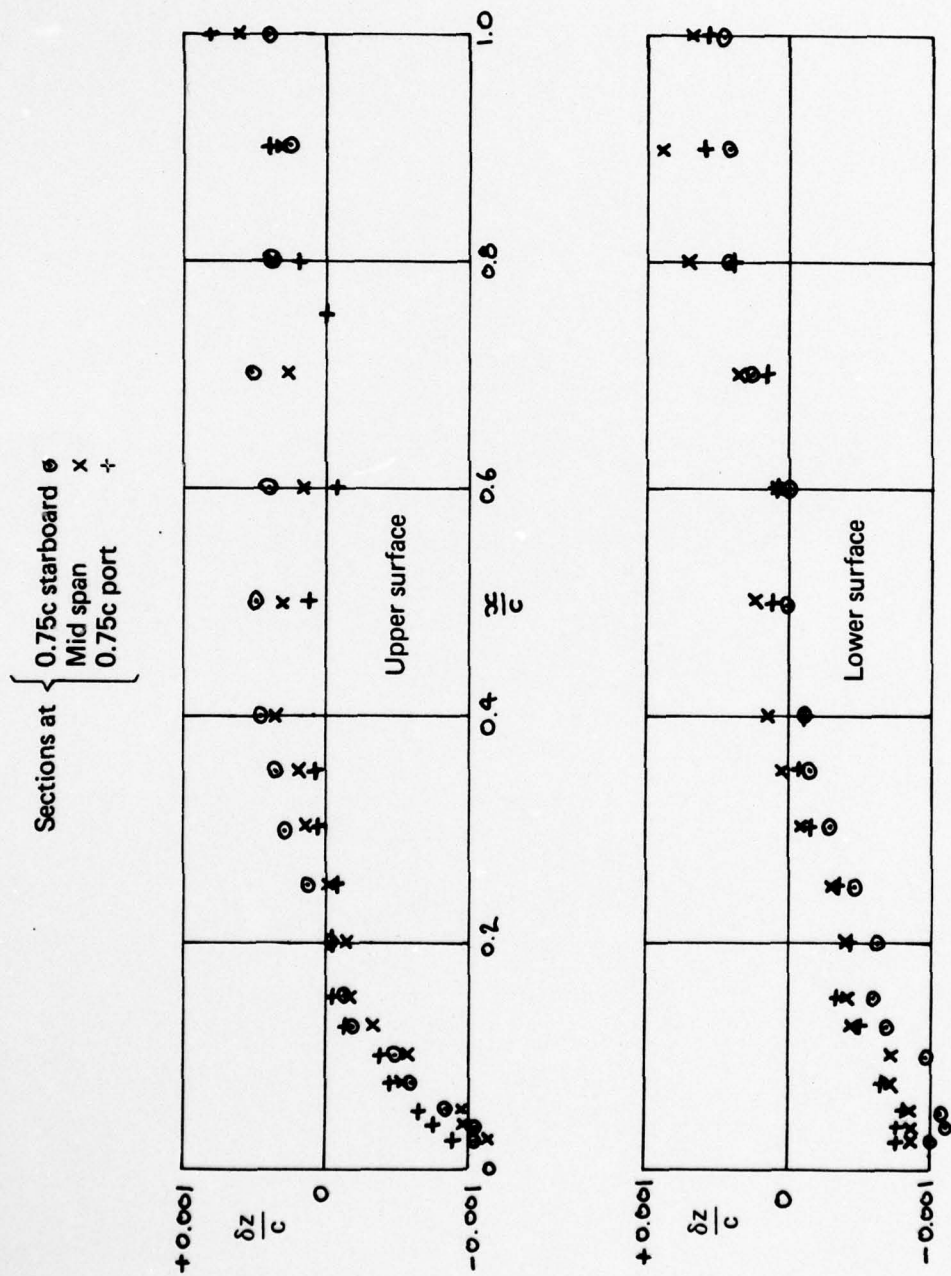


FIG. 4. PROFILE ERRORS - NACA-0012; 203.2 mm CHORD



δz = Measured ordinates — design ordinates
(positive for excess metal)

FIG. 5. PROFILE ERRORS — NACA-0012; 101.6 mm CHORD

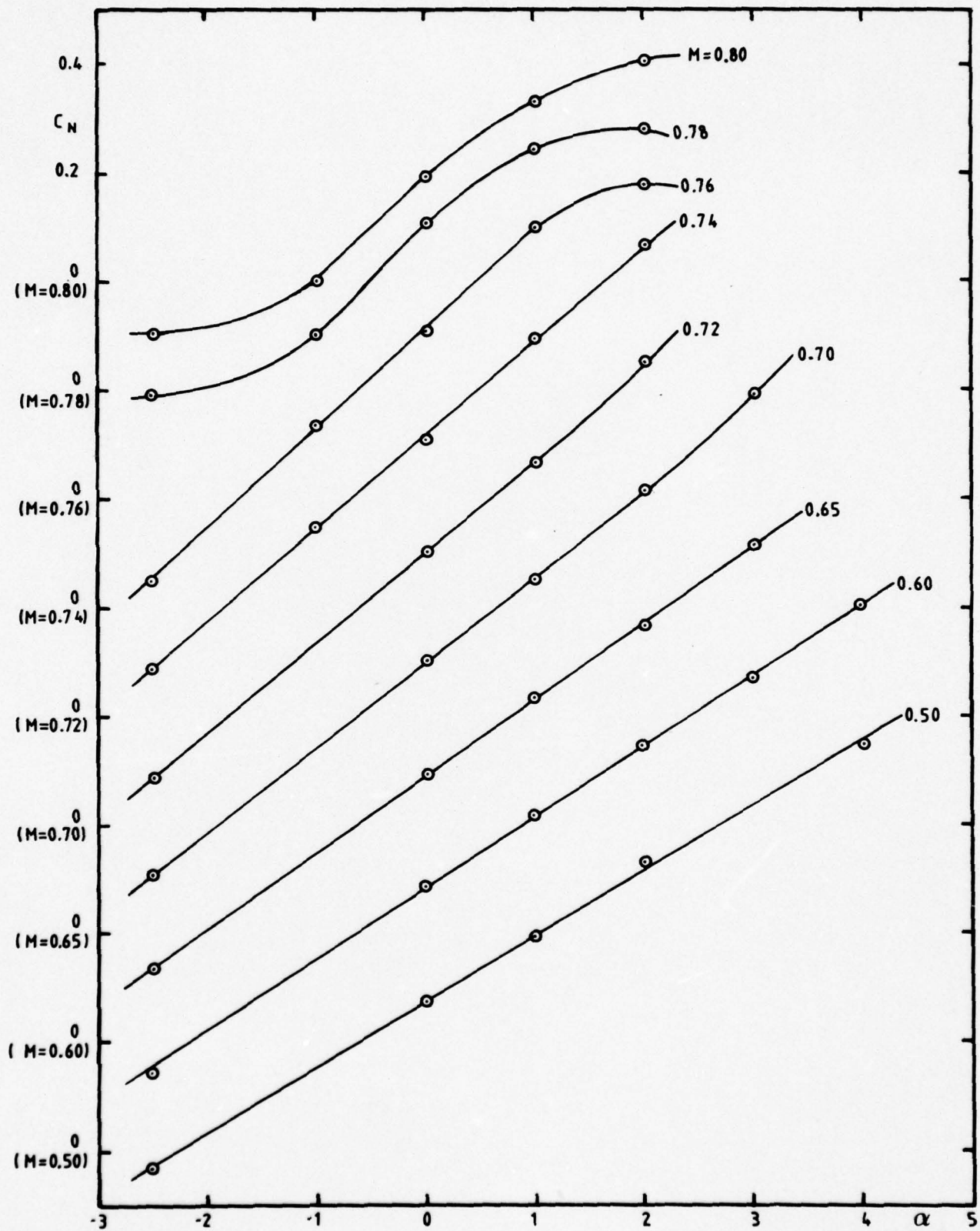


FIG. 6. VARIATION OF C_N WITH α - BGK-1; 203.2 mm CHORD

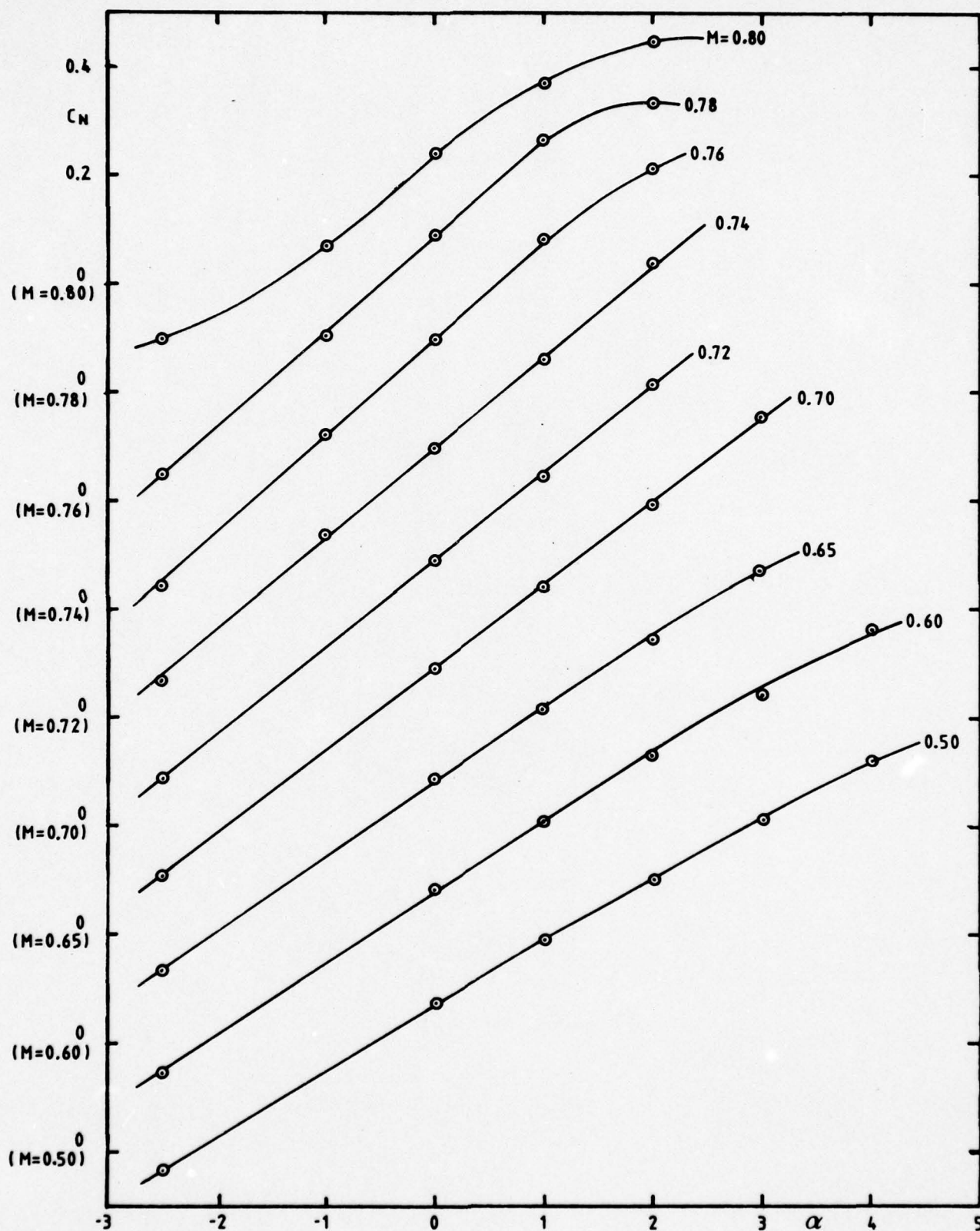


FIG. 7. VARIATION OF C_N WITH α - BGK-1; 101.6 mm CHORD

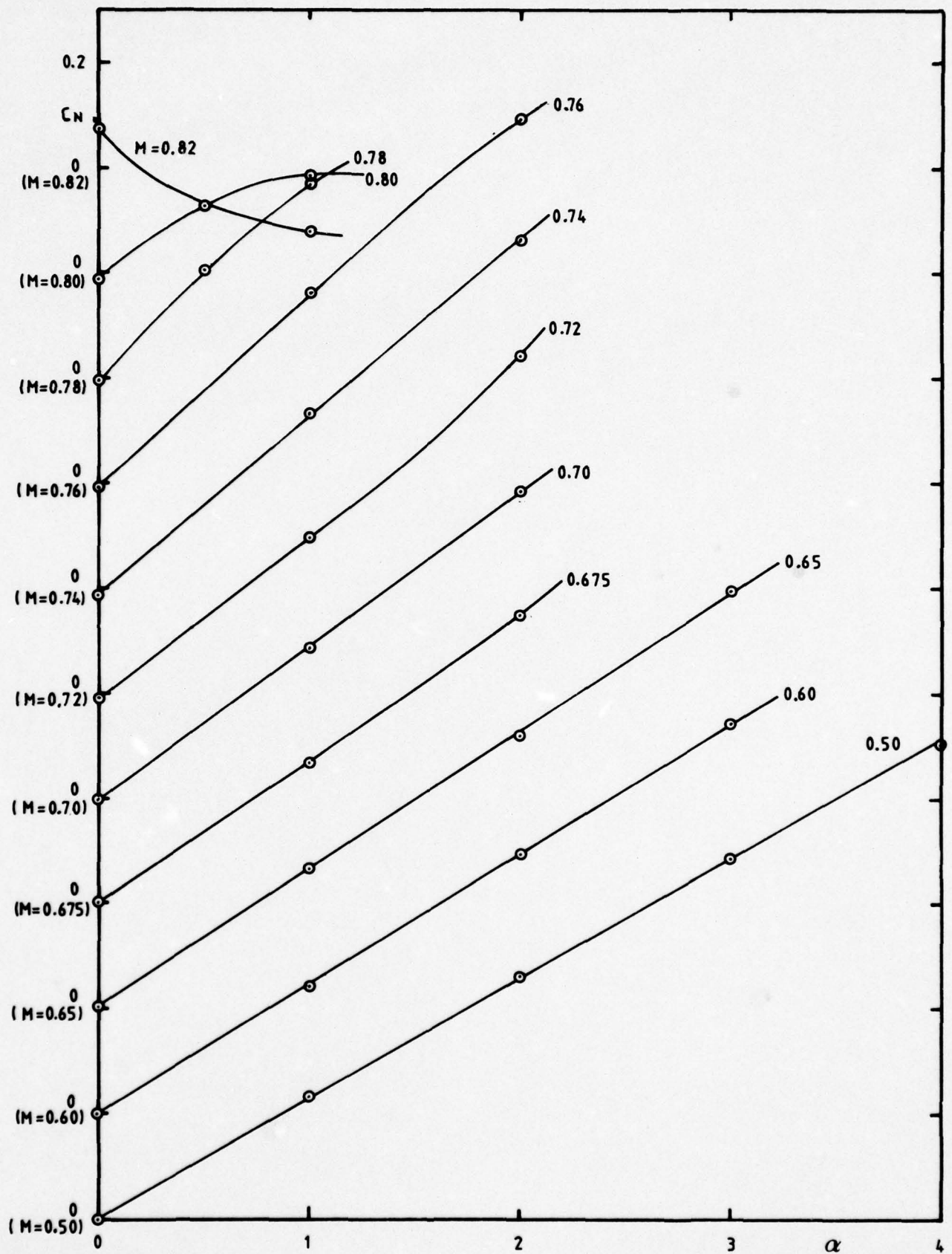


FIG. 8. VARIATION OF C_N WITH α - NACA-0012; 203.2 mm CHORD

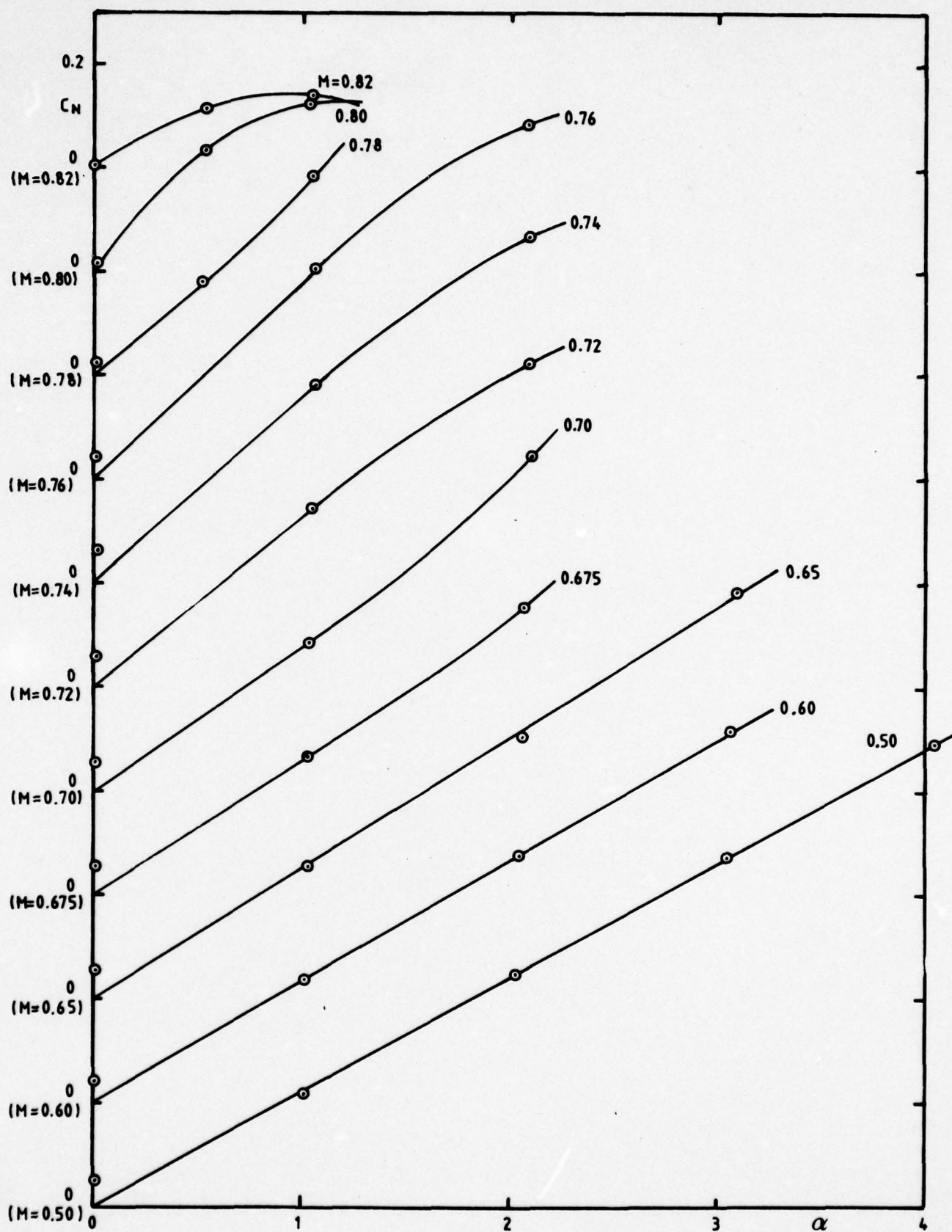


FIG. 9. VARIATION OF C_N WITH α - NACA-0012; 101.6 mm CHORD

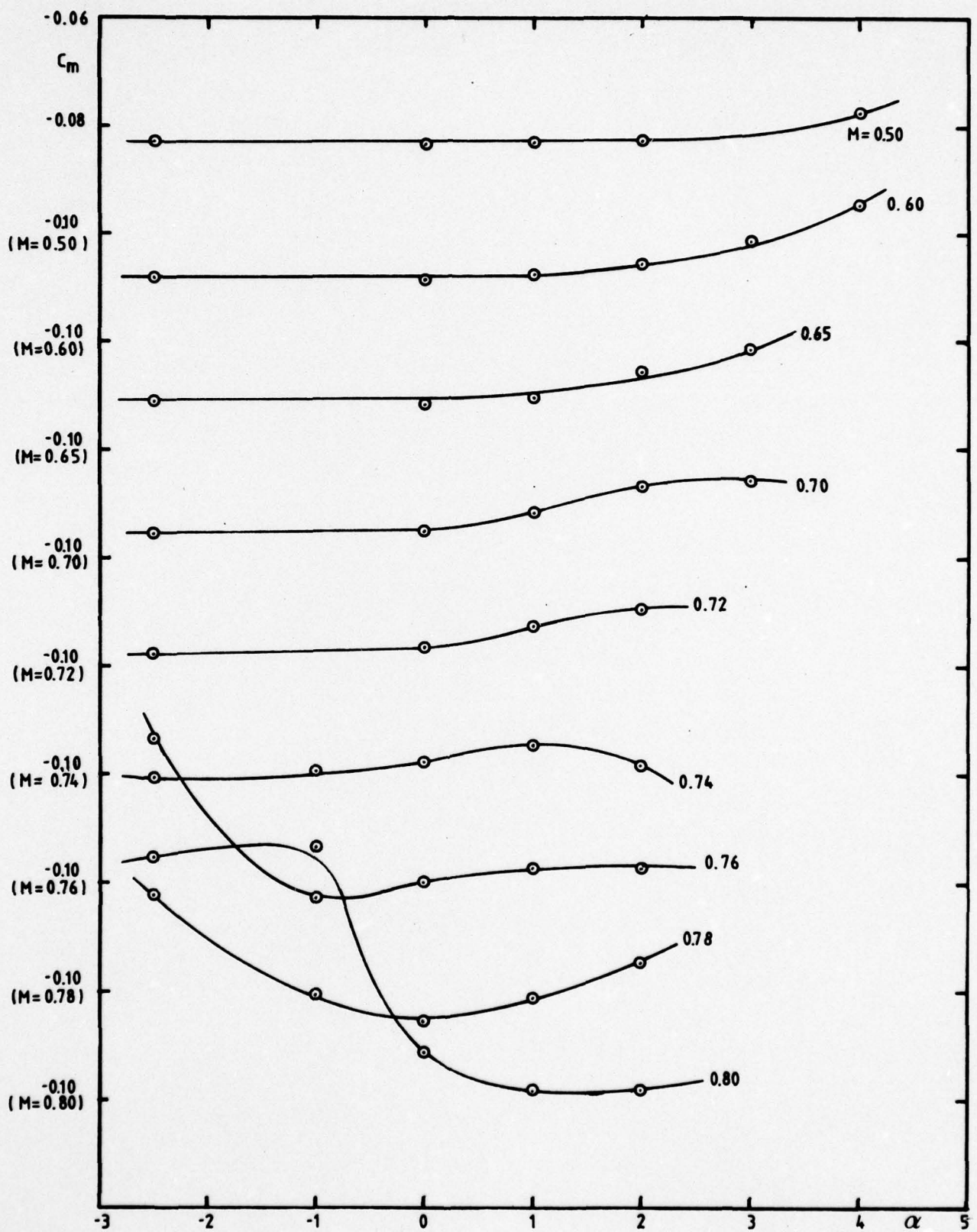


FIG. 10. VARIATION OF C_m WITH α - BGK-1; 203.2 mm CHORD

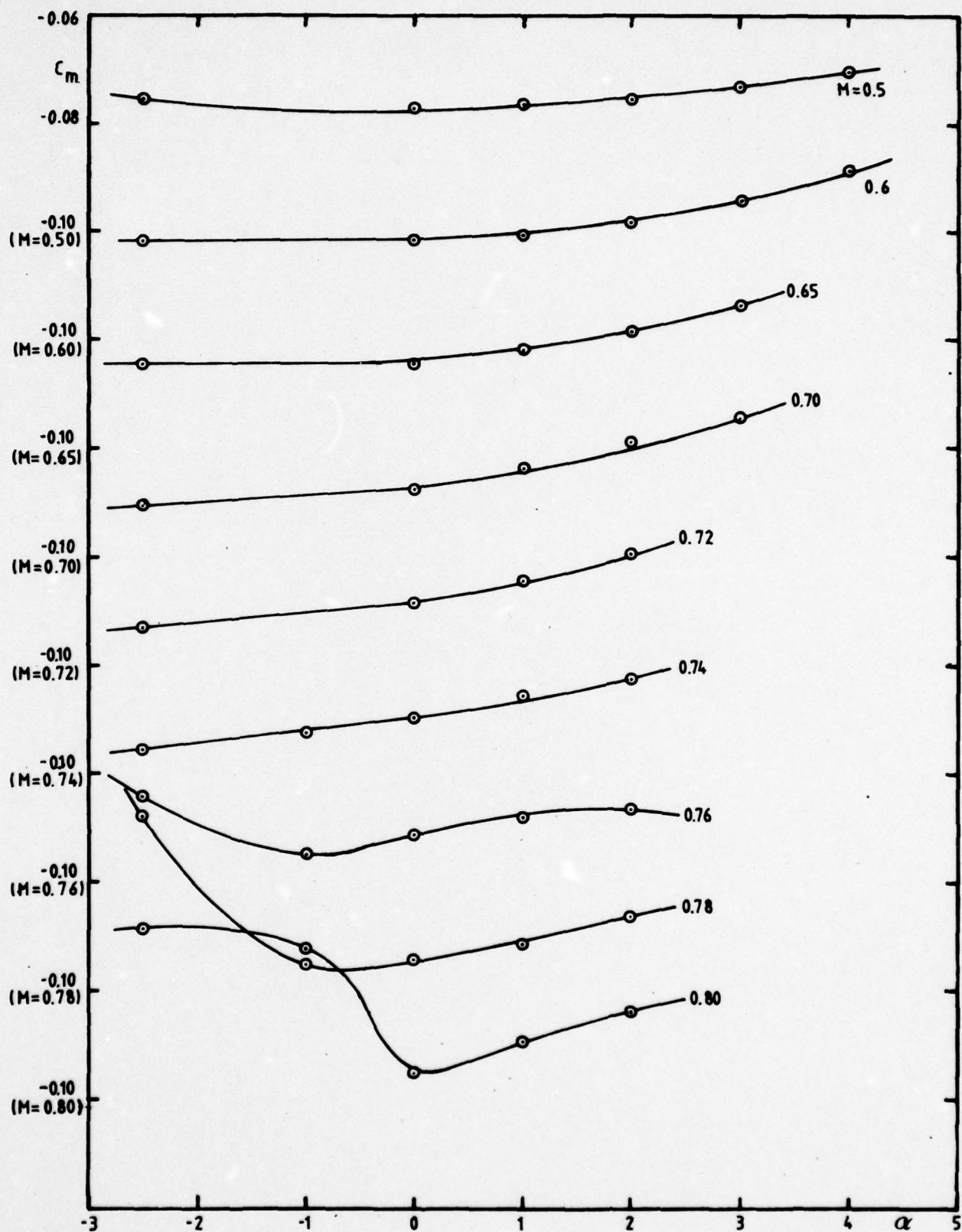


FIG. 11. VARIATION OF C_m WITH α - BGK-1; 101.6 mm CHORD

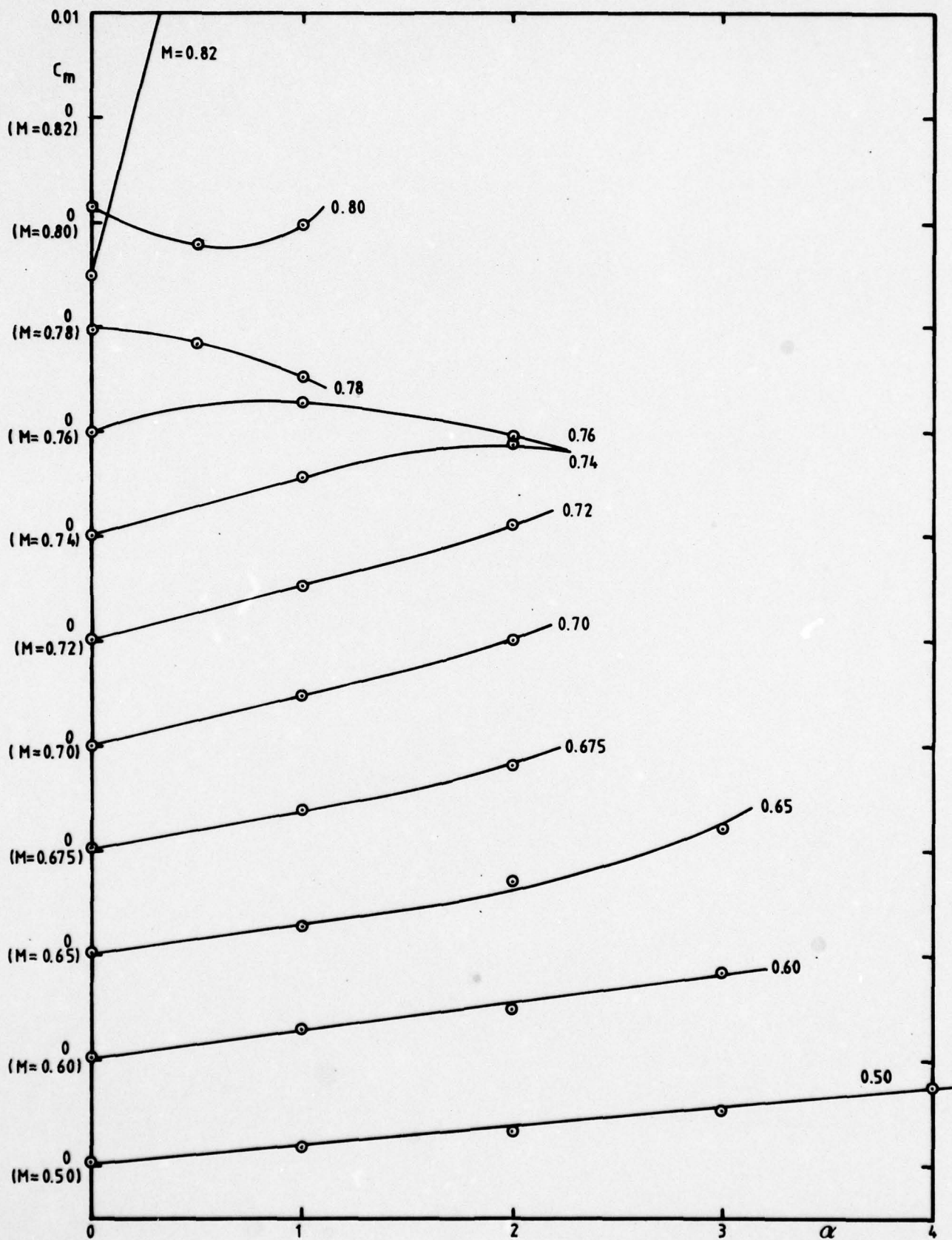


FIG. 12. VARIATION OF C_m WITH α - NACA-0012; 203.2 mm CHORD

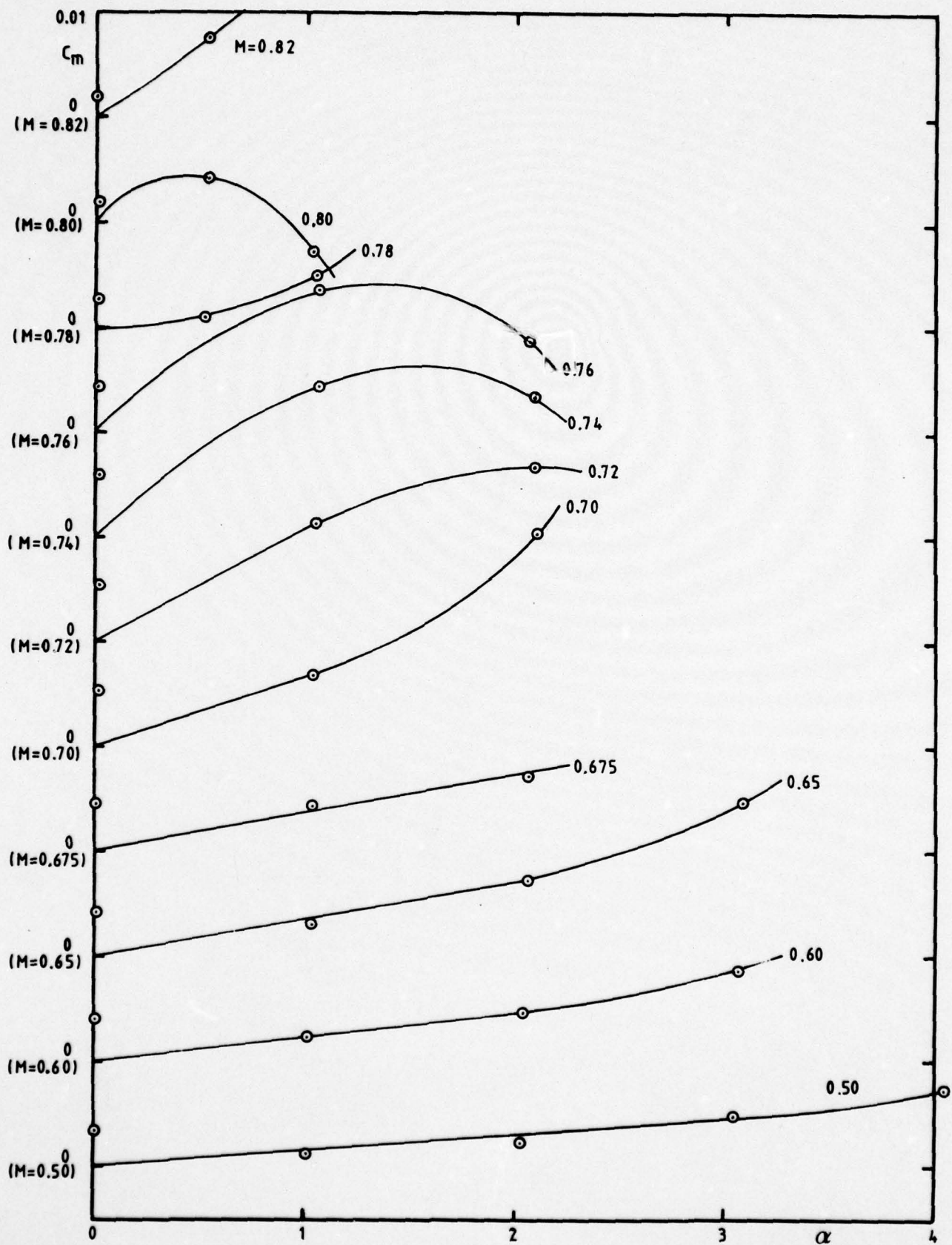


FIG. 13. VARIATION OF C_m WITH α - NACA-0012; 101.6 mm CHORD

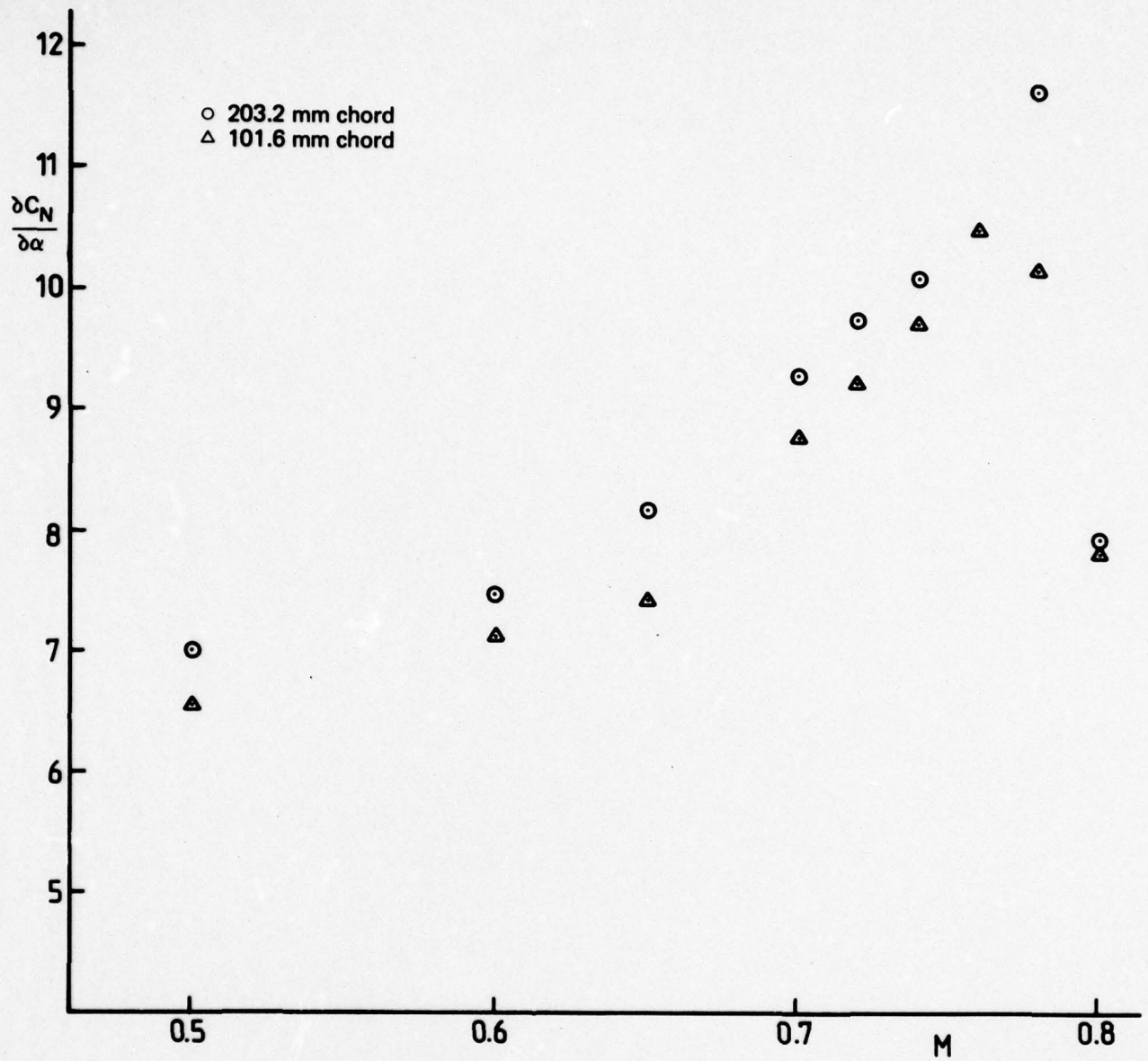


FIG. 14. VARIATION OF LIFT CURVE SLOPE WITH MACH NUMBER - BGK-1

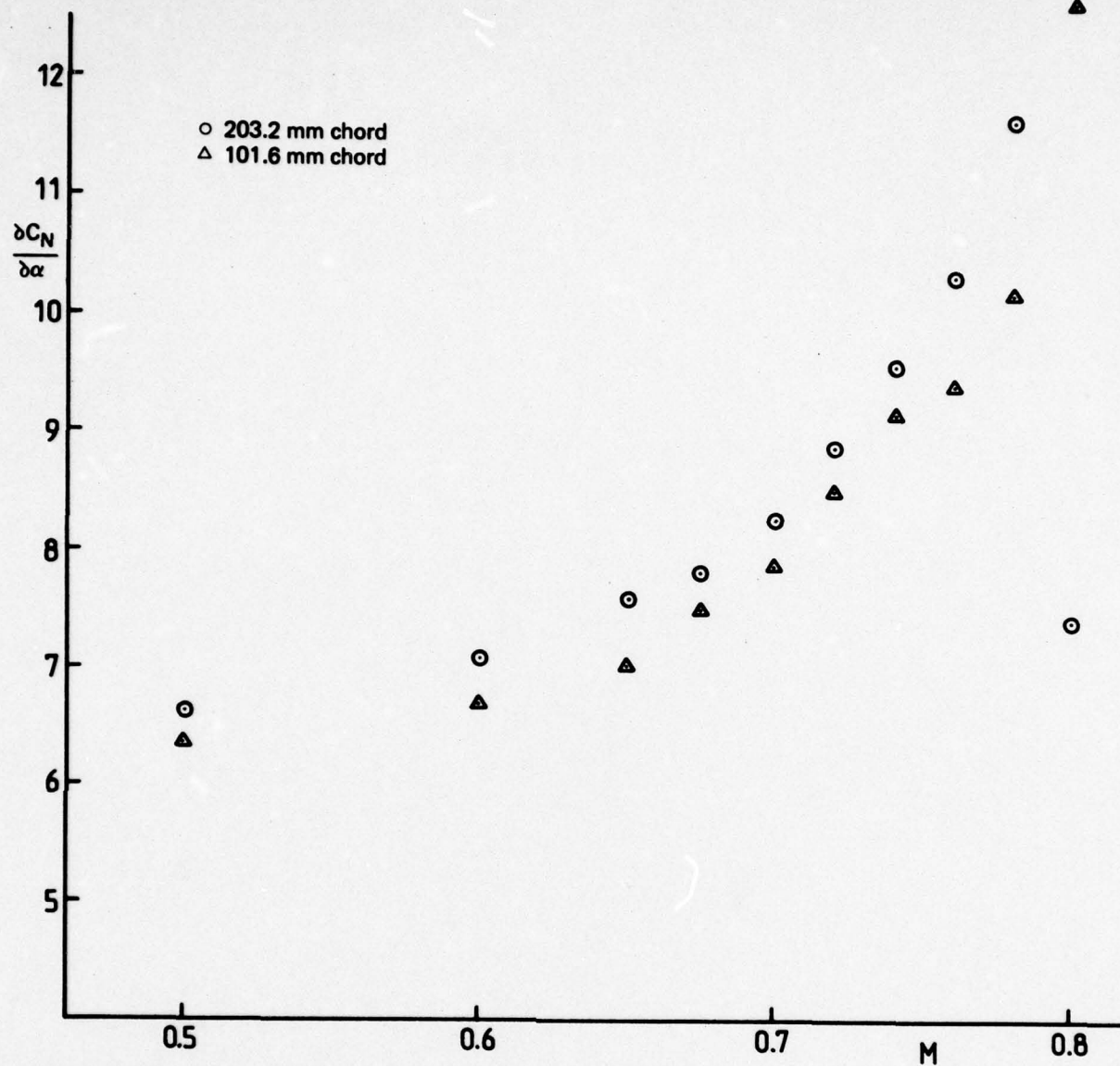
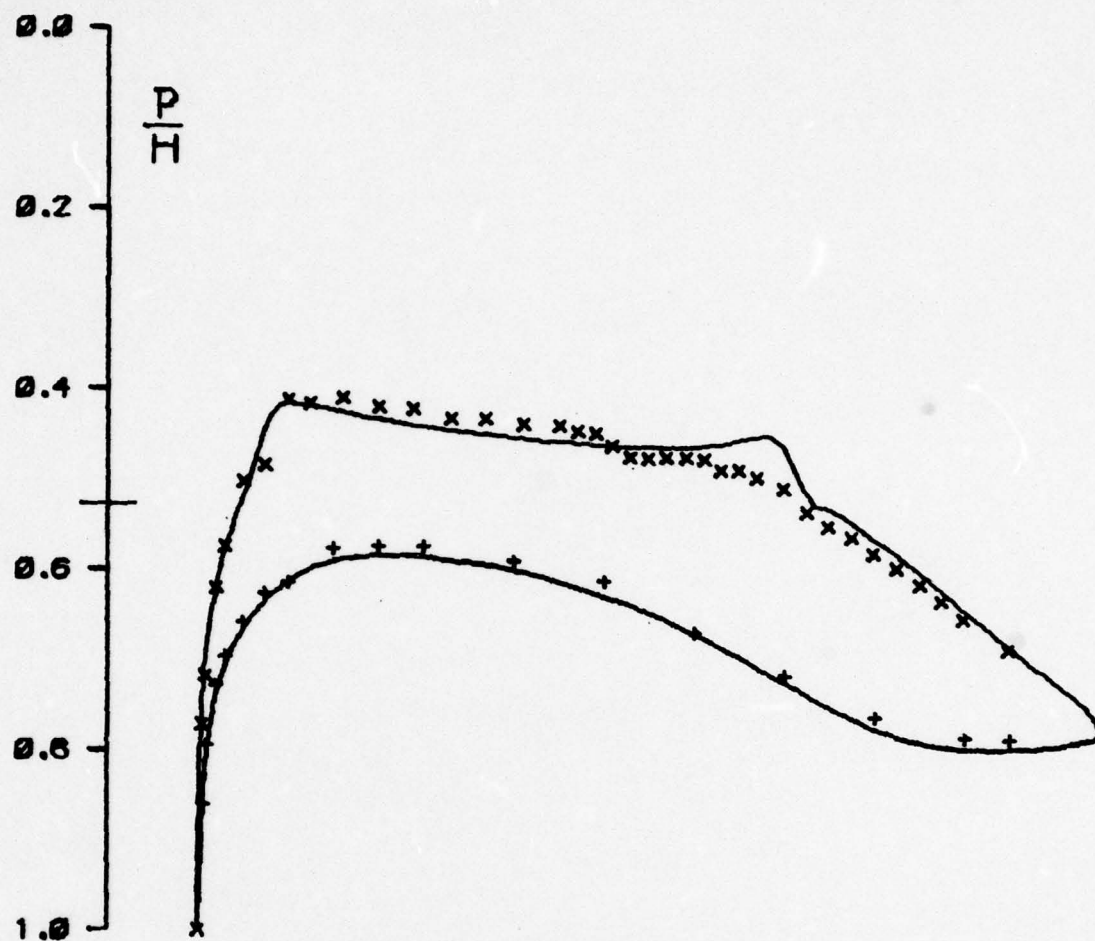


FIG. 15. VARIATION OF LIFT CURVE SLOPE WITH MACH NUMBER - NACA-0012



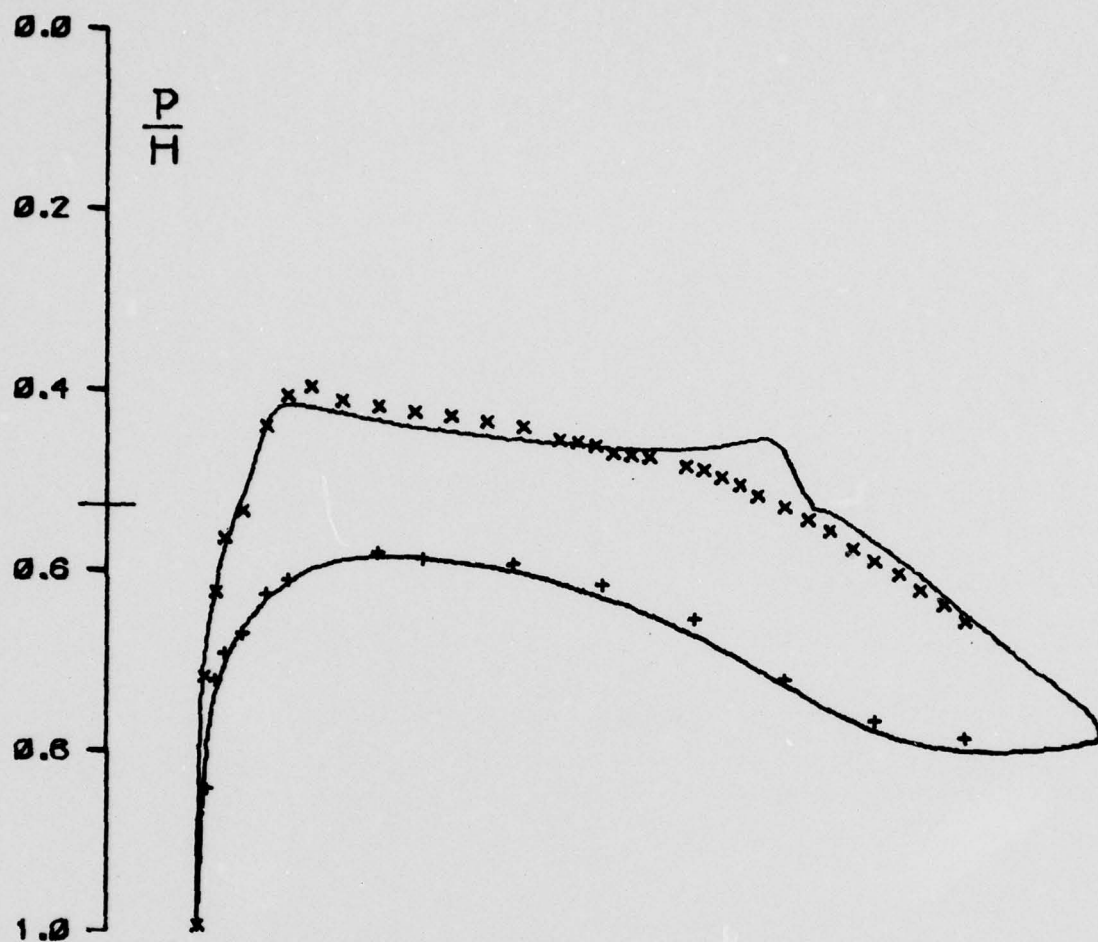
— THEORETICAL DESIGN DISTRIBUTION

$M = .750$ $AL = 0.00$ $CL = 0.612$ $CM = 0.147$ $CD = 0.002$ $NRN = 0$

+ x EXPERIMENT 203.2MM CHORD SOLID WALLS

$M = .755$ $AL = 0.56$ $CL = 0.536$ $CM = 0.118$ $R = 1.653$

FIG. 16



——— THEORETICAL DESIGN DISTRIBUTION
 M= .750 AL= 0.00 CL= 0.612 CM= -0.147 CD= 0.002 NR= 0
 + x EXPERIMENT 101.6MM CHORD SOLID WALLS
 M= .755 AL= 0.60 CM= 0.536 CN= -0.113 R= 0.632

FIG. 17

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ABSTRACT

Transonic wind tunnel tests are reported on a series of two-dimensional models in a test section equipped with four solid walls. The tests were conducted on two geometrically similar models of each of two aerofoil sections—NACA-0012 and BGK-1—and covered a range of Mach numbers between 0.5 and 0.82. The main purpose of the tests was to provide a body of data suitable for testing the validity of linear interference theory without the complication of the unknown boundary conditions associated with non-solid wind tunnel walls.

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APPENDIX A
TABLATED RESULTS

NOTATION

| | | |
|---------|--------------|--------------------------------------------------|
| ALPHA | (α) | Angle of incidence |
| CM | (C_m) | Pitching moment coefficient |
| CN | (C_N) | Normal force coefficient |
| CP | (C_p) | Pressure coefficient |
| MACH NO | (M) | Free stream Mach number |
| P H | (p/H) | Local static to free stream total pressure ratio |
| REY | (R) | Reynolds number based on chord |
| X C | (x/c) | Chordwise position |

Note: Symbols in brackets are the corresponding symbols used in the main part of the report.

CONTENTS

| No. | Model | M | n |
|-----|-----------------|-------|------|
| 1 | 203-2 mm BGK-1* | 0.750 | 0.55 |
| 2 | 203-2 mm BGK-1* | 0.749 | 0.55 |
| 3 | 203-2 mm BGK-1* | 0.749 | 0.55 |
| 4 | 203-2 mm BGK-1 | 0.50 | -2.5 |
| 5 | 203-2 mm BGK-1 | 0.50 | 0 |
| 6 | 203-2 mm BGK-1 | 0.50 | 1 |
| 7 | 203-2 mm BGK-1 | 0.50 | 2 |
| 8 | 203-2 mm BGK-1 | 0.50 | 3 |
| 9 | 203-2 mm BGK-1 | 0.50 | 4 |
| 10 | 203-2 mm BGK-1 | 0.60 | -2.5 |
| 11 | 203-2 mm BGK-1 | 0.60 | 0 |
| 12 | 203-2 mm BGK-1 | 0.60 | 1 |
| 13 | 203-2 mm BGK-1 | 0.60 | 2 |
| 14 | 203-2 mm BGK-1 | 0.60 | 3 |
| 15 | 203-2 mm BGK-1 | 0.60 | 4 |
| 16 | 203-2 mm BGK-1 | 0.65 | -2.4 |
| 17 | 203-2 mm BGK-1 | 0.65 | 0 |
| 18 | 203-2 mm BGK-1 | 0.65 | 1 |
| 19 | 203-2 mm BGK-1 | 0.65 | 2 |
| 20 | 203-2 mm BGK-1 | 0.65 | 3 |
| 21 | 203-2 mm BGK-1 | 0.70 | -2.5 |
| 22 | 203-2 mm BGK-1 | 0.70 | 0 |
| 23 | 203-2 mm BGK-1 | 0.70 | 1 |
| 24 | 203-2 mm BGK-1 | 0.70 | 2 |
| 25 | 203-2 mm BGK-1 | 0.70 | 3 |
| 26 | 203-2 mm BGK-1 | 0.72 | -2.5 |
| 27 | 203-2 mm BGK-1 | 0.72 | 0 |
| 28 | 203-2 mm BGK-1 | 0.72 | 1 |
| 29 | 203-2 mm BGK-1 | 0.72 | 2 |
| 30 | 203-2 mm BGK-1 | 0.74 | -2.5 |
| 31 | 203-2 mm BGK-1 | 0.74 | -1 |
| 32 | 203-2 mm BGK-1 | 0.74 | 0 |
| 33 | 203-2 mm BGK-1 | 0.74 | 1 |
| 34 | 203-2 mm BGK-1 | 0.74 | 2 |
| 35 | 203-2 mm BGK-1 | 0.76 | -2.5 |
| 36 | 203-2 mm BGK-1 | 0.76 | -1 |
| 37 | 203-2 mm BGK-1 | 0.76 | 0 |
| 38 | 203-2 mm BGK-1 | 0.76 | 1 |
| 39 | 203-2 mm BGK-1 | 0.76 | 2 |
| 40 | 203-2 mm BGK-1 | 0.78 | -2.5 |
| 41 | 203-2 mm BGK-1 | 0.78 | -1 |
| 42 | 203-2 mm BGK-1 | 0.78 | 0 |
| 43 | 203-2 mm BGK-1 | 0.78 | 1 |
| 44 | 203-2 mm BGK-1 | 0.78 | 2 |
| 45 | 203-2 mm BGK-1 | 0.80 | -2.5 |
| 46 | 203-2 mm BGK-1 | 0.80 | -1 |
| 47 | 203-2 mm BGK-1 | 0.80 | 0 |
| 48 | 203-2 mm BGK-1 | 0.80 | 1 |
| 49 | 203-2 mm BGK-1 | 0.80 | 2 |
| 50 | 101-6 mm BGK-1* | 0.753 | 0.60 |
| 51 | 101-6 mm BGK-1* | 0.752 | 0.60 |

* Free transverse.

| No. | Model | M | n |
|-----|--------------------|-------|------|
| 52 | 101-6 mm BGK-1* | 0-755 | 0-60 |
| 53 | 101-6 mm BGK-1 | 0-50 | -2-5 |
| 54 | 101-6 mm BGK-1 | 0-50 | 0 |
| 55 | 101-6 mm BGK-1 | 0-50 | 1 |
| 56 | 101-6 mm BGK-1 | 0-50 | 2 |
| 57 | 101-6 mm BGK-1 | 0-50 | 3 |
| 58 | 101-6 mm BGK-1 | 0-50 | 4 |
| 59 | 101-6 mm BGK-1 | 0-660 | -2-5 |
| 60 | 101-6 mm BGK-1 | 0-50 | 0 |
| 61 | 101-6 mm BGK-1 | 0-60 | 1 |
| 62 | 101-6 mm BGK-1 | 0-60 | 2 |
| 63 | 101-6 mm BGK-1 | 0-60 | 3 |
| 64 | 101-6 mm BGK-1 | 0-60 | 4 |
| 65 | 101-6 mm BGK-1 | 0-65 | -2-5 |
| 66 | 101-6 mm BGK-1 | 0-65 | 0 |
| 67 | 101-6 mm BGK-1 | 0-65 | 1 |
| 68 | 101-6 mm BGK-1 | 0-65 | 2 |
| 69 | 101-6 mm BGK-1 | 0-65 | 3 |
| 70 | 101-6 mm BGK-1 | 0-70 | -2-5 |
| 71 | 101-6 mm BGK-1 | 0-70 | 0 |
| 72 | 101-6 mm BGK-1 | 0-70 | 1 |
| 73 | 101-6 mm BGK-1 | 0-70 | 2 |
| 74 | 101-6 mm BGK-1 | 0-70 | 3 |
| 75 | 101-6 mm BGK-1 | 0-72 | -2-5 |
| 76 | 101-6 mm BGK-1 | 0-72 | 0 |
| 77 | 101-6 mm BGK-1 | 0-72 | 1 |
| 78 | 101-6 mm BGK-1 | 0-72 | 2 |
| 79 | 101-6 mm BGK-1 | 0-74 | -2-5 |
| 80 | 101-6 mm BGK-1 | 0-74 | -1 |
| 81 | 101-6 mm BGK-1 | 0-74 | 0 |
| 82 | 101-6 mm BGK-1 | 0-74 | 1 |
| 83 | 101-6 mm BGK-1 | 0-74 | 2 |
| 84 | 101-6 mm BGK-1 | 0-76 | -2-5 |
| 85 | 101-6 mm BGK-1 | 0-76 | -1 |
| 86 | 101-6 mm BGK-1 | 0-76 | 0 |
| 87 | 101-6 mm BGK-1 | 0-76 | 1 |
| 88 | 101-6 mm BGK-1 | 0-76 | 2 |
| 89 | 101-6 mm BGK-1 | 0-78 | -2-5 |
| 90 | 101-6 mm BGK-1 | 0-78 | -1 |
| 91 | 101-6 mm BGK-1 | 0-78 | 0 |
| 92 | 101-6 mm BGK-1 | 0-78 | 1 |
| 93 | 101-6 mm BGK-1 | 0-78 | 2 |
| 94 | 101-6 mm BGK-1 | 0-80 | -2-5 |
| 95 | 101-6 mm BGK-1 | 0-80 | -1 |
| 96 | 101-6 mm BGK-1 | 0-80 | 0 |
| 97 | 101-6 mm BGK-1 | 0-80 | 1 |
| 98 | 101-6 mm BGK-1 | 0-80 | 2 |
| 99 | 203-2 mm NACA-0012 | 0-50 | 0 |
| 100 | 203-2 mm NACA-0012 | 0-50 | 1 |
| 101 | 203-2 mm NACA-0012 | 0-50 | 2 |
| 102 | 203-2 mm NACA-0012 | 0-50 | 3 |

* Free transition.

| No. | Model | M | n |
|-----|--------------------|-------|-----|
| 103 | 203-2 mm NACA-0012 | 0-50 | 4 |
| 104 | 203-2 mm NACA-0012 | 0-60 | 0 |
| 105 | 203-2 mm NACA-0012 | 0-60 | 1 |
| 106 | 203-2 mm NACA-0012 | 0-60 | 2 |
| 107 | 203-2 mm NACA-0012 | 0-60 | 3 |
| 108 | 203-2 mm NACA-0012 | 0-65 | 0 |
| 109 | 203-2 mm NACA-0012 | 0-65 | 1 |
| 110 | 203-2 mm NACA-0012 | 0-65 | 2 |
| 111 | 203-2 mm NACA-0012 | 0-65 | 3 |
| 112 | 203-2 mm NACA-0012 | 0-675 | 0 |
| 113 | 203-2 mm NACA-0012 | 0-675 | 1 |
| 114 | 203-2 mm NACA-0012 | 0-675 | 2 |
| 115 | 203-2 mm NACA-0012 | 0-70 | 0 |
| 116 | 203-2 mm NACA-0012 | 0-70 | 1 |
| 117 | 203-2 mm NACA-0012 | 0-70 | 2 |
| 118 | 203-2 mm NACA-0012 | 0-72 | 0 |
| 119 | 203-2 mm NACA-0012 | 0-72 | 1 |
| 120 | 203-2 mm NACA-0012 | 0-72 | 2 |
| 121 | 203-2 mm NACA-0012 | 0-74 | 0 |
| 122 | 203-2 mm NACA-0012 | 0-74 | 1 |
| 123 | 203-2 mm NACA-0012 | 0-74 | 2 |
| 124 | 203-2 mm NACA-0012 | 0-76 | 0 |
| 125 | 203-2 mm NACA-0012 | 0-76 | 1 |
| 126 | 203-2 mm NACA-0012 | 0-76 | 2 |
| 127 | 203-2 mm NACA-0012 | 0-78 | 0 |
| 128 | 203-2 mm NACA-0012 | 0-78 | 0-5 |
| 129 | 203-2 mm NACA-0012 | 0-78 | 1 |
| 130 | 203-2 mm NACA-0012 | 0-80 | 0 |
| 131 | 203-2 mm NACA-0012 | 0-80 | 0-5 |
| 132 | 203-2 mm NACA-0012 | 0-80 | 1 |
| 133 | 203-2 mm NACA-0012 | 0-82 | 0 |
| 134 | 203-2 mm NACA-0012 | 0-82 | 0-5 |
| 135 | 203-2 mm NACA-0012 | 0-82 | 1-0 |
| 136 | 101-6 mm NACA-0012 | 0-50 | 0 |
| 137 | 101-6 mm NACA-0012 | 0-50 | 1 |
| 138 | 101-6 mm NACA-0012 | 0-50 | 2 |
| 139 | 101-6 mm NACA-0012 | 0-50 | 3 |
| 140 | 101-6 mm NACA-0012 | 0-50 | 4 |
| 141 | 101-6 mm NACA-0012 | 0-60 | 0 |
| 142 | 101-6 mm NACA-0012 | 0-60 | 1 |
| 143 | 101-6 mm NACA-0012 | 0-60 | 2 |
| 144 | 101-6 mm NACA-0012 | 0-60 | 3 |
| 145 | 101-6 mm NACA-0012 | 0-65 | 0 |
| 146 | 101-6 mm NACA-0012 | 0-65 | 1 |
| 147 | 101-6 mm NACA-0012 | 0-65 | 2 |
| 148 | 101-6 mm NACA-0012 | 0-65 | 3 |
| 149 | 101-6 mm NACA-0012 | 0-675 | 0 |
| 150 | 101-6 mm NACA-0012 | 0-675 | 1 |
| 151 | 101-6 mm NACA-0012 | 0-675 | 2 |
| 152 | 101-6 mm NACA-0012 | 0-70 | 0 |
| 153 | 101-6 mm NACA-0012 | 0-70 | 1 |

| No. | Model | M | n |
|-----|--------------------|------|-----|
| 154 | 101-6 mm NACA-0012 | 0.70 | 2 |
| 155 | 101-6 mm NACA-0012 | 0.72 | 0 |
| 156 | 101-6 mm NACA-0012 | 0.72 | 1 |
| 157 | 101-6 mm NACA-0012 | 0.72 | 2 |
| 158 | 101-6 mm NACA-0012 | 0.74 | 0 |
| 159 | 101-6 mm NACA-0012 | 0.74 | 1 |
| 160 | 101-6 mm NACA-0012 | 0.74 | 2 |
| 161 | 101-6 mm NACA-0012 | 0.76 | 0 |
| 162 | 101-6 mm NACA-0012 | 0.76 | 1 |
| 163 | 101-6 mm NACA-0012 | 0.76 | 2 |
| 164 | 101-6 mm NACA-0012 | 0.78 | 0 |
| 165 | 101-6 mm NACA-0012 | 0.78 | 0.5 |
| 166 | 101-6 mm NACA-0012 | 0.78 | 1 |
| 167 | 101-6 mm NACA-0012 | 0.80 | 0 |
| 168 | 101-6 mm NACA-0012 | 0.80 | 0.5 |
| 169 | 101-6 mm NACA-0012 | 0.80 | 1 |
| 170 | 101-6 mm NACA-0012 | 0.82 | 0 |
| 171 | 101-6 mm NACA-0012 | 0.82 | 0.5 |
| 172 | 101-6 mm NACA-0012 | 0.82 | 1 |

SEC 1 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.752 ALPHA 0.55 REY 1.66×10^6

INTEGRATED FORCE COEFFICIENTS

$C_L = 2.5549$ $C_M = -3.1381$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 1.146 | 0.999 | 0.0000 | 1.146 | 0.999 |
| 0.0035 | 0.302 | 2.773 | 0.0035 | 0.633 | 2.853 |
| 0.0079 | 0.193 | 2.715 | 0.0093 | 2.391 | 2.744 |
| 0.0196 | -0.253 | 3.627 | 0.0201 | 0.146 | 2.728 |
| 0.0294 | -0.421 | 3.174 | 0.0306 | 0.028 | 2.696 |
| 0.0495 | -0.691 | 3.521 | 0.0520 | -0.114 | 2.657 |
| 0.0741 | -0.753 | 2.482 | 0.0749 | -0.238 | 2.626 |
| 0.0991 | -1.119 | 2.412 | 0.1003 | -0.275 | 2.614 |
| 0.1241 | -1.156 | 2.415 | 0.1498 | -0.479 | 2.577 |
| 0.1591 | -1.234 | 2.425 | 0.2300 | -0.426 | 2.573 |
| 0.1994 | -1.103 | 2.416 | 0.2499 | -0.426 | 2.573 |
| 0.2352 | -0.944 | 2.419 | 0.3502 | -0.358 | 2.558 |
| 0.2794 | -0.973 | 2.424 | 0.4520 | -0.256 | 2.610 |
| 0.3198 | -0.977 | 2.423 | 0.5490 | -0.093 | 2.606 |
| 0.3594 | -0.957 | 2.423 | 0.6492 | 0.099 | 2.715 |
| 0.3993 | -0.905 | 2.426 | 0.7498 | 0.268 | 2.761 |
| 0.4494 | -0.947 | 2.431 | 0.8499 | 2.353 | 2.744 |
| 0.4995 | -0.957 | 2.429 | 0.9000 | 0.358 | 2.745 |
| 0.5505 | -0.921 | 2.433 | | | |
| 0.6007 | -0.723 | 2.436 | | | |
| 0.6503 | -0.549 | 2.444 | | | |
| 0.7004 | -0.474 | 2.451 | | | |
| 0.7506 | -0.381 | 2.454 | | | |
| 0.8002 | -0.257 | 2.456 | | | |
| 0.8507 | -0.136 | 2.461 | | | |
| 0.9009 | -0.034 | 2.462 | | | |
| 0.9510 | -0.102 | 2.471 | | | |
| 1.0014 | -0.741 | 2.487 | | | |
| 1.0517 | -1.527 | 2.545 | | | |
| 1.1019 | -3.496 | 2.554 | | | |
| 1.1527 | -0.457 | 2.564 | | | |
| 1.2035 | -0.443 | 2.579 | | | |
| 1.2543 | -0.315 | 2.595 | | | |
| 1.3051 | -0.283 | 2.611 | | | |
| 1.3559 | -0.271 | 2.625 | | | |
| 1.4067 | -0.157 | 2.648 | | | |
| 1.4575 | -0.113 | 2.665 | | | |

MODEL 223.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

WACH NO. 4.749 ALPHA 0.55 REY 1.66*10⁶

INTEGRATED FORCE COEFFICIENTS

C_L = 0.5655 C_M = -0.1263

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.145 | 2.999 | 0.0000 | 1.145 | 2.999 |
| 0.0050 | 0.301 | 2.771 | 0.0051 | 0.626 | 0.859 |
| 0.0100 | 0.100 | 2.716 | 0.0100 | 2.393 | 2.705 |
| 0.0150 | -0.204 | 2.623 | 0.0201 | 2.143 | 2.728 |
| 0.0200 | -0.426 | 2.574 | 0.0300 | 2.028 | 2.675 |
| 0.0250 | -0.645 | 2.501 | 0.0400 | -0.110 | 2.607 |
| 0.0300 | -0.704 | 2.492 | 0.0500 | -0.234 | 2.626 |
| 0.0350 | -1.124 | 2.412 | 0.1000 | -0.280 | 2.614 |
| 0.0400 | -1.012 | 2.416 | 0.1498 | -0.417 | 2.576 |
| 0.0450 | -1.135 | 2.409 | 0.2000 | -0.437 | 2.573 |
| 0.0500 | -1.200 | 2.418 | 0.2499 | -0.429 | 2.573 |
| 0.0550 | -1.056 | 2.423 | 0.3500 | -0.369 | 2.509 |
| 0.0600 | -0.956 | 2.431 | 0.4500 | -0.290 | 2.611 |
| 0.0650 | -0.952 | 2.432 | 0.5499 | -0.083 | 2.607 |
| 0.0700 | -0.736 | 2.436 | 0.6492 | 0.097 | 2.715 |
| 0.0750 | -0.903 | 2.445 | 0.7498 | 0.267 | 2.751 |
| 0.0800 | -0.901 | 2.446 | 0.8499 | 0.353 | 2.755 |
| 0.0850 | -0.910 | 2.443 | 0.9000 | 0.358 | 2.706 |
| 0.0900 | -0.354 | 2.455 | | | |
| 0.0950 | -0.316 | 2.468 | | | |
| 0.1000 | -0.412 | 2.473 | | | |
| 0.1050 | -0.501 | 2.472 | | | |
| 0.1100 | -0.438 | 2.462 | | | |
| 0.1150 | -0.529 | 2.465 | | | |
| 0.1200 | -0.403 | 2.472 | | | |
| 0.1250 | -0.707 | 2.476 | | | |
| 0.1300 | -0.759 | 2.484 | | | |
| 0.1350 | -0.574 | 2.507 | | | |
| 0.1400 | -0.503 | 2.537 | | | |
| 0.1450 | -0.514 | 2.550 | | | |
| 0.1500 | -0.407 | 2.563 | | | |
| 0.1550 | -0.408 | 2.579 | | | |
| 0.1600 | -0.350 | 2.595 | | | |
| 0.1650 | -0.287 | 2.512 | | | |
| 0.1700 | -0.242 | 2.629 | | | |
| 0.1750 | -0.147 | 2.650 | | | |
| 0.1800 | -0.015 | 2.685 | | | |

SEC 1 223.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

SOLID WALLS

MACH NO. 0.749 ALPHA 0.55 REY 1.66×10^6

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.5455$ $C_M = -0.1321$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.145 | 2.999 | 0.0000 | 1.145 | 2.999 |
| 0.0035 | 0.296 | 2.762 | 0.0051 | 0.635 | 2.860 |
| 0.0079 | 0.348 | 2.715 | 0.0098 | 0.397 | 2.746 |
| 0.0196 | -0.298 | 2.519 | 0.0221 | 0.149 | 2.725 |
| 0.0294 | -0.429 | 2.572 | 0.0332 | 0.025 | 2.645 |
| 0.0495 | -0.645 | 2.520 | 0.0522 | -0.116 | 2.556 |
| 0.0741 | -0.753 | 2.482 | 0.0749 | -0.231 | 2.623 |
| 0.0991 | -1.023 | 2.411 | 0.1000 | -0.275 | 2.613 |
| 0.1241 | -1.309 | 2.415 | 0.1493 | -0.413 | 2.576 |
| 0.1491 | -1.596 | 2.438 | 0.2000 | -0.426 | 2.572 |
| 0.1994 | -1.704 | 2.417 | 0.2499 | -0.427 | 2.572 |
| 0.2342 | -0.946 | 2.418 | 0.3522 | -0.367 | 2.588 |
| 0.2794 | -0.972 | 2.425 | 0.4522 | -0.238 | 2.629 |
| 0.3155 | -0.977 | 2.424 | 0.5493 | -0.091 | 2.666 |
| 0.3594 | -0.959 | 2.428 | 0.6492 | 0.108 | 2.715 |
| 0.3993 | -0.954 | 2.427 | 0.7495 | 0.270 | 2.761 |
| 0.4494 | -0.956 | 2.429 | 0.8499 | 0.357 | 2.765 |
| 0.4995 | -0.961 | 2.428 | 0.9503 | 0.361 | 2.766 |
| 0.5495 | -0.964 | 2.435 | | | |
| 0.5987 | -0.962 | 2.438 | | | |
| 0.6483 | -0.960 | 2.439 | | | |
| 0.6984 | -0.958 | 2.445 | | | |
| 0.7486 | -0.945 | 2.451 | | | |
| 0.7992 | -0.939 | 2.455 | | | |
| 0.8497 | -0.930 | 2.457 | | | |
| 0.8999 | -0.943 | 2.460 | | | |
| 0.9503 | -0.962 | 2.473 | | | |
| 0.9994 | -0.976 | 2.493 | | | |
| 1.0497 | -1.529 | 2.544 | | | |
| 1.0999 | -0.464 | 2.557 | | | |
| 1.1499 | -0.449 | 2.566 | | | |
| 1.1999 | -0.393 | 2.581 | | | |
| 1.2499 | -0.339 | 2.596 | | | |
| 1.2999 | -0.291 | 2.612 | | | |
| 1.3499 | -0.241 | 2.625 | | | |
| 1.3999 | -0.187 | 2.648 | | | |
| 1.4499 | -0.114 | 2.684 | | | |

NO 1 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.508 ALPHA -2.92 REY 1.66e18⁶

INTEGRATED FORCE COEFFICIENTS

CX = -0.0313 CY = -0.0330

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 0.697 | 3.973 | 0.9360 | 0.857 | 3.972 |
| 0.0035 | 0.893 | 3.973 | 0.9351 | -0.441 | 0.778 |
| 0.0079 | 0.696 | 2.940 | 0.9399 | -0.672 | 3.744 |
| 0.0196 | 0.316 | 3.893 | 0.9281 | -0.727 | 0.736 |
| 0.0294 | 0.193 | 2.866 | 0.9387 | -0.666 | 0.742 |
| 0.0495 | 0.302 | 2.643 | 0.9502 | -0.736 | 0.739 |
| 0.0741 | -0.197 | 2.621 | 0.9749 | -0.772 | 0.742 |
| 0.0941 | -0.205 | 2.612 | 0.9880 | -0.652 | 0.747 |
| 0.1241 | -0.173 | 2.617 | 0.9498 | -0.657 | 0.746 |
| 0.1491 | -0.216 | 2.611 | 0.9262 | -0.589 | 0.756 |
| 0.1794 | -0.240 | 2.614 | 0.9499 | -0.542 | 0.751 |
| 0.2332 | -0.243 | 2.627 | 0.9542 | -0.441 | 0.779 |
| 0.2744 | -0.278 | 2.625 | 0.9508 | -0.328 | 3.745 |
| 0.3188 | -0.280 | 2.622 | 0.9490 | -0.177 | 0.817 |
| 0.3544 | -0.291 | 2.621 | 0.9492 | -0.072 | 0.843 |
| 0.3943 | -0.296 | 2.799 | 0.9498 | 0.145 | 0.855 |
| 0.4194 | -0.295 | 2.799 | 0.9479 | 0.272 | 0.876 |
| 0.4495 | -0.307 | 2.798 | 0.9320 | 0.339 | 0.872 |
| 0.4581 | -0.311 | 2.797 | | | |
| 0.4787 | -0.318 | 2.796 | | | |
| 0.4983 | -0.326 | 2.795 | | | |
| 0.5184 | -0.331 | 2.794 | | | |
| 0.5395 | -0.334 | 2.794 | | | |
| 0.5542 | -0.337 | 2.793 | | | |
| 0.5747 | -0.340 | 2.793 | | | |
| 0.5939 | -0.350 | 2.791 | | | |
| 0.6140 | -0.355 | 2.791 | | | |
| 0.6484 | -0.345 | 2.792 | | | |
| 0.6747 | -0.323 | 2.795 | | | |
| 0.6959 | -0.308 | 2.797 | | | |
| 0.7257 | -0.288 | 2.801 | | | |
| 0.7486 | -0.259 | 2.825 | | | |
| 0.7734 | -0.227 | 2.829 | | | |
| 0.7993 | -0.193 | 2.815 | | | |
| 0.8241 | -0.155 | 2.820 | | | |
| 0.8449 | -0.111 | 2.827 | | | |
| 0.8689 | -0.014 | 2.841 | | | |

PG 1 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

ACH NO. 0.541 ALP4A 0.02 REY 1.66×10^6

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.2763$ $C_M = -0.2835$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.0000 | 1.002 | 2.999 | 0.0000 | 1.052 | 2.998 |
| 2.0035 | 0.219 | 2.675 | 0.0051 | 0.456 | 0.910 |
| 2.0079 | 0.021 | 2.846 | 0.0098 | 0.206 | 0.873 |
| 2.0196 | -0.295 | 2.800 | 0.0201 | -0.905 | 0.842 |
| 2.0274 | -0.410 | 2.781 | 0.0300 | -0.068 | 0.833 |
| 2.0395 | -0.467 | 2.773 | 0.0500 | -0.195 | 0.814 |
| 2.0741 | -0.574 | 2.757 | 0.0749 | -0.272 | 0.802 |
| 2.0991 | -0.577 | 2.757 | 0.1000 | -0.283 | 0.801 |
| 2.1241 | -0.466 | 2.773 | 0.1498 | -0.355 | 0.770 |
| 2.1591 | -0.492 | 2.777 | 0.2000 | -0.344 | 0.772 |
| 2.1994 | -0.408 | 2.775 | 0.2499 | -0.337 | 0.773 |
| 2.2392 | -0.409 | 2.775 | 0.3002 | -0.290 | 0.806 |
| 2.2794 | -0.401 | 2.776 | 0.4508 | -0.220 | 0.812 |
| 2.3158 | -0.405 | 2.775 | 0.5498 | -0.102 | 0.828 |
| 2.3594 | -0.442 | 2.777 | 0.6492 | 0.046 | 0.849 |
| 2.3993 | -0.443 | 2.777 | 0.7498 | 0.189 | 0.871 |
| 2.4194 | -0.438 | 2.778 | 0.8499 | 0.258 | 0.891 |
| 2.4395 | -0.441 | 2.777 | 0.9000 | 0.267 | 0.892 |
| 2.4585 | -0.439 | 2.778 | | | |
| 2.4787 | -0.441 | 2.777 | | | |
| 2.4943 | -0.444 | 2.777 | | | |
| 2.5184 | -0.438 | 2.775 | | | |
| 2.5396 | -0.438 | 2.778 | | | |
| 2.5592 | -0.439 | 2.778 | | | |
| 2.5787 | -0.436 | 2.779 | | | |
| 2.5939 | -0.437 | 2.776 | | | |
| 2.6152 | -0.438 | 2.775 | | | |
| 2.6454 | -0.425 | 2.783 | | | |
| 2.6747 | -0.394 | 2.784 | | | |
| 2.6999 | -0.372 | 2.782 | | | |
| 2.7237 | -0.347 | 2.791 | | | |
| 2.7433 | -0.312 | 2.796 | | | |
| 2.7738 | -0.275 | 2.822 | | | |
| 2.7998 | -0.232 | 2.828 | | | |
| 2.8248 | -0.156 | 2.815 | | | |
| 2.8459 | -0.135 | 2.823 | | | |
| 2.8989 | -0.091 | 2.838 | | | |

PGC 1 233.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

SOLID WALLS

MACH NO. 0.499 ALPHA 1.82 REY 1.66*10⁶

INTEGRATED FORCE COEFFICIENTS

C_x = 0.3945 C_y = -2.0630

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.3322 | 1.338 | 2.996 | 0.8888 | 1.838 | 2.996 |
| 2.3335 | -2.162 | 2.519 | 0.8351 | 0.662 | 2.943 |
| 2.2379 | -2.314 | 2.796 | 0.8298 | 0.451 | 2.929 |
| 2.2196 | -0.505 | 2.756 | 0.8231 | 0.219 | 2.875 |
| 2.2294 | -2.651 | 2.742 | 0.8320 | 2.125 | 2.951 |
| 2.2495 | -0.715 | 2.737 | 0.8588 | -0.028 | 2.839 |
| 2.2741 | -3.756 | 2.732 | 0.8749 | -0.125 | 2.824 |
| 2.2991 | -2.740 | 2.734 | 0.1003 | -0.154 | 2.823 |
| 2.1241 | -0.616 | 2.752 | 0.1498 | -0.247 | 2.816 |
| 2.1591 | -3.585 | 2.753 | 0.2022 | -0.252 | 2.816 |
| 2.1394 | -0.557 | 2.761 | 0.2499 | -0.260 | 2.814 |
| 2.2382 | -2.544 | 2.762 | 0.3532 | -0.234 | 2.818 |
| 2.2794 | -1.531 | 2.764 | 0.4502 | -2.178 | 2.816 |
| 2.3198 | -2.567 | 2.765 | 0.5490 | -0.074 | 2.832 |
| 2.3594 | -0.5 | 2.768 | 0.6492 | 0.065 | 2.852 |
| 2.3993 | -0.501 | 2.767 | 0.7499 | 2.226 | 2.873 |
| 2.4194 | -3.491 | 2.773 | 0.8499 | 0.269 | 2.853 |
| 2.4395 | -3.493 | 2.772 | 0.9323 | 0.278 | 2.854 |
| 2.4595 | -2.454 | 2.771 | | | |
| 2.4797 | -2.464 | 2.772 | | | |
| 2.4953 | -2.483 | 2.771 | | | |
| 2.5184 | -2.484 | 2.771 | | | |
| 2.5396 | -2.482 | 2.772 | | | |
| 2.5592 | -2.483 | 2.772 | | | |
| 2.5787 | -2.475 | 2.772 | | | |
| 2.5989 | -2.475 | 2.772 | | | |
| 2.6183 | -2.472 | 2.772 | | | |
| 2.6484 | -2.451 | 2.776 | | | |
| 2.6747 | -2.417 | 2.781 | | | |
| 2.6959 | -2.393 | 2.784 | | | |
| 2.7237 | -2.363 | 2.789 | | | |
| 2.7488 | -2.325 | 2.795 | | | |
| 2.7733 | -2.292 | 2.801 | | | |
| 2.7992 | -2.241 | 2.807 | | | |
| 2.8243 | -2.193 | 2.814 | | | |
| 2.8489 | -2.144 | 2.822 | | | |
| 2.8989 | -2.339 | 2.837 | | | |

364 1 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

TACH NO. 0.479 ALPHA 2.86 REY 1.46E18⁶

INTEGRATED FORCE COEFFICIENTS

C_D = 0.5313 C_M = -0.3029

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 0.945 | 2.962 | 0.0000 | 0.945 | 0.962 |
| 0.0035 | -0.632 | 2.753 | 0.0051 | 0.886 | 0.974 |
| 0.0079 | -0.714 | 2.738 | 0.0098 | 0.691 | 0.945 |
| 0.0196 | -0.933 | 2.705 | 0.0201 | 0.498 | 0.929 |
| 0.0294 | -0.906 | 2.698 | 0.0302 | 0.340 | 0.893 |
| 0.0495 | -0.906 | 2.781 | 0.0520 | 0.165 | 2.947 |
| 0.0741 | -0.876 | 2.699 | 0.0749 | 0.047 | 0.858 |
| 0.0991 | -0.914 | 2.738 | 0.1078 | -0.072 | 0.845 |
| 0.1241 | -0.705 | 2.737 | 0.1498 | -0.113 | 0.827 |
| 0.1591 | -0.728 | 2.736 | 0.2000 | -0.139 | 0.823 |
| 0.1994 | -0.862 | 2.745 | 0.2499 | -0.159 | 0.822 |
| 0.2382 | -0.637 | 2.749 | 0.3522 | -0.157 | 0.828 |
| 0.2744 | -0.609 | 2.753 | 0.4520 | -0.119 | 0.826 |
| 0.3188 | -0.543 | 2.756 | 0.5490 | -0.031 | 0.839 |
| 0.3594 | -0.509 | 2.759 | 0.6492 | 0.097 | 0.858 |
| 0.3983 | -0.557 | 2.761 | 0.7498 | 0.228 | 0.877 |
| 0.4194 | -0.548 | 2.762 | 0.8499 | 0.287 | 0.886 |
| 0.4195 | -0.545 | 2.763 | 0.9300 | 0.298 | 0.886 |
| 0.4585 | -0.541 | 2.763 | | | |
| 0.4757 | -0.535 | 2.764 | | | |
| 0.4963 | -0.534 | 2.764 | | | |
| 0.5194 | -0.528 | 2.765 | | | |
| 0.5395 | -0.520 | 2.766 | | | |
| 0.5592 | -0.515 | 2.767 | | | |
| 0.5757 | -0.513 | 2.767 | | | |
| 0.5939 | -0.512 | 2.768 | | | |
| 0.6190 | -0.506 | 2.768 | | | |
| 0.6454 | -0.491 | 2.772 | | | |
| 0.6747 | -0.445 | 2.777 | | | |
| 0.6999 | -0.412 | 2.782 | | | |
| 0.7237 | -0.392 | 2.787 | | | |
| 0.7455 | -0.337 | 2.793 | | | |
| 0.7735 | -0.294 | 2.803 | | | |
| 0.7998 | -0.248 | 2.806 | | | |
| 0.8243 | -0.199 | 2.814 | | | |
| 0.8450 | -0.154 | 2.822 | | | |
| 0.8789 | -0.248 | 2.837 | | | |

#C< 1 283.2 MM CHORD
 EXPERIMENTAL PRESSURE DISTRIBUTION
 SOLID WALLS

MACH NO. 0.479 ALPHA 4.00 REY 1.66e18⁶

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.7533$ $C_M = -0.0775$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 0.636 | 2.937 | 0.0000 | 0.636 | 2.937 |
| 0.0035 | -1.829 | 2.572 | 0.0051 | 1.834 | 2.996 |
| 0.0079 | -1.691 | 2.596 | 0.0090 | 0.933 | 2.901 |
| 0.0146 | -1.748 | 2.579 | 0.0201 | 0.713 | 2.946 |
| 0.0294 | -1.676 | 2.597 | 0.0380 | 0.592 | 2.938 |
| 0.0495 | -1.552 | 2.613 | 0.0503 | 0.401 | 2.902 |
| 0.0741 | -1.417 | 2.635 | 0.0749 | 0.261 | 2.902 |
| 0.0991 | -1.273 | 2.656 | 0.1000 | 0.100 | 2.871 |
| 0.1241 | -1.102 | 2.687 | 0.1498 | 0.049 | 2.850 |
| 0.1591 | -0.977 | 2.703 | 0.2000 | -0.034 | 2.842 |
| 0.1994 | -0.871 | 2.715 | 0.2499 | -0.037 | 2.837 |
| 0.2382 | -0.813 | 2.724 | 0.3572 | -0.055 | 2.833 |
| 0.2794 | -0.767 | 2.730 | 0.4500 | -0.050 | 2.835 |
| 0.3153 | -0.736 | 2.735 | 0.5490 | 0.017 | 2.845 |
| 0.3594 | -0.697 | 2.741 | 0.6492 | 0.120 | 2.802 |
| 0.3993 | -0.669 | 2.745 | 0.7498 | 0.253 | 2.800 |
| 0.4194 | -0.653 | 2.747 | 0.8499 | 0.303 | 2.837 |
| 0.4395 | -0.642 | 2.749 | 0.9300 | 0.302 | 2.857 |
| 0.4565 | -0.632 | 2.753 | | | |
| 0.4757 | -0.622 | 2.752 | | | |
| 0.4953 | -0.618 | 2.752 | | | |
| 0.5154 | -0.605 | 2.754 | | | |
| 0.5396 | -0.545 | 2.756 | | | |
| 0.5592 | -0.503 | 2.758 | | | |
| 0.5797 | -0.573 | 2.759 | | | |
| 0.5999 | -0.506 | 2.760 | | | |
| 0.6103 | -0.508 | 2.761 | | | |
| 0.6454 | -0.526 | 2.766 | | | |
| 0.6747 | -0.403 | 2.772 | | | |
| 0.6989 | -0.449 | 2.777 | | | |
| 0.7237 | -0.410 | 2.783 | | | |
| 0.7488 | -0.300 | 2.790 | | | |
| 0.7738 | -0.311 | 2.794 | | | |
| 0.7990 | -0.259 | 2.805 | | | |
| 0.8240 | -0.245 | 2.813 | | | |
| 0.8489 | -0.149 | 2.821 | | | |
| 0.8690 | -0.050 | 2.836 | | | |

#64 1 233.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.593 ALPHA -2.53 REYNOLDS 1,640,120

INTEGRATED FORCE COEFFICIENTS

C_D = -3.8529 C_M = -8.8861

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 0.889 | 2.960 | 0.0000 | 0.889 | 0.953 |
| 0.0035 | 0.885 | 2.963 | 0.0051 | -0.435 | 2.699 |
| 0.0079 | 0.899 | 2.922 | 0.0098 | -0.733 | 0.646 |
| 0.0196 | 0.360 | 2.856 | 0.0231 | -0.785 | 2.632 |
| 0.0294 | 0.198 | 2.822 | 0.0328 | -0.748 | 2.637 |
| 0.0495 | 0.025 | 2.793 | 0.0503 | -0.797 | 0.623 |
| 0.0741 | -0.143 | 2.757 | 0.0749 | -0.799 | 0.627 |
| 0.0991 | -0.211 | 2.743 | 0.1000 | -0.742 | 2.638 |
| 0.1241 | -0.176 | 2.750 | 0.1498 | -0.756 | 0.636 |
| 0.1591 | -0.227 | 2.740 | 0.2000 | -0.672 | 0.652 |
| 0.1994 | -0.227 | 2.743 | 0.2499 | -0.612 | 0.664 |
| 0.2382 | -0.255 | 2.734 | 0.3502 | -0.492 | 2.658 |
| 0.2794 | -0.273 | 2.731 | 0.4500 | -0.368 | 2.714 |
| 0.3158 | -0.242 | 2.727 | 0.5498 | -0.190 | 2.747 |
| 0.3594 | -0.247 | 2.726 | 0.6492 | -0.003 | 0.754 |
| 0.3993 | -0.317 | 2.722 | 0.7498 | 0.145 | 0.813 |
| 0.4194 | -0.318 | 2.722 | 0.8499 | 0.225 | 2.829 |
| 0.4395 | -0.331 | 2.719 | 0.9288 | 0.244 | 0.833 |
| 0.4585 | -0.334 | 2.719 | | | |
| 0.4787 | -0.343 | 2.717 | | | |
| 0.4933 | -0.354 | 2.715 | | | |
| 0.5184 | -0.362 | 2.714 | | | |
| 0.5396 | -0.363 | 2.713 | | | |
| 0.5592 | -0.369 | 2.712 | | | |
| 0.5787 | -0.371 | 2.711 | | | |
| 0.5989 | -0.379 | 2.710 | | | |
| 0.6183 | -0.381 | 2.709 | | | |
| 0.6484 | -0.376 | 2.713 | | | |
| 0.6747 | -0.351 | 2.715 | | | |
| 0.6989 | -0.333 | 2.719 | | | |
| 0.7237 | -0.312 | 2.723 | | | |
| 0.7488 | -0.278 | 2.733 | | | |
| 0.7738 | -0.241 | 2.737 | | | |
| 0.7992 | -0.227 | 2.744 | | | |
| 0.8243 | -0.162 | 2.753 | | | |
| 0.8489 | -0.114 | 2.762 | | | |
| 0.8999 | -0.019 | 2.783 | | | |

SEC 1 283.2 MM CHORD
 EXPERIMENTAL PRESSURE DISTRIBUTION
 SOLID WALLS

WACH NO. P.598 ALPHA 0.80 REV 1.64018⁶

INTEGRATED FORCE COEFFICIENTS

C_x = 0.2893 C_y = -0.0086

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 1.390 | 2.999 | 0.2000 | 1.000 | 2.999 |
| 0.0035 | 2.302 | 2.844 | 0.2051 | 0.464 | 2.876 |
| 0.0079 | 2.206 | 2.822 | 0.2098 | 0.210 | 2.826 |
| 0.0196 | -0.292 | 2.735 | 0.2221 | -0.010 | 2.753 |
| 0.0294 | -0.195 | 2.725 | 0.2300 | -0.076 | 2.772 |
| 0.0495 | -0.495 | 2.657 | 0.2510 | -0.210 | 2.741 |
| 0.0741 | -0.524 | 2.652 | 0.2749 | -0.323 | 2.725 |
| 0.0991 | -0.532 | 2.663 | 0.3020 | -0.317 | 2.722 |
| 0.1241 | -0.532 | 2.683 | 0.3193 | -0.428 | 2.725 |
| 0.1501 | -0.530 | 2.673 | 0.3200 | -0.305 | 2.729 |
| 0.1994 | -0.503 | 2.686 | 0.3499 | -0.375 | 2.711 |
| 0.2362 | -0.499 | 2.686 | 0.3502 | -0.324 | 2.726 |
| 0.2794 | -0.491 | 2.683 | 0.4520 | -0.246 | 2.736 |
| 0.3158 | -0.497 | 2.687 | 0.5492 | -0.114 | 2.762 |
| 0.3594 | -0.495 | 2.683 | 0.5492 | 0.045 | 2.753 |
| 0.3993 | -0.495 | 2.680 | 0.7498 | 0.194 | 2.823 |
| 0.4194 | -0.494 | 2.689 | 0.8499 | 0.268 | 2.837 |
| 0.4395 | -0.488 | 2.683 | 0.9000 | 0.277 | 2.839 |
| 0.4595 | -0.485 | 2.680 | | | |
| 0.4797 | -0.486 | 2.689 | | | |
| 0.4993 | -0.491 | 2.683 | | | |
| 0.5194 | -0.498 | 2.683 | | | |
| 0.5396 | -0.485 | 2.683 | | | |
| 0.5592 | -0.489 | 2.680 | | | |
| 0.5787 | -0.492 | 2.689 | | | |
| 0.5989 | -0.484 | 2.689 | | | |
| 0.6193 | -0.493 | 2.689 | | | |
| 0.6454 | -0.493 | 2.693 | | | |
| 0.6747 | -0.433 | 2.783 | | | |
| 0.6999 | -0.440 | 2.705 | | | |
| 0.7237 | -0.373 | 2.711 | | | |
| 0.7455 | -0.329 | 2.723 | | | |
| 0.7735 | -0.286 | 2.723 | | | |
| 0.7993 | -0.243 | 2.737 | | | |
| 0.8243 | -0.198 | 2.747 | | | |
| 0.8469 | -0.153 | 2.758 | | | |
| 0.8689 | -0.220 | 2.779 | | | |

FIG 1 233.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.631 ALPHA 1.00 REV 1.65012⁶

INTEGRATED FORCE COEFFICIENTS
C_D = 2.4126 C_M = -0.0675

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 1.175 | 3.996 | 0.8830 | 1.875 | 0.976 |
| 0.0035 | -0.356 | 3.772 | 0.8851 | 2.693 | 3.921 |
| 0.0079 | -0.234 | 3.737 | 0.8898 | 0.461 | 0.875 |
| 0.0196 | -0.594 | 3.673 | 0.8921 | 0.222 | 3.827 |
| 0.0294 | -0.479 | 3.649 | 0.8980 | 3.131 | 3.859 |
| 0.0495 | -0.744 | 3.636 | 0.8983 | -0.834 | 0.776 |
| 0.0741 | -0.946 | 3.616 | 0.8749 | -0.133 | 3.756 |
| 0.0991 | -0.921 | 3.621 | 0.1022 | -0.175 | 3.749 |
| 0.1241 | -0.886 | 3.647 | 0.1498 | -0.277 | 3.728 |
| 0.1591 | -0.865 | 3.651 | 0.2000 | -0.234 | 3.727 |
| 0.1994 | -0.612 | 3.662 | 0.2499 | -0.237 | 3.727 |
| 0.2352 | -0.549 | 3.665 | 0.3502 | -0.240 | 3.732 |
| 0.2794 | -0.591 | 3.663 | 0.4502 | -0.197 | 3.744 |
| 0.3158 | -0.576 | 3.669 | 0.5492 | -0.332 | 3.757 |
| 0.3594 | -0.592 | 3.672 | 0.6492 | 0.368 | 3.757 |
| 0.3993 | -0.594 | 3.673 | 0.7498 | 0.211 | 3.825 |
| 0.4194 | -0.545 | 3.675 | 0.8499 | 0.252 | 0.839 |
| 0.4395 | -0.547 | 3.675 | 0.9230 | 0.245 | 3.843 |
| 0.4585 | -0.541 | 3.674 | | | |
| 0.4787 | -0.537 | 3.677 | | | |
| 0.4983 | -0.539 | 3.676 | | | |
| 0.5184 | -0.532 | 3.673 | | | |
| 0.5396 | -0.529 | 3.673 | | | |
| 0.5592 | -0.523 | 3.680 | | | |
| 0.5797 | -0.521 | 3.680 | | | |
| 0.5989 | -0.521 | 3.680 | | | |
| 0.6182 | -0.517 | 3.681 | | | |
| 0.6494 | -0.491 | 3.686 | | | |
| 0.6747 | -0.494 | 3.693 | | | |
| 0.6989 | -0.422 | 3.720 | | | |
| 0.7237 | -0.359 | 3.736 | | | |
| 0.7458 | -0.342 | 3.716 | | | |
| 0.7736 | -0.296 | 3.725 | | | |
| 0.7990 | -0.245 | 3.735 | | | |
| 0.8243 | -0.193 | 3.745 | | | |
| 0.8489 | -0.136 | 3.756 | | | |
| 0.8989 | -0.332 | 3.777 | | | |

FIG 1 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.599 ALPHA 2.82 REY 1.64×10^6

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.5486$ $C_M = -0.0855$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.022 | 2.986 | 0.0000 | 1.022 | 2.956 |
| 0.0035 | -0.497 | 2.688 | 0.0051 | 0.465 | 0.955 |
| 0.0079 | -0.679 | 2.664 | 0.0098 | 0.664 | 0.915 |
| 0.0196 | -0.923 | 2.634 | 0.0231 | 0.415 | 0.866 |
| 0.0294 | -0.997 | 2.567 | 0.0300 | 0.335 | 0.844 |
| 0.0445 | -1.022 | 2.582 | 0.0500 | 0.126 | 0.839 |
| 0.0741 | -1.113 | 2.565 | 0.0749 | 0.076 | 0.755 |
| 0.0991 | -1.022 | 2.582 | 0.1000 | -0.345 | 0.775 |
| 0.1241 | -0.849 | 2.615 | 0.1498 | -0.164 | 0.752 |
| 0.1591 | -0.849 | 2.624 | 0.2002 | -0.169 | 0.747 |
| 0.1994 | -0.732 | 2.639 | 0.2499 | -0.278 | 0.743 |
| 0.2392 | -0.705 | 2.645 | 0.3502 | -0.199 | 0.745 |
| 0.2794 | -0.677 | 2.650 | 0.4508 | -0.155 | 0.754 |
| 0.3185 | -0.652 | 2.654 | 0.5492 | -0.051 | 0.774 |
| 0.3594 | -0.632 | 2.659 | 0.6492 | 0.037 | 0.801 |
| 0.3993 | -0.617 | 2.662 | 0.7498 | 0.228 | 0.829 |
| 0.4194 | -0.603 | 2.665 | 0.8499 | 0.242 | 0.841 |
| 0.4395 | -0.592 | 2.665 | 0.9300 | 0.293 | 0.842 |
| 0.4585 | -0.584 | 2.667 | | | |
| 0.4737 | -0.579 | 2.668 | | | |
| 0.4953 | -0.569 | 2.668 | | | |
| 0.5184 | -0.563 | 2.669 | | | |
| 0.5395 | -0.574 | 2.673 | | | |
| 0.5592 | -0.586 | 2.673 | | | |
| 0.5737 | -0.557 | 2.674 | | | |
| 0.5969 | -0.554 | 2.675 | | | |
| 0.6192 | -0.547 | 2.676 | | | |
| 0.6434 | -0.518 | 2.682 | | | |
| 0.6747 | -0.477 | 2.690 | | | |
| 0.6989 | -0.442 | 2.697 | | | |
| 0.7237 | -0.424 | 2.704 | | | |
| 0.7485 | -0.394 | 2.714 | | | |
| 0.7738 | -0.364 | 2.724 | | | |
| 0.7982 | -0.293 | 2.735 | | | |
| 0.8242 | -0.195 | 2.746 | | | |
| 0.8469 | -0.138 | 2.757 | | | |
| 0.8759 | -0.335 | 2.777 | | | |

SG4 1 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

WACH NO. 0.630 ALPHA 3.22 REV 1.65-12⁶

INTEGRATED FORCE COEFFICIENTS

CV = 0.6714 CM = -0.3811

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 0.911 | 2.964 | 0.0000 | 0.911 | 2.964 |
| 0.0055 | -0.968 | 2.592 | 0.0051 | 0.976 | 0.977 |
| 0.0079 | -0.967 | 2.589 | 0.0098 | 0.813 | 0.945 |
| 0.0136 | -1.318 | 2.523 | 0.0201 | 0.571 | 0.877 |
| 0.0244 | -1.335 | 2.520 | 0.0302 | 0.454 | 0.874 |
| 0.0495 | -1.332 | 2.520 | 0.0502 | 0.265 | 0.806 |
| 0.0741 | -1.504 | 2.486 | 0.0749 | 0.132 | 0.810 |
| 0.0991 | -1.241 | 2.534 | 0.1000 | 0.270 | 0.798 |
| 0.1241 | -1.020 | 2.582 | 0.1498 | -0.661 | 0.772 |
| 0.1591 | -0.946 | 2.597 | 0.2000 | -0.103 | 0.754 |
| 0.2044 | -0.943 | 2.617 | 0.2499 | -0.132 | 0.758 |
| 0.2382 | -0.799 | 2.626 | 0.3502 | -0.143 | 0.756 |
| 0.2794 | -0.756 | 2.635 | 0.4500 | -0.110 | 0.763 |
| 0.3148 | -0.729 | 2.640 | 0.5498 | -0.024 | 0.702 |
| 0.3544 | -0.693 | 2.647 | 0.6492 | 0.107 | 0.805 |
| 0.3993 | -0.670 | 2.652 | 0.7498 | 0.241 | 0.852 |
| 0.4144 | -0.657 | 2.654 | 0.8499 | 0.312 | 0.843 |
| 0.4395 | -0.651 | 2.656 | 0.9300 | 0.299 | 0.843 |
| 0.4585 | -0.642 | 2.657 | | | |
| 0.4767 | -0.634 | 2.659 | | | |
| 0.4953 | -0.627 | 2.663 | | | |
| 0.5144 | -0.614 | 2.663 | | | |
| 0.5335 | -0.604 | 2.665 | | | |
| 0.5522 | -0.597 | 2.667 | | | |
| 0.5707 | -0.590 | 2.668 | | | |
| 0.5899 | -0.579 | 2.670 | | | |
| 0.6193 | -0.567 | 2.672 | | | |
| 0.6484 | -0.552 | 2.679 | | | |
| 0.6747 | -0.486 | 2.683 | | | |
| 0.6949 | -0.446 | 2.690 | | | |
| 0.7237 | -0.405 | 2.704 | | | |
| 0.7469 | -0.355 | 2.714 | | | |
| 0.7733 | -0.304 | 2.724 | | | |
| 0.7993 | -0.249 | 2.735 | | | |
| 0.8243 | -0.194 | 2.746 | | | |
| 0.8489 | -0.138 | 2.757 | | | |
| 0.8749 | -0.093 | 2.776 | | | |

WGL 1 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.620 ALPHA 4.83 REYNOLDS 1.05E+10⁶

INTEGRATED FORCE COEFFICIENTS

CX = 2.8234 CY = -2.0747

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 0.774 | 2.937 | 0.0000 | 0.774 | 2.937 |
| 0.2235 | -1.257 | 2.496 | 0.2235 | 1.241 | 2.923 |
| 0.2279 | -1.245 | 2.528 | 0.2298 | 0.924 | 2.957 |
| 0.2196 | -1.952 | 2.396 | 0.2241 | 0.708 | 2.922 |
| 0.2294 | -1.658 | 2.453 | 0.0300 | 0.581 | 2.899 |
| 0.2495 | -1.724 | 2.443 | 0.0500 | 0.388 | 2.861 |
| 0.2741 | -1.768 | 2.431 | 0.2749 | 0.252 | 2.833 |
| 0.2991 | -2.256 | 2.373 | 0.1020 | 0.175 | 2.819 |
| 0.1241 | -1.198 | 2.553 | 0.1498 | 0.232 | 2.752 |
| 0.1591 | -1.732 | 2.583 | 0.2020 | -0.019 | 2.768 |
| 0.1994 | -2.951 | 2.430 | 0.2499 | -0.055 | 2.774 |
| 0.2352 | -2.479 | 2.620 | 0.3502 | -0.184 | 2.768 |
| 0.2794 | -2.634 | 2.619 | 0.4502 | -0.369 | 2.771 |
| 0.3158 | -2.797 | 2.626 | 0.5492 | 0.024 | 2.755 |
| 0.3594 | -2.755 | 2.635 | 0.6492 | 0.127 | 2.812 |
| 0.3999 | -2.725 | 2.641 | 0.7498 | 0.254 | 2.835 |
| 0.4194 | -2.739 | 2.644 | 0.8499 | 0.328 | 2.845 |
| 0.4395 | -2.698 | 2.646 | 0.9302 | 0.325 | 2.845 |
| 0.4565 | -2.688 | 2.646 | | | |
| 0.4787 | -2.673 | 2.651 | | | |
| 0.4983 | -2.667 | 2.652 | | | |
| 0.5184 | -2.652 | 2.655 | | | |
| 0.5395 | -2.640 | 2.657 | | | |
| 0.5592 | -2.625 | 2.663 | | | |
| 0.5787 | -2.613 | 2.663 | | | |
| 0.5989 | -2.601 | 2.665 | | | |
| 0.6188 | -2.589 | 2.663 | | | |
| 0.6484 | -2.549 | 2.675 | | | |
| 0.6747 | -2.502 | 2.685 | | | |
| 0.6989 | -2.457 | 2.694 | | | |
| 0.7237 | -2.414 | 2.722 | | | |
| 0.7488 | -2.356 | 2.713 | | | |
| 0.7733 | -2.304 | 2.724 | | | |
| 0.7992 | -2.247 | 2.735 | | | |
| 0.8242 | -2.192 | 2.746 | | | |
| 0.8489 | -2.136 | 2.757 | | | |
| 0.8689 | -2.076 | 2.775 | | | |

SEC 1 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 2.621 ALPHA -2.52 REY 1.63⁶10

INTEGRATED FORCE COEFFICIENTS

CX = -2.8634 CM = -3.8912

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 2.8822 | 2.922 | 2.958 | 0.2222 | 0.922 | 2.958 |
| 2.7355 | 3.918 | 2.957 | 0.2251 | -3.376 | 0.669 |
| 2.2779 | 2.711 | 2.911 | 0.3098 | -0.674 | 0.613 |
| 2.2196 | 2.376 | 2.836 | 0.2201 | -0.794 | 2.576 |
| 2.2244 | 2.262 | 2.797 | 0.3300 | -0.756 | 2.534 |
| 2.2495 | 2.224 | 2.759 | 0.2502 | -0.866 | 0.542 |
| 2.2741 | -2.146 | 2.722 | 2.2749 | -0.862 | 0.561 |
| 2.2991 | -2.216 | 2.704 | 0.1022 | -0.823 | 2.574 |
| 2.1241 | -2.101 | 2.712 | 2.1496 | -0.831 | 0.568 |
| 2.1591 | -2.233 | 2.700 | 2.2022 | -0.732 | 2.592 |
| 2.1944 | -2.240 | 2.699 | 2.2499 | -0.666 | 2.614 |
| 2.2352 | -2.271 | 2.692 | 2.3502 | -0.527 | 0.635 |
| 2.2794 | -2.289 | 2.688 | 2.4502 | -2.378 | 2.669 |
| 2.3155 | -2.313 | 2.683 | 2.5492 | -0.196 | 0.719 |
| 2.3594 | -2.319 | 2.681 | 2.6492 | 2.222 | 2.753 |
| 2.3923 | -2.337 | 2.677 | 2.7496 | 2.144 | 2.755 |
| 2.4194 | -2.339 | 2.677 | 2.8499 | 0.225 | 0.823 |
| 2.4395 | -2.353 | 2.674 | 2.9202 | 0.243 | 0.827 |
| 2.4585 | -2.357 | 2.673 | | | |
| 2.4787 | -2.366 | 2.671 | | | |
| 2.4983 | -2.377 | 2.669 | | | |
| 2.5184 | -2.382 | 2.668 | | | |
| 2.5394 | -2.383 | 2.667 | | | |
| 2.5592 | -2.388 | 2.666 | | | |
| 2.5787 | -2.394 | 2.665 | | | |
| 2.5989 | -2.405 | 2.662 | | | |
| 2.6182 | -2.409 | 2.661 | | | |
| 2.6484 | -2.402 | 2.663 | | | |
| 2.6747 | -2.373 | 2.669 | | | |
| 2.6989 | -2.352 | 2.674 | | | |
| 2.7237 | -2.327 | 2.680 | | | |
| 2.7488 | -2.299 | 2.688 | | | |
| 2.7735 | -2.250 | 2.697 | | | |
| 2.7992 | -2.208 | 2.706 | | | |
| 2.8240 | -2.160 | 2.717 | | | |
| 2.8489 | -2.109 | 2.728 | | | |
| 2.8999 | -2.003 | 2.752 | | | |

364 1 283.2 MM CHORD
 EXPERIMENTAL PRESSURE DISTRIBUTION
 SOLID WALLS

MACH NO. 0.652 ALPHA 8.38 REV 1.6401F⁶

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.2963$ $C_M = -0.0917$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.2338 | 1.104 | 2.999 | 0.2338 | 1.104 | 0.929 |
| 0.2335 | 0.344 | 3.029 | 0.2351 | 0.471 | 0.857 |
| 0.2379 | 0.133 | 2.781 | 0.2396 | 0.215 | 0.803 |
| 0.2196 | -0.217 | 2.733 | 0.2281 | -0.028 | 0.758 |
| 0.2294 | -0.372 | 2.669 | 0.2308 | -0.077 | 0.735 |
| 0.2095 | -0.406 | 2.643 | 0.2582 | -0.233 | 0.702 |
| 0.2741 | -0.501 | 2.624 | 0.2749 | -0.325 | 0.679 |
| 0.2691 | -0.671 | 2.622 | 0.1302 | -0.344 | 0.675 |
| 0.1241 | -0.507 | 2.625 | 0.1496 | -0.438 | 0.654 |
| 0.1591 | -0.571 | 2.624 | 0.2082 | -0.425 | 0.657 |
| 0.1994 | -0.535 | 2.632 | 0.2499 | -0.414 | 0.660 |
| 0.2382 | -0.535 | 2.633 | 0.3502 | -0.356 | 0.672 |
| 0.2744 | -0.528 | 2.634 | 0.4502 | -0.265 | 0.673 |
| 0.3128 | -0.531 | 2.633 | 0.5493 | -0.122 | 0.725 |
| 0.3594 | -0.523 | 2.634 | 0.6492 | 0.047 | 0.763 |
| 0.3993 | -0.522 | 2.635 | 0.7498 | 0.199 | 0.776 |
| 0.4194 | -0.519 | 2.635 | 0.8499 | 0.273 | 0.813 |
| 0.4395 | -0.523 | 2.635 | 0.9380 | 0.252 | 0.815 |
| 0.4565 | -0.523 | 2.636 | | | |
| 0.4787 | -0.528 | 2.636 | | | |
| 0.4943 | -0.522 | 2.635 | | | |
| 0.5184 | -0.523 | 2.636 | | | |
| 0.5396 | -0.513 | 2.636 | | | |
| 0.5542 | -0.517 | 2.637 | | | |
| 0.5747 | -0.516 | 2.637 | | | |
| 0.5959 | -0.514 | 2.635 | | | |
| 0.6183 | -0.516 | 2.637 | | | |
| 0.6454 | -0.494 | 2.642 | | | |
| 0.6747 | -0.494 | 2.651 | | | |
| 0.6949 | -0.422 | 2.654 | | | |
| 0.7237 | -0.397 | 2.666 | | | |
| 0.7486 | -0.337 | 2.677 | | | |
| 0.7739 | -0.293 | 2.687 | | | |
| 0.7992 | -0.299 | 2.697 | | | |
| 0.8248 | -0.174 | 2.711 | | | |
| 0.8489 | -0.126 | 2.724 | | | |
| 0.8689 | -0.119 | 2.743 | | | |

HGK 1 283.2 MM CHORD
 EXPERIMENTAL PRESSURE DISTRIBUTION
 SOLID WALLS

MACH NO. 0.649 ALPHA 1.83 REY 1.63×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.4358$ $C_M = -0.8931$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.899 | 2.997 | 0.0000 | 1.899 | 2.997 |
| 0.0035 | 2.318 | 2.755 | 0.0035 | 0.695 | 2.927 |
| 0.0079 | -2.171 | 2.715 | 0.0079 | 0.461 | 0.855 |
| 0.0196 | -2.512 | 3.639 | 0.0196 | 0.219 | 0.831 |
| 0.0294 | -2.655 | 2.637 | 0.0294 | 0.125 | 3.788 |
| 0.0495 | -2.741 | 2.588 | 0.0495 | -0.045 | 2.742 |
| 0.0741 | -2.793 | 2.544 | 0.0741 | -0.155 | 0.717 |
| 0.0991 | -2.843 | 2.554 | 0.1000 | -0.192 | 0.719 |
| 0.1241 | -2.743 | 2.565 | 0.1498 | -2.384 | 2.684 |
| 0.1591 | -2.723 | 2.592 | 0.2000 | -0.312 | 0.633 |
| 0.1994 | -2.682 | 2.626 | 0.2499 | -0.317 | 0.652 |
| 0.2392 | -2.648 | 2.639 | 0.3502 | -0.284 | 0.639 |
| 0.2794 | -2.628 | 2.613 | 0.4500 | -0.213 | 3.725 |
| 0.3186 | -2.622 | 2.615 | 0.5498 | -0.898 | 0.752 |
| 0.3594 | -2.642 | 3.619 | 0.6492 | 0.068 | 0.768 |
| 0.3993 | -2.595 | 2.621 | 0.7498 | 0.215 | 0.812 |
| 0.4194 | -2.596 | 2.623 | 0.8499 | 0.285 | 3.816 |
| 0.4395 | -2.598 | 2.622 | 0.9382 | 0.293 | 0.817 |
| 0.4585 | -2.592 | 2.623 | | | |
| 0.4787 | -2.576 | 2.624 | | | |
| 0.4983 | -2.578 | 2.624 | | | |
| 0.5184 | -2.573 | 2.626 | | | |
| 0.5396 | -2.567 | 2.627 | | | |
| 0.5592 | -2.562 | 2.627 | | | |
| 0.5787 | -2.557 | 2.628 | | | |
| 0.5989 | -2.553 | 2.629 | | | |
| 0.6182 | -2.545 | 2.631 | | | |
| 0.6484 | -2.518 | 2.637 | | | |
| 0.6747 | -2.476 | 2.647 | | | |
| 0.6959 | -2.439 | 2.655 | | | |
| 0.7237 | -2.461 | 2.663 | | | |
| 0.7488 | -2.349 | 2.675 | | | |
| 0.7738 | -2.298 | 2.686 | | | |
| 0.7992 | -2.243 | 2.698 | | | |
| 0.8248 | -2.187 | 2.711 | | | |
| 0.8489 | -2.127 | 2.724 | | | |
| 0.8789 | -2.026 | 2.746 | | | |

SEC 1 223.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.649 ALPHA 2.02 REV 1.63+13⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.5686 CM = -0.0857

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | Cp | P/H | X/C | Cp | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 1.000 | 2.987 | 0.0000 | 1.053 | 3.947 |
| 0.0335 | -0.305 | 2.674 | 0.0351 | 0.861 | 3.945 |
| 0.0379 | -0.405 | 2.645 | 0.0398 | 0.654 | 3.899 |
| 0.0196 | -0.827 | 2.569 | 0.0201 | 0.437 | 3.844 |
| 0.0294 | -0.908 | 2.545 | 0.0302 | 0.299 | 3.822 |
| 0.0495 | -0.964 | 2.534 | 0.0500 | 0.115 | 3.779 |
| 0.0741 | -1.309 | 2.462 | 0.0749 | -0.207 | 3.752 |
| 0.0991 | -1.409 | 2.443 | 0.1000 | -0.259 | 3.743 |
| 0.1241 | -1.899 | 2.553 | 0.1498 | -0.156 | 3.712 |
| 0.1591 | -1.503 | 2.562 | 0.2000 | -0.212 | 3.716 |
| 0.1994 | -0.779 | 2.583 | 0.2499 | -0.213 | 3.712 |
| 0.2382 | -0.703 | 2.587 | 0.3502 | -0.223 | 3.724 |
| 0.2794 | -0.719 | 2.594 | 0.4500 | -0.171 | 3.715 |
| 0.3185 | -0.701 | 2.593 | 0.5498 | -0.261 | 3.743 |
| 0.3594 | -1.671 | 2.624 | 0.6492 | 0.257 | 3.773 |
| 0.3993 | -0.674 | 2.628 | 0.7498 | 0.238 | 3.815 |
| 0.4194 | -0.641 | 2.611 | 0.8499 | 0.274 | 3.819 |
| 0.4395 | -1.638 | 2.612 | 0.9000 | 0.296 | 3.819 |
| 0.4565 | -0.632 | 2.613 | | | |
| 0.4767 | -0.626 | 2.614 | | | |
| 0.4983 | -0.624 | 2.615 | | | |
| 0.5184 | -0.613 | 2.617 | | | |
| 0.5396 | -0.603 | 2.619 | | | |
| 0.5592 | -0.594 | 2.621 | | | |
| 0.5767 | -0.585 | 2.623 | | | |
| 0.5969 | -0.579 | 2.625 | | | |
| 0.6151 | -0.571 | 2.627 | | | |
| 0.6484 | -0.567 | 2.634 | | | |
| 0.6747 | -0.491 | 2.644 | | | |
| 0.6969 | -0.451 | 2.653 | | | |
| 0.7237 | -0.406 | 2.663 | | | |
| 0.7488 | -0.358 | 2.676 | | | |
| 0.7733 | -0.298 | 2.687 | | | |
| 0.7992 | -0.212 | 2.723 | | | |
| 0.8248 | -0.195 | 2.712 | | | |
| 0.8489 | -0.125 | 2.725 | | | |
| 0.8989 | -0.753 | 2.746 | | | |

FIG 1 223.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

1ACH NO. 0.649 ALPHA 3.28 REY 1.54×10^6

INTEGRATED FORCE COEFFICIENTS

$C_D = 0.7185$ $C_M = -0.3813$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 0.973 | 2.969 | 0.0000 | 0.973 | 2.969 |
| 0.0035 | -0.753 | 2.586 | 0.0035 | 0.973 | 2.972 |
| 0.0079 | -0.758 | 2.578 | 0.0079 | 0.864 | 2.932 |
| 0.0196 | -1.150 | 2.491 | 0.0196 | 0.563 | 2.378 |
| 0.0294 | -1.157 | 2.489 | 0.0294 | 0.448 | 2.853 |
| 0.0495 | -1.365 | 2.459 | 0.0495 | 0.259 | 2.811 |
| 0.0741 | -1.483 | 2.423 | 0.0741 | 0.124 | 2.751 |
| 0.0991 | -1.747 | 2.365 | 0.0991 | 0.059 | 2.756 |
| 0.1241 | -1.553 | 2.386 | 0.1241 | -0.075 | 2.737 |
| 0.1591 | -1.295 | 2.472 | 0.1591 | -0.115 | 2.727 |
| 0.1994 | -0.835 | 2.567 | 0.1994 | -0.149 | 2.720 |
| 0.2352 | -0.564 | 2.574 | 0.2352 | -0.159 | 2.718 |
| 0.2794 | -0.758 | 2.579 | 0.2794 | -0.124 | 2.726 |
| 0.3158 | -0.757 | 2.583 | 0.3158 | -0.329 | 2.747 |
| 0.3594 | -0.754 | 2.590 | 0.3594 | 0.128 | 2.777 |
| 0.3993 | -0.715 | 2.594 | 0.3993 | 0.246 | 2.808 |
| 0.4194 | -0.702 | 2.597 | 0.4194 | 0.325 | 2.821 |
| 0.4395 | -0.696 | 2.598 | 0.4395 | 0.335 | 2.821 |
| 0.4555 | -0.687 | 2.601 | | | |
| 0.4747 | -0.675 | 2.603 | | | |
| 0.4953 | -0.668 | 2.605 | | | |
| 0.5154 | -0.655 | 2.624 | | | |
| 0.5396 | -0.643 | 2.613 | | | |
| 0.5592 | -0.629 | 2.613 | | | |
| 0.5787 | -0.617 | 2.614 | | | |
| 0.5989 | -0.609 | 2.615 | | | |
| 0.6182 | -0.598 | 2.622 | | | |
| 0.6454 | -0.588 | 2.629 | | | |
| 0.6747 | -0.575 | 2.643 | | | |
| 0.6989 | -0.483 | 2.653 | | | |
| 0.7237 | -0.415 | 2.663 | | | |
| 0.7486 | -0.351 | 2.673 | | | |
| 0.7738 | -0.307 | 2.685 | | | |
| 0.7992 | -0.249 | 2.693 | | | |
| 0.8243 | -0.192 | 2.711 | | | |
| 0.8489 | -0.135 | 2.723 | | | |
| 0.8789 | -0.142 | 2.744 | | | |

FIG 1 253.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.698 ALPHA -2.58 REY 1.63×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = -0.8915$ $C_M = -0.3954$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.2323 | 2.946 | 2.955 | 2.8888 | 0.946 | 2.955 |
| 2.2335 | 2.936 | 2.953 | 2.8851 | -0.339 | 2.645 |
| 2.2379 | 0.735 | 2.983 | 2.8898 | -0.619 | 2.569 |
| 2.2196 | 2.402 | 2.821 | 2.8281 | -0.767 | 2.532 |
| 2.2294 | 2.227 | 2.775 | 2.8332 | -0.726 | 2.542 |
| 2.2495 | 2.251 | 2.734 | 2.8530 | -0.994 | 2.479 |
| 2.2741 | -2.136 | 2.585 | 2.8749 | -1.251 | 2.452 |
| 2.2991 | -2.215 | 2.669 | 2.1880 | -0.983 | 2.499 |
| 2.1241 | -2.191 | 2.577 | 2.1498 | -1.829 | 2.466 |
| 2.1591 | -2.243 | 2.662 | 2.2080 | -0.841 | 2.514 |
| 2.1994 | -2.249 | 2.663 | 2.2499 | -0.752 | 2.537 |
| 2.2352 | -2.293 | 2.652 | 2.3532 | -0.573 | 2.552 |
| 2.2794 | -2.324 | 2.647 | 2.4528 | -0.398 | 2.623 |
| 2.3186 | -2.337 | 2.642 | 2.5490 | -0.228 | 2.672 |
| 2.3594 | -2.343 | 2.638 | 2.6492 | 0.071 | 2.722 |
| 2.3993 | -2.351 | 2.633 | 2.7498 | 0.139 | 2.756 |
| 2.4191 | -2.365 | 2.632 | 2.8499 | 0.221 | 2.776 |
| 2.4395 | -2.368 | 2.629 | 2.9330 | 0.242 | 2.781 |
| 2.4585 | -2.354 | 2.627 | | | |
| 2.4757 | -2.344 | 2.625 | | | |
| 2.4943 | -2.424 | 2.622 | | | |
| 2.5184 | -2.418 | 2.621 | | | |
| 2.5396 | -2.417 | 2.619 | | | |
| 2.5592 | -2.425 | 2.617 | | | |
| 2.5787 | -2.429 | 2.616 | | | |
| 2.5959 | -2.439 | 2.613 | | | |
| 2.6153 | -2.442 | 2.613 | | | |
| 2.6454 | -2.431 | 2.615 | | | |
| 2.6747 | -2.349 | 2.623 | | | |
| 2.6959 | -2.374 | 2.629 | | | |
| 2.7237 | -2.344 | 2.637 | | | |
| 2.7456 | -2.302 | 2.647 | | | |
| 2.7735 | -2.298 | 2.658 | | | |
| 2.7990 | -2.213 | 2.669 | | | |
| 2.8248 | -2.108 | 2.682 | | | |
| 2.8489 | -2.105 | 2.686 | | | |
| 2.8799 | 2.213 | 2.725 | | | |

26X 1 283.2 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

SOLID WALLS

MACH NO. 4.638 ALPHA -2.58 REYNOLDS 1.63*10⁶

INTEGRATED FORCE COEFFICIENTS

C_x = -0.3915 C_m = -0.3954

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.8333 | 3.946 | 3.955 | 0.2068 | 0.946 | 0.955 |
| 2.8335 | 3.936 | 3.953 | 0.8251 | -0.329 | 0.645 |
| 2.8379 | 3.735 | 3.933 | 0.8890 | -0.619 | 0.569 |
| 0.2196 | 3.402 | 3.821 | 0.8281 | -0.767 | 0.532 |
| 2.8214 | 3.247 | 2.778 | 0.8308 | -0.728 | 0.542 |
| 0.2495 | 3.751 | 3.734 | 0.8588 | -0.984 | 0.479 |
| 0.2741 | -2.136 | 2.653 | 0.8749 | -1.851 | 0.452 |
| 0.2991 | -2.215 | 2.669 | 0.1388 | -0.903 | 0.499 |
| 2.1241 | -2.171 | 2.677 | 0.1498 | -1.829 | 2.468 |
| 2.1571 | -2.242 | 2.662 | 2.2382 | -0.841 | 2.514 |
| 2.1974 | -2.249 | 2.663 | 0.2499 | -0.753 | 0.537 |
| 2.2352 | -2.253 | 2.652 | 0.3502 | -0.573 | 0.568 |
| 2.2794 | -2.374 | 2.647 | 0.4588 | -0.398 | 0.623 |
| 2.3188 | -2.382 | 2.643 | 0.5498 | -0.228 | 0.672 |
| 2.3594 | -2.343 | 2.638 | 0.6492 | 0.881 | 0.722 |
| 2.3993 | -2.351 | 2.633 | 0.7498 | 0.139 | 0.756 |
| 2.4194 | -2.385 | 2.532 | 0.8499 | 0.221 | 2.776 |
| 2.4375 | -2.382 | 2.629 | 0.9388 | 0.243 | 2.781 |
| 2.4555 | -2.384 | 2.627 | | | |
| 2.4757 | -2.374 | 2.625 | | | |
| 2.4953 | -2.424 | 2.622 | | | |
| 2.5154 | -2.412 | 2.621 | | | |
| 0.5396 | -2.417 | 2.619 | | | |
| 0.5592 | -2.425 | 2.617 | | | |
| 2.5737 | -2.429 | 2.616 | | | |
| 2.5959 | -2.439 | 2.613 | | | |
| 2.6153 | -2.442 | 2.613 | | | |
| 2.6454 | -2.431 | 2.615 | | | |
| 0.6747 | -2.376 | 2.623 | | | |
| 2.6949 | -2.374 | 2.629 | | | |
| 0.7237 | -2.344 | 2.637 | | | |
| 2.7436 | -2.342 | 2.647 | | | |
| 2.7738 | -2.298 | 2.658 | | | |
| 0.7993 | -2.213 | 2.669 | | | |
| 2.8248 | -2.152 | 2.682 | | | |
| 0.8489 | -2.175 | 2.696 | | | |
| 2.8989 | 2.213 | 2.725 | | | |

CGC 1 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.721 ALPHA 0.86 REY 1.63e10⁶

INTEGRATED FORCE COEFFICIENTS

CV = 0.3716 CH = -0.9953

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | PPH | X/C | CP | PPH |
|--------|--------|-------|--------|--------|-------|
| 0.2780 | 1.123 | 2.999 | 0.8088 | 1.123 | 0.999 |
| 0.2835 | 2.417 | 2.922 | 0.8251 | 0.476 | 0.839 |
| 0.2879 | 2.142 | 2.768 | 0.8396 | 0.217 | 0.775 |
| 0.296 | -2.154 | 2.683 | 0.8521 | -0.811 | 0.719 |
| 0.2994 | -1.326 | 2.639 | 0.8688 | -0.882 | 0.721 |
| 0.3095 | -2.453 | 2.686 | 0.8883 | -0.252 | 0.659 |
| 0.3141 | -2.499 | 2.547 | 0.8749 | -0.352 | 0.634 |
| 0.3191 | -2.720 | 2.542 | 0.8888 | -0.376 | 0.629 |
| 0.3241 | -2.403 | 2.571 | 0.8998 | -0.493 | 0.603 |
| 0.3291 | -2.411 | 2.569 | 0.9088 | -0.474 | 0.604 |
| 0.3394 | -2.571 | 2.579 | 0.9299 | -0.461 | 0.607 |
| 0.3462 | -2.576 | 2.578 | 0.9382 | -0.394 | 0.629 |
| 0.3494 | -2.574 | 2.573 | 0.9488 | -0.280 | 0.650 |
| 0.3535 | -2.579 | 2.577 | 0.9498 | -0.133 | 0.689 |
| 0.3594 | -2.566 | 2.581 | 0.9492 | 0.846 | 0.733 |
| 0.3693 | -2.563 | 2.582 | 0.9498 | 0.199 | 0.771 |
| 0.3794 | -2.563 | 2.581 | 0.9499 | 0.275 | 0.789 |
| 0.3895 | -2.571 | 2.579 | 0.9808 | 0.284 | 0.792 |
| 0.3985 | -2.568 | 2.583 | | | |
| 0.4087 | -2.568 | 2.583 | | | |
| 0.4183 | -2.578 | 2.582 | | | |
| 0.4284 | -2.567 | 2.580 | | | |
| 0.4396 | -2.567 | 2.583 | | | |
| 0.4492 | -2.565 | 2.582 | | | |
| 0.4587 | -2.563 | 2.583 | | | |
| 0.4689 | -2.561 | 2.582 | | | |
| 0.4788 | -2.565 | 2.584 | | | |
| 0.4884 | -2.567 | 2.591 | | | |
| 0.4977 | -2.493 | 2.603 | | | |
| 0.5089 | -2.442 | 2.612 | | | |
| 0.5187 | -2.397 | 2.623 | | | |
| 0.5288 | -2.343 | 2.636 | | | |
| 0.5383 | -2.291 | 2.649 | | | |
| 0.5493 | -2.235 | 2.663 | | | |
| 0.5588 | -2.176 | 2.678 | | | |
| 0.5689 | -2.116 | 2.693 | | | |
| 0.5789 | -2.053 | 2.718 | | | |

#C4 1 283.2 MM CHORD
 EXPERIMENTAL PRESSURE DISTRIBUTION
 SOLID WALLS

MACH NO. 0.771 ALPHA 1.00 REV 1.63*1A⁶

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.4519$ $C_M = -0.3917$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | C _p | P/H | X/C | C _p | P/H |
|--------|----------------|-------|--------|----------------|-------|
| 0.0223 | 1.121 | 3.998 | 0.8888 | 1.121 | 3.998 |
| 0.0335 | 0.127 | 3.752 | 0.8851 | 0.685 | 3.898 |
| 0.0379 | -0.701 | 3.735 | 0.8898 | 0.452 | 3.833 |
| 0.0196 | -2.14 | 3.618 | 0.8201 | 0.238 | 3.773 |
| 0.0294 | -7.504 | 3.580 | 0.8382 | 0.116 | 3.758 |
| 0.0495 | -0.555 | 3.558 | 0.8582 | -0.068 | 3.726 |
| 0.0741 | -0.503 | 3.481 | 0.8749 | -0.174 | 3.679 |
| 0.0991 | -1.732 | 3.415 | 0.1383 | -0.216 | 3.658 |
| 0.1241 | -2.909 | 3.495 | 0.1498 | -0.341 | 3.637 |
| 0.1591 | -8.744 | 3.536 | 0.2388 | -0.333 | 3.639 |
| 0.1934 | -1.705 | 3.545 | 0.2499 | -0.359 | 3.632 |
| 0.2332 | -2.549 | 3.548 | 0.3522 | -0.315 | 3.643 |
| 0.2794 | -3.636 | 3.551 | 0.4502 | -0.243 | 3.661 |
| 0.3148 | -3.692 | 3.552 | 0.5493 | -0.105 | 3.695 |
| 0.3594 | -3.501 | 3.557 | 0.6492 | 0.063 | 3.737 |
| 0.3993 | -2.524 | 3.559 | 0.7498 | 0.215 | 3.774 |
| 0.4194 | -3.644 | 3.561 | 0.8499 | 0.286 | 3.792 |
| 0.4395 | -2.624 | 3.561 | 0.9022 | 0.292 | 3.793 |
| 0.4535 | -2.435 | 3.563 | | | |
| 0.4737 | -3.533 | 3.564 | | | |
| 0.4933 | -0.631 | 3.564 | | | |
| 0.5184 | -3.622 | 3.566 | | | |
| 0.5396 | -3.517 | 3.566 | | | |
| 0.5592 | -3.512 | 3.573 | | | |
| 0.5787 | -2.603 | 3.572 | | | |
| 0.5989 | -2.547 | 3.573 | | | |
| 0.6188 | -2.503 | 3.577 | | | |
| 0.6454 | -2.547 | 3.586 | | | |
| 0.6747 | -2.494 | 3.599 | | | |
| 0.6989 | -2.428 | 3.613 | | | |
| 0.7237 | -2.404 | 3.621 | | | |
| 0.7488 | -2.345 | 3.635 | | | |
| 0.7738 | -2.298 | 3.649 | | | |
| 0.7992 | -2.232 | 3.664 | | | |
| 0.8243 | -2.173 | 3.678 | | | |
| 0.8489 | -2.116 | 3.693 | | | |
| 0.8789 | -2.022 | 3.716 | | | |

364 1 233.2 MM CHORD
 EXPERIMENTAL PRESSURE DISTRIBUTION
 SOLID WALLS

MACH NO. 4.723 ALPHA 2.80 REY 1.53*10⁶

INTEGRATED FORCE COEFFICIENTS

C_L = 0.6194 C_M = -0.2869

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.2270 | 1.343 | 2.991 | 0.2288 | 1.893 | 3.931 |
| 2.2235 | -2.199 | 2.679 | 0.2351 | 0.845 | 3.932 |
| 2.2279 | -2.313 | 2.643 | 0.2398 | 0.637 | 3.879 |
| 2.2196 | -2.503 | 2.557 | 0.2421 | 0.391 | 3.818 |
| 2.2294 | -2.794 | 2.527 | 0.2328 | 0.286 | 3.792 |
| 2.2495 | -2.936 | 2.514 | 0.2530 | 0.181 | 3.746 |
| 2.2741 | -1.198 | 2.434 | 0.2749 | -0.025 | 3.715 |
| 2.2941 | -1.341 | 2.377 | 0.1000 | -0.279 | 3.712 |
| 2.1241 | -1.335 | 2.391 | 0.1496 | -0.214 | 2.653 |
| 2.1591 | -1.357 | 2.353 | 0.2008 | -0.241 | 2.652 |
| 2.1944 | -1.318 | 2.397 | 0.2499 | -0.262 | 2.657 |
| 2.2392 | -1.248 | 2.467 | 0.3502 | -0.250 | 2.606 |
| 2.2794 | -2.645 | 2.549 | 0.4572 | -0.192 | 2.674 |
| 2.3139 | -2.990 | 2.551 | 0.5493 | -0.371 | 2.714 |
| 2.3594 | -2.888 | 2.551 | 0.6492 | 0.386 | 2.743 |
| 2.3993 | -2.493 | 2.551 | 0.7496 | 0.233 | 2.779 |
| 2.4194 | -2.504 | 2.552 | 0.8499 | 0.339 | 2.795 |
| 2.4395 | -2.607 | 2.551 | 0.9300 | 0.302 | 2.796 |
| 2.4595 | -2.677 | 2.553 | | | |
| 2.4787 | -2.674 | 2.554 | | | |
| 2.4933 | -2.668 | 2.556 | | | |
| 2.5194 | -2.656 | 2.559 | | | |
| 2.5396 | -2.647 | 2.561 | | | |
| 2.5592 | -2.638 | 2.563 | | | |
| 2.5797 | -2.628 | 2.566 | | | |
| 2.5999 | -2.620 | 2.568 | | | |
| 2.6190 | -2.605 | 2.572 | | | |
| 2.6494 | -2.584 | 2.582 | | | |
| 2.6747 | -2.589 | 2.595 | | | |
| 2.6999 | -2.461 | 2.607 | | | |
| 2.7237 | -2.411 | 2.623 | | | |
| 2.7468 | -2.392 | 2.634 | | | |
| 2.7738 | -2.295 | 2.648 | | | |
| 2.7993 | -2.296 | 2.663 | | | |
| 2.8248 | -2.177 | 2.677 | | | |
| 2.8499 | -2.141 | 2.691 | | | |
| 2.8989 | -2.330 | 2.714 | | | |

SEC 1 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.679 ALPHA 3.82 REV 1.63*12⁶

INTEGRATED FORCE COEFFICIENTS

C_L = 0.7968 C_M = -0.3854

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.2000 | 1.336 | 2.977 | 0.8000 | 1.836 | 0.977 |
| 0.3000 | -0.459 | 2.836 | 0.8051 | 0.953 | 0.957 |
| 0.3079 | -0.547 | 2.597 | 0.8098 | 0.776 | 0.913 |
| 0.3196 | -0.953 | 2.492 | 0.8201 | 0.534 | 0.853 |
| 0.3294 | -0.958 | 2.485 | 0.8302 | 0.421 | 0.825 |
| 0.3405 | -1.123 | 2.444 | 0.8503 | 0.231 | 0.779 |
| 0.3741 | -1.244 | 2.414 | 0.8749 | 0.047 | 0.745 |
| 0.3991 | -1.529 | 2.349 | 0.1022 | 0.031 | 0.729 |
| 0.4241 | -1.453 | 2.356 | 0.1498 | -0.117 | 0.694 |
| 0.4591 | -1.505 | 2.353 | 0.2002 | -0.151 | 0.654 |
| 0.4994 | -1.476 | 2.362 | 0.2499 | -0.183 | 0.676 |
| 0.5382 | -1.455 | 2.354 | 0.3502 | -0.192 | 0.674 |
| 0.5794 | -1.449 | 2.363 | 0.4508 | -0.152 | 0.654 |
| 0.6168 | -1.453 | 2.363 | 0.5498 | -0.042 | 0.711 |
| 0.6354 | -1.172 | 2.437 | 0.6492 | 0.135 | 0.747 |
| 0.6993 | -0.779 | 2.529 | 0.7498 | 0.247 | 0.752 |
| 0.74194 | -0.716 | 2.544 | 0.8499 | 0.312 | 0.796 |
| 0.74395 | -0.678 | 2.556 | 0.9002 | 0.313 | 0.798 |
| 0.74555 | -0.639 | 2.564 | | | |
| 0.74757 | -0.622 | 2.568 | | | |
| 0.74983 | -0.611 | 2.571 | | | |
| 0.75184 | -0.603 | 2.572 | | | |
| 0.75396 | -0.599 | 2.573 | | | |
| 0.75592 | -0.594 | 2.574 | | | |
| 0.75787 | -0.587 | 2.576 | | | |
| 0.75959 | -0.582 | 2.577 | | | |
| 0.76152 | -0.571 | 2.583 | | | |
| 0.76484 | -0.539 | 2.588 | | | |
| 0.76747 | -0.491 | 2.620 | | | |
| 0.76989 | -0.449 | 2.610 | | | |
| 0.77237 | -0.433 | 2.627 | | | |
| 0.77458 | -0.398 | 2.635 | | | |
| 0.77735 | -0.272 | 2.649 | | | |
| 0.77993 | -0.239 | 2.662 | | | |
| 0.8242 | -0.194 | 2.676 | | | |
| 0.8489 | -0.131 | 2.689 | | | |
| 0.8969 | -0.338 | 2.712 | | | |

364 1 283.2 MM CHORD
 EXPERIMENTAL PRESSURE DISTRIBUTION
 SOLID WALLS

NACH NO. 0.728 ALPHA -2.50 REY 1.64*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = -0.1116 CM = -0.0979

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.2382 | 0.967 | 0.956 | 0.0000 | 0.967 | 0.956 |
| 2.2335 | 0.948 | 0.949 | 0.0051 | -0.247 | 0.644 |
| 2.2279 | 0.736 | 0.897 | 0.0098 | -0.556 | 0.365 |
| 2.2196 | 0.484 | 0.811 | 0.0201 | -0.716 | 0.523 |
| 2.2094 | 0.234 | 0.767 | 0.0300 | -0.674 | 0.534 |
| 2.2495 | 0.355 | 0.721 | 0.0500 | -0.967 | 0.459 |
| 2.2741 | -0.141 | 0.671 | 0.0749 | -1.049 | 0.438 |
| 2.2971 | -0.243 | 0.651 | 0.1000 | -1.021 | 0.445 |
| 2.1241 | -0.107 | 0.659 | 0.1498 | -0.925 | 0.478 |
| 2.1591 | -0.245 | 0.644 | 0.2000 | -1.161 | 0.429 |
| 0.1994 | -0.253 | 0.642 | 0.2499 | -0.965 | 0.468 |
| 2.2392 | -0.293 | 0.632 | 0.3502 | -0.571 | 0.561 |
| 2.2794 | -0.313 | 0.627 | 0.4500 | -0.398 | 0.625 |
| 2.3188 | -0.343 | 0.619 | 0.5498 | -0.199 | 0.656 |
| 2.3594 | -0.351 | 0.617 | 0.6492 | 0.001 | 0.708 |
| 2.3993 | -0.376 | 0.613 | 0.7498 | 0.130 | 0.743 |
| 2.4194 | -0.379 | 0.613 | 0.8499 | 0.222 | 0.764 |
| 2.4395 | -0.393 | 0.636 | 0.9888 | 0.248 | 0.769 |
| 2.4585 | -0.396 | 0.685 | | | |
| 2.4787 | -0.409 | 0.682 | | | |
| 2.4943 | -0.423 | 0.599 | | | |
| 2.5184 | -0.427 | 0.597 | | | |
| 2.5396 | -0.434 | 0.595 | | | |
| 2.5592 | -0.444 | 0.593 | | | |
| 2.5787 | -0.449 | 0.592 | | | |
| 2.5989 | -0.461 | 0.589 | | | |
| 2.6188 | -0.464 | 0.588 | | | |
| 2.6484 | -0.451 | 0.591 | | | |
| 2.6747 | -0.414 | 0.631 | | | |
| 2.6989 | -0.385 | 0.638 | | | |
| 2.7237 | -0.354 | 0.616 | | | |
| 2.7488 | -0.307 | 0.628 | | | |
| 2.7733 | -0.202 | 0.642 | | | |
| 2.7993 | -0.215 | 0.652 | | | |
| 2.8243 | -0.101 | 0.666 | | | |
| 2.8489 | -0.103 | 0.681 | | | |
| 2.8769 | 0.006 | 0.718 | | | |

264 1 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.728 ALPHA 0.00 REY 1.65×10^6

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.3060$ $C_M = -0.0966$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.2300 | 1.130 | 0.990 | 0.0000 | 1.130 | 0.990 |
| 2.2335 | 0.443 | 0.822 | 0.0051 | 0.474 | 0.838 |
| 2.2379 | 0.225 | 0.766 | 0.0090 | 0.215 | 0.753 |
| 2.2406 | -0.131 | 0.674 | 0.0201 | -0.014 | 0.724 |
| 2.2294 | -0.299 | 0.631 | 0.0300 | -0.005 | 0.686 |
| 2.2405 | -0.444 | 0.596 | 0.0500 | -0.263 | 0.640 |
| 2.2741 | -0.722 | 0.522 | 0.0749 | -0.366 | 0.614 |
| 2.2991 | -0.792 | 0.514 | 0.1000 | -0.391 | 0.617 |
| 2.1241 | -0.616 | 0.549 | 0.1498 | -0.516 | 0.575 |
| 2.1591 | -0.633 | 0.545 | 0.2000 | -0.499 | 0.579 |
| 2.1994 | -0.594 | 0.555 | 0.2499 | -0.466 | 0.563 |
| 2.2382 | -0.600 | 0.554 | 0.3502 | -0.413 | 0.622 |
| 2.2794 | -0.598 | 0.554 | 0.4500 | -0.298 | 0.631 |
| 2.3188 | -0.607 | 0.552 | 0.5490 | -0.137 | 0.673 |
| 2.3594 | -0.598 | 0.554 | 0.6492 | 0.045 | 0.719 |
| 2.3993 | -0.603 | 0.553 | 0.7490 | 0.190 | 0.759 |
| 2.4194 | -0.593 | 0.554 | 0.8499 | 0.275 | 0.770 |
| 2.4395 | -0.605 | 0.552 | 0.9000 | 0.203 | 0.731 |
| 2.4505 | -0.599 | 0.554 | | | |
| 2.4707 | -0.599 | 0.554 | | | |
| 2.4903 | -0.602 | 0.553 | | | |
| 2.5104 | -0.599 | 0.554 | | | |
| 2.5396 | -0.596 | 0.555 | | | |
| 2.5592 | -0.593 | 0.555 | | | |
| 2.5787 | -0.568 | 0.557 | | | |
| 2.5989 | -0.509 | 0.556 | | | |
| 2.6182 | -0.503 | 0.559 | | | |
| 2.6404 | -0.545 | 0.560 | | | |
| 2.6747 | -0.492 | 0.581 | | | |
| 2.6960 | -0.445 | 0.593 | | | |
| 2.7237 | -0.399 | 0.605 | | | |
| 2.7455 | -0.343 | 0.620 | | | |
| 2.7733 | -0.297 | 0.634 | | | |
| 2.7993 | -0.227 | 0.649 | | | |
| 2.8243 | -0.100 | 0.665 | | | |
| 2.8459 | -0.100 | 0.682 | | | |
| 2.8980 | -0.009 | 0.705 | | | |

8C4 1 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.721 ALPHA 1.83 REY 1.65*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.4674 CM = -0.8927

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.8388 | 1.127 | 0.997 | 0.0000 | 1.127 | 0.997 |
| 2.8335 | 0.186 | 0.755 | 0.0051 | 0.674 | 0.888 |
| 2.8279 | -0.306 | 0.706 | 0.0098 | 0.437 | 0.819 |
| 2.8196 | -0.355 | 0.616 | 0.0201 | 0.199 | 0.757 |
| 2.8294 | -0.509 | 0.576 | 0.0308 | 0.198 | 0.734 |
| 2.8495 | -0.547 | 0.553 | 0.0508 | -0.071 | 0.689 |
| 2.8741 | -0.921 | 0.470 | 0.0749 | -0.188 | 0.656 |
| 2.8991 | -1.156 | 0.401 | 0.1000 | -0.230 | 0.647 |
| 2.1241 | -1.294 | 0.428 | 0.1498 | -0.362 | 0.613 |
| 2.1591 | -1.103 | 0.423 | 0.2008 | -0.374 | 0.612 |
| 2.1994 | -0.728 | 0.519 | 0.2499 | -0.361 | 0.638 |
| 2.2382 | -0.676 | 0.532 | 0.3502 | -0.349 | 0.619 |
| 2.2794 | -0.701 | 0.526 | 0.4508 | -0.253 | 0.641 |
| 2.3188 | -0.713 | 0.523 | 0.5498 | -0.118 | 0.678 |
| 2.3594 | -0.696 | 0.527 | 0.6492 | 0.064 | 0.723 |
| 2.3993 | -0.692 | 0.528 | 0.7498 | 0.215 | 0.762 |
| 2.4395 | -0.686 | 0.530 | 0.8499 | 0.287 | 0.761 |
| 2.4585 | -0.678 | 0.532 | 0.9888 | 0.291 | 0.782 |
| 2.4787 | -0.675 | 0.533 | | | |
| 2.4983 | -0.669 | 0.534 | | | |
| 2.5194 | -0.659 | 0.537 | | | |
| 2.5396 | -0.653 | 0.538 | | | |
| 2.5592 | -0.643 | 0.540 | | | |
| 2.5787 | -0.634 | 0.543 | | | |
| 2.5997 | -0.626 | 0.545 | | | |
| 2.6188 | -0.609 | 0.549 | | | |
| 2.6484 | -0.582 | 0.561 | | | |
| 2.6747 | -0.584 | 0.576 | | | |
| 2.6989 | -0.472 | 0.589 | | | |
| 0.7237 | -0.402 | 0.682 | | | |
| 0.7488 | -0.343 | 0.618 | | | |
| 2.7738 | -0.282 | 0.633 | | | |
| 2.7993 | -0.222 | 0.649 | | | |
| 2.8248 | -0.184 | 0.664 | | | |
| 2.8489 | -0.186 | 0.679 | | | |
| 2.8989 | -0.318 | 0.732 | | | |

BGK 1 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.719 ALPHA 2.00 REY 1.65×10^6

INTEGRATED FORCE COEFFICIENTS
C_N = 0.6516 C_M = -0.0896

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 2.0388 | 1.106 | 2.992 | 0.0000 | 1.106 | 0.992 |
| 2.0335 | -2.308 | 2.686 | 0.0051 | 0.831 | 0.922 |
| 2.0379 | -2.241 | 0.646 | 0.0098 | 0.623 | 0.860 |
| 0.0196 | -3.506 | 2.550 | 0.0201 | 0.376 | 0.805 |
| 0.0294 | -3.712 | 2.526 | 0.0300 | 0.272 | 0.770 |
| 2.0495 | -3.766 | 2.512 | 0.0500 | 0.085 | 0.730 |
| 2.0741 | -1.004 | 2.430 | 0.0749 | -0.041 | 0.690 |
| 2.0991 | -1.316 | 2.370 | 0.1000 | -0.097 | 0.653 |
| 2.1241 | -1.263 | 2.384 | 0.1498 | -0.235 | 0.646 |
| 2.1591 | -1.308 | 2.372 | 0.2000 | -0.263 | 0.641 |
| 2.1994 | -1.271 | 2.382 | 0.2499 | -0.284 | 0.635 |
| 0.2352 | -1.276 | 0.301 | 0.3502 | -0.271 | 0.639 |
| 0.2794 | -1.239 | 0.398 | 0.4500 | -0.285 | 0.656 |
| 0.3158 | -1.232 | 0.392 | 0.5498 | -0.077 | 0.690 |
| 0.3594 | -0.994 | 2.456 | 0.6492 | 0.005 | 0.730 |
| 0.3993 | -0.659 | 2.539 | 0.7490 | 0.234 | 0.760 |
| 0.4194 | -0.625 | 2.548 | 0.8499 | 0.301 | 0.756 |
| 0.4395 | -0.618 | 2.550 | 0.9000 | 0.303 | 0.756 |
| 0.4585 | -0.618 | 2.550 | | | |
| 0.4787 | -0.623 | 2.548 | | | |
| 0.4983 | -0.623 | 2.548 | | | |
| 0.5184 | -0.623 | 2.548 | | | |
| 0.5396 | -0.623 | 2.548 | | | |
| 0.5592 | -0.621 | 2.549 | | | |
| 0.5787 | -0.610 | 2.552 | | | |
| 0.5989 | -0.611 | 2.552 | | | |
| 0.6180 | -0.598 | 2.555 | | | |
| 0.6484 | -0.557 | 2.565 | | | |
| 0.6747 | -0.502 | 2.579 | | | |
| 0.6989 | -0.453 | 2.592 | | | |
| 0.7237 | -0.403 | 2.625 | | | |
| 0.7488 | -0.343 | 2.628 | | | |
| 0.7738 | -0.287 | 2.635 | | | |
| 0.7993 | -0.228 | 2.650 | | | |
| 0.8243 | -0.172 | 2.664 | | | |
| 0.8489 | -0.116 | 2.678 | | | |
| 0.8989 | -0.031 | 2.700 | | | |

AGC 1 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.745 ALPHA -2.58 REY 1.64×10^6

INTEGRATED FORCE COEFFICIENTS
C_N = -0.1132 C_M = -0.1000

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 2.0303 | 0.991 | 2.959 | 0.0000 | 0.991 | 0.959 |
| 2.0335 | 0.930 | 2.945 | 0.0051 | -0.103 | 0.646 |
| 2.0379 | 0.739 | 0.892 | 0.0090 | -0.498 | 0.564 |
| 2.0496 | 0.406 | 0.083 | 0.0201 | -0.655 | 0.520 |
| 2.0294 | 0.229 | 0.756 | 0.0300 | -0.616 | 0.531 |
| 2.0495 | 0.254 | 0.709 | 0.0500 | -0.916 | 0.451 |
| 2.0741 | -0.147 | 0.655 | 0.0749 | -0.984 | 0.432 |
| 2.0991 | -0.229 | 0.634 | 0.1000 | -1.021 | 0.423 |
| 2.1241 | -0.195 | 0.643 | 0.1498 | -1.049 | 0.415 |
| 2.1591 | -0.254 | 0.627 | 0.2000 | -1.140 | 0.391 |
| 2.1994 | -0.206 | 0.624 | 0.2499 | -1.167 | 0.384 |
| 2.2382 | -0.303 | 0.614 | 0.3502 | -0.617 | 0.538 |
| 2.2794 | -0.320 | 0.607 | 0.4500 | -0.367 | 0.597 |
| 2.3109 | -0.301 | 0.599 | 0.5498 | -0.186 | 0.645 |
| 2.3594 | -0.371 | 0.596 | 0.6492 | 0.001 | 0.695 |
| 2.3993 | -0.398 | 0.589 | 0.7498 | 0.134 | 0.731 |
| 2.4194 | -0.403 | 0.587 | 0.8499 | 0.221 | 0.754 |
| 2.4395 | -0.420 | 0.583 | 0.9000 | 0.240 | 0.759 |
| 2.4505 | -0.426 | 0.581 | | | |
| 2.4707 | -0.438 | 0.578 | | | |
| 2.4903 | -0.450 | 0.575 | | | |
| 2.5104 | -0.456 | 0.573 | | | |
| 2.5396 | -0.463 | 0.571 | | | |
| 2.5592 | -0.473 | 0.569 | | | |
| 2.5787 | -0.473 | 0.567 | | | |
| 2.5989 | -0.492 | 0.564 | | | |
| 2.6150 | -0.495 | 0.563 | | | |
| 2.6434 | -0.408 | 0.567 | | | |
| 2.6747 | -0.438 | 0.576 | | | |
| 2.6989 | -0.404 | 0.587 | | | |
| 2.7237 | -0.369 | 0.597 | | | |
| 2.7490 | -0.319 | 0.610 | | | |
| 2.7733 | -0.269 | 0.623 | | | |
| 2.7993 | -0.219 | 0.637 | | | |
| 2.8243 | -0.162 | 0.652 | | | |
| 2.8489 | -0.102 | 0.665 | | | |
| 2.8999 | 0.009 | 0.697 | | | |

BC 1 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.742 ALPHA -1.00 REY 1.64*10⁶

INTEGRATED FORCE COEFFICIENTS
C_x = 0.1448 C_y = -0.8994

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 2.0303 | 1.104 | 0.989 | 0.0000 | 1.104 | 0.989 |
| 2.0335 | 0.698 | 0.881 | 0.0051 | 0.221 | 0.754 |
| 2.0379 | 0.477 | 0.822 | 0.0098 | -0.855 | 0.632 |
| 0.0196 | 0.121 | 0.727 | 0.0201 | -0.263 | 0.625 |
| 2.0294 | -0.353 | 0.681 | 0.0300 | -0.306 | 0.613 |
| 0.2495 | -0.211 | 0.638 | 0.0500 | -0.581 | 0.561 |
| 2.0741 | -0.447 | 0.575 | 0.0749 | -0.597 | 0.536 |
| 2.0991 | -0.513 | 0.559 | 0.1000 | -0.684 | 0.534 |
| 2.1241 | -0.455 | 0.579 | 0.1498 | -0.773 | 0.489 |
| 2.1591 | -0.477 | 0.567 | 0.2000 | -0.786 | 0.547 |
| 2.1994 | -0.464 | 0.571 | 0.2499 | -0.659 | 0.519 |
| 2.2352 | -0.488 | 0.564 | 0.3502 | -0.515 | 0.558 |
| 2.2794 | -0.499 | 0.561 | 0.4500 | -0.357 | 0.600 |
| 2.3189 | -0.519 | 0.556 | 0.5498 | -0.169 | 0.650 |
| 2.3594 | -0.518 | 0.556 | 0.6492 | 0.026 | 0.722 |
| 2.3993 | -0.535 | 0.552 | 0.7498 | 0.174 | 0.741 |
| 2.4104 | -0.535 | 0.552 | 0.8499 | 0.254 | 0.763 |
| 2.4355 | -0.547 | 0.549 | 0.9000 | 0.269 | 0.767 |
| 2.4585 | -0.548 | 0.543 | | | |
| 2.4787 | -0.556 | 0.546 | | | |
| 2.4983 | -0.562 | 0.544 | | | |
| 2.5194 | -0.564 | 0.544 | | | |
| 2.5396 | -0.565 | 0.544 | | | |
| 2.5592 | -0.567 | 0.543 | | | |
| 2.5787 | -0.570 | 0.543 | | | |
| 2.5989 | -0.576 | 0.541 | | | |
| 2.6184 | -0.577 | 0.541 | | | |
| 2.6484 | -0.585 | 0.552 | | | |
| 2.6747 | -0.481 | 0.566 | | | |
| 2.6989 | -0.438 | 0.578 | | | |
| 2.7237 | -0.392 | 0.599 | | | |
| 2.7495 | -0.355 | 0.625 | | | |
| 2.7738 | -0.289 | 0.628 | | | |
| 2.7993 | -0.221 | 0.636 | | | |
| 2.8243 | -0.163 | 0.651 | | | |
| 2.8489 | -0.102 | 0.663 | | | |
| 2.8980 | 0.303 | 0.698 | | | |

SEC 1 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.741 ALPHA 0.00 REY 1.65⁶13

INTEGRATED FORCE COEFFICIENTS

CX = 0.3062 CH = -0.0978

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.7223 | 1.138 | 0.993 | 0.0000 | 1.138 | 0.998 |
| 2.2335 | 0.471 | 0.823 | 0.0051 | 0.472 | 0.821 |
| 0.2379 | 0.203 | 0.764 | 0.0098 | 0.212 | 0.752 |
| 2.2196 | -0.394 | 0.659 | 0.0291 | -0.017 | 0.691 |
| 0.0294 | -0.205 | 0.624 | 0.0380 | -0.087 | 0.672 |
| 2.2435 | -0.349 | 0.588 | 0.0500 | -0.273 | 0.623 |
| 2.2741 | -0.739 | 0.497 | 0.0749 | -0.383 | 0.595 |
| 2.3991 | -0.873 | 0.462 | 0.1000 | -0.414 | 0.555 |
| 2.1241 | -0.609 | 0.532 | 0.1498 | -0.558 | 0.547 |
| 2.1591 | -0.652 | 0.521 | 0.2000 | -0.546 | 0.550 |
| 2.1994 | -0.617 | 0.533 | 0.2499 | -0.531 | 0.553 |
| 2.2352 | -0.631 | 0.527 | 0.3502 | -0.443 | 0.577 |
| 2.2794 | -0.633 | 0.526 | 0.4500 | -0.317 | 0.610 |
| 0.3168 | -0.642 | 0.523 | 0.5498 | -0.247 | 0.656 |
| 2.3594 | -0.634 | 0.526 | 0.6492 | 0.042 | 0.726 |
| 2.3993 | -0.639 | 0.524 | 0.7498 | 0.195 | 0.747 |
| 2.4194 | -0.633 | 0.526 | 0.8499 | 0.272 | 0.758 |
| 2.4395 | -0.641 | 0.524 | 0.9000 | 0.281 | 0.770 |
| 0.4555 | -0.638 | 0.525 | | | |
| 2.4797 | -0.641 | 0.524 | | | |
| 2.4993 | -0.640 | 0.524 | | | |
| 2.5134 | -0.635 | 0.525 | | | |
| 2.5396 | -0.632 | 0.526 | | | |
| 2.5592 | -0.632 | 0.527 | | | |
| 2.5787 | -0.623 | 0.529 | | | |
| 2.5983 | -0.619 | 0.530 | | | |
| 2.6193 | -0.607 | 0.533 | | | |
| 2.6494 | -0.581 | 0.546 | | | |
| 2.6747 | -0.499 | 0.562 | | | |
| 2.6999 | -0.447 | 0.576 | | | |
| 2.7237 | -0.393 | 0.589 | | | |
| 2.7438 | -0.336 | 0.605 | | | |
| 2.7733 | -0.280 | 0.620 | | | |
| 2.7993 | -0.219 | 0.637 | | | |
| 2.8248 | -0.190 | 0.652 | | | |
| 2.8489 | -0.141 | 0.668 | | | |
| 2.8999 | -0.009 | 0.693 | | | |

3C4 1 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.748 ALPHA 1.88 REY 1.64×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.4939$ $C_M = -0.8944$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.2328 | 1.142 | 0.999 | 0.0000 | 1.142 | 0.999 |
| 2.2335 | 0.244 | 0.768 | 0.0051 | 0.668 | 0.873 |
| 2.2379 | 0.258 | 0.788 | 0.0098 | 0.432 | 0.829 |
| 2.2196 | -2.299 | 2.615 | 0.0201 | 0.189 | 0.745 |
| 2.2294 | -3.451 | 2.574 | 0.0308 | 0.188 | 0.721 |
| 2.2495 | -3.542 | 2.558 | 0.0500 | -0.001 | 2.672 |
| 2.2741 | -3.872 | 2.462 | 0.0745 | -0.283 | 0.643 |
| 2.2991 | -1.118 | 2.397 | 0.1088 | -0.248 | 0.626 |
| 2.1241 | -1.358 | 2.412 | 0.1498 | -0.389 | 0.598 |
| 2.1591 | -1.378 | 2.409 | 0.2000 | -0.482 | 0.597 |
| 2.1974 | -1.348 | 2.415 | 0.2499 | -0.427 | 0.595 |
| 2.2382 | -1.312 | 2.425 | 0.3582 | -0.365 | 0.597 |
| 2.2794 | -2.973 | 2.435 | 0.4588 | -0.268 | 0.622 |
| 2.3158 | -3.807 | 2.479 | 0.5492 | -0.116 | 0.663 |
| 2.3594 | -0.636 | 2.524 | 0.6492 | 0.062 | 0.711 |
| 2.3993 | -0.606 | 2.517 | 0.7498 | 0.215 | 0.751 |
| 2.4194 | -3.681 | 2.513 | 0.8499 | 0.287 | 0.771 |
| 2.4395 | -3.782 | 2.587 | 0.9888 | 0.291 | 0.772 |
| 2.4585 | -0.704 | 2.586 | | | |
| 2.4787 | -0.788 | 2.585 | | | |
| 2.4983 | -3.785 | 2.586 | | | |
| 2.5184 | -2.695 | 2.589 | | | |
| 2.5396 | -2.686 | 2.511 | | | |
| 2.5592 | -2.674 | 2.514 | | | |
| 2.5787 | -2.691 | 2.517 | | | |
| 2.5989 | -2.658 | 2.528 | | | |
| 2.6188 | -2.627 | 2.526 | | | |
| 2.6484 | -4.572 | 2.541 | | | |
| 2.6747 | -4.585 | 2.558 | | | |
| 2.6989 | -0.451 | 2.573 | | | |
| 2.7237 | -0.396 | 2.588 | | | |
| 2.7486 | -2.334 | 2.605 | | | |
| 2.7738 | -2.275 | 2.621 | | | |
| 2.7992 | -2.215 | 2.637 | | | |
| 2.8248 | -2.157 | 2.652 | | | |
| 2.8499 | -2.182 | 2.667 | | | |
| 2.8789 | -2.328 | 2.689 | | | |

364 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.742 ALPHA 2.00 REY 1.64*10⁶

INTEGRATED FORCE COEFFICIENTS

C_x = 0.6265 C_m = -0.8982

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.2222 | 1.123 | 2.994 | 0.0000 | 1.123 | 0.994 |
| 2.2035 | 0.826 | 2.781 | 0.0051 | 0.821 | 0.988 |
| 2.2379 | -3.131 | 2.659 | 0.0098 | 0.589 | 0.851 |
| 2.2196 | -3.473 | 2.568 | 0.0201 | 0.347 | 0.786 |
| 2.2294 | -3.674 | 2.533 | 0.0300 | 0.247 | 0.762 |
| 2.2495 | -3.605 | 2.516 | 0.0500 | 0.068 | 0.718 |
| 2.2741 | -3.971 | 2.431 | 0.0749 | -0.066 | 0.676 |
| 2.2991 | -1.289 | 2.371 | 0.1008 | -0.121 | 0.661 |
| 2.1241 | -1.195 | 2.383 | 0.1498 | -0.267 | 0.622 |
| 2.1501 | -1.215 | 2.373 | 0.2002 | -0.297 | 0.614 |
| 2.1994 | -1.133 | 2.377 | 0.2499 | -0.328 | 0.639 |
| 2.2382 | -1.201 | 2.372 | 0.3502 | -0.386 | 0.612 |
| 2.2794 | -1.183 | 2.378 | 0.4500 | -0.235 | 0.631 |
| 2.3188 | -1.188 | 2.376 | 0.5498 | -0.148 | 0.667 |
| 2.3594 | -1.191 | 2.378 | 0.6492 | 0.069 | 0.712 |
| 2.3993 | -1.206 | 2.371 | 0.7498 | 0.221 | 0.753 |
| 2.4194 | -1.192 | 2.375 | 0.8499 | 0.298 | 0.771 |
| 2.4495 | -1.218 | 2.378 | 0.9002 | 0.298 | 0.771 |
| 2.4787 | -1.198 | 2.373 | | | |
| 2.4983 | -1.129 | 2.368 | | | |
| 2.5184 | -0.939 | 2.392 | | | |
| 2.5393 | -0.607 | 2.469 | | | |
| 2.5592 | -0.595 | 2.515 | | | |
| 2.5797 | -0.593 | 2.534 | | | |
| 2.5989 | -0.518 | 2.546 | | | |
| 2.6193 | -0.498 | 2.555 | | | |
| 2.6484 | -0.498 | 2.563 | | | |
| 2.6747 | -0.406 | 2.573 | | | |
| 2.6989 | -0.307 | 2.585 | | | |
| 2.7237 | -0.327 | 2.595 | | | |
| 2.7488 | -0.278 | 2.686 | | | |
| 2.7738 | -0.237 | 2.619 | | | |
| 2.7990 | -0.191 | 0.630 | | | |
| 2.8248 | -0.171 | 0.643 | | | |
| 2.8489 | -0.115 | 0.653 | | | |
| 2.8989 | -0.056 | 0.663 | | | |
| | | 0.678 | | | |

36X 1 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

WACH NO. W.729 ALP4A -2.50 REY 1.63*10⁶

INTEGRATED FORCE COEFFICIENTS
C_x = -0.1525 C_m = -0.0735

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.2320 | 1.316 | 2.962 | 0.8882 | 1.816 | 0.952 |
| 0.2335 | 2.938 | 2.941 | 0.8851 | -0.121 | 0.649 |
| 0.2379 | 2.735 | 2.885 | 0.8898 | -0.421 | 0.556 |
| 0.2156 | 0.406 | 2.794 | 0.8281 | -0.588 | 0.528 |
| 0.2244 | 0.260 | 2.746 | 0.8300 | -0.552 | 0.532 |
| 0.2495 | 0.354 | 2.697 | 0.8508 | -0.853 | 0.447 |
| 0.2741 | -0.195 | 2.539 | 0.8749 | -0.910 | 0.431 |
| 0.2991 | -0.243 | 2.616 | 0.1380 | -0.968 | 0.415 |
| 0.1241 | -0.287 | 2.625 | 0.1498 | -1.211 | 0.423 |
| 0.1591 | -0.272 | 2.624 | 0.2303 | -1.208 | 0.352 |
| 0.1994 | -0.253 | 2.624 | 0.2499 | -1.155 | 0.353 |
| 0.2362 | -0.322 | 2.593 | 0.3582 | -0.696 | 0.442 |
| 0.2794 | -0.349 | 2.586 | 0.4580 | -0.418 | 0.557 |
| 0.3188 | -0.353 | 2.576 | 0.5492 | -0.265 | 0.609 |
| 0.3594 | -0.392 | 2.574 | 0.6492 | -0.140 | 0.643 |
| 0.3973 | -0.425 | 2.565 | 0.7498 | -0.239 | 0.671 |
| 0.4194 | -0.432 | 2.563 | 0.8499 | 0.257 | 0.677 |
| 0.4395 | -0.452 | 2.558 | 0.9808 | 0.108 | 0.712 |
| 0.4555 | -0.459 | 2.556 | | | |
| 0.4787 | -0.474 | 2.551 | | | |
| 0.4993 | -0.489 | 2.547 | | | |
| 0.5184 | -0.500 | 2.544 | | | |
| 0.5396 | -0.512 | 2.542 | | | |
| 0.5592 | -0.523 | 2.538 | | | |
| 0.5787 | -0.514 | 2.535 | | | |
| 0.5989 | -0.548 | 2.531 | | | |
| 0.6183 | -0.549 | 2.531 | | | |
| 0.6484 | -0.523 | 2.538 | | | |
| 0.6747 | -0.473 | 2.552 | | | |
| 0.6959 | -0.433 | 2.563 | | | |
| 0.7237 | -0.392 | 2.575 | | | |
| 0.7488 | -0.358 | 2.589 | | | |
| 0.7735 | -0.295 | 2.623 | | | |
| 0.7990 | -0.251 | 2.618 | | | |
| 0.8243 | -0.175 | 2.634 | | | |
| 0.8489 | -0.113 | 2.651 | | | |
| 0.8989 | -0.202 | 2.682 | | | |

BGK 1 283.2 MM CHORD
 EXPERIMENTAL PRESSURE DISTRIBUTION
 SOLID WALLS

MACH NO. 0.759 ALPHA -1.00 REY $1.63 \cdot 10^6$

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.1335$ $C_M = -0.1038$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|------------------|----------------------|--------|------------------|
| X/C | CP | P/P _∞ | X/C | CP | P/P _∞ |
| 0.0700 | 1.114 | 0.989 | 0.0000 | 1.114 | 0.989 |
| 0.0835 | 2.718 | 2.088 | 0.0051 | 0.233 | 0.746 |
| 0.0879 | 0.508 | 2.628 | 0.0098 | -0.040 | 0.670 |
| 0.0196 | 0.116 | 2.722 | 0.0201 | -0.250 | 0.612 |
| 0.0294 | -0.249 | 2.674 | 0.0308 | -0.209 | 0.641 |
| 0.0495 | -0.105 | 2.638 | 0.0500 | -0.504 | 0.542 |
| 0.0741 | -0.444 | 2.559 | 0.0749 | -0.603 | 0.515 |
| 0.0991 | -0.520 | 2.533 | 0.1002 | -0.616 | 0.511 |
| 0.1241 | -0.430 | 2.561 | 0.1498 | -0.765 | 0.478 |
| 0.1591 | -0.406 | 2.547 | 0.2000 | -0.875 | 0.439 |
| 0.1994 | -0.479 | 2.554 | 0.2499 | -0.930 | 0.424 |
| 0.2382 | -0.505 | 2.541 | 0.3502 | -0.585 | 0.542 |
| 0.2794 | -0.523 | 2.537 | 0.4500 | -0.360 | 0.562 |
| 0.3188 | -0.549 | 2.529 | 0.5498 | -0.170 | 0.634 |
| 0.3594 | -0.549 | 2.529 | 0.6492 | 0.027 | 0.639 |
| 0.3993 | -0.575 | 2.522 | 0.7498 | 0.172 | 0.729 |
| 0.4194 | -0.578 | 2.521 | 0.8499 | 0.293 | 0.751 |
| 0.4395 | -0.592 | 2.517 | 0.9000 | 0.265 | 0.754 |
| 0.4585 | -0.595 | 2.517 | | | |
| 0.4787 | -0.607 | 2.513 | | | |
| 0.4983 | -0.617 | 2.511 | | | |
| 0.5184 | -0.618 | 2.517 | | | |
| 0.5396 | -0.623 | 2.519 | | | |
| 0.5592 | -0.627 | 2.524 | | | |
| 0.5787 | -0.625 | 2.528 | | | |
| 0.5989 | -0.630 | 2.527 | | | |
| 0.6187 | -0.610 | 2.511 | | | |
| 0.6484 | -0.592 | 2.524 | | | |
| 0.6747 | -0.474 | 2.545 | | | |
| 0.6999 | -0.442 | 2.559 | | | |
| 0.7237 | -0.398 | 2.573 | | | |
| 0.7488 | -0.320 | 2.591 | | | |
| 0.7738 | -0.273 | 2.605 | | | |
| 0.7990 | -0.212 | 2.621 | | | |
| 0.8240 | -0.173 | 2.639 | | | |
| 0.8489 | -0.292 | 2.656 | | | |
| 0.8989 | 0.305 | 2.693 | | | |

UGC 1 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.760 ALPHA 0.20 REY 1.63×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.3056$ $C_M = -0.1000$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.145 | 0.998 | 0.0000 | 1.145 | 0.998 |
| 0.0035 | 0.511 | 0.823 | 0.0051 | 0.466 | 0.811 |
| 0.0079 | 0.348 | 0.765 | 0.0090 | 0.284 | 0.739 |
| 0.0196 | -0.351 | 0.668 | 0.0201 | -0.825 | 0.676 |
| 0.0294 | -0.241 | 0.621 | 0.0300 | -0.893 | 0.657 |
| 0.0495 | -0.355 | 0.584 | 0.0500 | -0.285 | 0.624 |
| 0.0741 | -0.688 | 0.493 | 0.0749 | -0.481 | 0.572 |
| 0.0991 | -0.552 | 0.447 | 0.1000 | -0.433 | 0.563 |
| 0.1241 | -0.771 | 0.470 | 0.1498 | -0.687 | 0.516 |
| 0.1591 | -0.610 | 0.514 | 0.2000 | -0.691 | 0.516 |
| 0.1994 | -0.540 | 0.508 | 0.2499 | -0.550 | 0.522 |
| 0.2382 | -0.654 | 0.502 | 0.3502 | -0.401 | 0.530 |
| 0.2794 | -0.608 | 0.498 | 0.4500 | -0.335 | 0.540 |
| 0.3168 | -0.694 | 0.491 | 0.5493 | -0.154 | 0.642 |
| 0.3594 | -0.608 | 0.493 | 0.6492 | 0.040 | 0.693 |
| 0.3993 | -0.709 | 0.487 | 0.7490 | 0.191 | 0.735 |
| 0.4194 | -0.705 | 0.488 | 0.8499 | 0.268 | 0.756 |
| 0.4395 | -0.714 | 0.486 | 0.9000 | 0.276 | 0.759 |
| 0.4585 | -0.707 | 0.488 | | | |
| 0.4757 | -0.704 | 0.489 | | | |
| 0.4983 | -0.697 | 0.493 | | | |
| 0.5184 | -0.694 | 0.491 | | | |
| 0.5396 | -0.601 | 0.495 | | | |
| 0.5592 | -0.671 | 0.498 | | | |
| 0.5787 | -0.650 | 0.501 | | | |
| 0.5989 | -0.634 | 0.508 | | | |
| 0.6188 | -0.507 | 0.527 | | | |
| 0.6494 | -0.447 | 0.546 | | | |
| 0.6747 | -0.441 | 0.561 | | | |
| 0.6959 | -0.387 | 0.576 | | | |
| 0.7237 | -0.325 | 0.593 | | | |
| 0.7488 | -0.265 | 0.610 | | | |
| 0.7735 | -0.205 | 0.626 | | | |
| 0.7998 | -0.147 | 0.642 | | | |
| 0.8243 | -0.091 | 0.658 | | | |
| 0.8489 | -0.088 | 0.680 | | | |
| 0.8999 | | | | | |

804 1 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.759 ALPHA 1.20 REY 1.63×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.4987$ $C_M = -0.8972$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 1.198 | 2.999 | 0.0000 | 1.148 | 2.999 |
| 0.0035 | 0.302 | 2.766 | 0.0051 | 0.652 | 2.852 |
| 0.0079 | 0.112 | 2.713 | 0.0098 | 0.413 | 2.776 |
| 0.0196 | -0.232 | 2.619 | 0.0201 | 0.172 | 2.730 |
| 0.0294 | -0.304 | 2.577 | 0.0300 | 0.005 | 2.736 |
| 0.0495 | -0.479 | 2.551 | 0.0508 | -0.101 | 2.855 |
| 0.0741 | -0.803 | 2.461 | 0.0749 | -0.225 | 2.621 |
| 0.0991 | -1.144 | 2.395 | 0.1002 | -0.271 | 2.618 |
| 0.1241 | -1.997 | 2.488 | 0.1498 | -0.426 | 2.566 |
| 0.1591 | -1.311 | 2.484 | 0.2000 | -0.443 | 2.561 |
| 0.1994 | -1.302 | 2.427 | 0.2499 | -0.453 | 2.558 |
| 0.2352 | -1.964 | 2.412 | 0.3502 | -0.481 | 2.572 |
| 0.2754 | -1.969 | 2.416 | 0.4500 | -0.292 | 2.602 |
| 0.3188 | -1.968 | 2.411 | 0.5492 | -0.131 | 2.647 |
| 0.3594 | -1.971 | 2.415 | 0.6492 | 0.054 | 2.698 |
| 0.3993 | -1.980 | 2.413 | 0.7496 | 0.207 | 2.740 |
| 0.4194 | -1.971 | 2.415 | 0.8495 | 0.288 | 2.768 |
| 0.4335 | -1.966 | 2.411 | 0.9022 | 0.284 | 2.761 |
| 0.4545 | -1.966 | 2.411 | | | |
| 0.4737 | -1.968 | 2.411 | | | |
| 0.4953 | -1.959 | 2.416 | | | |
| 0.5134 | -1.954 | 2.453 | | | |
| 0.5376 | -1.651 | 2.534 | | | |
| 0.5592 | -1.577 | 2.524 | | | |
| 0.5757 | -1.553 | 2.530 | | | |
| 0.5939 | -1.546 | 2.532 | | | |
| 0.6130 | -1.537 | 2.535 | | | |
| 0.6404 | -1.503 | 2.544 | | | |
| 0.6747 | -1.448 | 2.559 | | | |
| 0.6989 | -1.403 | 2.572 | | | |
| 0.7237 | -1.352 | 2.586 | | | |
| 0.7448 | -1.296 | 2.601 | | | |
| 0.7738 | -1.244 | 2.616 | | | |
| 0.7990 | -1.188 | 2.631 | | | |
| 0.8243 | -1.137 | 2.645 | | | |
| 0.8499 | -1.090 | 2.658 | | | |
| 0.8999 | -1.027 | 2.675 | | | |

364 1 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.768 ALP4A 2.88 REY 1.63e6

INTEGRATED FORCE COEFFICIENTS

C_L = 0.5750 C_M = -0.0973

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 1.143 | 0.997 | 0.0000 | 1.143 | 0.997 |
| 0.0335 | 0.165 | 0.727 | 0.0335 | 0.756 | 0.890 |
| 0.0679 | -0.318 | 0.670 | 0.0679 | 0.535 | 0.838 |
| 0.1019 | -0.358 | 0.585 | 0.1019 | 0.298 | 0.762 |
| 0.1359 | -0.409 | 0.546 | 0.1359 | 0.195 | 0.735 |
| 0.1699 | -0.558 | 0.527 | 0.1699 | 0.084 | 0.653 |
| 0.2039 | -0.606 | 0.437 | 0.2039 | -0.126 | 0.647 |
| 0.2379 | -1.046 | 0.379 | 0.2379 | -0.162 | 0.631 |
| 0.2719 | -1.063 | 0.380 | 0.2719 | -0.340 | 0.550 |
| 0.3059 | -1.114 | 0.374 | 0.3059 | -0.373 | 0.579 |
| 0.3399 | -1.044 | 0.379 | 0.3399 | -0.399 | 0.572 |
| 0.3739 | -1.109 | 0.375 | 0.3739 | -0.374 | 0.578 |
| 0.4079 | -1.393 | 0.380 | 0.4079 | -0.293 | 0.601 |
| 0.4419 | -1.154 | 0.377 | 0.4419 | -0.144 | 0.642 |
| 0.4759 | -1.107 | 0.376 | 0.4759 | 0.033 | 0.691 |
| 0.5099 | -1.152 | 0.369 | 0.5099 | 0.196 | 0.733 |
| 0.5439 | -1.122 | 0.371 | 0.5439 | 0.252 | 0.751 |
| 0.5779 | -1.143 | 0.366 | 0.5779 | 0.245 | 0.750 |
| 0.6119 | -1.134 | 0.368 | | | |
| 0.6459 | -1.155 | 0.362 | | | |
| 0.6799 | -1.014 | 0.421 | | | |
| 0.7139 | -0.705 | 0.465 | | | |
| 0.7479 | -0.625 | 0.529 | | | |
| 0.7819 | -0.533 | 0.533 | | | |
| 0.8159 | -0.498 | 0.544 | | | |
| 0.8499 | -0.402 | 0.554 | | | |
| 0.8839 | -0.435 | 0.561 | | | |
| 0.9179 | -0.395 | 0.572 | | | |
| 0.9519 | -0.353 | 0.584 | | | |
| 0.9859 | -0.327 | 0.591 | | | |
| 1.0199 | -0.305 | 0.597 | | | |
| 1.0539 | -0.256 | 0.602 | | | |
| 1.0879 | -0.208 | 0.607 | | | |
| 1.1219 | -0.251 | 0.612 | | | |
| 1.1559 | -0.234 | 0.617 | | | |
| 1.1899 | -0.214 | 0.623 | | | |
| 1.2239 | -0.194 | 0.623 | | | |

FIG 1 293.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.792 ALPHA -2.50 REY 1.61⁶10

INTEGRATED FORCE COEFFICIENTS

C_x = -0.0130 C_m = -0.0023

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.396 | 3.981 | 0.0000 | 1.096 | 3.961 |
| 0.0035 | 0.815 | 3.901 | 0.0051 | 0.150 | 0.710 |
| 0.0079 | 0.549 | 3.839 | 0.0098 | -0.125 | 0.631 |
| 0.0196 | 0.299 | 3.741 | 0.0201 | -0.321 | 0.575 |
| 0.0294 | 0.081 | 3.691 | 0.0302 | -0.337 | 0.573 |
| 0.0495 | -0.204 | 3.644 | 0.0503 | -0.584 | 0.520 |
| 0.0741 | -0.328 | 3.574 | 0.0749 | -0.672 | 0.474 |
| 0.0991 | -0.416 | 3.548 | 0.1002 | -0.663 | 0.471 |
| 0.1241 | -0.392 | 3.567 | 0.1249 | -0.760 | 0.443 |
| 0.1591 | -0.411 | 3.550 | 0.2002 | -0.957 | 0.343 |
| 0.1994 | -0.413 | 3.549 | 0.2499 | -0.976 | 0.350 |
| 0.2392 | -0.450 | 3.538 | 0.3502 | -1.001 | 0.358 |
| 0.2794 | -0.476 | 3.531 | 0.4503 | -0.390 | 0.556 |
| 0.3198 | -0.512 | 3.520 | 0.5498 | -0.214 | 0.606 |
| 0.3594 | -0.523 | 3.517 | 0.6492 | -0.054 | 0.649 |
| 0.3993 | -0.595 | 3.528 | 0.7498 | 0.050 | 0.602 |
| 0.4194 | -0.500 | 3.587 | 0.8499 | 0.159 | 0.713 |
| 0.4395 | -0.502 | 3.580 | 0.9000 | 0.193 | 0.722 |
| 0.4595 | -0.542 | 3.498 | | | |
| 0.4767 | -0.612 | 3.492 | | | |
| 0.4981 | -0.627 | 3.488 | | | |
| 0.5184 | -0.642 | 3.484 | | | |
| 0.5396 | -0.648 | 3.482 | | | |
| 0.5592 | -0.666 | 3.476 | | | |
| 0.5787 | -0.678 | 3.473 | | | |
| 0.5989 | -0.704 | 3.465 | | | |
| 0.6183 | -0.713 | 3.463 | | | |
| 0.6484 | -0.617 | 3.490 | | | |
| 0.6747 | -0.405 | 3.528 | | | |
| 0.6989 | -0.423 | 3.546 | | | |
| 0.7237 | -0.370 | 3.561 | | | |
| 0.7488 | -0.318 | 3.578 | | | |
| 0.7738 | -0.292 | 3.595 | | | |
| 0.7990 | -0.192 | 3.612 | | | |
| 0.8240 | -0.133 | 3.629 | | | |
| 0.8489 | -0.175 | 3.645 | | | |
| 0.8989 | 0.019 | 3.672 | | | |

PG 1 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.753 ALPHA -1.02 REY 1.62×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.2995$ $C_M = -0.1029$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 1.121 | 2.988 | 0.8000 | 1.121 | 0.990 |
| 0.0035 | 0.745 | 2.581 | 0.8051 | 0.246 | 0.738 |
| 0.0079 | 0.527 | 2.619 | 0.8098 | -0.821 | 0.662 |
| 0.0196 | 0.119 | 2.720 | 0.8231 | -0.229 | 0.643 |
| 0.0294 | 0.002 | 2.669 | 0.8383 | -0.265 | 0.592 |
| 0.0495 | -0.157 | 2.624 | 0.8533 | -0.489 | 0.528 |
| 0.0741 | -0.423 | 2.543 | 0.8749 | -0.502 | 0.502 |
| 0.0991 | -0.516 | 2.522 | 0.1000 | -0.615 | 0.492 |
| 0.1241 | -0.438 | 2.546 | 0.1498 | -0.762 | 0.452 |
| 0.1591 | -0.487 | 2.533 | 0.2088 | -0.833 | 0.432 |
| 0.1994 | -0.479 | 2.532 | 0.2499 | -0.926 | 0.413 |
| 0.2392 | -0.513 | 2.522 | 0.3582 | -0.997 | 0.363 |
| 0.2794 | -0.536 | 2.515 | 0.4520 | -0.339 | 2.571 |
| 0.3128 | -0.570 | 2.505 | 0.5493 | -0.157 | 2.623 |
| 0.3594 | -0.581 | 2.502 | 0.6492 | 0.021 | 2.674 |
| 0.3993 | -0.513 | 2.493 | 0.7498 | 0.155 | 0.712 |
| 0.4194 | -0.617 | 2.492 | 0.8499 | 0.241 | 0.737 |
| 0.4395 | -0.538 | 2.486 | 0.9020 | 0.256 | 0.741 |
| 0.4585 | -0.644 | 2.484 | | | |
| 0.4787 | -0.656 | 2.475 | | | |
| 0.4983 | -0.678 | 2.475 | | | |
| 0.5184 | -0.694 | 2.477 | | | |
| 0.5396 | -0.702 | 2.468 | | | |
| 0.5592 | -0.719 | 2.463 | | | |
| 0.5787 | -0.731 | 2.459 | | | |
| 0.5989 | -0.763 | 2.453 | | | |
| 0.6180 | -0.791 | 2.445 | | | |
| 0.6484 | -0.855 | 2.473 | | | |
| 0.6747 | -0.481 | 2.533 | | | |
| 0.6989 | -0.417 | 2.549 | | | |
| 0.7237 | -0.393 | 2.564 | | | |
| 0.7488 | -0.384 | 2.581 | | | |
| 0.7738 | -0.297 | 2.593 | | | |
| 0.7998 | -0.187 | 2.615 | | | |
| 0.8248 | -0.131 | 2.631 | | | |
| 0.8459 | -0.076 | 2.646 | | | |
| 0.8699 | 0.208 | 2.670 | | | |

SEC 1 233.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.778 ALPHA 0.00 REY 1.60×10^6

INTEGRATED FORCE COEFFICIENTS
C_L = 0.3088 C_D = -0.1059

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.122 | 3.997 | 0.0000 | 1.152 | 2.997 |
| 0.0035 | 0.506 | 2.831 | 0.0051 | 0.445 | 2.795 |
| 0.0079 | 0.357 | 2.771 | 0.0098 | 0.188 | 2.722 |
| 0.0125 | 0.313 | 2.673 | 0.0231 | -0.237 | 2.658 |
| 0.0294 | -0.191 | 2.624 | 0.0370 | -0.102 | 2.642 |
| 0.0495 | -0.298 | 2.585 | 0.0502 | -0.308 | 2.553 |
| 0.0741 | -0.625 | 2.492 | 0.0749 | -0.428 | 2.549 |
| 0.0991 | -0.795 | 2.444 | 0.1003 | -0.452 | 2.539 |
| 0.1241 | -0.763 | 2.453 | 0.1498 | -0.642 | 2.456 |
| 0.1591 | -0.794 | 2.455 | 0.2020 | -0.726 | 2.461 |
| 0.1994 | -0.545 | 2.506 | 0.2499 | -0.723 | 2.451 |
| 0.2352 | -0.614 | 2.495 | 0.3502 | -0.498 | 2.526 |
| 0.2794 | -1.654 | 2.484 | 0.4502 | -0.356 | 2.506 |
| 0.3138 | -0.698 | 2.471 | 0.5492 | -0.167 | 2.622 |
| 0.3594 | -0.714 | 2.467 | 0.6492 | 0.029 | 2.676 |
| 0.3993 | -0.745 | 2.459 | 0.7498 | 0.179 | 2.719 |
| 0.4394 | -0.751 | 2.455 | 0.8499 | 0.257 | 2.741 |
| 0.4395 | -0.774 | 2.453 | 0.9200 | 0.264 | 2.743 |
| 0.4565 | -0.782 | 2.448 | | | |
| 0.4787 | -0.805 | 2.441 | | | |
| 0.4983 | -0.819 | 2.437 | | | |
| 0.5144 | -0.838 | 2.432 | | | |
| 0.5396 | -0.847 | 2.429 | | | |
| 0.5592 | -0.864 | 2.424 | | | |
| 0.5757 | -0.874 | 2.421 | | | |
| 0.5945 | -0.906 | 2.412 | | | |
| 0.6133 | -0.905 | 2.412 | | | |
| 0.6444 | -0.625 | 2.492 | | | |
| 0.6747 | -0.444 | 2.543 | | | |
| 0.6989 | -0.372 | 2.563 | | | |
| 0.7237 | -0.317 | 2.579 | | | |
| 0.7465 | -0.281 | 2.595 | | | |
| 0.7738 | -0.212 | 2.629 | | | |
| 0.7992 | -0.194 | 2.623 | | | |
| 0.8243 | -0.142 | 2.635 | | | |
| 0.8459 | -0.083 | 2.645 | | | |
| 0.8959 | -0.333 | 2.659 | | | |

SEC 1 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.792 ALPHA 1.00 REY 1.61e10⁶

INTEGRATED FORCE COEFFICIENTS

C_x = 0.4383 C_y = -0.1813

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0232 | 1.124 | 2.993 | 0.0380 | 1.154 | 0.958 |
| 0.0335 | 3.417 | 2.787 | 0.0551 | 0.634 | 2.841 |
| 0.0379 | 3.216 | 2.732 | 0.0698 | 0.363 | 0.772 |
| 0.0496 | -0.123 | 2.633 | 0.0801 | 0.127 | 0.745 |
| 0.0594 | -0.202 | 2.568 | 0.0930 | 0.045 | 0.632 |
| 0.0695 | -0.357 | 2.558 | 0.1000 | -0.148 | 0.627 |
| 0.0741 | -0.393 | 2.471 | 0.1074 | -0.273 | 0.541 |
| 0.0791 | -0.392 | 2.420 | 0.1100 | -0.321 | 0.577 |
| 0.1241 | -0.389 | 2.415 | 0.1498 | -0.522 | 0.526 |
| 0.1591 | -0.915 | 2.428 | 0.2000 | -0.531 | 0.517 |
| 0.1794 | -0.717 | 2.439 | 0.2499 | -0.551 | 0.513 |
| 0.2352 | -0.398 | 2.412 | 0.3582 | -0.475 | 0.534 |
| 0.2794 | -0.491 | 2.414 | 0.4502 | -0.338 | 0.573 |
| 0.3155 | -0.315 | 2.437 | 0.5490 | -0.161 | 0.624 |
| 0.3594 | -0.907 | 2.410 | 0.6492 | 0.030 | 0.678 |
| 0.3993 | -0.927 | 2.424 | 0.7498 | 0.183 | 0.722 |
| 0.4194 | -0.923 | 2.425 | 0.8499 | 0.257 | 0.743 |
| 0.4395 | -0.943 | 2.428 | 0.9388 | 0.256 | 0.742 |
| 0.4555 | -0.949 | 2.398 | | | |
| 0.4737 | -0.959 | 2.392 | | | |
| 0.4953 | -0.975 | 2.389 | | | |
| 0.5154 | -1.001 | 2.383 | | | |
| 0.5396 | -1.007 | 2.381 | | | |
| 0.5592 | -1.014 | 2.379 | | | |
| 0.5737 | -0.920 | 2.426 | | | |
| 0.5969 | -0.674 | 2.475 | | | |
| 0.6180 | -0.525 | 2.519 | | | |
| 0.6454 | -0.401 | 2.552 | | | |
| 0.6747 | -0.342 | 2.571 | | | |
| 0.6959 | -0.297 | 2.584 | | | |
| 0.7237 | -0.266 | 2.593 | | | |
| 0.7488 | -0.233 | 2.602 | | | |
| 0.7733 | -0.206 | 2.610 | | | |
| 0.7990 | -0.177 | 2.613 | | | |
| 0.8248 | -0.157 | 2.624 | | | |
| 0.8489 | -0.134 | 2.631 | | | |
| 0.8759 | -0.113 | 2.637 | | | |

PG 1 233.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.731 ALPHA 2.32 REY 1.61×10^6

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.4755$ $C_M = -0.0949$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.195 | 2.993 | 0.0000 | 1.155 | 3.993 |
| 0.0335 | 0.205 | 2.750 | 0.0351 | 0.703 | 3.870 |
| 0.0670 | 0.102 | 2.697 | 0.0698 | 0.472 | 2.814 |
| 0.1005 | -0.236 | 2.681 | 0.1041 | 0.233 | 0.715 |
| 0.1340 | -0.301 | 2.559 | 0.1380 | 0.140 | 3.712 |
| 0.1675 | -0.403 | 2.535 | 0.1720 | -0.053 | 3.655 |
| 0.2010 | -0.709 | 2.443 | 0.2060 | -0.186 | 2.617 |
| 0.2345 | -0.902 | 2.368 | 0.2400 | -0.242 | 2.601 |
| 0.2680 | -0.905 | 2.393 | 0.2740 | -0.421 | 2.503 |
| 0.3015 | -1.007 | 2.381 | 0.3080 | -0.466 | 2.537 |
| 0.3350 | -0.999 | 2.383 | 0.3420 | -0.505 | 2.526 |
| 0.3685 | -1.017 | 2.379 | 0.3760 | -0.479 | 2.534 |
| 0.4020 | -1.002 | 2.383 | 0.4100 | -0.363 | 2.557 |
| 0.4355 | -1.016 | 2.379 | 0.4440 | -0.191 | 2.616 |
| 0.4690 | -1.024 | 2.377 | 0.4780 | 0.001 | 2.672 |
| 0.5025 | -1.008 | 2.369 | 0.5120 | 0.153 | 2.713 |
| 0.5360 | -1.005 | 2.371 | 0.5460 | 0.217 | 2.732 |
| 0.5695 | -1.005 | 2.365 | 0.5800 | 0.227 | 2.729 |
| 0.6030 | -1.001 | 2.360 | | | |
| 0.6365 | -1.002 | 2.363 | | | |
| 0.6700 | -0.993 | 2.422 | | | |
| 0.7035 | -0.992 | 2.481 | | | |
| 0.7370 | -0.999 | 2.509 | | | |
| 0.7705 | -0.993 | 2.526 | | | |
| 0.8040 | -0.994 | 2.540 | | | |
| 0.8375 | -0.994 | 2.549 | | | |
| 0.8710 | -0.992 | 2.555 | | | |
| 0.9045 | -0.996 | 2.562 | | | |
| 0.9380 | -0.995 | 2.563 | | | |
| 0.9715 | -0.991 | 2.572 | | | |
| 1.0050 | -0.992 | 2.575 | | | |
| 1.0385 | -0.990 | 2.573 | | | |
| 1.0720 | -0.989 | 2.581 | | | |
| 1.1055 | -0.993 | 2.593 | | | |
| 1.1390 | -0.997 | 2.583 | | | |
| 1.1725 | -0.998 | 2.591 | | | |
| 1.2060 | -0.991 | 2.593 | | | |

PG 1 233.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.843 ALPHA -2.50 REYNOLDS 1.64e10⁶

INTEGRATED FORCE COEFFICIENTS

C_x = -0.1215 C_y = -0.0552

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0328 | 1.374 | 2.972 | 0.0302 | 1.074 | 0.972 |
| 0.0335 | 1.909 | 2.923 | 0.0351 | 0.052 | 2.671 |
| 0.0379 | 2.701 | 2.562 | 0.0098 | -0.226 | 2.539 |
| 0.0196 | 2.374 | 2.766 | 0.0201 | -0.406 | 2.536 |
| 0.0294 | 2.198 | 2.714 | 0.0302 | -0.398 | 2.541 |
| 0.0495 | 2.222 | 2.662 | 0.0508 | -0.667 | 2.453 |
| 0.0741 | -2.208 | 2.595 | 0.0749 | -0.761 | 2.432 |
| 0.0991 | -2.302 | 2.567 | 0.1000 | -0.811 | 2.417 |
| 0.1241 | -2.296 | 2.581 | 0.1498 | -0.874 | 2.399 |
| 0.1591 | -2.525 | 2.560 | 0.2002 | -0.951 | 2.376 |
| 0.1994 | -2.339 | 2.555 | 0.2490 | -1.033 | 2.351 |
| 0.2382 | -2.396 | 2.542 | 0.3502 | -0.720 | 2.443 |
| 0.2794 | -2.419 | 2.533 | 0.4500 | -0.443 | 2.525 |
| 0.3188 | -2.403 | 2.529 | 0.5490 | -0.401 | 2.537 |
| 0.3594 | -2.478 | 2.515 | 0.6492 | -0.363 | 2.548 |
| 0.3993 | -2.524 | 2.502 | 0.7498 | -0.326 | 2.559 |
| 0.4194 | -2.533 | 2.499 | 0.8499 | -0.267 | 2.577 |
| 0.4395 | -2.501 | 2.491 | 0.9002 | -0.222 | 2.593 |
| 0.4595 | -2.573 | 2.485 | | | |
| 0.4757 | -2.598 | 2.482 | | | |
| 0.4943 | -2.618 | 2.474 | | | |
| 0.5184 | -2.640 | 2.465 | | | |
| 0.5396 | -2.657 | 2.463 | | | |
| 0.5592 | -2.681 | 2.456 | | | |
| 0.5787 | -2.700 | 2.452 | | | |
| 0.5989 | -2.741 | 2.433 | | | |
| 0.6180 | -2.777 | 2.427 | | | |
| 0.6434 | -2.840 | 2.430 | | | |
| 0.6747 | -2.846 | 2.407 | | | |
| 0.6939 | -2.798 | 2.433 | | | |
| 0.7237 | -2.628 | 2.471 | | | |
| 0.7488 | -2.507 | 2.507 | | | |
| 0.7738 | -2.430 | 2.533 | | | |
| 0.7990 | -2.394 | 2.549 | | | |
| 0.8240 | -2.306 | 2.566 | | | |
| 0.8489 | -2.201 | 2.577 | | | |
| 0.8939 | -2.204 | 2.594 | | | |

AGC 1 283.2 MM CM34C
 EXPERIMENTAL PRESSURE DISTRIBUTION
 SOLID WALLS

MACH NO. 0.799 ALPHA 1.82 REY 1.63×10^6

INTEGRATED FORCE COEFFICIENTS

$C_L = -2.0339$ $C_M = -2.3534$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 1.127 | 2.987 | 0.0000 | 1.127 | 2.907 |
| 0.0035 | 0.705 | 2.881 | 0.0051 | 2.257 | 2.732 |
| 0.0079 | 2.549 | 2.817 | 0.0098 | -2.027 | 2.654 |
| 0.0196 | 0.229 | 2.718 | 0.0201 | -2.213 | 0.594 |
| 0.0294 | 0.034 | 2.666 | 0.0308 | -2.244 | 0.565 |
| 0.0495 | -0.123 | 2.628 | 0.0508 | -2.473 | 0.517 |
| 0.0741 | -0.336 | 2.548 | 0.0749 | -0.552 | 0.434 |
| 0.0991 | -0.502 | 2.518 | 0.1002 | -0.619 | 0.475 |
| 0.1241 | -0.408 | 2.536 | 0.1498 | -0.718 | 0.445 |
| 0.1591 | -0.470 | 2.518 | 0.2002 | -0.676 | 0.339 |
| 0.1994 | -0.404 | 2.520 | 0.2496 | -0.910 | 0.389 |
| 0.2392 | -0.501 | 2.539 | 0.3502 | -1.020 | 0.356 |
| 0.2794 | -0.500 | 2.501 | 0.4502 | -0.414 | 0.534 |
| 0.3193 | -0.508 | 2.489 | 0.5498 | -0.299 | 0.566 |
| 0.3594 | -0.504 | 2.485 | 0.6492 | -0.213 | 0.574 |
| 0.3993 | -0.620 | 2.474 | 0.7498 | -0.120 | 0.619 |
| 0.4194 | -0.627 | 2.472 | 0.8499 | -0.215 | 0.632 |
| 0.4395 | -0.621 | 2.465 | 0.9000 | 0.039 | 0.626 |
| 0.4585 | -0.629 | 2.463 | | | |
| 0.4757 | -0.602 | 2.456 | | | |
| 0.4955 | -0.607 | 2.452 | | | |
| 0.5154 | -0.717 | 2.446 | | | |
| 0.5396 | -0.727 | 2.443 | | | |
| 0.5592 | -0.746 | 2.437 | | | |
| 0.5797 | -0.763 | 2.432 | | | |
| 0.5959 | -0.797 | 2.422 | | | |
| 0.6160 | -0.801 | 2.412 | | | |
| 0.6454 | -0.809 | 2.401 | | | |
| 0.6747 | -0.592 | 2.483 | | | |
| 0.6949 | -0.336 | 2.540 | | | |
| 0.7237 | -0.213 | 2.565 | | | |
| 0.7455 | -0.244 | 2.595 | | | |
| 0.7735 | -0.190 | 2.620 | | | |
| 0.7993 | -0.102 | 2.615 | | | |
| 0.8248 | -0.095 | 2.628 | | | |
| 0.8489 | -0.090 | 2.639 | | | |
| 0.8989 | -0.013 | 2.653 | | | |

AD-A076 131

AERONAUTICAL RESEARCH LABS MELBOURNE (AUSTRALIA)

F/G 20/4

TRANSONIC WIND TUNNEL TESTS ON A SERIES OF TWO-DIMENSIONAL AERO--ETC(U)

JAN 79 B D FAIRLIE , N POLLOCK

UNCLASSIFIED

ARL/AERO NOTE-384

NL

2 OF 5
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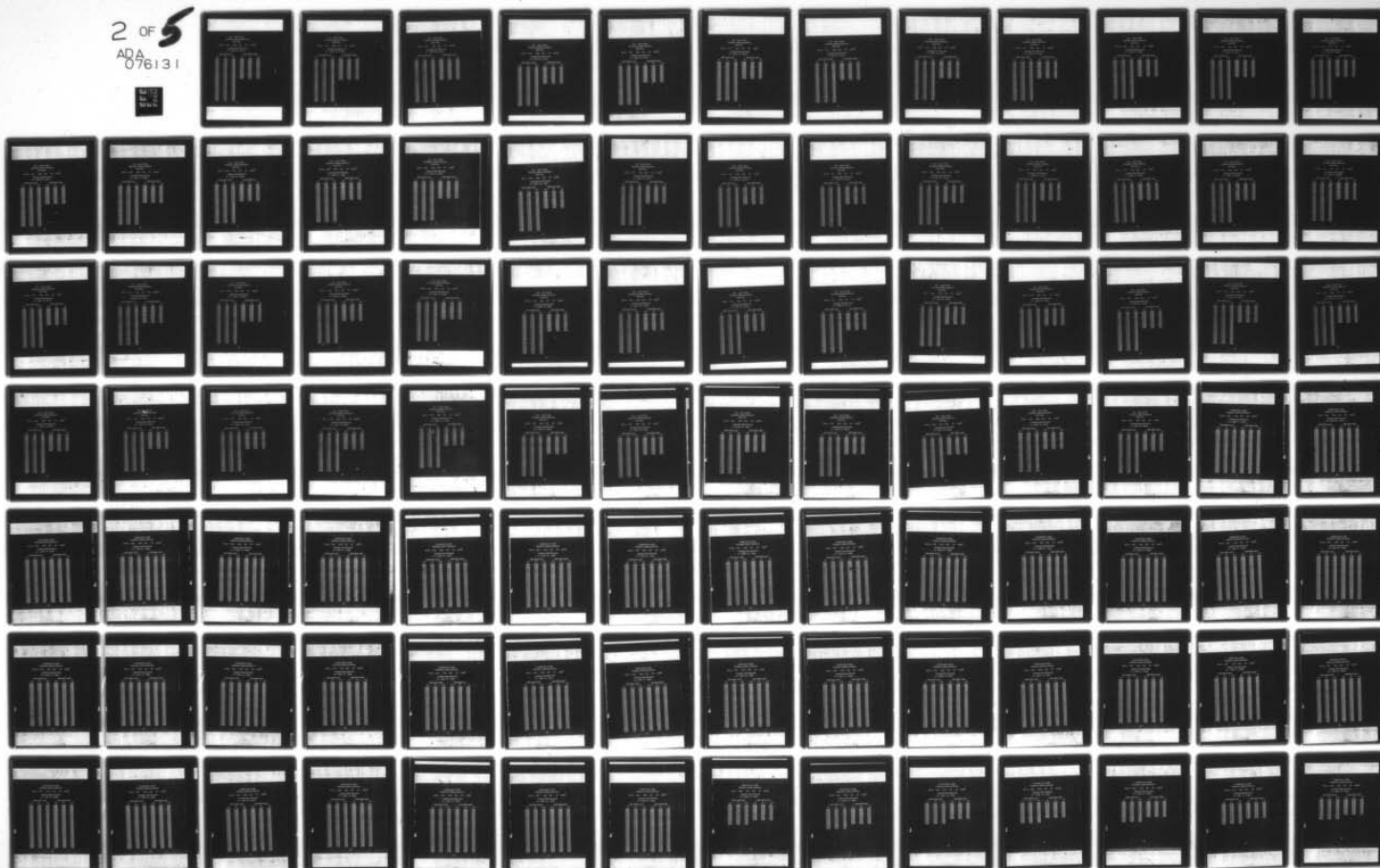


FIG 1 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.799 ALPHA 0.82 REY 1.03×10^6

INTEGRATED FORCE COEFFICIENTS

$C_D = 0.1387$ $C_M = -2.3914$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.2328 | 1.127 | 3.996 | 2.8320 | 1.157 | 0.996 |
| 2.2335 | 3.631 | 3.541 | 2.8351 | 0.425 | 0.791 |
| 2.2379 | 3.422 | 3.783 | 2.8398 | 0.164 | 0.704 |
| 2.2496 | 3.377 | 3.679 | 2.8421 | -0.054 | 0.640 |
| 2.2524 | -0.243 | 3.633 | 2.8448 | -0.113 | 0.622 |
| 2.2495 | -0.229 | 3.589 | 2.8500 | -0.322 | 0.561 |
| 2.2741 | -0.592 | 3.494 | 2.8749 | -0.441 | 0.526 |
| 2.2951 | -0.717 | 3.445 | 2.8903 | -0.473 | 0.518 |
| 2.3241 | -0.698 | 3.454 | 2.9198 | -0.631 | 0.473 |
| 2.3591 | -0.669 | 3.454 | 2.9383 | -0.754 | 0.434 |
| 2.3994 | -0.603 | 3.453 | 2.9499 | -0.823 | 0.412 |
| 2.4352 | -0.649 | 3.477 | 2.9502 | -0.903 | 0.389 |
| 2.4794 | -0.603 | 3.479 | 2.9502 | -0.496 | 0.529 |
| 2.5133 | -0.643 | 3.466 | 2.9493 | -0.193 | 0.577 |
| 2.5594 | -0.677 | 3.457 | 2.9492 | -0.025 | 0.648 |
| 2.5993 | -0.704 | 3.449 | 2.9498 | 0.107 | 0.657 |
| 2.6194 | -0.709 | 3.448 | 2.9499 | 0.270 | 0.714 |
| 2.6395 | -0.735 | 3.443 | 2.9382 | 0.211 | 0.717 |
| 2.6505 | -0.745 | 3.437 | | | |
| 2.6757 | -0.772 | 3.429 | | | |
| 2.6953 | -0.797 | 3.422 | | | |
| 2.7154 | -0.816 | 3.416 | | | |
| 2.7396 | -0.827 | 3.413 | | | |
| 2.7592 | -0.850 | 3.407 | | | |
| 2.7757 | -0.864 | 3.402 | | | |
| 2.7999 | -0.900 | 3.392 | | | |
| 2.8153 | -0.921 | 3.386 | | | |
| 2.8444 | -0.901 | 3.456 | | | |
| 2.8747 | -0.910 | 3.536 | | | |
| 2.8999 | -0.926 | 3.563 | | | |
| 2.9237 | -0.973 | 3.576 | | | |
| 2.9455 | -0.927 | 3.589 | | | |
| 2.9755 | -0.916 | 3.593 | | | |
| 2.9990 | -0.917 | 3.606 | | | |
| 3.0240 | -0.919 | 3.612 | | | |
| 3.0449 | -0.932 | 3.617 | | | |
| 3.0649 | -0.917 | 3.624 | | | |

PGK 1 283.2 MM CHORD
 EXPERIMENTAL PRESSURE DISTRIBUTION
 SOLID WALLS

MACH NO. 0.931 ALPHA 1.03 REY 1.64*10⁶

INTEGRATED FORCE COEFFICIENTS

C_x = 0.3249 C_y = -0.3985

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | C _p | P/H | X/C | C _p | P/H |
|--------|----------------|-------|--------|----------------|-------|
| 0.0000 | 1.107 | 2.999 | 0.0000 | 1.167 | 0.999 |
| 0.0035 | 0.511 | 2.885 | 0.2051 | 0.551 | 0.817 |
| 0.0079 | 0.308 | 2.746 | 0.3098 | 0.308 | 0.746 |
| 0.0196 | -0.229 | 2.646 | 0.3221 | 0.076 | 0.678 |
| 0.0294 | -0.125 | 2.599 | 0.3302 | 0.035 | 0.657 |
| 0.0405 | -0.303 | 2.565 | 0.3520 | -0.196 | 0.598 |
| 0.0741 | -0.556 | 2.482 | 0.3749 | -0.323 | 0.568 |
| 0.0991 | -0.857 | 2.424 | 0.1000 | -0.366 | 0.548 |
| 0.1241 | -0.777 | 2.426 | 0.1498 | -0.568 | 0.492 |
| 0.1591 | -0.626 | 2.411 | 0.2000 | -0.653 | 0.463 |
| 0.1994 | -0.516 | 2.414 | 0.2499 | -0.675 | 0.457 |
| 0.2382 | -0.311 | 2.415 | 0.3502 | -0.788 | 0.423 |
| 0.2794 | -0.610 | 2.416 | 0.4500 | -0.375 | 0.545 |
| 0.3138 | -0.807 | 2.408 | 0.5492 | -0.193 | 0.599 |
| 0.3594 | -0.532 | 2.410 | 0.6492 | -0.033 | 0.654 |
| 0.3993 | -0.807 | 2.432 | 0.7498 | 0.140 | 0.696 |
| 0.4194 | -0.853 | 2.404 | 0.8499 | 0.213 | 0.718 |
| 0.4395 | -0.879 | 2.396 | 0.9320 | 0.209 | 0.717 |
| 0.4585 | -0.604 | 2.394 | | | |
| 0.4787 | -0.925 | 2.383 | | | |
| 0.4983 | -0.917 | 2.385 | | | |
| 0.5184 | -0.942 | 2.377 | | | |
| 0.5396 | -0.946 | 2.376 | | | |
| 0.5592 | -0.942 | 2.378 | | | |
| 0.5787 | -0.936 | 2.409 | | | |
| 0.5989 | -0.621 | 2.472 | | | |
| 0.6182 | -0.474 | 2.516 | | | |
| 0.6484 | -0.376 | 2.544 | | | |
| 0.6747 | -0.319 | 2.561 | | | |
| 0.6989 | -0.341 | 2.504 | | | |
| 0.7237 | -0.243 | 2.572 | | | |
| 0.7488 | -0.276 | 2.574 | | | |
| 0.7738 | -0.253 | 2.578 | | | |
| 0.7992 | -0.258 | 2.582 | | | |
| 0.8248 | -0.239 | 2.585 | | | |
| 0.8489 | -0.233 | 2.588 | | | |
| 0.8989 | -0.214 | 2.592 | | | |

PGK 1 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.796 ALPHA 2.00 REV 1.53*12⁶

INTEGRATED FORCE COEFFICIENTS

C_x = 0.3996 C_y = -0.3952

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 2.0323 | 1.166 | 2.999 | 2.8038 | 1.166 | 2.999 |
| 2.0335 | 1.363 | 2.767 | 2.8051 | 0.657 | 0.532 |
| 2.0379 | 1.193 | 2.714 | 2.8098 | 0.423 | 0.751 |
| 2.0196 | -2.142 | 2.615 | 2.8261 | 0.185 | 0.712 |
| 2.0294 | -0.292 | 2.572 | 2.8338 | 0.898 | 0.606 |
| 2.0495 | -0.381 | 2.545 | 2.8528 | -0.128 | 0.428 |
| 2.0741 | -0.784 | 2.451 | 2.8749 | -0.233 | 0.559 |
| 2.0991 | -0.911 | 2.393 | 2.8888 | -0.207 | 2.573 |
| 2.1241 | -0.808 | 2.397 | 2.8998 | -0.488 | 0.514 |
| 2.1591 | -2.914 | 2.389 | 2.9282 | -0.548 | 0.457 |
| 2.1994 | -2.917 | 2.388 | 2.9499 | -0.631 | 0.472 |
| 2.2332 | -2.932 | 2.384 | 2.9522 | -2.716 | 0.447 |
| 2.2794 | -2.922 | 2.367 | 2.9582 | -3.398 | 0.542 |
| 2.3138 | -2.946 | 2.393 | 2.9492 | -0.227 | 2.591 |
| 2.3594 | -2.991 | 2.378 | 2.9492 | -0.831 | 2.648 |
| 2.3993 | -2.978 | 2.771 | 2.9492 | 2.121 | 0.693 |
| 2.4194 | -2.974 | 2.372 | 2.9499 | 0.189 | 0.713 |
| 2.4395 | -2.995 | 2.366 | 2.9888 | 0.179 | 0.712 |
| 2.4585 | -2.994 | 2.366 | | | |
| 2.4767 | -1.319 | 2.358 | | | |
| 2.4933 | -1.326 | 2.356 | | | |
| 2.5134 | -1.289 | 2.362 | | | |
| 2.5396 | -2.792 | 2.425 | | | |
| 2.5592 | -2.592 | 2.487 | | | |
| 2.5797 | -2.450 | 2.517 | | | |
| 2.5999 | -2.448 | 2.528 | | | |
| 2.6153 | -2.403 | 2.532 | | | |
| 2.6384 | -2.398 | 2.543 | | | |
| 2.6747 | -2.374 | 2.548 | | | |
| 2.6999 | -2.365 | 2.550 | | | |
| 2.7237 | -2.355 | 2.553 | | | |
| 2.7498 | -2.349 | 2.555 | | | |
| 2.7733 | -2.338 | 2.558 | | | |
| 2.7993 | -2.336 | 2.559 | | | |
| 2.8243 | -2.333 | 2.563 | | | |
| 2.8499 | -2.327 | 2.561 | | | |
| 2.8999 | -2.322 | 2.563 | | | |

BOX 1 131.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.753 ALPHA 0.68 REY 8.83×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.5444$ $C_M = -0.1146$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.124 | 2.993 | 0.0000 | 1.124 | 0.993 |
| 0.0389 | 0.114 | 2.718 | 0.0389 | 0.570 | 2.842 |
| 0.0786 | -0.234 | 2.623 | 0.0786 | 0.130 | 0.722 |
| 0.1186 | -0.445 | 2.565 | 0.1186 | 0.017 | 0.691 |
| 0.1584 | -0.563 | 2.533 | 0.1584 | -0.063 | 2.659 |
| 0.1982 | -0.906 | 2.443 | 0.1982 | -0.223 | 2.626 |
| 0.2380 | -1.334 | 2.485 | 0.2380 | -0.292 | 2.610 |
| 0.2778 | -1.369 | 2.395 | 0.2778 | -0.390 | 2.588 |
| 0.3176 | -1.008 | 2.412 | 0.3176 | -0.365 | 2.537 |
| 0.3574 | -0.989 | 2.417 | 0.3574 | -0.347 | 2.592 |
| 0.3972 | -0.963 | 2.424 | 0.3972 | -0.265 | 2.614 |
| 0.4370 | -0.953 | 2.427 | 0.4370 | -0.128 | 2.651 |
| 0.4768 | -0.931 | 2.433 | 0.4768 | 0.112 | 2.717 |
| 0.5166 | -0.903 | 2.443 | 0.5166 | 0.203 | 2.764 |
| 0.5564 | -0.898 | 2.453 | 0.5564 | 0.348 | 2.781 |
| 0.5962 | -0.844 | 2.456 | | | |
| 0.6360 | -0.829 | 2.463 | | | |
| 0.6758 | -0.805 | 2.467 | | | |
| 0.7156 | -0.794 | 2.478 | | | |
| 0.7554 | -0.787 | 2.472 | | | |
| 0.7952 | -0.792 | 2.481 | | | |
| 0.8350 | -0.796 | 2.486 | | | |
| 0.8748 | -0.704 | 2.494 | | | |
| 0.9146 | -0.676 | 2.582 | | | |
| 0.9544 | -0.631 | 2.514 | | | |
| 0.9942 | -0.591 | 2.525 | | | |
| 1.0340 | -0.537 | 2.543 | | | |
| 1.0738 | -0.497 | 2.551 | | | |
| 1.1136 | -0.421 | 2.572 | | | |
| 1.1534 | -0.307 | 2.586 | | | |
| 1.1932 | -0.316 | 2.688 | | | |
| 1.2330 | -0.290 | 2.618 | | | |
| 1.2728 | -0.142 | 2.634 | | | |
| 1.3126 | -0.132 | 2.651 | | | |

BCK 1 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.752 ALPHA 0.60 REY 0.03018

INTEGRATED FORCE COEFFICIENTS

CN = 0.5464 CM = -0.1145

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0320 | 1.122 | 0.992 | 0.0000 | 1.122 | 0.992 |
| 0.0369 | 0.113 | 0.719 | 0.0096 | 0.569 | 0.842 |
| 0.0426 | -0.238 | 0.622 | 0.0285 | 0.133 | 0.723 |
| 0.0486 | -0.447 | 0.565 | 0.0386 | 0.018 | 0.692 |
| 0.0504 | -0.561 | 0.534 | 0.0506 | -0.060 | 0.670 |
| 0.0752 | -0.907 | 0.448 | 0.0754 | -0.221 | 0.627 |
| 0.0999 | -1.295 | 0.495 | 0.1001 | -0.278 | 0.611 |
| 0.1252 | -1.371 | 0.395 | 0.2000 | -0.388 | 0.591 |
| 0.1599 | -1.309 | 0.412 | 0.2496 | -0.362 | 0.568 |
| 0.2030 | -0.992 | 0.417 | 0.3495 | -0.348 | 0.592 |
| 0.2407 | -0.995 | 0.424 | 0.4486 | -0.262 | 0.615 |
| 0.2798 | -0.955 | 0.427 | 0.5493 | -0.137 | 0.649 |
| 0.3202 | -0.929 | 0.434 | 0.6499 | 0.111 | 0.717 |
| 0.3603 | -0.904 | 0.448 | 0.7499 | 0.201 | 0.763 |
| 0.4021 | -0.892 | 0.455 | 0.8502 | 0.346 | 0.781 |
| 0.4198 | -0.891 | 0.455 | | | |
| 0.4432 | -0.894 | 0.460 | | | |
| 0.4599 | -0.812 | 0.466 | | | |
| 0.4796 | -0.809 | 0.467 | | | |
| 0.4996 | -0.802 | 0.468 | | | |
| 0.5485 | -0.776 | 0.475 | | | |
| 0.5602 | -0.747 | 0.483 | | | |
| 0.5803 | -0.711 | 0.493 | | | |
| 0.6004 | -0.686 | 0.503 | | | |
| 0.6203 | -0.654 | 0.514 | | | |
| 0.6505 | -0.589 | 0.526 | | | |
| 0.6754 | -0.556 | 0.540 | | | |
| 0.7085 | -0.498 | 0.551 | | | |
| 0.7255 | -0.419 | 0.573 | | | |
| 0.7584 | -0.306 | 0.587 | | | |
| 0.7753 | -0.315 | 0.601 | | | |
| 0.8005 | -0.258 | 0.619 | | | |
| 0.8255 | -0.190 | 0.635 | | | |
| 0.8503 | -0.133 | 0.651 | | | |

BCK 1 181.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.734 ALPHA 0.68 REY 1.48×10^6

INTEGRATED FORCE COEFFICIENTS
CY = 0.5478 CM = -0.1152

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 2.2888 | 1.123 | 2.992 | 0.8888 | 1.123 | 2.992 |
| 2.2889 | 0.116 | 2.718 | 0.8896 | 0.566 | 2.841 |
| 2.4286 | -0.233 | 2.622 | 0.8285 | 0.131 | 2.722 |
| 2.5336 | -2.444 | 2.565 | 0.8386 | 0.015 | 2.692 |
| 2.5534 | -0.558 | 2.534 | 0.8526 | -0.862 | 2.659 |
| 2.8752 | -0.903 | 2.439 | 0.8754 | -0.221 | 2.626 |
| 2.8999 | -1.731 | 2.484 | 0.1081 | -0.281 | 2.649 |
| 2.1252 | -1.267 | 2.395 | 0.2888 | -0.389 | 2.538 |
| 2.1599 | -1.307 | 2.411 | 0.2496 | -0.363 | 2.537 |
| 2.2088 | -0.991 | 2.415 | 0.3495 | -0.347 | 2.591 |
| 2.2437 | -2.964 | 2.423 | 0.4486 | -0.263 | 2.614 |
| 2.2798 | -0.956 | 2.425 | 0.5453 | -0.132 | 2.658 |
| 2.3282 | -0.943 | 2.428 | 0.6499 | 0.112 | 2.717 |
| 2.3653 | -2.903 | 2.439 | 0.7499 | 0.283 | 2.763 |
| 2.4231 | -3.856 | 2.452 | 0.8588 | 0.348 | 2.751 |
| 2.4198 | -2.852 | 2.453 | | | |
| 2.4432 | -2.832 | 2.459 | | | |
| 2.4599 | -2.817 | 2.463 | | | |
| 2.4796 | -2.813 | 2.465 | | | |
| 2.4896 | -2.797 | 2.468 | | | |
| 2.5485 | -2.788 | 2.473 | | | |
| 2.5682 | -2.774 | 2.475 | | | |
| 2.5883 | -2.741 | 2.484 | | | |
| 2.6084 | -2.713 | 2.492 | | | |
| 2.6283 | -2.634 | 2.513 | | | |
| 2.6585 | -2.584 | 2.526 | | | |
| 2.6754 | -2.526 | 2.542 | | | |
| 2.7885 | -2.498 | 2.552 | | | |
| 2.7255 | -2.414 | 2.573 | | | |
| 2.7584 | -2.384 | 2.587 | | | |
| 2.7753 | -2.313 | 2.608 | | | |
| 2.8885 | -2.248 | 2.618 | | | |
| 2.8255 | -2.190 | 2.634 | | | |
| 2.8583 | -2.131 | 2.650 | | | |

BGK 1 131.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.502 ALPHA -2.50 REYNOLDS 0.83×10^6

INTEGRATED FORCE COEFFICIENTS
C_D = -0.0336 C_M = -0.0753

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.947 | 2.977 | 0.0000 | 0.987 | 2.977 |
| 0.0089 | 2.702 | 2.946 | 0.0096 | -0.476 | 2.771 |
| 0.0206 | 2.343 | 2.893 | 0.0285 | -0.837 | 2.718 |
| 0.0386 | 2.159 | 2.865 | 0.0386 | -0.737 | 2.732 |
| 0.0524 | -0.321 | 2.839 | 0.0526 | -0.719 | 2.735 |
| 0.0752 | -0.175 | 2.816 | 0.0754 | -0.647 | 2.746 |
| 0.0999 | -0.211 | 2.810 | 0.1001 | -0.668 | 2.743 |
| 0.1252 | -0.143 | 2.813 | 0.1202 | -0.571 | 2.758 |
| 0.1599 | -0.104 | 2.814 | 0.1496 | -0.499 | 2.768 |
| 0.2020 | -0.210 | 2.813 | 0.1495 | -0.436 | 2.778 |
| 0.2427 | -0.203 | 2.807 | 0.1486 | -0.310 | 2.796 |
| 0.2798 | -0.255 | 2.804 | 0.1493 | -0.172 | 2.827 |
| 0.3222 | -0.209 | 2.802 | 0.1499 | -0.012 | 2.848 |
| 0.3683 | -0.277 | 2.801 | 0.1499 | 0.134 | 2.852 |
| 0.4081 | -0.281 | 2.800 | 0.1502 | 0.243 | 2.872 |
| 0.4198 | -0.292 | 2.798 | | | |
| 0.4422 | -0.291 | 2.798 | | | |
| 0.4599 | -0.296 | 2.797 | | | |
| 0.4796 | -0.304 | 2.796 | | | |
| 0.4996 | -0.311 | 2.796 | | | |
| 0.5405 | -0.312 | 2.795 | | | |
| 0.5682 | -0.325 | 2.793 | | | |
| 0.5933 | -0.328 | 2.793 | | | |
| 0.6204 | -0.329 | 2.793 | | | |
| 0.6283 | -0.335 | 2.792 | | | |
| 0.6585 | -0.332 | 2.792 | | | |
| 0.6754 | -0.321 | 2.794 | | | |
| 0.7035 | -0.307 | 2.796 | | | |
| 0.7255 | -0.288 | 2.802 | | | |
| 0.7534 | -0.245 | 2.806 | | | |
| 0.7753 | -0.214 | 2.813 | | | |
| 0.8025 | -0.180 | 2.815 | | | |
| 0.8255 | -0.143 | 2.821 | | | |
| 0.8503 | -0.105 | 2.826 | | | |

BOX 1 121.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.510 ALPHA 0.00 REY 0.63×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.2752$ $C_M = -0.0773$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 2.2338 | 1.242 | 2.997 | 0.0303 | 1.042 | 0.997 |
| 2.3089 | 0.841 | 2.849 | 0.0396 | 0.378 | 0.896 |
| 2.3206 | -0.271 | 2.683 | 0.0205 | -0.026 | 0.838 |
| 2.3326 | -0.409 | 2.782 | 0.0326 | -0.060 | 0.833 |
| 2.3504 | -0.534 | 2.764 | 0.0506 | -0.174 | 0.916 |
| 2.2752 | -0.623 | 2.750 | 2.0754 | -0.222 | 0.809 |
| 2.2999 | -0.547 | 2.755 | 0.1001 | -0.291 | 0.799 |
| 2.1252 | -0.518 | 2.766 | 0.2300 | -0.321 | 0.795 |
| 2.1599 | -0.491 | 2.775 | 0.2496 | -0.295 | 0.798 |
| 2.2200 | -0.452 | 2.776 | 0.3495 | -0.288 | 0.799 |
| 2.2407 | -0.452 | 2.776 | 0.4486 | -0.200 | 0.812 |
| 2.2798 | -0.455 | 2.776 | 0.5493 | -0.099 | 0.827 |
| 2.3202 | -0.453 | 2.776 | 0.6499 | 0.038 | 0.848 |
| 2.3603 | -0.440 | 2.778 | 0.7499 | 0.180 | 0.859 |
| 2.4001 | -0.430 | 2.779 | 0.8500 | 0.245 | 0.970 |
| 2.4198 | -0.432 | 2.779 | | | |
| 2.4402 | -0.428 | 2.780 | | | |
| 2.4599 | -0.430 | 2.779 | | | |
| 2.4796 | -0.434 | 2.779 | | | |
| 2.4996 | -0.433 | 2.778 | | | |
| 2.5105 | -0.423 | 2.780 | | | |
| 2.5402 | -0.428 | 2.779 | | | |
| 2.5803 | -0.426 | 2.779 | | | |
| 2.6004 | -0.428 | 2.779 | | | |
| 2.6203 | -0.424 | 2.780 | | | |
| 2.6405 | -0.408 | 2.782 | | | |
| 2.6704 | -0.390 | 2.785 | | | |
| 2.7005 | -0.371 | 2.788 | | | |
| 2.7255 | -0.326 | 2.794 | | | |
| 2.7504 | -0.294 | 2.798 | | | |
| 2.7753 | -0.225 | 2.804 | | | |
| 2.8005 | -0.216 | 2.810 | | | |
| 2.8255 | -0.172 | 2.817 | | | |
| 2.8503 | -0.100 | 2.823 | | | |

BGK 1 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

TECH NO. 0.500 ALPHA 1.00 REY 0.83⁶10

INTEGRATED FORCE COEFFICIENTS

CV = 0.3910 CM = -0.2764

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.012 | 2.992 | 0.0000 | 1.012 | 0.992 |
| 0.0009 | -0.324 | 2.795 | 0.0009 | 0.607 | 2.932 |
| 0.0036 | -0.512 | 2.758 | 0.0036 | 0.226 | 0.076 |
| 0.0066 | -0.661 | 2.742 | 0.0066 | 0.143 | 2.804 |
| 0.0104 | -0.767 | 2.729 | 0.0104 | 0.083 | 2.843 |
| 0.0152 | -0.825 | 2.721 | 0.0152 | -0.077 | 0.831 |
| 0.0209 | -0.797 | 2.731 | 0.0209 | -0.158 | 0.819 |
| 0.0252 | -0.693 | 2.746 | 0.0252 | -0.237 | 2.677 |
| 0.0299 | -0.514 | 2.759 | 0.0299 | -0.218 | 2.610 |
| 0.0333 | -0.590 | 2.761 | 0.0333 | -0.230 | 2.608 |
| 0.0367 | -0.536 | 2.763 | 0.0367 | -0.155 | 2.020 |
| 0.0398 | -0.331 | 2.764 | 0.0398 | -0.068 | 2.032 |
| 0.0422 | -0.521 | 2.765 | 0.0422 | 0.061 | 2.052 |
| 0.0433 | -0.505 | 2.763 | 0.0433 | 0.195 | 0.071 |
| 0.0430 | -0.487 | 2.771 | 0.0430 | 0.257 | 0.641 |
| 0.0419 | -0.485 | 2.771 | | | |
| 0.0422 | -0.477 | 2.772 | | | |
| 0.0459 | -0.476 | 2.772 | | | |
| 0.0476 | -0.475 | 2.772 | | | |
| 0.0496 | -0.473 | 2.772 | | | |
| 0.0505 | -0.458 | 2.775 | | | |
| 0.0502 | -0.463 | 2.774 | | | |
| 0.0503 | -0.459 | 2.775 | | | |
| 0.0504 | -0.458 | 2.775 | | | |
| 0.0523 | -0.447 | 2.776 | | | |
| 0.0525 | -0.431 | 2.779 | | | |
| 0.0524 | -0.410 | 2.782 | | | |
| 0.0535 | -0.359 | 2.785 | | | |
| 0.0535 | -0.341 | 2.792 | | | |
| 0.0524 | -0.308 | 2.797 | | | |
| 0.0523 | -0.265 | 2.803 | | | |
| 0.0525 | -0.227 | 2.809 | | | |
| 0.0525 | -0.179 | 2.816 | | | |
| 0.0523 | -0.136 | 2.822 | | | |

80K 1 101.6 MM CHOPD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.530 ALPHA 2.03 REY 3.83×10^6

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.5337$ $C_M = -0.3753$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.0233 | 3.932 | 2.983 | 0.2000 | 0.932 | 2.9-2 |
| 2.0259 | -2.738 | 2.733 | 0.2096 | 0.785 | 3.959 |
| 2.0286 | -2.949 | 2.713 | 0.2205 | 0.425 | 3.9-5 |
| 2.0326 | -2.906 | 2.703 | 0.0306 | 0.321 | 3.892 |
| 2.0524 | -1.716 | 2.692 | 0.0506 | 0.159 | 2.8-6 |
| 2.0752 | -1.753 | 2.692 | 0.0754 | 0.052 | 0.853 |
| 0.2999 | -2.728 | 2.725 | 0.1301 | -0.039 | 0.837 |
| 2.1252 | -2.745 | 2.725 | 0.2000 | -0.151 | 0.522 |
| 2.1599 | -2.643 | 2.741 | 0.2496 | -0.147 | 0.621 |
| 0.2330 | -2.653 | 2.746 | 0.3495 | -0.174 | 0.617 |
| 0.2407 | -2.622 | 2.751 | 0.4486 | -0.116 | 0.625 |
| 0.2796 | -2.603 | 2.753 | 0.5493 | -0.043 | 2.837 |
| 2.3222 | -2.583 | 2.756 | 0.6499 | 0.083 | 2.854 |
| 2.3623 | -2.558 | 2.760 | 0.7499 | 0.213 | 0.874 |
| 2.4271 | -2.538 | 2.763 | 0.8522 | 0.269 | 3.6-2 |
| 2.4198 | -2.533 | 2.764 | | | |
| 2.4402 | -2.526 | 2.765 | | | |
| 2.4599 | -2.524 | 2.765 | | | |
| 2.4796 | -2.522 | 2.765 | | | |
| 2.4996 | -2.517 | 2.766 | | | |
| 2.5425 | -2.497 | 2.769 | | | |
| 2.5522 | -2.498 | 2.769 | | | |
| 2.5533 | -2.492 | 2.770 | | | |
| 2.5324 | -2.488 | 2.770 | | | |
| 0.6223 | -2.478 | 2.772 | | | |
| 0.615 | -2.458 | 2.775 | | | |
| 0.6794 | -2.432 | 2.779 | | | |
| 0.7225 | -2.407 | 2.782 | | | |
| 0.7255 | -2.398 | 2.793 | | | |
| 0.7524 | -2.321 | 2.795 | | | |
| 0.7753 | -2.277 | 2.802 | | | |
| 0.8025 | -2.231 | 2.809 | | | |
| 0.8255 | -2.183 | 2.816 | | | |
| 2.5533 | -2.137 | 2.822 | | | |

B2K 1 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

TACH NO. 0.542 ALPHA 3.28 REV 0.64+15⁶

INTEGRATED FORCE COEFFICIENTS
C_x = 0.6148 C_m = -0.0731

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|----------------|-------|----------------------|----------------|-------|
| X/C | C _p | P/H | X/C | C _p | P/H |
| 0.0000 | 2.741 | 2.960 | 0.0000 | 0.791 | 0.960 |
| 0.0050 | -1.220 | 2.661 | 0.0096 | 0.910 | 2.977 |
| 0.0100 | -1.299 | 2.653 | 0.0205 | 2.596 | 2.931 |
| 0.0150 | -1.259 | 2.650 | 0.0306 | 0.475 | 0.913 |
| 0.0200 | -1.274 | 2.650 | 0.0506 | 0.378 | 0.807 |
| 0.0250 | -1.255 | 2.655 | 0.0754 | 0.179 | 0.870 |
| 0.0300 | -1.101 | 2.673 | 0.1001 | 0.077 | 0.855 |
| 0.0350 | -0.930 | 2.724 | 0.2000 | -0.069 | 0.833 |
| 0.0400 | -0.802 | 2.723 | 0.2496 | -0.073 | 0.832 |
| 0.0450 | -0.746 | 2.731 | 0.3495 | -0.118 | 0.826 |
| 0.0500 | -0.711 | 2.737 | 0.4486 | -0.074 | 0.832 |
| 0.0550 | -0.601 | 2.741 | 0.5493 | -0.039 | 0.842 |
| 0.0600 | -0.553 | 2.745 | 0.6499 | 0.102 | 0.858 |
| 0.0650 | -0.615 | 2.751 | 0.7499 | 0.229 | 0.877 |
| 0.0700 | -0.586 | 2.755 | 0.8502 | 0.281 | 0.885 |
| 0.0750 | -0.576 | 2.756 | | | |
| 0.0800 | -0.505 | 2.758 | | | |
| 0.0850 | -0.501 | 2.759 | | | |
| 0.0900 | -0.542 | 2.761 | | | |
| 0.0950 | -0.556 | 2.760 | | | |
| 0.1000 | -0.552 | 2.764 | | | |
| 0.1050 | -0.550 | 2.764 | | | |
| 0.1100 | -0.541 | 2.765 | | | |
| 0.1150 | -0.512 | 2.767 | | | |
| 0.1200 | -0.497 | 2.769 | | | |
| 0.1250 | -0.471 | 2.773 | | | |
| 0.1300 | -0.442 | 2.777 | | | |
| 0.1350 | -0.414 | 2.781 | | | |
| 0.1400 | -0.363 | 2.789 | | | |
| 0.1450 | -0.326 | 2.795 | | | |
| 0.1500 | -0.273 | 2.821 | | | |
| 0.1550 | -0.236 | 2.823 | | | |
| 0.1600 | -0.196 | 2.815 | | | |
| 0.1650 | -0.138 | 2.822 | | | |

BCK 1 121.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 1.449 ALPHA 4.88 REY 2.83e12⁶

INTEGRATED FORCE COEFFICIENTS

CL = 0.7239 CM = -2.2725

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.0000 | 0.606 | 2.932 | 2.0000 | 0.606 | 2.932 |
| 2.0059 | -1.760 | 2.581 | 2.0096 | 0.994 | 2.942 |
| 2.0226 | -1.660 | 2.595 | 2.0205 | 0.731 | 2.951 |
| 2.0326 | -1.647 | 2.623 | 2.0306 | 0.676 | 2.932 |
| 2.0524 | -1.598 | 2.639 | 2.0526 | 0.421 | 2.945 |
| 2.0752 | -1.496 | 2.624 | 2.0754 | 0.288 | 2.856 |
| 2.0999 | -1.277 | 2.655 | 2.1001 | 0.179 | 2.859 |
| 2.1252 | -1.216 | 2.684 | 2.1250 | 0.009 | 2.744 |
| 2.1599 | -0.942 | 2.727 | 2.1496 | -0.011 | 2.641 |
| 2.1926 | -0.641 | 2.719 | 2.1745 | -0.057 | 2.833 |
| 2.2227 | -0.740 | 2.727 | 2.1986 | -0.036 | 2.837 |
| 2.2493 | -0.753 | 2.732 | 2.2243 | 0.019 | 2.846 |
| 2.2722 | -0.740 | 2.737 | 2.2499 | 0.121 | 2.801 |
| 2.2973 | -0.676 | 2.744 | 2.2749 | 0.241 | 2.879 |
| 2.3171 | -0.643 | 2.749 | 2.2980 | 0.288 | 2.856 |
| 2.3413 | -0.632 | 2.753 | | | |
| 2.3612 | -0.618 | 2.752 | | | |
| 2.3899 | -0.610 | 2.753 | | | |
| 2.4146 | -0.600 | 2.755 | | | |
| 2.4396 | -0.590 | 2.756 | | | |
| 2.4625 | -0.583 | 2.760 | | | |
| 2.4872 | -0.597 | 2.761 | | | |
| 2.5123 | -0.546 | 2.763 | | | |
| 2.5374 | -0.535 | 2.764 | | | |
| 2.5623 | -0.541 | 2.766 | | | |
| 2.5825 | -0.491 | 2.771 | | | |
| 2.6054 | -0.459 | 2.776 | | | |
| 2.6285 | -0.427 | 2.780 | | | |
| 2.6525 | -0.375 | 2.788 | | | |
| 2.6754 | -0.332 | 2.794 | | | |
| 2.6953 | -0.295 | 2.801 | | | |
| 2.7175 | -0.238 | 2.828 | | | |
| 2.7325 | -0.107 | 2.815 | | | |
| 2.7523 | -0.141 | 2.822 | | | |

80K 1 121.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.631 ALPHA -2.50 REY 0.81e10⁶

INTEGRATED FORCE COEFFICIENTS
C_x = -0.0512 C_y = -0.0010

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 2.0000 | 0.948 | 2.971 | 0.0000 | 0.948 | 0.971 |
| 2.0089 | 0.740 | 2.930 | 0.0096 | -0.440 | 0.694 |
| 2.2236 | 0.384 | 0.859 | 0.0205 | -0.891 | 0.626 |
| 0.0306 | 0.198 | 0.023 | 0.0306 | -0.806 | 0.622 |
| 2.0504 | -0.001 | 2.703 | 0.0506 | -0.816 | 0.620 |
| 2.0752 | -0.174 | 0.749 | 0.0754 | -0.761 | 0.631 |
| 2.0999 | -0.210 | 0.740 | 0.1001 | -0.763 | 0.633 |
| 2.1252 | -0.202 | 0.743 | 0.2000 | -0.653 | 0.652 |
| 2.1599 | -0.194 | 0.745 | 0.2406 | -0.570 | 0.669 |
| 2.2000 | -0.242 | 0.739 | 0.3495 | -0.494 | 0.684 |
| 2.2427 | -0.249 | 0.734 | 0.4486 | -0.343 | 0.714 |
| 2.2798 | -0.273 | 0.729 | 0.5493 | -0.182 | 0.746 |
| 2.3202 | -0.290 | 0.726 | 0.6499 | -0.010 | 0.780 |
| 2.3603 | -0.290 | 0.724 | 0.7499 | 0.130 | 0.812 |
| 2.4001 | -0.307 | 0.722 | 0.8500 | 0.210 | 0.924 |
| 2.4198 | -0.317 | 0.720 | | | |
| 2.4402 | -0.318 | 0.723 | | | |
| 2.4599 | -0.326 | 0.719 | | | |
| 2.4796 | -0.333 | 0.717 | | | |
| 2.4996 | -0.341 | 0.716 | | | |
| 2.5405 | -0.341 | 0.715 | | | |
| 2.5602 | -0.356 | 0.712 | | | |
| 2.5823 | -0.361 | 0.711 | | | |
| 2.6004 | -0.370 | 0.710 | | | |
| 2.6203 | -0.370 | 0.719 | | | |
| 2.6505 | -0.364 | 0.711 | | | |
| 2.6754 | -0.349 | 0.714 | | | |
| 2.7005 | -0.336 | 0.716 | | | |
| 2.7255 | -0.293 | 0.725 | | | |
| 2.7504 | -0.265 | 2.732 | | | |
| 0.7753 | -0.234 | 0.736 | | | |
| 0.8005 | -0.194 | 0.744 | | | |
| 0.8255 | -0.151 | 2.753 | | | |
| 0.8505 | -0.110 | 2.761 | | | |

BCK 1 181.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.638 ALPHA 0.88 REY 0.81*10⁶

INTEGRATED FORCE COEFFICIENTS

C_N = 0.2846 C_M = -0.8814

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| Z/C | CP | P/H | X/C | CP | P/H |
| 2.2322 | 1.071 | 0.995 | 0.0000 | 1.071 | 0.995 |
| 2.2389 | 0.112 | 0.886 | 0.0396 | 0.787 | 0.660 |
| 2.2226 | -0.221 | 0.748 | 0.3205 | -0.032 | 0.776 |
| 2.2326 | -0.380 | 0.708 | 0.8386 | -0.067 | 0.769 |
| 2.2504 | -0.545 | 0.675 | 0.0506 | -0.196 | 0.744 |
| 2.2752 | -0.678 | 0.649 | 0.0754 | -0.251 | 0.733 |
| 2.2999 | -0.648 | 0.655 | 0.1001 | -0.325 | 0.710 |
| 2.1252 | -0.543 | 0.672 | 0.2000 | -0.365 | 0.710 |
| 2.1599 | -0.501 | 0.684 | 0.2496 | -0.329 | 0.717 |
| 2.2080 | -0.496 | 0.687 | 0.3495 | -0.320 | 0.719 |
| 2.2407 | -0.490 | 0.686 | 0.4486 | -0.223 | 0.730 |
| 2.2798 | -0.487 | 0.686 | 0.5493 | -0.185 | 0.762 |
| 2.3232 | -0.485 | 0.687 | 0.6499 | 0.042 | 0.791 |
| 2.3603 | -0.471 | 0.689 | 0.7499 | 0.188 | 0.820 |
| 2.4001 | -0.468 | 0.690 | 0.8500 | 0.252 | 0.833 |
| 2.4193 | -0.469 | 0.690 | | | |
| 2.4422 | -0.464 | 0.691 | | | |
| 2.4599 | -0.467 | 0.690 | | | |
| 2.4796 | -0.470 | 0.690 | | | |
| 2.4996 | -0.469 | 0.690 | | | |
| 2.5485 | -0.457 | 0.692 | | | |
| 2.5602 | -0.464 | 0.691 | | | |
| 2.5803 | -0.463 | 0.691 | | | |
| 2.6002 | -0.465 | 0.690 | | | |
| 2.6203 | -0.457 | 0.692 | | | |
| 2.6505 | -0.441 | 0.695 | | | |
| 2.6754 | -0.418 | 0.700 | | | |
| 2.7025 | -0.394 | 0.704 | | | |
| 2.7255 | -0.345 | 0.714 | | | |
| 2.7504 | -0.300 | 0.722 | | | |
| 2.7753 | -0.267 | 0.730 | | | |
| 2.8005 | -0.221 | 0.739 | | | |
| 2.8255 | -0.172 | 0.749 | | | |
| 2.8503 | -0.127 | 0.758 | | | |

BSX 1 181.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.638 ALPHA 1.03 REY 8.81×10^6

INTEGRATED FORCE COEFFICIENTS
C_N = 0.4113 C_M = -0.0684

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 2.8888 | 1.851 | 3.992 | 8.8888 | 1.851 | 0.992 |
| 2.8889 | -2.228 | 3.739 | 8.8896 | 0.619 | 0.936 |
| 2.8886 | -0.521 | 0.681 | 8.8285 | 0.228 | 0.827 |
| 2.8386 | -2.655 | 3.654 | 8.8386 | 0.142 | 0.812 |
| 2.8524 | -2.802 | 3.625 | 8.8586 | -0.889 | 0.782 |
| 2.8752 | -0.912 | 3.683 | 8.8754 | -0.896 | 0.765 |
| 2.8979 | -0.841 | 3.617 | 8.8881 | -0.183 | 0.747 |
| 8.1252 | -0.718 | 3.642 | 8.2888 | -0.268 | 0.731 |
| 8.1599 | -2.638 | 3.659 | 8.2496 | -0.258 | 0.734 |
| 2.2888 | -2.688 | 3.665 | 8.3495 | -0.257 | 0.733 |
| 2.2487 | -3.594 | 3.686 | 8.4486 | -0.176 | 0.749 |
| 2.2798 | -3.582 | 3.668 | 8.5493 | -0.875 | 0.769 |
| 2.3282 | -3.568 | 3.671 | 8.6499 | 0.863 | 0.796 |
| 2.3683 | -3.544 | 3.676 | 8.7499 | 0.287 | 0.825 |
| 2.4281 | -3.534 | 3.678 | 8.8588 | 0.267 | 0.836 |
| 2.4198 | -0.531 | 3.678 | | | |
| 2.4422 | -2.521 | 3.683 | | | |
| 2.4599 | -0.515 | 3.682 | | | |
| 2.4776 | -2.514 | 3.682 | | | |
| 2.4.76 | -2.508 | 3.683 | | | |
| 2.5435 | -2.496 | 3.685 | | | |
| 2.5632 | -2.503 | 3.684 | | | |
| 2.5823 | -2.497 | 3.685 | | | |
| 2.6284 | -0.495 | 3.685 | | | |
| 2.6223 | -2.485 | 3.687 | | | |
| 2.6535 | -0.483 | 3.691 | | | |
| 2.6754 | -2.439 | 3.696 | | | |
| 2.7885 | -0.414 | 3.781 | | | |
| 2.7255 | -0.361 | 3.712 | | | |
| 2.7534 | -0.328 | 3.728 | | | |
| 2.7753 | -0.276 | 3.729 | | | |
| 2.8225 | -0.225 | 3.737 | | | |
| 2.8255 | -0.178 | 3.748 | | | |
| 2.8583 | -0.152 | 3.757 | | | |

80: 1 181.6 MM CHORD
 EXPERIMENTAL PRESSURE DISTRIBUTION
 SOLID WALLS

MACH NO. 0.638 ALPHA 2.88 REY 0.61*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.5342 CH = -0.0701

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.0000 | 0.907 | 2.979 | 0.0000 | 0.907 | 0.979 |
| 2.0359 | -0.621 | 0.661 | 0.0896 | 0.792 | 0.948 |
| 2.0206 | -0.854 | 3.615 | 0.0205 | 0.426 | 0.650 |
| 2.0386 | -0.953 | 3.595 | 0.0306 | 0.327 | 0.646 |
| 2.2524 | -1.008 | 3.563 | 0.0506 | 0.154 | 0.614 |
| 0.0752 | -1.179 | 0.553 | 0.0754 | 0.043 | 0.792 |
| 2.7999 | -1.245 | 0.577 | 0.1001 | -0.052 | 0.773 |
| 2.1252 | -0.340 | 0.511 | 0.2000 | -0.174 | 0.749 |
| 2.1599 | -0.701 | 2.033 | 0.2496 | -0.108 | 0.751 |
| 2.2208 | -0.714 | 3.642 | 0.3495 | -0.194 | 0.745 |
| 2.2407 | -0.698 | 0.647 | 0.4406 | -0.132 | 0.750 |
| 2.2798 | -0.607 | 2.652 | 0.5113 | -0.045 | 0.775 |
| 2.3202 | -0.646 | 2.656 | 0.6499 | 0.004 | 0.808 |
| 2.3603 | -0.615 | 2.662 | 0.7499 | 0.220 | 0.827 |
| 2.4001 | -0.592 | 2.667 | 0.8500 | 0.276 | 0.838 |
| 2.4198 | -0.505 | 2.666 | | | |
| 2.4422 | -0.573 | 2.673 | | | |
| 2.4599 | -0.507 | 0.672 | | | |
| 2.4796 | -0.554 | 2.672 | | | |
| 2.4996 | -0.557 | 2.673 | | | |
| 2.5425 | -0.539 | 2.677 | | | |
| 2.5622 | -0.538 | 2.677 | | | |
| 2.5923 | -0.531 | 2.679 | | | |
| 2.6304 | -0.523 | 2.680 | | | |
| 2.6223 | -0.510 | 2.683 | | | |
| 2.6595 | -0.486 | 2.688 | | | |
| 2.6754 | -0.457 | 2.693 | | | |
| 2.7305 | -0.424 | 2.700 | | | |
| 2.7255 | -0.370 | 2.710 | | | |
| 0.7534 | -0.326 | 2.719 | | | |
| 0.7753 | -0.280 | 2.728 | | | |
| 0.8305 | -0.229 | 2.738 | | | |
| 2.8255 | -0.179 | 2.745 | | | |
| 0.8503 | -0.134 | 2.757 | | | |

BCK 1 181.6 MM CHORD
 EXPERIMENTAL PRESSURE DISTRIBUTION
 SOLID WALLS

MACH NO. 0.681 ALPHA 3.88 REY 0.82-18⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.6456 CN = -0.8741

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.8282 | 0.881 | 2.958 | 0.8828 | 0.881 | 2.958 |
| 2.8259 | -1.831 | 2.579 | 0.8896 | 0.914 | 2.965 |
| 2.8286 | -1.108 | 2.545 | 0.8285 | 0.587 | 6.988 |
| 1.8386 | -1.259 | 2.534 | 0.8336 | 0.475 | 6.878 |
| 2.8584 | -1.389 | 2.588 | 0.8526 | 0.294 | 0.842 |
| 2.8752 | -1.581 | 2.478 | 0.8754 | 0.165 | 0.816 |
| 2.8999 | -1.249 | 2.536 | 0.1801 | 0.862 | 0.796 |
| 2.1252 | -1.826 | 2.580 | 0.2808 | -0.898 | 0.766 |
| 2.1599 | -2.873 | 2.610 | 0.2196 | -0.898 | 0.764 |
| 2.2888 | -3.888 | 2.625 | 0.8465 | -0.144 | 2.755 |
| 2.2487 | -2.774 | 2.638 | 0.4486 | -0.894 | 2.765 |
| 2.2798 | -3.742 | 2.637 | 0.3493 | -0.818 | 0.788 |
| 2.3222 | -2.789 | 2.643 | 0.6499 | 0.183 | 0.834 |
| 2.3683 | -2.688 | 2.651 | 0.7495 | 0.235 | 0.838 |
| 2.4181 | -2.642 | 2.656 | 0.8588 | 0.288 | 0.841 |
| 2.4198 | -2.632 | 2.658 | | | |
| 2.4422 | -2.617 | 2.661 | | | |
| 2.4599 | -2.604 | 2.664 | | | |
| 2.4796 | -2.598 | 2.665 | | | |
| 2.4996 | -2.590 | 2.667 | | | |
| 2.5485 | -2.586 | 2.672 | | | |
| 2.5682 | -2.584 | 2.672 | | | |
| 2.5883 | -2.593 | 2.674 | | | |
| 2.6884 | -2.544 | 2.676 | | | |
| 2.6783 | -2.529 | 2.679 | | | |
| 2.6585 | -2.497 | 2.686 | | | |
| 2.6754 | -2.486 | 2.692 | | | |
| 2.7885 | -2.432 | 2.698 | | | |
| 2.7255 | -2.474 | 2.718 | | | |
| 2.7584 | -2.326 | 2.719 | | | |
| 2.7753 | -2.279 | 2.729 | | | |
| 2.8885 | -2.229 | 2.739 | | | |
| 2.8255 | -2.177 | 2.749 | | | |
| 2.8583 | -2.148 | 2.758 | | | |

BSK 1 181.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.598 ALPHA 4.88 REY 0.81*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.7653 CH = -0.8686

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 0.758 | 0.934 | 0.0000 | 0.758 | 0.934 |
| 0.0089 | -1.408 | 0.496 | 0.0096 | 1.002 | 0.902 |
| 0.0286 | -1.508 | 0.476 | 0.0285 | 0.722 | 0.927 |
| 0.0306 | -1.588 | 0.474 | 0.0306 | 0.602 | 0.903 |
| 0.0584 | -1.635 | 0.463 | 0.0506 | 0.416 | 0.866 |
| 0.0752 | -2.053 | 0.381 | 0.0754 | 0.201 | 0.839 |
| 0.0999 | -1.745 | 0.432 | 0.1001 | 0.169 | 0.617 |
| 0.1252 | -1.223 | 0.544 | 0.2300 | -0.010 | 0.732 |
| 0.1599 | -0.993 | 0.589 | 0.2496 | -0.029 | 0.770 |
| 0.2000 | -0.885 | 0.610 | 0.3495 | -0.007 | 0.767 |
| 0.2487 | -0.846 | 0.619 | 0.4486 | -0.053 | 0.773 |
| 0.2798 | -0.806 | 0.626 | 0.5493 | 0.009 | 0.786 |
| 0.3202 | -0.771 | 0.633 | 0.6499 | 0.122 | 0.808 |
| 0.3623 | -0.724 | 0.642 | 0.7499 | 0.250 | 0.833 |
| 0.4001 | -0.692 | 0.648 | 0.8500 | 0.298 | 0.843 |
| 0.4190 | -0.679 | 0.651 | | | |
| 0.4402 | -0.660 | 0.655 | | | |
| 0.4599 | -0.645 | 0.658 | | | |
| 0.4796 | -0.637 | 0.659 | | | |
| 0.4996 | -0.624 | 0.661 | | | |
| 0.5405 | -0.594 | 0.657 | | | |
| 0.5602 | -0.586 | 0.669 | | | |
| 0.5803 | -0.573 | 0.671 | | | |
| 0.6004 | -0.560 | 0.674 | | | |
| 0.6203 | -0.541 | 0.678 | | | |
| 0.6505 | -0.507 | 0.685 | | | |
| 0.6754 | -0.470 | 0.692 | | | |
| 0.7005 | -0.432 | 0.699 | | | |
| 0.7255 | -0.374 | 0.711 | | | |
| 0.7504 | -0.325 | 0.720 | | | |
| 0.7753 | -0.275 | 0.738 | | | |
| 0.8005 | -0.224 | 0.748 | | | |
| 0.8255 | -0.175 | 0.750 | | | |
| 0.8503 | -0.133 | 0.750 | | | |

B3K 1 181.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

SOLID WALLS

MACH NO. 0.649 ALPHA -2.58 REY 2.81e10

INTEGRATED FORCE COEFFICIENTS

Cx = -0.8663 Cy = -0.8846

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 0.968 | 2.968 | 0.0000 | 0.968 | 2.968 |
| 0.0099 | 0.761 | 2.922 | 0.0099 | -0.446 | 0.662 |
| 0.0206 | 0.401 | 2.842 | 0.0206 | -0.936 | 0.551 |
| 0.0306 | 0.215 | 2.831 | 0.0306 | -0.827 | 0.568 |
| 0.0514 | 0.113 | 2.756 | 0.0514 | -0.863 | 0.568 |
| 0.0752 | -0.111 | 2.715 | 0.0752 | -0.898 | 0.554 |
| 0.0999 | -0.248 | 2.724 | 0.0999 | -0.628 | 0.568 |
| 0.1252 | -0.205 | 2.727 | 0.1252 | -0.738 | 0.535 |
| 0.1509 | -0.197 | 2.729 | 0.1509 | -0.612 | 0.616 |
| 0.1733 | -0.229 | 2.722 | 0.1733 | -0.524 | 0.636 |
| 0.1937 | -0.229 | 2.695 | 0.1937 | -0.359 | 0.672 |
| 0.2193 | -0.207 | 2.689 | 0.2193 | -0.188 | 0.711 |
| 0.2422 | -0.106 | 2.665 | 0.2422 | -0.039 | 0.751 |
| 0.2623 | -0.313 | 2.683 | 0.2623 | 0.136 | 0.783 |
| 0.2821 | -0.320 | 2.681 | 0.2821 | 0.287 | 0.799 |
| 0.3018 | -0.332 | 2.679 | | | |
| 0.3212 | -0.333 | 2.679 | | | |
| 0.3459 | -0.342 | 2.677 | | | |
| 0.3706 | -0.352 | 2.674 | | | |
| 0.3996 | -0.359 | 2.672 | | | |
| 0.4235 | -0.361 | 2.672 | | | |
| 0.4522 | -0.376 | 2.669 | | | |
| 0.4823 | -0.376 | 2.665 | | | |
| 0.5084 | -0.386 | 2.666 | | | |
| 0.5283 | -0.384 | 2.667 | | | |
| 0.5505 | -0.373 | 2.668 | | | |
| 0.5754 | -0.384 | 2.671 | | | |
| 0.6005 | -0.350 | 2.674 | | | |
| 0.6255 | -0.384 | 2.685 | | | |
| 0.6524 | -0.274 | 2.691 | | | |
| 0.6753 | -0.237 | 2.723 | | | |
| 0.6975 | -0.196 | 2.729 | | | |
| 0.7255 | -0.151 | 2.719 | | | |
| 0.7533 | -0.108 | 2.723 | | | |

BSK 1 121.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 1.649 ALPHA 0.00 REY 3,81012⁶

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.2982$ $C_M = -0.8842$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.207 | 2.995 | 0.0000 | 1.037 | 0.995 |
| 0.0089 | 0.154 | 2.787 | 0.0096 | 0.396 | 0.841 |
| 0.0286 | -0.140 | 2.711 | 0.0225 | -0.032 | 0.746 |
| 0.0336 | -0.355 | 2.674 | 0.0386 | -0.073 | 0.737 |
| 0.0534 | -0.545 | 2.632 | 0.0586 | -0.226 | 0.727 |
| 0.0752 | -0.707 | 2.595 | 0.0754 | -0.271 | 0.692 |
| 0.0999 | -0.856 | 2.630 | 0.1031 | -0.354 | 0.674 |
| 0.1252 | -0.992 | 2.671 | 0.2030 | -0.400 | 0.664 |
| 0.1599 | -0.925 | 2.636 | 0.2496 | -0.362 | 0.672 |
| 0.2000 | -0.816 | 2.633 | 0.3495 | -0.347 | 0.676 |
| 0.2437 | -0.519 | 2.637 | 0.4466 | -0.242 | 0.699 |
| 0.2798 | -0.520 | 2.637 | 0.5493 | -0.144 | 0.727 |
| 0.3232 | -0.523 | 2.636 | 0.6499 | 0.043 | 0.762 |
| 0.3683 | -0.507 | 2.640 | 0.7499 | 0.192 | 0.776 |
| 0.4031 | -0.493 | 2.642 | 0.8520 | 0.257 | 0.813 |
| 0.4198 | -0.500 | 2.642 | | | |
| 0.4432 | -0.494 | 2.643 | | | |
| 0.4599 | -0.493 | 2.643 | | | |
| 0.4796 | -0.496 | 2.642 | | | |
| 0.4996 | -0.497 | 2.642 | | | |
| 0.5435 | -0.496 | 2.645 | | | |
| 0.5682 | -0.491 | 2.643 | | | |
| 0.5833 | -0.489 | 2.644 | | | |
| 0.6084 | -0.492 | 2.644 | | | |
| 0.6233 | -0.491 | 2.646 | | | |
| 0.6525 | -0.463 | 2.653 | | | |
| 0.6754 | -0.437 | 2.655 | | | |
| 0.7025 | -0.411 | 2.661 | | | |
| 0.7255 | -0.393 | 2.673 | | | |
| 0.7524 | -0.317 | 2.692 | | | |
| 0.7753 | -0.272 | 2.692 | | | |
| 0.8035 | -0.242 | 2.734 | | | |
| 0.8255 | -0.172 | 2.715 | | | |
| 0.8503 | -0.123 | 2.726 | | | |

B3K 1 121.6 MM CHORD

EXPERIMENTAL PRESSURE DISTRIBUTION

SOLID WALLS

MACH NO. 0.632 ALPHA 1.88 REY 0.61×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.4292$ $C_M = -0.0819$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 1.372 | 2.991 | 0.0000 | 1.372 | 2.991 |
| 0.0059 | -0.145 | 2.719 | 0.0059 | 0.617 | 2.853 |
| 0.0206 | -0.460 | 2.949 | 0.0206 | 0.214 | 2.853 |
| 0.0336 | -0.608 | 2.615 | 0.0336 | 0.139 | 2.753 |
| 0.0524 | -0.796 | 2.573 | 0.0524 | -0.013 | 2.743 |
| 0.0752 | -0.935 | 2.531 | 0.0752 | -0.145 | 2.729 |
| 0.0999 | -0.988 | 2.548 | 0.1001 | -0.197 | 2.719 |
| 0.1252 | -0.785 | 2.589 | 0.1252 | -0.291 | 2.653 |
| 0.1599 | -0.601 | 2.624 | 0.1599 | -0.270 | 2.652 |
| 0.2000 | -0.533 | 2.613 | 0.2000 | -0.282 | 2.642 |
| 0.2437 | -0.622 | 2.512 | 0.2437 | -0.195 | 2.719 |
| 0.2799 | -0.614 | 2.614 | 0.2799 | -0.036 | 2.714 |
| 0.3222 | -0.601 | 2.617 | 0.3222 | 0.063 | 2.757 |
| 0.3623 | -0.578 | 2.622 | 0.3623 | 0.237 | 2.779 |
| 0.4001 | -0.501 | 2.625 | 0.4001 | 0.270 | 2.823 |
| 0.4198 | -0.550 | 2.525 | | | |
| 0.4432 | -0.550 | 2.628 | | | |
| 0.4599 | -0.545 | 2.632 | | | |
| 0.4796 | -0.544 | 2.633 | | | |
| 0.4996 | -0.543 | 2.631 | | | |
| 0.5405 | -0.525 | 2.635 | | | |
| 0.5602 | -0.528 | 2.634 | | | |
| 0.5833 | -0.523 | 2.635 | | | |
| 0.6024 | -0.520 | 2.636 | | | |
| 0.6223 | -0.509 | 2.638 | | | |
| 0.6506 | -0.486 | 2.643 | | | |
| 0.6754 | -0.454 | 2.651 | | | |
| 0.7025 | -0.423 | 2.658 | | | |
| 0.7255 | -0.385 | 2.670 | | | |
| 0.7504 | -0.321 | 2.681 | | | |
| 0.7753 | -0.275 | 2.691 | | | |
| 0.8035 | -0.245 | 2.722 | | | |
| 0.8255 | -0.173 | 2.714 | | | |
| 0.8523 | -0.124 | 2.725 | | | |

BSK 1 131.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.649 ALPHA 2.00 REY 0.61×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.5474$ $C_M = -0.0783$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 1.342 | 0.980 | 0.0000 | 1.022 | 0.980 |
| 0.0089 | -2.441 | 2.444 | 0.0096 | 0.787 | 0.920 |
| 0.0226 | -3.794 | 2.585 | 0.0205 | 0.411 | 0.844 |
| 0.0326 | -3.865 | 2.561 | 0.0306 | 0.319 | 0.824 |
| 0.0504 | -1.856 | 2.518 | 0.0506 | 0.145 | 0.785 |
| 0.0752 | -1.418 | 2.437 | 0.0754 | 0.034 | 0.761 |
| 0.0992 | -1.216 | 2.449 | 0.1001 | -0.067 | 0.738 |
| 0.1252 | -1.043 | 2.548 | 0.2000 | -0.194 | 0.710 |
| 0.1599 | -0.793 | 2.577 | 0.2496 | -0.288 | 0.711 |
| 0.2000 | -0.738 | 2.589 | 0.3495 | -0.217 | 0.705 |
| 0.2407 | -0.725 | 2.592 | 0.4486 | -0.149 | 0.720 |
| 0.2798 | -0.703 | 2.596 | 0.5493 | -0.055 | 0.741 |
| 0.3202 | -0.653 | 2.581 | 0.6499 | 0.081 | 0.771 |
| 0.3603 | -0.647 | 2.609 | 0.7499 | 0.223 | 0.803 |
| 0.4001 | -0.624 | 2.614 | 0.8500 | 0.280 | 0.815 |
| 0.4195 | -0.616 | 2.516 | | | |
| 0.4472 | -0.604 | 2.519 | | | |
| 0.4599 | -0.596 | 2.620 | | | |
| 0.4796 | -0.592 | 2.621 | | | |
| 0.4996 | -0.586 | 2.623 | | | |
| 0.5425 | -0.564 | 2.627 | | | |
| 0.5602 | -0.559 | 2.629 | | | |
| 0.5683 | -0.552 | 2.630 | | | |
| 0.6124 | -0.543 | 2.632 | | | |
| 0.6203 | -0.530 | 2.635 | | | |
| 0.6505 | -0.531 | 2.641 | | | |
| 0.6754 | -0.498 | 2.649 | | | |
| 0.7005 | -0.430 | 2.657 | | | |
| 0.7255 | -0.373 | 2.670 | | | |
| 0.7504 | -0.325 | 2.681 | | | |
| 0.7753 | -0.276 | 2.691 | | | |
| 0.8005 | -0.223 | 2.703 | | | |
| 0.8255 | -0.174 | 2.714 | | | |
| 0.8503 | -0.124 | 2.725 | | | |

BSK 1 121.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 4.650 ALPHA 3.00 REY 3.61×10^6

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.6736$ $C_M = -0.2733$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 2.942 | 2.962 | 0.0000 | 0.943 | 2.902 |
| 0.0049 | -2.921 | 2.969 | 0.0096 | 0.935 | 2.904 |
| 0.0226 | -1.323 | 2.925 | 0.0205 | 0.570 | 2.879 |
| 0.0306 | -1.206 | 2.913 | 0.0306 | 0.461 | 2.855 |
| 0.0524 | -1.243 | 2.475 | 0.0536 | 0.253 | 2.815 |
| 0.0752 | -1.633 | 2.383 | 0.0754 | 0.157 | 2.757 |
| 0.0999 | -1.692 | 2.375 | 0.1001 | 0.053 | 2.703 |
| 0.1252 | -1.673 | 2.379 | 0.1200 | -0.104 | 2.729 |
| 0.1599 | -2.977 | 2.532 | 0.1496 | -0.113 | 2.727 |
| 0.2000 | -2.604 | 2.573 | 0.1795 | -0.159 | 2.717 |
| 0.2427 | -2.772 | 2.583 | 0.2086 | -0.104 | 2.709 |
| 0.2793 | -2.754 | 2.584 | 0.2393 | -0.027 | 2.716 |
| 0.3222 | -2.733 | 2.589 | 0.2699 | 0.101 | 2.775 |
| 0.3603 | -2.694 | 2.597 | 0.2999 | 0.238 | 2.805 |
| 0.4021 | -2.658 | 2.623 | 0.3500 | 0.292 | 2.817 |
| 0.4498 | -2.682 | 2.625 | | | |
| 0.4402 | -2.644 | 2.623 | | | |
| 0.4599 | -2.636 | 2.613 | | | |
| 0.4796 | -2.629 | 2.612 | | | |
| 0.4996 | -2.621 | 2.614 | | | |
| 0.5405 | -2.592 | 2.623 | | | |
| 0.5632 | -2.597 | 2.621 | | | |
| 0.5803 | -2.575 | 2.624 | | | |
| 0.6004 | -2.564 | 2.626 | | | |
| 0.6203 | -2.593 | 2.631 | | | |
| 0.6325 | -2.569 | 2.639 | | | |
| 0.6704 | -2.473 | 2.647 | | | |
| 0.7005 | -2.435 | 2.655 | | | |
| 0.7205 | -2.375 | 2.669 | | | |
| 0.7524 | -2.327 | 2.679 | | | |
| 0.7753 | -2.277 | 2.693 | | | |
| 0.8005 | -2.224 | 2.702 | | | |
| 0.8205 | -2.171 | 2.714 | | | |
| 0.8523 | -2.126 | 2.724 | | | |

BOX 1 121.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

SOLID WALLS

MACH. NO. 0.740 ALP-4 -2.52 REY 2.31*10⁶

INTEGRATED FORCE COEFFICIENTS

C_x = -0.8657 C_y = -0.8907

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 0.997 | 2.967 | 0.0000 | 0.997 | 2.967 |
| 0.0050 | 0.791 | 2.914 | 0.0096 | -0.317 | 2.641 |
| 0.0100 | 0.426 | 2.626 | 0.0205 | -0.860 | 2.506 |
| 0.0150 | 0.241 | 2.782 | 0.0306 | -3.793 | 2.524 |
| 0.0200 | 0.027 | 2.727 | 0.0506 | -0.876 | 2.512 |
| 0.0250 | -0.168 | 2.679 | 0.0754 | -1.183 | 2.445 |
| 0.0300 | -0.225 | 2.665 | 0.1001 | -0.993 | 2.473 |
| 0.0350 | -0.206 | 2.669 | 0.2000 | -0.820 | 2.516 |
| 0.0400 | -0.201 | 2.672 | 0.2496 | -0.675 | 2.552 |
| 0.0450 | -0.206 | 2.652 | 0.3495 | -0.560 | 2.500 |
| 0.0500 | -0.213 | 2.653 | 0.4466 | -0.370 | 2.627 |
| 0.0550 | -0.304 | 2.645 | 0.5493 | -0.166 | 2.673 |
| 0.0600 | -0.327 | 2.639 | 0.6499 | -0.005 | 2.718 |
| 0.0650 | -0.334 | 2.635 | 0.7499 | 0.136 | 2.753 |
| 0.0700 | -0.345 | 2.635 | 0.8500 | 0.205 | 2.772 |
| 0.0750 | -0.357 | 2.632 | | | |
| 0.0800 | -0.350 | 2.631 | | | |
| 0.0850 | -0.359 | 2.629 | | | |
| 0.0900 | -0.379 | 2.627 | | | |
| 0.0950 | -0.355 | 2.624 | | | |
| 0.1000 | -0.390 | 2.623 | | | |
| 0.1050 | -0.407 | 2.619 | | | |
| 0.1100 | -0.412 | 2.614 | | | |
| 0.1150 | -0.422 | 2.615 | | | |
| 0.1200 | -0.423 | 2.614 | | | |
| 0.1250 | -0.413 | 2.613 | | | |
| 0.1300 | -0.393 | 2.623 | | | |
| 0.1350 | -0.374 | 2.627 | | | |
| 0.1400 | -0.361 | 2.643 | | | |
| 0.1450 | -0.265 | 2.649 | | | |
| 0.1500 | -0.244 | 2.659 | | | |
| 0.1550 | -0.197 | 2.671 | | | |
| 0.1600 | -0.148 | 2.683 | | | |
| 0.1650 | -0.104 | 2.694 | | | |

BK 1 131.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.713 ALPHA 2.88 REY 2.51*10⁶

INTEGRATED FORCE COEFFICIENTS
C_L = 0.2941 C_M = -0.0877

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.107 | 0.995 | 0.0000 | 1.137 | 2.925 |
| 0.0050 | 0.227 | 0.777 | 0.0096 | 0.423 | 0.821 |
| 0.0206 | -0.125 | 0.693 | 0.0265 | -0.036 | 2.712 |
| 0.0326 | -0.249 | 0.646 | 0.0326 | -0.076 | 0.702 |
| 0.0504 | -0.515 | 0.593 | 0.0506 | -0.223 | 0.656 |
| 0.0752 | -0.721 | 0.535 | 0.0754 | -0.336 | 0.645 |
| 0.0999 | -0.740 | 0.537 | 0.1021 | -0.369 | 0.625 |
| 0.1252 | -0.626 | 0.565 | 0.2000 | -0.445 | 0.611 |
| 0.1599 | -0.550 | 0.584 | 0.2496 | -0.424 | 0.621 |
| 0.2000 | -0.545 | 0.586 | 0.3495 | -0.386 | 0.626 |
| 0.2407 | -0.556 | 0.583 | 0.4486 | -0.263 | 0.606 |
| 0.2799 | -0.563 | 0.581 | 0.5493 | -0.124 | 0.607 |
| 0.3202 | -0.565 | 0.581 | 0.6499 | 0.042 | 0.731 |
| 0.3603 | -0.545 | 0.585 | 0.7499 | 0.195 | 0.769 |
| 0.4001 | -0.539 | 0.587 | 0.8502 | 0.260 | 0.705 |
| 0.4198 | -0.542 | 0.586 | | | |
| 0.4432 | -0.538 | 0.587 | | | |
| 0.4609 | -0.537 | 0.588 | | | |
| 0.4796 | -0.538 | 0.587 | | | |
| 0.4996 | -0.539 | 0.587 | | | |
| 0.5205 | -0.527 | 0.590 | | | |
| 0.5422 | -0.533 | 0.589 | | | |
| 0.5601 | -0.529 | 0.590 | | | |
| 0.5804 | -0.528 | 0.590 | | | |
| 0.6003 | -0.516 | 0.593 | | | |
| 0.6203 | -0.491 | 0.599 | | | |
| 0.6403 | -0.461 | 0.607 | | | |
| 0.6754 | -0.424 | 0.615 | | | |
| 0.7025 | -0.388 | 0.630 | | | |
| 0.7255 | -0.341 | 0.642 | | | |
| 0.7504 | -0.273 | 0.653 | | | |
| 0.7753 | -0.219 | 0.667 | | | |
| 0.8005 | -0.157 | 0.680 | | | |
| 0.8255 | -0.116 | 0.692 | | | |
| 0.8503 | | | | | |

BGX 1 171.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.701 ALPHA 1.30 REY 3.51×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.4415$ $C_M = -0.2638$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 1.244 | 2.991 | 0.0000 | 1.094 | 3.931 |
| 0.0059 | -0.346 | 2.788 | 0.0096 | 0.611 | 3.871 |
| 0.0226 | -0.307 | 2.629 | 0.0205 | 0.205 | 2.771 |
| 0.0326 | -0.519 | 2.591 | 0.0306 | 0.134 | 2.753 |
| 0.0524 | -0.725 | 2.543 | 0.0506 | -0.028 | 2.713 |
| 0.0752 | -1.116 | 2.443 | 0.0754 | -0.121 | 2.673 |
| 0.0999 | -1.198 | 2.423 | 0.1021 | -0.221 | 2.656 |
| 0.1252 | -0.932 | 2.489 | 0.1202 | -0.321 | 2.641 |
| 0.1599 | -0.557 | 2.553 | 0.1496 | -0.382 | 2.646 |
| 0.2000 | -0.654 | 2.555 | 0.1795 | -0.308 | 2.644 |
| 0.2437 | -0.675 | 2.552 | 0.2486 | -0.211 | 2.658 |
| 0.2798 | -0.673 | 2.553 | 0.3493 | -0.091 | 2.678 |
| 0.3222 | -0.659 | 2.554 | 0.4499 | 0.063 | 2.736 |
| 0.3623 | -0.636 | 2.562 | 0.5499 | 0.212 | 2.773 |
| 0.4021 | -0.619 | 2.566 | 0.6502 | 0.274 | 2.738 |
| 0.4495 | -0.612 | 2.565 | | | |
| 0.4422 | -0.603 | 2.572 | | | |
| 0.4599 | -0.597 | 2.572 | | | |
| 0.4796 | -0.594 | 2.572 | | | |
| 0.4996 | -0.591 | 2.573 | | | |
| 0.5425 | -0.573 | 2.578 | | | |
| 0.5622 | -0.570 | 2.578 | | | |
| 0.5833 | -0.562 | 2.583 | | | |
| 0.6034 | -0.557 | 2.582 | | | |
| 0.6223 | -0.542 | 2.585 | | | |
| 0.6525 | -0.529 | 2.594 | | | |
| 0.6754 | -0.472 | 2.623 | | | |
| 0.7025 | -0.431 | 2.613 | | | |
| 0.7255 | -0.372 | 2.628 | | | |
| 0.7524 | -0.319 | 2.643 | | | |
| 0.7753 | -0.269 | 2.653 | | | |
| 0.8005 | -0.213 | 2.667 | | | |
| 0.8255 | -0.161 | 2.683 | | | |
| 0.8523 | -0.112 | 2.692 | | | |

BK 1 131.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.639 ALPHA 2.00 REY 2.81×10^6

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.5933$ $C_M = -0.0789$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 2.2033 | 1.263 | 2.954 | 0.0338 | 1.063 | 2.954 |
| 2.2089 | -0.319 | 2.643 | 0.0396 | 0.776 | 2.913 |
| 2.2206 | -0.594 | 2.574 | 0.0205 | 0.488 | 0.819 |
| 2.2336 | -0.710 | 2.546 | 0.0306 | 0.398 | 0.796 |
| 2.2524 | -0.912 | 2.496 | 0.0536 | 0.135 | 0.753 |
| 2.2752 | -1.239 | 2.415 | 0.0754 | 0.025 | 0.726 |
| 2.2999 | -1.382 | 2.383 | 0.1001 | -0.085 | 0.699 |
| 2.3222 | -1.433 | 2.367 | 0.2328 | -0.223 | 0.665 |
| 2.3599 | -1.355 | 2.387 | 0.2496 | -0.214 | 0.657 |
| 2.4223 | -1.163 | 2.434 | 0.3495 | -0.245 | 0.659 |
| 2.4277 | -0.734 | 2.543 | 0.4486 | -0.166 | 0.679 |
| 2.4798 | -0.686 | 2.552 | 0.5493 | -0.062 | 0.774 |
| 2.5222 | -0.688 | 2.551 | 0.6499 | 0.093 | 0.743 |
| 2.5633 | -0.670 | 2.556 | 0.7499 | 0.230 | 0.777 |
| 2.6031 | -0.653 | 2.559 | 0.8503 | 0.289 | 0.791 |
| 2.6496 | -0.654 | 2.559 | | | |
| 2.6472 | -0.643 | 2.562 | | | |
| 2.6599 | -0.635 | 2.564 | | | |
| 2.6796 | -0.633 | 2.565 | | | |
| 2.6996 | -0.625 | 2.566 | | | |
| 2.7435 | -0.608 | 2.572 | | | |
| 2.7632 | -0.598 | 2.573 | | | |
| 2.7833 | -0.588 | 2.575 | | | |
| 2.8034 | -0.577 | 2.578 | | | |
| 2.8223 | -0.568 | 2.582 | | | |
| 2.8525 | -0.523 | 2.591 | | | |
| 2.8754 | -0.485 | 2.601 | | | |
| 2.9035 | -0.444 | 2.611 | | | |
| 2.9295 | -0.381 | 2.626 | | | |
| 2.9534 | -0.327 | 2.639 | | | |
| 2.9753 | -0.274 | 2.652 | | | |
| 2.9985 | -0.216 | 2.667 | | | |
| 2.8295 | -0.184 | 2.679 | | | |
| 2.8523 | -0.115 | 2.692 | | | |

BSK 1 121.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

WACH NO. 4.700 ALPHA 3.00 REY 0.31×10^6

INTEGRATED FORCE COEFFICIENTS

$C_F = 0.7559$ $C_M = -0.0743$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 0.949 | 2.963 | 0.0000 | 0.999 | 2.948 |
| 0.0049 | -0.547 | 2.575 | 0.0096 | 0.891 | 2.941 |
| 0.0206 | -0.314 | 2.519 | 0.0205 | 0.553 | 2.858 |
| 0.0306 | -0.306 | 2.581 | 0.0306 | 0.449 | 2.832 |
| 0.0524 | -1.028 | 2.466 | 0.0506 | 0.271 | 2.758 |
| 0.0752 | -1.439 | 2.364 | 0.0754 | 0.148 | 2.758 |
| 0.0999 | -1.514 | 2.346 | 0.1001 | 0.037 | 2.758 |
| 0.1252 | -1.523 | 2.343 | 0.1202 | -0.122 | 2.691 |
| 0.1599 | -1.445 | 2.353 | 0.1496 | -0.128 | 2.659 |
| 0.2000 | -1.502 | 2.349 | 0.1795 | -0.175 | 2.678 |
| 0.2407 | -1.455 | 2.360 | 0.2096 | -0.116 | 2.692 |
| 0.2793 | -1.454 | 2.361 | 0.2493 | -0.027 | 2.724 |
| 0.3202 | -1.195 | 2.432 | 0.2899 | 0.146 | 2.747 |
| 0.3603 | -0.755 | 2.534 | 0.3299 | 0.246 | 2.752 |
| 0.4001 | -0.651 | 2.564 | 0.3600 | 0.301 | 2.745 |
| 0.4198 | -0.617 | 2.568 | | | |
| 0.4402 | -0.599 | 2.572 | | | |
| 0.4599 | -0.584 | 2.576 | | | |
| 0.4796 | -0.561 | 2.577 | | | |
| 0.4996 | -0.576 | 2.573 | | | |
| 0.5105 | -0.561 | 2.582 | | | |
| 0.5202 | -0.553 | 2.583 | | | |
| 0.5303 | -0.553 | 2.584 | | | |
| 0.5404 | -0.543 | 2.586 | | | |
| 0.5507 | -0.527 | 2.593 | | | |
| 0.5605 | -0.494 | 2.593 | | | |
| 0.5754 | -0.462 | 2.607 | | | |
| 0.5805 | -0.422 | 2.616 | | | |
| 0.5905 | -0.384 | 2.633 | | | |
| 0.6004 | -0.314 | 2.643 | | | |
| 0.6103 | -0.264 | 2.655 | | | |
| 0.6205 | -0.212 | 2.663 | | | |
| 0.6305 | -0.193 | 2.681 | | | |
| 0.6403 | -0.118 | 2.692 | | | |

BOX 1 121.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 4.722 ALPHA -2.58 REY 2.81×10^6

INTEGRATED FORCE COEFFICIENTS

CV = -0.1786 CM = -0.3931

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.000 | 2.967 | 0.0000 | 1.000 | 2.967 |
| 0.0050 | 0.765 | 2.929 | 0.0050 | -0.273 | 2.637 |
| 0.0100 | 0.436 | 2.819 | 0.0100 | -0.818 | 2.406 |
| 0.0150 | 0.248 | 2.777 | 0.0150 | -0.753 | 2.513 |
| 0.0200 | 0.334 | 2.715 | 0.0200 | -0.652 | 2.408 |
| 0.0250 | -0.196 | 2.664 | 0.0250 | -1.033 | 2.428 |
| 0.0300 | -0.224 | 2.549 | 0.0300 | -1.087 | 2.427 |
| 0.0350 | -0.267 | 2.653 | 0.0350 | -1.031 | 2.449 |
| 0.0400 | -0.199 | 2.655 | 0.0400 | -0.685 | 2.532 |
| 0.0450 | -0.235 | 2.646 | 0.0450 | -0.572 | 2.563 |
| 0.0500 | -0.273 | 2.636 | 0.0500 | -0.376 | 2.613 |
| 0.0550 | -0.306 | 2.627 | 0.0550 | -0.185 | 2.659 |
| 0.0600 | -0.332 | 2.621 | 0.0600 | -0.033 | 2.706 |
| 0.0650 | -0.359 | 2.619 | 0.0650 | 0.137 | 2.742 |
| 0.0700 | -0.391 | 2.616 | 0.0700 | 0.234 | 2.752 |
| 0.0750 | -0.363 | 2.613 | | | |
| 0.0800 | -0.337 | 2.612 | | | |
| 0.0850 | -0.377 | 2.609 | | | |
| 0.0900 | -0.337 | 2.607 | | | |
| 0.0950 | -0.337 | 2.604 | | | |
| 0.1000 | -0.402 | 2.603 | | | |
| 0.1050 | -0.421 | 2.598 | | | |
| 0.1100 | -0.426 | 2.597 | | | |
| 0.1150 | -0.435 | 2.594 | | | |
| 0.1200 | -0.435 | 2.594 | | | |
| 0.1250 | -0.424 | 2.597 | | | |
| 0.1300 | -0.404 | 2.603 | | | |
| 0.1350 | -0.385 | 2.608 | | | |
| 0.1400 | -0.328 | 2.622 | | | |
| 0.1450 | -0.259 | 2.633 | | | |
| 0.1500 | -0.247 | 2.643 | | | |
| 0.1550 | -0.198 | 2.655 | | | |
| 0.1600 | -0.148 | 2.669 | | | |
| 0.1650 | -0.101 | 2.681 | | | |

BSK 1 121.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.722 ALPHA 0.30 REYNOLDS 2.81×10^6

INTEGRATED FORCE COEFFICIENTS

$C_D = 0.2943$ $C_M = -0.3886$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 1.112 | 2.994 | 0.0000 | 1.112 | 2.994 |
| 0.0059 | 1.205 | 2.775 | 0.0096 | 0.488 | 2.811 |
| 0.0206 | -0.387 | 2.954 | 0.0285 | -0.838 | 2.678 |
| 0.0326 | -0.284 | 2.838 | 0.0386 | -0.879 | 2.628 |
| 0.0534 | -0.492 | 2.582 | 0.0526 | -0.232 | 2.649 |
| 0.0752 | -0.767 | 2.528 | 0.0754 | -0.324 | 2.625 |
| 0.0999 | -0.777 | 2.525 | 0.1081 | -0.487 | 2.624 |
| 0.1252 | -0.643 | 2.543 | 0.2080 | -0.472 | 2.557 |
| 0.1599 | -0.591 | 2.561 | 0.2496 | -0.427 | 2.538 |
| 0.2038 | -0.594 | 2.561 | 0.3495 | -0.444 | 2.624 |
| 0.2407 | -0.578 | 2.557 | 0.4486 | -0.273 | 2.638 |
| 0.2798 | -0.559 | 2.554 | 0.5493 | -0.127 | 2.675 |
| 0.3202 | -0.544 | 2.553 | 0.6499 | 0.042 | 2.719 |
| 0.3583 | -0.572 | 2.559 | 0.7499 | 0.194 | 2.759 |
| 0.4001 | -0.591 | 2.562 | 0.8502 | 0.263 | 2.775 |
| 0.4194 | -0.584 | 2.561 | | | |
| 0.4422 | -0.598 | 2.562 | | | |
| 0.4599 | -0.597 | 2.562 | | | |
| 0.4796 | -0.597 | 2.562 | | | |
| 0.4996 | -0.599 | 2.563 | | | |
| 0.5425 | -0.544 | 2.567 | | | |
| 0.5682 | -0.591 | 2.565 | | | |
| 0.5833 | -0.547 | 2.565 | | | |
| 0.6024 | -0.543 | 2.567 | | | |
| 0.6223 | -0.582 | 2.573 | | | |
| 0.6525 | -0.543 | 2.577 | | | |
| 0.6754 | -0.498 | 2.586 | | | |
| 0.7035 | -0.432 | 2.596 | | | |
| 0.7255 | -0.398 | 2.612 | | | |
| 0.7504 | -0.319 | 2.626 | | | |
| 0.7753 | -0.209 | 2.639 | | | |
| 0.8025 | -0.213 | 2.653 | | | |
| 0.8255 | -0.199 | 2.667 | | | |
| 0.8523 | -0.167 | 2.683 | | | |

GSK 1 121.9 MM CHORD
 EXPERIMENTAL PRESSURE DISTRIBUTION
 SOLID WALLS

MACH NO. 0.721 ALPHA 1.02 REY 2.81×10^6

INTEGRATED FORCE COEFFICIENTS
 $C_F = 0.4489$ $C_M = -0.0845$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 2.0000 | 1.147 | 2.092 | 0.0000 | 1.147 | 2.092 |
| 2.0050 | 1.215 | 2.711 | 0.0096 | 0.718 | 2.854 |
| 2.0225 | -0.312 | 2.626 | 0.0205 | 2.201 | 2.759 |
| 2.0335 | -0.405 | 2.587 | 0.0305 | 0.138 | 2.742 |
| 2.0504 | -0.572 | 2.534 | 0.0506 | -0.035 | 2.698 |
| 2.0752 | -1.343 | 2.438 | 0.0754 | -0.134 | 2.673 |
| 2.0999 | -1.172 | 2.485 | 0.1001 | -0.236 | 2.647 |
| 2.1252 | -1.103 | 2.437 | 0.1200 | -0.346 | 2.610 |
| 2.1509 | -0.911 | 2.472 | 0.1494 | -0.323 | 2.624 |
| 2.1733 | -0.843 | 2.541 | 0.1795 | -0.333 | 2.622 |
| 2.1927 | -0.695 | 2.535 | 0.1986 | -0.226 | 2.649 |
| 2.2193 | -0.686 | 2.533 | 0.1993 | -0.078 | 2.662 |
| 2.2322 | -0.695 | 2.527 | 0.1999 | 0.062 | 2.723 |
| 2.2505 | -0.691 | 2.536 | 0.1999 | 0.212 | 2.762 |
| 2.2601 | -0.690 | 2.542 | 0.2000 | 0.275 | 2.778 |
| 2.2619 | -0.686 | 2.542 | | | |
| 2.2622 | -0.620 | 2.545 | | | |
| 2.2699 | -0.621 | 2.547 | | | |
| 2.2796 | -0.621 | 2.547 | | | |
| 2.2896 | -0.618 | 2.547 | | | |
| 2.2935 | -0.594 | 2.554 | | | |
| 2.2932 | -0.593 | 2.554 | | | |
| 2.2923 | -0.593 | 2.555 | | | |
| 2.2984 | -0.575 | 2.559 | | | |
| 2.2933 | -0.556 | 2.563 | | | |
| 2.2935 | -0.519 | 2.573 | | | |
| 2.2954 | -0.479 | 2.583 | | | |
| 2.2935 | -0.418 | 2.594 | | | |
| 2.2955 | -0.372 | 2.611 | | | |
| 2.2934 | -0.317 | 2.625 | | | |
| 2.2953 | -0.266 | 2.635 | | | |
| 2.2935 | -0.269 | 2.653 | | | |
| 2.2955 | -0.196 | 2.667 | | | |
| 2.2933 | -0.148 | 2.677 | | | |

BSK 1 131.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.719 ALPHA 2.38 REY 0.81013

INTEGRATED FORCE COEFFICIENTS

CV = 0.6172 CM = -0.0795

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 1.202 | 2.985 | 0.0000 | 1.002 | 2.956 |
| 0.0259 | -0.241 | 2.845 | 0.0259 | 0.765 | 2.925 |
| 0.0518 | -0.521 | 2.575 | 0.0518 | 0.335 | 2.857 |
| 0.0777 | -0.641 | 2.544 | 0.0777 | 0.296 | 2.754 |
| 0.1036 | -0.437 | 2.403 | 0.1036 | 0.121 | 2.719 |
| 0.1295 | -1.157 | 2.411 | 0.1295 | 0.014 | 2.712 |
| 0.1554 | -1.302 | 2.375 | 0.1554 | -0.096 | 2.654 |
| 0.1813 | -1.373 | 2.361 | 0.1813 | -0.236 | 2.649 |
| 0.2072 | -1.302 | 2.374 | 0.2072 | -0.229 | 2.649 |
| 0.2331 | -1.201 | 2.383 | 0.2331 | -0.259 | 2.642 |
| 0.2590 | -1.215 | 2.396 | 0.2590 | -0.175 | 2.663 |
| 0.2849 | -1.102 | 2.405 | 0.2849 | -0.066 | 2.691 |
| 0.3108 | -0.893 | 2.479 | 0.3108 | 0.005 | 2.753 |
| 0.3367 | -0.636 | 2.545 | 0.3367 | 0.232 | 2.758 |
| 0.3626 | -0.607 | 2.552 | 0.3626 | 0.291 | 2.753 |
| 0.3885 | -0.608 | 2.552 | | | |
| 0.4144 | -0.602 | 2.554 | | | |
| 0.4403 | -0.602 | 2.554 | | | |
| 0.4662 | -0.605 | 2.553 | | | |
| 0.4921 | -0.604 | 2.553 | | | |
| 0.5180 | -0.508 | 2.557 | | | |
| 0.5439 | -0.509 | 2.557 | | | |
| 0.5698 | -0.551 | 2.559 | | | |
| 0.5957 | -0.573 | 2.561 | | | |
| 0.6216 | -0.505 | 2.566 | | | |
| 0.6475 | -0.517 | 2.576 | | | |
| 0.6734 | -0.473 | 2.585 | | | |
| 0.6993 | -0.435 | 2.597 | | | |
| 0.7252 | -0.371 | 2.613 | | | |
| 0.7511 | -0.317 | 2.627 | | | |
| 0.7770 | -0.254 | 2.641 | | | |
| 0.8029 | -0.208 | 2.653 | | | |
| 0.8288 | -0.176 | 2.663 | | | |
| 0.8547 | -0.109 | 2.693 | | | |

80K 1 121.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.739 ALPHA -2.52 REY 8.31e12

INTEGRATED FORCE COEFFICIENTS

CL = -2.1294 CM = -0.0954

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.722 | 2.967 | 0.0000 | 1.022 | 2.967 |
| 0.0089 | 0.799 | 2.935 | 0.0096 | -0.226 | 2.636 |
| 0.0226 | 0.439 | 2.812 | 0.0205 | -0.772 | 2.491 |
| 0.0306 | 0.292 | 2.763 | 0.0306 | -0.709 | 2.508 |
| 0.0584 | 0.339 | 2.736 | 0.0506 | -0.814 | 2.432 |
| 0.0752 | -0.126 | 2.651 | 0.0754 | -1.036 | 2.421 |
| 0.0999 | -0.226 | 2.635 | 0.1001 | -1.053 | 2.417 |
| 0.1252 | -0.212 | 2.643 | 0.1203 | -1.137 | 2.394 |
| 0.1599 | -0.202 | 2.642 | 0.1496 | -1.136 | 2.394 |
| 0.1700 | -0.243 | 2.631 | 0.1495 | -0.532 | 2.555 |
| 0.1807 | -0.261 | 2.621 | 0.1486 | -0.361 | 2.613 |
| 0.1798 | -0.326 | 2.612 | 0.1493 | -0.176 | 2.652 |
| 0.1302 | -0.343 | 2.624 | 0.1499 | 0.000 | 2.646 |
| 0.1003 | -0.321 | 2.622 | 0.1499 | 0.133 | 2.733 |
| 0.1001 | -0.392 | 2.599 | 0.0500 | 0.245 | 2.751 |
| 0.1198 | -0.374 | 2.596 | | | |
| 0.1432 | -0.367 | 2.595 | | | |
| 0.1499 | -0.391 | 2.592 | | | |
| 0.1796 | -0.402 | 2.589 | | | |
| 0.1996 | -0.413 | 2.586 | | | |
| 0.1585 | -0.417 | 2.585 | | | |
| 0.1602 | -0.405 | 2.588 | | | |
| 0.1533 | -0.442 | 2.578 | | | |
| 0.1604 | -0.452 | 2.576 | | | |
| 0.1623 | -0.450 | 2.576 | | | |
| 0.1635 | -0.438 | 2.579 | | | |
| 0.1654 | -0.416 | 2.585 | | | |
| 0.1705 | -0.394 | 2.591 | | | |
| 0.1755 | -0.363 | 2.607 | | | |
| 0.1784 | -0.292 | 2.616 | | | |
| 0.1753 | -0.249 | 2.630 | | | |
| 0.1635 | -0.197 | 2.644 | | | |
| 0.1625 | -0.146 | 2.657 | | | |
| 0.1583 | -0.098 | 2.672 | | | |

BOX 1 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.740 ALPHA -1.00 REV 0.01018

INTEGRATED FORCE COEFFICIENTS

CV = 0.1397 CM = -0.0923

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.105 | 2.989 | 0.0000 | 1.105 | 0.959 |
| 0.0089 | 0.527 | 2.632 | 0.0089 | 0.160 | 2.737 |
| 0.0206 | 0.102 | 2.735 | 0.0206 | -0.315 | 2.610 |
| 0.0306 | -0.000 | 2.686 | 0.0306 | -0.316 | 2.610 |
| 0.0334 | -0.202 | 2.627 | 0.0334 | -0.457 | 2.573 |
| 0.0352 | -0.405 | 2.565 | 0.0352 | -0.573 | 2.542 |
| 0.0399 | -0.527 | 2.554 | 0.0399 | -0.626 | 2.527 |
| 0.0425 | -0.400 | 2.572 | 0.0425 | -0.681 | 2.513 |
| 0.0459 | -0.425 | 2.564 | 0.0459 | -0.580 | 2.542 |
| 0.0470 | -0.437 | 2.578 | 0.0470 | -0.510 | 2.529 |
| 0.0477 | -0.406 | 2.573 | 0.0477 | -0.332 | 2.606 |
| 0.0478 | -0.492 | 2.563 | 0.0478 | -0.158 | 2.652 |
| 0.0482 | -0.510 | 2.559 | 0.0482 | 0.023 | 2.721 |
| 0.0483 | -0.502 | 2.561 | 0.0483 | 0.172 | 2.741 |
| 0.0484 | -0.500 | 2.561 | 0.0484 | 0.240 | 2.750 |
| 0.0486 | -0.507 | 2.559 | | | |
| 0.0487 | -0.507 | 2.559 | | | |
| 0.0489 | -0.511 | 2.554 | | | |
| 0.0496 | -0.520 | 2.554 | | | |
| 0.0496 | -0.526 | 2.554 | | | |
| 0.0495 | -0.519 | 2.556 | | | |
| 0.0502 | -0.532 | 2.553 | | | |
| 0.0503 | -0.503 | 2.552 | | | |
| 0.0504 | -0.505 | 2.552 | | | |
| 0.0503 | -0.527 | 2.554 | | | |
| 0.0505 | -0.499 | 2.561 | | | |
| 0.0504 | -0.406 | 2.570 | | | |
| 0.0505 | -0.431 | 2.583 | | | |
| 0.0505 | -0.385 | 2.597 | | | |
| 0.0504 | -0.314 | 2.611 | | | |
| 0.0503 | -0.294 | 2.624 | | | |
| 0.0505 | -0.267 | 2.639 | | | |
| 0.0505 | -0.192 | 2.654 | | | |
| 0.0503 | -0.103 | 2.667 | | | |

BGK 1 181.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.740 ALPHA 0.03 REY 0.01012⁶

INTEGRATED FORCE COEFFICIENTS

C_x = 2.2998 C_m = -0.0007

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 1.120 | 2.993 | 0.2000 | 1.120 | 0.993 |
| 0.0089 | 0.296 | 2.774 | 0.2095 | 0.401 | 2.821 |
| 0.0206 | -0.296 | 2.683 | 0.2205 | -0.043 | 2.634 |
| 0.0336 | -0.232 | 2.633 | 0.2336 | -0.079 | 2.673 |
| 0.0504 | -0.451 | 2.574 | 0.2506 | -0.233 | 2.632 |
| 0.0752 | -0.625 | 2.574 | 0.2754 | -0.336 | 2.625 |
| 0.0999 | -0.545 | 2.469 | 0.3001 | -0.422 | 2.552 |
| 0.1252 | -0.674 | 2.515 | 0.3248 | -0.501 | 2.561 |
| 0.1599 | -0.569 | 2.543 | 0.3496 | -0.450 | 2.574 |
| 0.2000 | -0.582 | 2.539 | 0.3743 | -0.426 | 2.581 |
| 0.2247 | -0.603 | 2.535 | 0.4000 | -0.285 | 2.619 |
| 0.2498 | -0.621 | 2.529 | 0.4247 | -0.131 | 2.659 |
| 0.2752 | -0.634 | 2.525 | 0.4499 | 0.042 | 2.706 |
| 0.3000 | -0.605 | 2.533 | 0.4749 | 0.193 | 2.746 |
| 0.3247 | -0.592 | 2.537 | 0.5000 | 0.258 | 2.763 |
| 0.3498 | -0.594 | 2.536 | | | |
| 0.3743 | -0.589 | 2.537 | | | |
| 0.4000 | -0.587 | 2.538 | | | |
| 0.4247 | -0.592 | 2.537 | | | |
| 0.4498 | -0.594 | 2.536 | | | |
| 0.4743 | -0.576 | 2.541 | | | |
| 0.5000 | -0.552 | 2.539 | | | |
| 0.5247 | -0.577 | 2.541 | | | |
| 0.5500 | -0.571 | 2.542 | | | |
| 0.5743 | -0.555 | 2.546 | | | |
| 0.6000 | -0.518 | 2.556 | | | |
| 0.6247 | -0.477 | 2.567 | | | |
| 0.6500 | -0.434 | 2.579 | | | |
| 0.6743 | -0.367 | 2.597 | | | |
| 0.7000 | -0.315 | 2.618 | | | |
| 0.7247 | -0.201 | 2.625 | | | |
| 0.7500 | -0.204 | 2.643 | | | |
| 0.7743 | -0.198 | 2.655 | | | |
| 0.8000 | -0.209 | 2.668 | | | |

95K 1 121.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.736 ALPHA 1.00 REY 2.51e6

INTEGRATED FORCE COEFFICIENTS

C_L = 0.4627 C_M = -0.2654

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 1.119 | 2.993 | 0.0000 | 1.119 | 2.993 |
| 0.0050 | 0.360 | 2.712 | 0.0050 | 0.598 | 2.855 |
| 0.0100 | -0.251 | 2.626 | 0.0100 | 0.102 | 2.744 |
| 0.0150 | -0.413 | 2.586 | 0.0150 | 0.115 | 2.725 |
| 0.0200 | -0.614 | 2.532 | 0.0200 | -0.049 | 2.682 |
| 0.0250 | -0.976 | 2.436 | 0.0250 | -0.153 | 2.654 |
| 0.0300 | -1.114 | 2.400 | 0.0300 | -0.253 | 2.628 |
| 0.0350 | -1.133 | 2.394 | 0.0350 | -0.347 | 2.597 |
| 0.0400 | -1.033 | 2.421 | 0.0400 | -0.343 | 2.624 |
| 0.0450 | -1.005 | 2.428 | 0.0450 | -0.353 | 2.632 |
| 0.0500 | -0.938 | 2.473 | 0.0500 | -0.236 | 2.632 |
| 0.0550 | -0.823 | 2.523 | 0.0550 | -0.132 | 2.668 |
| 0.0600 | -0.674 | 2.517 | 0.0600 | 0.291 | 2.711 |
| 0.0650 | -0.573 | 2.518 | 0.0650 | 0.213 | 2.752 |
| 0.0700 | -0.424 | 2.522 | 0.0700 | 0.275 | 2.768 |
| 0.0750 | -0.293 | 2.522 | | | |
| 0.0800 | -0.168 | 2.524 | | | |
| 0.0850 | -0.044 | 2.524 | | | |
| 0.0900 | -0.044 | 2.524 | | | |
| 0.0950 | -0.044 | 2.525 | | | |
| 0.1000 | -0.017 | 2.532 | | | |
| 0.1050 | -0.018 | 2.531 | | | |
| 0.1100 | -0.008 | 2.534 | | | |
| 0.1150 | -0.006 | 2.537 | | | |
| 0.1200 | -0.005 | 2.543 | | | |
| 0.1250 | -0.003 | 2.555 | | | |
| 0.1300 | -0.005 | 2.567 | | | |
| 0.1350 | -0.009 | 2.579 | | | |
| 0.1400 | -0.010 | 2.597 | | | |
| 0.1450 | -0.013 | 2.612 | | | |
| 0.1500 | -0.029 | 2.626 | | | |
| 0.1550 | -0.202 | 2.642 | | | |
| 0.1600 | -0.151 | 2.655 | | | |
| 0.1650 | -0.101 | 2.669 | | | |

BOK 1 121.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.759 ALPHA 2.02 REY 0.01012⁶

INTEGRATED FORCE COEFFICIENTS

C_x = 0.6425 C_m = -0.2827

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 1.000 | 2.936 | 0.0000 | 1.000 | 0.986 |
| 0.0089 | -0.104 | 2.652 | 0.0089 | 0.753 | 0.696 |
| 0.0236 | -0.443 | 2.577 | 0.0236 | 0.370 | 0.794 |
| 0.0386 | -0.504 | 2.545 | 0.0386 | 0.285 | 0.771 |
| 0.0524 | -0.750 | 2.493 | 0.0524 | 0.113 | 0.725 |
| 0.0752 | -1.077 | 2.429 | 0.0752 | 0.001 | 0.695 |
| 0.0999 | -1.214 | 2.372 | 0.1001 | -0.109 | 0.666 |
| 0.1252 | -1.209 | 2.357 | 0.1252 | -0.252 | 0.628 |
| 0.1599 | -1.227 | 2.369 | 0.1599 | -0.243 | 0.632 |
| 0.2000 | -1.217 | 2.371 | 0.2000 | -0.274 | 0.622 |
| 0.2437 | -1.100 | 2.331 | 0.2437 | -0.197 | 0.645 |
| 0.2796 | -1.108 | 2.304 | 0.2796 | -0.071 | 0.676 |
| 0.3252 | -1.104 | 2.360 | 0.3252 | 0.001 | 0.717 |
| 0.3603 | -1.105 | 2.379 | 0.3603 | 0.237 | 0.756 |
| 0.4001 | -1.179 | 2.381 | 0.4001 | 0.299 | 0.772 |
| 0.4198 | -1.125 | 2.307 | | | |
| 0.4402 | -0.926 | 2.440 | | | |
| 0.4599 | -0.700 | 2.539 | | | |
| 0.4796 | -0.627 | 2.528 | | | |
| 0.4996 | -0.573 | 2.542 | | | |
| 0.5175 | -0.522 | 2.556 | | | |
| 0.5302 | -0.513 | 2.559 | | | |
| 0.5483 | -0.506 | 2.561 | | | |
| 0.5624 | -0.498 | 2.562 | | | |
| 0.5723 | -0.404 | 2.566 | | | |
| 0.5855 | -0.402 | 2.575 | | | |
| 0.5954 | -0.421 | 2.583 | | | |
| 0.6085 | -0.302 | 2.593 | | | |
| 0.6255 | -0.327 | 2.608 | | | |
| 0.6384 | -0.200 | 2.620 | | | |
| 0.6553 | -0.202 | 2.633 | | | |
| 0.6705 | -0.101 | 2.647 | | | |
| 0.6855 | -0.137 | 2.653 | | | |
| 0.6953 | -0.206 | 2.669 | | | |

BSK 1 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.721 ALPHA -2.52 REV 2.81012⁶

INTEGRATED FORCE COEFFICIENTS

C_x = -0.1624 C_y = -0.0841

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.2280 | 1.346 | 2.971 | 0.3380 | 1.248 | 2.971 |
| 0.2389 | 3.767 | 2.697 | 0.3396 | -0.145 | 2.841 |
| 0.2426 | 0.438 | 2.922 | 0.3425 | -0.637 | 2.443 |
| 0.2326 | 0.221 | 2.751 | 0.3386 | -0.626 | 2.528 |
| 0.2524 | 0.334 | 2.691 | 0.3536 | -0.745 | 2.475 |
| 0.2752 | -0.174 | 2.633 | 0.3754 | -0.954 | 2.417 |
| 0.2999 | -0.237 | 2.615 | 0.3881 | -0.976 | 2.411 |
| 0.3252 | -0.223 | 2.621 | 0.3982 | -1.112 | 2.373 |
| 0.3599 | -0.212 | 2.623 | 0.4246 | -1.119 | 2.371 |
| 0.3850 | -0.255 | 2.611 | 0.4425 | -1.035 | 2.433 |
| 0.4207 | -0.295 | 2.597 | 0.4436 | -0.359 | 2.579 |
| 0.4798 | -0.336 | 2.585 | 0.5493 | -0.131 | 2.631 |
| 0.5282 | -0.307 | 2.587 | 0.6499 | -0.236 | 2.671 |
| 0.5633 | -0.377 | 2.577 | 0.7499 | 0.231 | 2.703 |
| 0.6021 | -0.393 | 2.574 | 0.8500 | 0.154 | 2.724 |
| 0.6199 | -0.404 | 2.573 | | | |
| 0.6432 | -0.411 | 2.568 | | | |
| 0.6599 | -0.423 | 2.564 | | | |
| 0.6796 | -0.436 | 2.561 | | | |
| 0.6996 | -0.450 | 2.556 | | | |
| 0.7435 | -0.455 | 2.555 | | | |
| 0.7622 | -0.477 | 2.549 | | | |
| 0.7583 | -0.465 | 2.547 | | | |
| 0.6824 | -0.446 | 2.544 | | | |
| 0.6233 | -0.493 | 2.545 | | | |
| 0.6585 | -0.473 | 2.553 | | | |
| 0.6754 | -0.442 | 2.558 | | | |
| 0.7285 | -0.414 | 2.566 | | | |
| 0.7255 | -0.348 | 2.584 | | | |
| 0.7524 | -0.381 | 2.593 | | | |
| 0.7753 | -0.255 | 2.612 | | | |
| 0.8085 | -0.281 | 2.625 | | | |
| 0.8255 | -0.146 | 2.643 | | | |
| 0.8583 | -0.394 | 2.655 | | | |

BGM 1 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.702 ALPHA -1.00 REYNOLDS 2.81×10^6

INTEGRATED FORCE COEFFICIENTS
 $C_L = 0.1241$ $C_M = -0.0250$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.114 | 2.989 | 0.0000 | 1.114 | 2.989 |
| 0.0089 | 0.539 | 2.832 | 0.0096 | 0.177 | 2.732 |
| 0.0206 | 0.177 | 2.732 | 0.0225 | -0.302 | 2.577 |
| 0.0326 | -0.304 | 2.682 | 0.0306 | -0.302 | 2.596 |
| 0.0504 | -0.292 | 2.617 | 0.0526 | -0.445 | 2.578 |
| 0.0752 | -0.401 | 2.548 | 0.0754 | -0.619 | 2.519 |
| 0.0999 | -0.537 | 2.532 | 0.1021 | -0.626 | 2.507 |
| 0.1252 | -0.408 | 2.551 | 0.1200 | -0.815 | 2.455 |
| 0.1509 | -0.418 | 2.565 | 0.1496 | -0.732 | 2.478 |
| 0.1763 | -0.440 | 2.557 | 0.1795 | -0.534 | 2.513 |
| 0.2007 | -0.476 | 2.548 | 0.2086 | -0.342 | 2.577 |
| 0.2294 | -0.512 | 2.539 | 0.2393 | -0.158 | 2.637 |
| 0.2522 | -0.537 | 2.532 | 0.2699 | 0.025 | 2.647 |
| 0.2773 | -0.528 | 2.534 | 0.2999 | 0.171 | 2.728 |
| 0.3031 | -0.524 | 2.536 | 0.3270 | 0.237 | 2.746 |
| 0.3298 | -0.535 | 2.533 | | | |
| 0.3532 | -0.539 | 2.531 | | | |
| 0.3797 | -0.546 | 2.530 | | | |
| 0.4046 | -0.556 | 2.527 | | | |
| 0.4296 | -0.565 | 2.524 | | | |
| 0.4545 | -0.566 | 2.527 | | | |
| 0.4792 | -0.570 | 2.523 | | | |
| 0.5038 | -0.573 | 2.522 | | | |
| 0.5284 | -0.575 | 2.522 | | | |
| 0.5523 | -0.562 | 2.525 | | | |
| 0.5765 | -0.521 | 2.537 | | | |
| 0.6004 | -0.476 | 2.549 | | | |
| 0.6245 | -0.434 | 2.561 | | | |
| 0.6485 | -0.365 | 2.580 | | | |
| 0.6724 | -0.313 | 2.595 | | | |
| 0.6963 | -0.258 | 2.604 | | | |
| 0.7205 | -0.199 | 2.626 | | | |
| 0.7445 | -0.145 | 2.641 | | | |
| 0.7685 | -0.093 | 2.655 | | | |

BSK 1 131.5 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.750 / ALPHA 0.20 REYN 3.81×10^6

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.3301$ $C_M = -0.0912$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.127 | 2.993 | 0.0000 | 1.127 | 2.993 |
| 0.0089 | 0.304 | 2.774 | 0.0096 | 0.414 | 2.793 |
| 0.0226 | -0.313 | 2.677 | 0.0205 | -0.239 | 2.671 |
| 0.0306 | -0.190 | 2.629 | 0.0306 | -0.278 | 2.650 |
| 0.0504 | -0.410 | 2.568 | 0.0506 | -0.238 | 2.616 |
| 0.0752 | -0.508 | 2.453 | 0.0754 | -0.349 | 2.585 |
| 0.0999 | -0.524 | 2.454 | 0.1001 | -0.441 | 2.500 |
| 0.1252 | -0.525 | 2.453 | 0.1200 | -0.555 | 2.528 |
| 0.1599 | -0.524 | 2.529 | 0.1496 | -0.490 | 2.546 |
| 0.1800 | -0.504 | 2.520 | 0.1695 | -0.463 | 2.524 |
| 0.2007 | -0.512 | 2.512 | 0.1886 | -0.371 | 2.549 |
| 0.2298 | -0.506 | 2.503 | 0.2093 | -0.133 | 2.643 |
| 0.2502 | -0.500 | 2.489 | 0.2299 | 0.039 | 2.642 |
| 0.2603 | -0.503 | 2.490 | 0.2499 | 0.191 | 2.734 |
| 0.2801 | -0.502 | 2.587 | 0.2600 | 0.257 | 2.753 |
| 0.2999 | -0.525 | 2.589 | | | |
| 0.3102 | -0.520 | 2.513 | | | |
| 0.3299 | -0.524 | 2.529 | | | |
| 0.3496 | -0.543 | 2.537 | | | |
| 0.3695 | -0.539 | 2.505 | | | |
| 0.3805 | -0.514 | 2.512 | | | |
| 0.3902 | -0.520 | 2.517 | | | |
| 0.4003 | -0.512 | 2.512 | | | |
| 0.4104 | -0.503 | 2.515 | | | |
| 0.4203 | -0.500 | 2.521 | | | |
| 0.4305 | -0.500 | 2.535 | | | |
| 0.4454 | -0.451 | 2.549 | | | |
| 0.4505 | -0.433 | 2.562 | | | |
| 0.4605 | -0.363 | 2.581 | | | |
| 0.4704 | -0.306 | 2.597 | | | |
| 0.4793 | -0.223 | 2.612 | | | |
| 0.4805 | -0.194 | 2.628 | | | |
| 0.4905 | -0.143 | 2.643 | | | |
| 0.4903 | -0.092 | 2.656 | | | |

BGK 1 181.6 MM C-ORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.751 ALPHA 1.00 REV 0.01-10⁶

INTEGRATED FORCE COEFFICIENTS

C_L = 0.4864 C_M = -0.0888

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.129 | 2.993 | 0.2325 | 1.129 | 2.993 |
| 0.0099 | 0.101 | 2.717 | 0.2336 | 0.599 | 2.844 |
| 0.0206 | -0.109 | 2.629 | 0.2345 | 0.173 | 2.729 |
| 0.0336 | -0.344 | 2.566 | 0.2366 | 0.127 | 2.711 |
| 0.0534 | -0.545 | 2.531 | 0.2526 | -0.859 | 2.665 |
| 0.0752 | -0.898 | 2.433 | 0.2754 | -0.171 | 2.614 |
| 0.0999 | -1.306 | 2.395 | 0.2821 | -0.272 | 2.616 |
| 0.1252 | -1.705 | 2.307 | 0.2882 | -0.393 | 2.572 |
| 0.1599 | -2.079 | 2.411 | 0.2496 | -0.370 | 2.579 |
| 0.2000 | -2.367 | 2.414 | 0.3405 | -0.376 | 2.577 |
| 0.2427 | -2.557 | 2.417 | 0.4436 | -0.251 | 2.612 |
| 0.2798 | -2.660 | 2.416 | 0.5493 | -0.126 | 2.651 |
| 0.3222 | -2.689 | 2.414 | 0.6499 | 0.060 | 2.693 |
| 0.3653 | -2.672 | 2.413 | 0.7490 | 0.213 | 2.742 |
| 0.4001 | -2.653 | 2.418 | 0.8500 | 0.276 | 2.757 |
| 0.4198 | -2.644 | 2.421 | | | |
| 0.4432 | -2.605 | 2.431 | | | |
| 0.4599 | -2.563 | 2.465 | | | |
| 0.4796 | -2.528 | 2.499 | | | |
| 0.4996 | -2.507 | 2.517 | | | |
| 0.5475 | -2.501 | 2.529 | | | |
| 0.5632 | -2.506 | 2.526 | | | |
| 0.5803 | -2.503 | 2.529 | | | |
| 0.6004 | -2.500 | 2.529 | | | |
| 0.6203 | -2.504 | 2.534 | | | |
| 0.6535 | -2.500 | 2.543 | | | |
| 0.6754 | -2.427 | 2.555 | | | |
| 0.7025 | -2.410 | 2.566 | | | |
| 0.7255 | -2.346 | 2.586 | | | |
| 0.7524 | -2.292 | 2.601 | | | |
| 0.7753 | -2.209 | 2.615 | | | |
| 0.8035 | -2.103 | 2.631 | | | |
| 0.8255 | -2.133 | 2.645 | | | |
| 0.8503 | -2.007 | 2.657 | | | |

BSK 1 121.6 MM CHORD
 EXPERIMENTAL PRESSURE DISTRIBUTION
 SOLID WALLS

MACH NO. 0.728 ALPHA 2.80 REY 3.81+13⁶

INTEGRATED FORCE COEFFICIENTS

C_x = 2.6132 C_y = -2.3857

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | C _p | P/H | X/C | C _p | P/H |
|--------|----------------|-------|--------|----------------|-------|
| 0.0000 | 1.112 | 2.989 | 0.0000 | 1.112 | 2.989 |
| 0.0049 | -0.349 | 2.673 | 0.0049 | 0.721 | 2.851 |
| 0.0206 | -0.340 | 2.593 | 0.0206 | 0.338 | 2.774 |
| 0.0305 | -0.472 | 2.553 | 0.0306 | 0.248 | 2.751 |
| 0.0504 | -0.670 | 2.499 | 0.0506 | 0.076 | 2.734 |
| 0.0752 | -0.946 | 2.412 | 0.0754 | -0.036 | 2.673 |
| 0.0949 | -1.127 | 2.373 | 0.1001 | -0.148 | 2.642 |
| 0.1252 | -1.172 | 2.353 | 0.2302 | -0.328 | 2.621 |
| 0.1599 | -1.142 | 2.369 | 0.2496 | -0.293 | 2.653 |
| 0.2004 | -1.134 | 2.371 | 0.3495 | -0.322 | 2.595 |
| 0.2407 | -1.104 | 2.379 | 0.4406 | -0.225 | 2.621 |
| 0.2798 | -1.143 | 2.380 | 0.5493 | -0.181 | 2.655 |
| 0.3202 | -1.141 | 2.375 | 0.6495 | 0.060 | 2.738 |
| 0.3603 | -1.133 | 2.372 | 0.7496 | 0.211 | 2.741 |
| 0.4001 | -1.131 | 2.372 | 0.8498 | 0.271 | 2.758 |
| 0.4198 | -1.144 | 2.368 | | | |
| 0.4482 | -1.137 | 2.371 | | | |
| 0.4599 | -1.141 | 2.369 | | | |
| 0.4796 | -1.131 | 2.372 | | | |
| 0.4906 | -1.173 | 2.388 | | | |
| 0.5405 | -0.603 | 2.531 | | | |
| 0.5602 | -0.502 | 2.529 | | | |
| 0.5803 | -0.508 | 2.543 | | | |
| 0.6304 | -0.468 | 2.555 | | | |
| 0.6703 | -0.434 | 2.564 | | | |
| 0.6805 | -0.396 | 2.577 | | | |
| 0.6754 | -0.350 | 2.587 | | | |
| 0.7035 | -0.310 | 2.598 | | | |
| 0.7205 | -0.270 | 2.609 | | | |
| 0.7524 | -0.234 | 2.619 | | | |
| 0.7753 | -0.196 | 2.629 | | | |
| 0.8025 | -0.161 | 2.639 | | | |
| 0.8295 | -0.125 | 2.646 | | | |
| 0.8533 | -0.110 | 2.653 | | | |

82K 1 121.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.779 ALPHA -2.52 REYN 3.79×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = -0.1503$ $C_M = -0.3680$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.369 | 2.973 | 0.0000 | 1.269 | 2.973 |
| 0.0039 | 0.774 | 2.593 | 0.0039 | -0.068 | 2.648 |
| 0.0226 | 0.425 | 2.793 | 0.0226 | -0.539 | 2.499 |
| 0.0336 | 2.245 | 2.739 | 0.0336 | -0.543 | 2.513 |
| 0.0534 | 0.021 | 2.675 | 0.0534 | -0.657 | 2.477 |
| 0.0752 | -0.196 | 2.613 | 0.0752 | -0.875 | 2.413 |
| 0.0999 | -0.266 | 2.594 | 0.0999 | -0.903 | 2.419 |
| 0.1292 | -0.246 | 2.599 | 0.1292 | -1.051 | 2.307 |
| 0.1599 | -0.235 | 2.602 | 0.1599 | -1.204 | 2.303 |
| 0.1900 | -0.260 | 2.589 | 0.1900 | -0.941 | 2.375 |
| 0.2207 | -0.321 | 2.577 | 0.2207 | -0.354 | 2.504 |
| 0.2508 | -0.355 | 2.565 | 0.2508 | -0.266 | 2.592 |
| 0.2822 | -0.403 | 2.554 | 0.2822 | -0.165 | 2.621 |
| 0.3133 | -0.415 | 2.550 | 0.3133 | -0.059 | 2.601 |
| 0.3431 | -0.425 | 2.543 | 0.3431 | 0.019 | 2.673 |
| 0.3719 | -0.443 | 2.542 | | | |
| 0.4022 | -0.457 | 2.539 | | | |
| 0.4329 | -0.474 | 2.534 | | | |
| 0.4636 | -0.495 | 2.525 | | | |
| 0.4936 | -0.513 | 2.522 | | | |
| 0.5235 | -0.523 | 2.519 | | | |
| 0.5532 | -0.526 | 2.520 | | | |
| 0.5833 | -0.574 | 2.525 | | | |
| 0.6134 | -0.593 | 2.499 | | | |
| 0.6433 | -0.594 | 2.499 | | | |
| 0.6735 | -0.584 | 2.527 | | | |
| 0.7034 | -0.587 | 2.524 | | | |
| 0.7335 | -0.429 | 2.537 | | | |
| 0.7635 | -0.382 | 2.559 | | | |
| 0.7934 | -0.338 | 2.574 | | | |
| 0.8233 | -0.270 | 2.588 | | | |
| 0.8535 | -0.223 | 2.634 | | | |
| 0.8835 | -0.168 | 2.623 | | | |
| 0.9133 | -0.116 | 2.635 | | | |

SSK 1 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

SACH NO. 0.793 ALPHA -1.00 REV 0.79017⁶

INTEGRATED FORCE COEFFICIENTS

C_D = 0.1046 C_M = -0.3952

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.2033 | 1.120 | 2.983 | 0.2038 | 1.128 | 2.9-8 |
| 0.2069 | 0.502 | 2.623 | 0.2096 | 0.233 | 2.725 |
| 0.2206 | 0.206 | 2.726 | 0.2225 | -0.209 | 2.598 |
| 0.2306 | 0.027 | 2.675 | 0.2336 | -0.271 | 2.549 |
| 0.2524 | -0.205 | 2.608 | 0.2536 | -0.415 | 2.548 |
| 0.2752 | -0.471 | 2.532 | 0.2754 | -0.547 | 2.452 |
| 0.2999 | -0.548 | 2.510 | 0.2921 | -0.596 | 2.447 |
| 0.3252 | -0.473 | 2.532 | 0.2932 | -0.832 | 2.429 |
| 0.3599 | -0.417 | 2.548 | 0.2496 | -0.881 | 2.415 |
| 0.2033 | -0.453 | 2.537 | 0.3495 | -0.814 | 2.414 |
| 0.2407 | -0.467 | 2.525 | 0.4486 | -0.318 | 2.576 |
| 0.2748 | -0.529 | 2.515 | 0.5493 | -0.147 | 2.625 |
| 0.3222 | -0.507 | 2.499 | 0.6499 | 0.026 | 2.675 |
| 0.3623 | -0.504 | 2.523 | 0.7499 | 0.193 | 2.715 |
| 0.4031 | -0.553 | 2.539 | 0.8530 | 0.235 | 2.735 |
| 0.4198 | -0.558 | 2.587 | | | |
| 0.4432 | -0.507 | 2.535 | | | |
| 0.4599 | -0.501 | 2.581 | | | |
| 0.4796 | -0.602 | 2.495 | | | |
| 0.4996 | -0.626 | 2.488 | | | |
| 0.5425 | -0.621 | 2.489 | | | |
| 0.5602 | -0.636 | 2.484 | | | |
| 0.5833 | -0.646 | 2.482 | | | |
| 0.6034 | -0.648 | 2.482 | | | |
| 0.6223 | -0.619 | 2.490 | | | |
| 0.6525 | -0.540 | 2.512 | | | |
| 0.6754 | -0.477 | 2.531 | | | |
| 0.7025 | -0.424 | 2.546 | | | |
| 0.7255 | -0.352 | 2.566 | | | |
| 0.7534 | -0.295 | 2.583 | | | |
| 0.7753 | -0.242 | 2.598 | | | |
| 0.8005 | -0.183 | 2.615 | | | |
| 0.8255 | -0.129 | 2.630 | | | |
| 0.8523 | -0.279 | 2.644 | | | |

SCM * 18.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.782 ALPHA 8.89 REY 0.72*10⁶

INTEGRATED FORCE COEFFICIENTS

CV = 0.2934 CM = -0.3945

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.2000 | 1.136 | 2.992 | 0.2000 | 1.136 | 2.992 |
| 0.2089 | 0.384 | 2.777 | 0.3096 | 0.401 | 2.712 |
| 0.2206 | 0.736 | 2.678 | 0.3205 | -0.041 | 2.556 |
| 0.2306 | -0.127 | 2.631 | 0.3306 | -0.261 | 2.645 |
| 0.2504 | -0.354 | 2.566 | 0.3526 | -0.242 | 2.598 |
| 0.2752 | -0.741 | 2.455 | 0.2754 | -0.364 | 2.564 |
| 0.2999 | -0.779 | 2.445 | 0.1881 | -0.456 | 2.538 |
| 0.1252 | -0.819 | 2.433 | 0.2000 | -0.642 | 2.485 |
| 0.1599 | -0.730 | 2.459 | 0.2496 | -0.630 | 2.497 |
| 0.2223 | -0.527 | 2.517 | 0.3495 | -0.516 | 2.523 |
| 0.2467 | -0.573 | 2.594 | 0.4486 | -0.319 | 2.577 |
| 0.2798 | -0.621 | 2.493 | 0.5493 | -0.145 | 2.626 |
| 0.3282 | -0.606 | 2.471 | 0.6499 | 0.035 | 2.672 |
| 0.3685 | -0.723 | 2.461 | 0.7499 | 0.186 | 2.702 |
| 0.4081 | -0.751 | 2.453 | 0.8500 | 0.253 | 2.742 |
| 0.4198 | -0.743 | 2.455 | | | |
| 0.4432 | -0.749 | 2.453 | | | |
| 0.4599 | -0.785 | 2.447 | | | |
| 0.4796 | -0.777 | 2.445 | | | |
| 0.4996 | -0.803 | 2.438 | | | |
| 0.5485 | -0.796 | 2.443 | | | |
| 0.5682 | -0.793 | 2.441 | | | |
| 0.5883 | -0.794 | 2.452 | | | |
| 0.6494 | -0.677 | 2.474 | | | |
| 0.6283 | -0.564 | 2.506 | | | |
| 0.6501 | -0.483 | 2.529 | | | |
| 0.6754 | -0.437 | 2.542 | | | |
| 0.7035 | -0.391 | 2.556 | | | |
| 0.7255 | -0.327 | 2.574 | | | |
| 0.7584 | -0.273 | 2.594 | | | |
| 0.7753 | -0.242 | 2.684 | | | |
| 0.8085 | -0.199 | 2.619 | | | |
| 0.8255 | -0.123 | 2.633 | | | |
| 0.8503 | -0.076 | 2.646 | | | |

BSK 1 121.6 MM C43RD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.778 ALPHA 1.22 REY 0.78028

INTEGRATED FORCE COEFFICIENTS

CN = 0.4684 CM = -0.3915

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0000 | 1.139 | 3.993 | 0.0000 | 1.138 | 3.993 |
| 0.0287 | 0.195 | 3.726 | 0.0286 | 0.567 | 3.831 |
| 0.0574 | -0.121 | 3.636 | 0.0573 | 0.144 | 3.711 |
| 0.0861 | -0.276 | 3.592 | 0.0860 | 0.033 | 3.653 |
| 0.1148 | -0.451 | 3.533 | 0.1147 | -0.082 | 3.646 |
| 0.1435 | -0.652 | 3.434 | 0.1434 | -0.197 | 3.613 |
| 0.1722 | -0.873 | 3.396 | 0.1721 | -0.332 | 3.554 |
| 0.2009 | -1.097 | 3.367 | 0.2008 | -0.459 | 3.519 |
| 0.2296 | -1.319 | 3.429 | 0.2295 | -0.423 | 3.549 |
| 0.2583 | -1.514 | 3.413 | 0.2582 | -0.434 | 3.546 |
| 0.2870 | -1.698 | 3.412 | 0.2869 | -0.285 | 3.589 |
| 0.3157 | -1.861 | 3.428 | 0.3156 | -0.131 | 3.632 |
| 0.3444 | -2.035 | 3.424 | 0.3443 | 0.044 | 3.682 |
| 0.3731 | -2.248 | 3.421 | 0.3730 | 0.198 | 3.726 |
| 0.4018 | -2.498 | 3.421 | 0.4017 | 0.263 | 3.744 |
| 0.4305 | -2.797 | 3.395 | | | |
| 0.4592 | -3.054 | 3.399 | | | |
| 0.4879 | -3.257 | 3.393 | | | |
| 0.5166 | -3.456 | 3.396 | | | |
| 0.5453 | -3.658 | 3.389 | | | |
| 0.5740 | -3.872 | 3.394 | | | |
| 0.6027 | -4.081 | 3.414 | | | |
| 0.6314 | -4.280 | 3.468 | | | |
| 0.6601 | -4.475 | 3.512 | | | |
| 0.6888 | -4.679 | 3.534 | | | |
| 0.7175 | -4.872 | 3.554 | | | |
| 0.7462 | -5.064 | 3.567 | | | |
| 0.7749 | -5.253 | 3.578 | | | |
| 0.8036 | -5.444 | 3.592 | | | |
| 0.8323 | -5.630 | 3.604 | | | |
| 0.8610 | -5.819 | 3.616 | | | |
| 0.8897 | -6.014 | 3.623 | | | |
| 0.9184 | -6.214 | 3.637 | | | |
| 0.9471 | -6.418 | 3.646 | | | |

95K 1 171.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.779 ALPHA 2.88 REY 0.79+13

INTEGRATED FORCE COEFFICIENTS

CV = 0.5375 CM = -0.0862

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0223 | 1.143 | 2.991 | 0.0030 | 1.130 | 0.991 |
| 0.0389 | 0.864 | 2.685 | 0.0096 | 0.677 | 2.063 |
| 0.0526 | -0.237 | 2.602 | 0.0205 | 0.276 | 2.749 |
| 0.0636 | -0.373 | 2.564 | 0.0326 | 0.221 | 2.729 |
| 0.0752 | -0.512 | 2.507 | 0.0536 | 0.027 | 2.676 |
| 0.0875 | -0.645 | 2.415 | 0.0754 | -0.092 | 2.645 |
| 0.0999 | -1.026 | 2.374 | 0.1001 | -0.200 | 2.614 |
| 0.1252 | -1.079 | 2.363 | 0.1200 | -0.367 | 2.567 |
| 0.1599 | -1.044 | 2.373 | 0.1496 | -0.354 | 2.573 |
| 0.2000 | -1.039 | 2.375 | 0.1495 | -0.393 | 2.559 |
| 0.2437 | -1.011 | 2.383 | 0.1486 | -0.276 | 2.593 |
| 0.2793 | -1.020 | 2.381 | 0.1493 | -0.139 | 2.632 |
| 0.3222 | -1.041 | 2.375 | 0.1499 | 0.030 | 2.679 |
| 0.3633 | -1.029 | 2.370 | 0.1499 | 0.196 | 2.724 |
| 0.4021 | -1.001 | 2.369 | 0.1500 | 0.244 | 2.740 |
| 0.4198 | -1.075 | 2.365 | | | |
| 0.4422 | -1.072 | 2.366 | | | |
| 0.4599 | -1.077 | 2.364 | | | |
| 0.4796 | -1.078 | 2.364 | | | |
| 0.4996 | -1.005 | 2.362 | | | |
| 0.5125 | -0.747 | 2.453 | | | |
| 0.5622 | -0.506 | 2.524 | | | |
| 0.5833 | -0.503 | 2.527 | | | |
| 0.6074 | -0.403 | 2.542 | | | |
| 0.6283 | -0.413 | 2.553 | | | |
| 0.6525 | -0.302 | 2.567 | | | |
| 0.6754 | -0.367 | 2.577 | | | |
| 0.7035 | -0.246 | 2.584 | | | |
| 0.7255 | -0.200 | 2.594 | | | |
| 0.7524 | -0.222 | 2.622 | | | |
| 0.7753 | -0.216 | 2.629 | | | |
| 0.8005 | -0.149 | 2.614 | | | |
| 0.8254 | -0.101 | 2.619 | | | |
| 0.8533 | -0.106 | 2.623 | | | |

BSK 1 181.6 MM CHORD
 EXPERIMENTAL PRESSURE DISTRIBUTION
 SOLID WALLS

MACH NO. 0.838 ALPHA -2.50 REV 8.78*10⁶

INTEGRATED FORCE COEFFICIENTS

C_N = -0.1088 C_M = -0.0687

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0388 | 1.392 | 3.976 | 0.0880 | 1.892 | 0.976 |
| 0.0389 | 0.753 | 3.877 | 0.0896 | 0.016 | 0.659 |
| 0.0226 | 0.406 | 3.775 | 0.0205 | -0.480 | 0.511 |
| 0.0386 | 2.222 | 3.721 | 0.0306 | -0.447 | 0.523 |
| 0.0584 | 3.803 | 3.656 | 0.0506 | -0.574 | 0.485 |
| 0.0752 | -3.227 | 3.589 | 0.0754 | -0.797 | 0.428 |
| 0.0999 | -2.301 | 3.567 | 0.1001 | -0.825 | 0.411 |
| 0.1252 | -3.275 | 3.575 | 0.2008 | -0.979 | 0.366 |
| 0.1599 | -3.257 | 3.588 | 0.2496 | -0.997 | 0.361 |
| 0.2038 | -2.306 | 3.565 | 0.3495 | -0.957 | 0.373 |
| 0.2407 | -0.351 | 3.552 | 0.4486 | -0.361 | 0.548 |
| 0.2798 | -3.403 | 3.537 | 0.5493 | -0.298 | 0.567 |
| 0.3202 | -3.455 | 3.522 | 0.6499 | -0.235 | 0.585 |
| 0.3603 | -0.405 | 3.518 | 0.7499 | -0.141 | 0.613 |
| 0.4001 | -0.406 | 3.510 | 0.8503 | -0.080 | 0.631 |
| 0.4198 | -2.453 | 3.513 | | | |
| 0.4482 | -3.447 | 3.509 | | | |
| 0.4599 | -1.516 | 3.504 | | | |
| 0.4796 | -2.543 | 3.496 | | | |
| 0.4996 | -3.573 | 3.486 | | | |
| 0.5405 | -3.587 | 3.482 | | | |
| 0.5602 | -2.619 | 3.473 | | | |
| 0.5803 | -3.648 | 3.464 | | | |
| 0.6204 | -3.643 | 3.452 | | | |
| 0.6283 | -3.728 | 3.443 | | | |
| 0.6535 | -2.776 | 3.426 | | | |
| 0.6754 | -3.804 | 3.419 | | | |
| 0.7035 | -0.822 | 3.413 | | | |
| 0.7255 | -3.585 | 3.403 | | | |
| 0.7534 | -0.377 | 3.393 | | | |
| 0.7753 | -3.245 | 3.386 | | | |
| 0.8035 | -3.234 | 3.386 | | | |
| 0.8255 | -3.141 | 3.381 | | | |
| 0.8503 | -2.133 | 3.615 | | | |

BGK 1 181.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.821 ALPHA -2.88 REY 2.79×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.0723$ $C_M = -0.3837$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.8388 | 1.138 | 0.988 | 0.0000 | 1.138 | 0.988 |
| 2.8089 | 2.584 | 0.827 | 0.0396 | 0.222 | 0.721 |
| 2.8286 | 0.292 | 0.723 | 0.0285 | -0.247 | 0.553 |
| 2.8336 | 0.348 | 0.459 | 0.0386 | -0.248 | 0.543 |
| 0.8584 | -3.171 | 0.685 | 0.0586 | -0.398 | 0.541 |
| 2.8752 | -1.446 | 0.524 | 0.0754 | -0.654 | 0.464 |
| 0.8999 | -0.543 | 0.495 | 0.1001 | -0.561 | 0.491 |
| 2.1252 | -3.401 | 0.519 | 0.2000 | -0.817 | 0.416 |
| 2.1599 | -0.402 | 0.537 | 0.2496 | -0.871 | 0.402 |
| 2.2208 | -3.438 | 0.527 | 0.3495 | -0.946 | 0.378 |
| 2.2437 | -0.474 | 0.516 | 0.4486 | -0.428 | 0.533 |
| 2.2798 | -0.517 | 0.583 | 0.5493 | -0.174 | 0.625 |
| 2.3282 | -0.577 | 0.486 | 0.6499 | -0.824 | 0.649 |
| 0.3635 | -2.611 | 0.476 | 0.7499 | 0.297 | 0.655 |
| 0.4001 | -0.623 | 0.472 | 0.8588 | 0.177 | 0.726 |
| 2.4195 | -2.632 | 0.473 | | | |
| 2.4432 | -0.634 | 0.469 | | | |
| 2.4599 | -0.542 | 0.467 | | | |
| 2.4796 | -0.651 | 0.464 | | | |
| 2.4996 | -0.674 | 0.458 | | | |
| 2.5125 | -0.673 | 0.458 | | | |
| 2.5632 | -0.687 | 0.454 | | | |
| 2.5833 | -0.700 | 0.453 | | | |
| 2.6234 | -2.726 | 0.442 | | | |
| 2.6273 | -2.792 | 0.438 | | | |
| 2.6525 | -2.646 | 0.466 | | | |
| 2.6754 | -3.469 | 0.518 | | | |
| 2.7025 | -2.355 | 0.543 | | | |
| 0.7255 | -0.316 | 0.563 | | | |
| 2.7534 | -2.201 | 0.579 | | | |
| 2.7793 | -0.289 | 0.595 | | | |
| 2.8025 | -0.195 | 0.618 | | | |
| 2.8225 | -0.145 | 0.625 | | | |
| 2.8533 | -0.308 | 0.638 | | | |

BCK 1 181.6 MM CHORD
 EXPERIMENTAL PRESSURE DISTRIBUTION
 SOLID WALLS

MACH NO. 0.779 ALPHA 0.88 REY 2,780,186

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.2447$ $C_M = -0.3951$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 2.3320 | 1.148 | 2.993 | 0.8000 | 1.148 | 2.993 |
| 2.3349 | 0.433 | 2.784 | 0.8396 | 0.356 | 2.759 |
| 2.3726 | 0.089 | 2.662 | 0.8285 | -0.268 | 2.638 |
| 2.3336 | -0.371 | 2.635 | 0.8326 | -0.893 | 2.629 |
| 2.3534 | -0.297 | 2.569 | 0.0526 | -0.253 | 2.562 |
| 2.3752 | -0.601 | 2.456 | 0.8754 | -0.397 | 2.539 |
| 2.3999 | -0.716 | 2.446 | 0.1001 | -0.464 | 2.523 |
| 2.1252 | -2.705 | 2.432 | 0.2308 | -0.691 | 2.453 |
| 2.1599 | -2.708 | 2.451 | 0.2496 | -0.747 | 2.436 |
| 2.2820 | -3.644 | 2.467 | 0.3495 | -0.829 | 2.418 |
| 2.2487 | -3.529 | 2.581 | 0.4466 | -0.316 | 2.553 |
| 2.2798 | -3.571 | 2.469 | 0.5493 | -0.146 | 2.613 |
| 2.3232 | -2.639 | 2.469 | 0.6499 | 0.829 | 2.554 |
| 2.3633 | -2.555 | 2.455 | 0.7499 | 0.174 | 2.7-7 |
| 2.4031 | -3.697 | 2.452 | 0.8500 | 0.238 | 2.726 |
| 2.4198 | -3.713 | 2.447 | | | |
| 2.4482 | -3.724 | 2.444 | | | |
| 2.4599 | -2.742 | 2.439 | | | |
| 2.4796 | -2.759 | 2.434 | | | |
| 2.4996 | -2.791 | 2.424 | | | |
| 2.5435 | -2.807 | 2.419 | | | |
| 2.5682 | -2.829 | 2.413 | | | |
| 2.5883 | -2.849 | 2.437 | | | |
| 2.6034 | -2.879 | 2.398 | | | |
| 2.6283 | -2.876 | 2.399 | | | |
| 2.6525 | -2.862 | 2.462 | | | |
| 2.6754 | -2.437 | 2.528 | | | |
| 2.7025 | -2.345 | 2.555 | | | |
| 2.7255 | -2.281 | 2.574 | | | |
| 2.7534 | -2.227 | 2.589 | | | |
| 2.7793 | -2.184 | 2.632 | | | |
| 2.8085 | -2.142 | 2.614 | | | |
| 2.8255 | -2.147 | 2.625 | | | |
| 2.8583 | -2.776 | 2.634 | | | |

BK 1 - 181.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.749 ALPHA 1.00 REY 0.79*10⁶

INTEGRATED FORCE COEFFICIENTS

C_L = 0.3734 C_M = -0.0094

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0000 | 1.146 | 0.993 | 0.0000 | 1.146 | 0.993 |
| 0.0000 | 2.304 | 0.745 | 0.0000 | 0.527 | 0.011 |
| 0.0000 | -0.024 | 0.649 | 0.0000 | 0.131 | 0.056 |
| 0.0000 | -0.172 | 0.685 | 0.0000 | 0.044 | 0.009 |
| 0.0000 | -0.307 | 0.542 | 0.0000 | -0.120 | 0.021 |
| 0.0000 | -0.744 | 0.437 | 0.0000 | -0.242 | 0.055 |
| 0.0000 | -0.865 | 0.422 | 0.0000 | -0.348 | 0.054 |
| 0.0000 | -0.891 | 0.394 | 0.0000 | -0.563 | 0.490 |
| 0.0000 | -0.824 | 0.414 | 0.0000 | -0.599 | 0.432 |
| 0.0000 | -0.824 | 0.414 | 0.0000 | -0.678 | 0.477 |
| 0.0000 | -0.922 | 0.414 | 0.0000 | -0.333 | 0.558 |
| 0.0000 | -0.836 | 0.413 | 0.0000 | -0.104 | 0.008 |
| 0.0000 | -0.853 | 0.425 | 0.0000 | 0.017 | 0.001 |
| 0.0000 | -0.674 | 0.399 | 0.0000 | 0.178 | 0.746 |
| 0.0000 | -0.879 | 0.398 | 0.0000 | 0.232 | 0.724 |
| 0.0000 | -0.891 | 0.394 | | | |
| 0.0000 | -0.892 | 0.394 | | | |
| 0.0000 | -0.894 | 0.392 | | | |
| 0.0000 | -0.909 | 0.389 | | | |
| 0.0000 | -0.935 | 0.381 | | | |
| 0.0000 | -0.942 | 0.379 | | | |
| 0.0000 | -0.951 | 0.377 | | | |
| 0.0000 | -0.943 | 0.394 | | | |
| 0.0000 | -0.954 | 0.455 | | | |
| 0.0000 | -0.957 | 0.507 | | | |
| 0.0000 | -0.952 | 0.544 | | | |
| 0.0000 | -0.923 | 0.563 | | | |
| 0.0000 | -0.926 | 0.572 | | | |
| 0.0000 | -0.949 | 0.563 | | | |
| 0.0000 | -0.919 | 0.592 | | | |
| 0.0000 | -0.912 | 0.600 | | | |
| 0.0000 | -0.910 | 0.585 | | | |
| 0.0000 | -0.913 | 0.612 | | | |
| 0.0000 | -0.917 | 0.616 | | | |

BSK 1 181.6 MM C4GRD
EXPERIMENTAL PRESSURE DISTRIBUTION
SOLID WALLS

MACH NO. 0.991 ALPHA 2.28 REY 8.79*10⁶

INTEGRATED FORCE COEFFICIENTS

CX = 0.4498 CY = -0.8039

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.2333 | 1.138 | 2.991 | 0.8888 | 1.138 | 0.991 |
| 0.2389 | 0.163 | 2.789 | 0.8896 | 0.435 | 0.844 |
| 0.2406 | -0.121 | 2.619 | 0.8905 | 0.238 | 0.725 |
| 0.2436 | -0.257 | 2.579 | 0.8906 | 0.158 | 0.714 |
| 0.2534 | -0.469 | 2.517 | 0.8906 | -0.012 | 0.654 |
| 0.2752 | -0.791 | 2.422 | 0.8754 | -0.132 | 0.628 |
| 0.2999 | -0.923 | 2.384 | 0.1381 | -0.243 | 0.598 |
| 0.3252 | -0.973 | 2.368 | 0.2388 | -0.443 | 0.529 |
| 0.3599 | -0.939 | 2.378 | 0.2496 | -0.423 | 0.535 |
| 0.3833 | -0.933 | 2.388 | 0.3495 | -0.455 | 0.517 |
| 0.4037 | -0.938 | 2.388 | 0.4481 | -0.327 | 0.563 |
| 0.4298 | -0.929 | 2.382 | 0.5493 | -0.174 | 0.628 |
| 0.4522 | -0.955 | 2.375 | 0.6499 | 0.035 | 0.668 |
| 0.4683 | -0.978 | 2.368 | 0.7499 | 0.161 | 0.716 |
| 0.4861 | -0.981 | 2.367 | 0.8588 | 0.218 | 0.723 |
| 0.4991 | -0.985 | 2.363 | | | |
| 0.4472 | -0.994 | 2.363 | | | |
| 0.4599 | -1.002 | 2.361 | | | |
| 0.4796 | -1.044 | 2.363 | | | |
| 0.4996 | -1.019 | 2.358 | | | |
| 0.5425 | -0.731 | 2.442 | | | |
| 0.5581 | -0.566 | 2.491 | | | |
| 0.5583 | -0.470 | 2.519 | | | |
| 0.6224 | -0.419 | 2.534 | | | |
| 0.6223 | -0.391 | 2.542 | | | |
| 0.6525 | -0.343 | 2.556 | | | |
| 0.6754 | -0.319 | 2.563 | | | |
| 0.7375 | -0.302 | 2.568 | | | |
| 0.7255 | -0.291 | 2.574 | | | |
| 0.7504 | -0.293 | 2.581 | | | |
| 0.7753 | -0.293 | 2.585 | | | |
| 0.8285 | -0.236 | 2.589 | | | |
| 0.8255 | -0.221 | 2.593 | | | |
| 0.8523 | -0.213 | 2.596 | | | |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.502 ALPHA 0.00 REY 1.67×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = -0.0008$ $C_M = 0.0004$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | 0.100 | 0.057 | 0.0119 | 0.100 | 0.057 |
| 0.0371 | -0.203 | 0.093 | 0.0371 | -0.250 | 0.094 |
| 0.0623 | -0.409 | 0.760 | 0.0623 | -0.497 | 0.760 |
| 0.0872 | -0.463 | 0.773 | 0.0872 | -0.459 | 0.774 |
| 0.1122 | -0.464 | 0.773 | 0.1122 | -0.461 | 0.774 |
| 0.1372 | -0.454 | 0.774 | 0.1372 | -0.454 | 0.775 |
| 0.1623 | -0.461 | 0.773 | 0.1623 | -0.465 | 0.773 |
| 0.1872 | -0.459 | 0.774 | 0.1872 | -0.461 | 0.774 |
| 0.2122 | -0.452 | 0.775 | 0.2122 | -0.457 | 0.774 |
| 0.2372 | -0.449 | 0.777 | 0.2372 | -0.440 | 0.777 |
| 0.2620 | -0.418 | 0.780 | 0.2620 | -0.421 | 0.768 |
| 0.2872 | -0.396 | 0.783 | 0.2872 | -0.397 | 0.763 |
| 0.3122 | -0.378 | 0.786 | 0.3122 | -0.383 | 0.765 |
| 0.3375 | -0.346 | 0.790 | 0.3375 | -0.347 | 0.791 |
| 0.3618 | -0.312 | 0.795 | 0.3618 | -0.318 | 0.795 |
| 0.3873 | -0.324 | 0.794 | 0.3873 | -0.324 | 0.794 |
| 0.4124 | -0.315 | 0.795 | 0.4124 | -0.311 | 0.796 |
| 0.4371 | -0.294 | 0.798 | 0.4371 | -0.289 | 0.799 |
| 0.4621 | -0.284 | 0.800 | 0.4621 | -0.283 | 0.800 |
| 0.4871 | -0.269 | 0.802 | 0.4871 | -0.267 | 0.803 |
| 0.5120 | -0.252 | 0.804 | 0.5120 | -0.253 | 0.805 |
| 0.5371 | -0.228 | 0.808 | 0.5371 | -0.231 | 0.808 |
| 0.5621 | -0.193 | 0.813 | 0.5621 | -0.197 | 0.813 |
| 0.5870 | -0.196 | 0.813 | 0.5870 | -0.198 | 0.813 |
| 0.6122 | -0.153 | 0.819 | 0.6122 | -0.156 | 0.819 |
| 0.6371 | -0.157 | 0.819 | 0.6371 | -0.150 | 0.819 |
| 0.6661 | -0.129 | 0.823 | 0.6661 | -0.129 | 0.823 |
| 0.6869 | -0.127 | 0.823 | 0.6869 | -0.129 | 0.823 |
| 0.7121 | -0.107 | 0.826 | 0.7121 | -0.110 | 0.826 |
| 0.7369 | -0.089 | 0.829 | 0.7369 | -0.092 | 0.828 |
| 0.7626 | -0.075 | 0.831 | 0.7626 | -0.075 | 0.831 |
| 0.7870 | -0.053 | 0.834 | 0.7870 | -0.054 | 0.834 |
| 0.8117 | -0.035 | 0.837 | 0.8117 | -0.036 | 0.837 |
| 0.8370 | -0.016 | 0.839 | 0.8370 | -0.018 | 0.839 |
| 0.8619 | 0.049 | 0.843 | 0.8619 | 0.046 | 0.843 |
| 0.8869 | 0.081 | 0.846 | 0.8869 | 0.079 | 0.846 |
| 0.9120 | 0.099 | 0.851 | 0.9120 | 0.058 | 0.851 |
| 0.9416 | 0.087 | 0.855 | 0.9416 | 0.046 | 0.855 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.500 ALPHA 1.00 REYNOLDS 1.66E10⁶

INTEGRATED FORCE COEFFICIENTS

C_L = 0.1154 C_D = 0.0017

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0119 | -0.175 | 0.817 | 0.0119 | 0.343 | 0.094 |
| 0.0371 | -0.403 | 0.772 | 0.0371 | -0.055 | 0.035 |
| 0.0623 | -0.692 | 0.741 | 0.0623 | -0.311 | 0.797 |
| 0.0872 | -0.628 | 0.752 | 0.0872 | -0.306 | 0.798 |
| 0.1122 | -0.599 | 0.755 | 0.1122 | -0.324 | 0.795 |
| 0.1372 | -0.578 | 0.758 | 0.1372 | -0.333 | 0.794 |
| 0.1623 | -0.574 | 0.758 | 0.1623 | -0.352 | 0.791 |
| 0.1872 | -0.563 | 0.768 | 0.1872 | -0.368 | 0.798 |
| 0.2122 | -0.558 | 0.762 | 0.2122 | -0.363 | 0.769 |
| 0.2372 | -0.527 | 0.765 | 0.2372 | -0.355 | 0.791 |
| 0.2620 | -0.498 | 0.769 | 0.2620 | -0.348 | 0.793 |
| 0.2872 | -0.471 | 0.773 | 0.2872 | -0.324 | 0.795 |
| 0.3122 | -0.458 | 0.774 | 0.3122 | -0.312 | 0.797 |
| 0.3375 | -0.411 | 0.782 | 0.3375 | -0.281 | 0.801 |
| 0.3618 | -0.374 | 0.783 | 0.3618 | -0.255 | 0.805 |
| 0.3873 | -0.388 | 0.787 | 0.3873 | -0.271 | 0.823 |
| 0.4124 | -0.365 | 0.789 | 0.4124 | -0.267 | 0.824 |
| 0.4371 | -0.336 | 0.793 | 0.4371 | -0.246 | 0.867 |
| 0.4621 | -0.323 | 0.795 | 0.4621 | -0.248 | 0.878 |
| 0.4871 | -0.305 | 0.797 | 0.4871 | -0.225 | 0.818 |
| 0.5120 | -0.289 | 0.800 | 0.5120 | -0.214 | 0.811 |
| 0.5371 | -0.264 | 0.803 | 0.5371 | -0.194 | 0.814 |
| 0.5621 | -0.227 | 0.809 | 0.5621 | -0.161 | 0.819 |
| 0.5870 | -0.225 | 0.809 | 0.5870 | -0.165 | 0.819 |
| 0.6122 | -0.183 | 0.815 | 0.6122 | -0.123 | 0.825 |
| 0.6371 | -0.188 | 0.816 | 0.6371 | -0.131 | 0.824 |
| 0.6651 | -0.149 | 0.820 | 0.6651 | -0.186 | 0.827 |
| 0.6969 | -0.145 | 0.821 | 0.6969 | -0.188 | 0.827 |
| 0.7121 | -0.124 | 0.824 | 0.7121 | -0.092 | 0.829 |
| 0.7369 | -0.104 | 0.827 | 0.7369 | -0.076 | 0.832 |
| 0.7620 | -0.086 | 0.829 | 0.7620 | -0.063 | 0.834 |
| 0.7870 | -0.062 | 0.833 | 0.7870 | -0.044 | 0.837 |
| 0.8117 | -0.043 | 0.834 | 0.8117 | -0.027 | 0.839 |
| 0.8370 | -0.022 | 0.839 | 0.8370 | -0.008 | 0.842 |
| 0.8619 | 0.004 | 0.843 | 0.8619 | 0.013 | 0.845 |
| 0.8869 | 0.029 | 0.847 | 0.8869 | 0.035 | 0.848 |
| 0.9120 | 0.057 | 0.851 | 0.9120 | 0.059 | 0.852 |
| 0.9416 | 0.086 | 0.855 | 0.9416 | 0.086 | 0.856 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 2.88 REY 1.66e12⁶

INTEGRATED FORCE COEFFICIENTS

C_L = 0.2384 C_D = 0.0032

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | -0.506 | 0.760 | 0.0119 | 0.547 | 0.924 |
| 0.0371 | -0.739 | 0.734 | 0.0371 | 0.125 | 0.861 |
| 0.0623 | -0.907 | 0.709 | 0.0623 | -0.143 | 0.822 |
| 0.0872 | -0.797 | 0.725 | 0.0872 | -0.164 | 0.819 |
| 0.1122 | -0.791 | 0.732 | 0.1122 | -0.201 | 0.813 |
| 0.1372 | -0.768 | 0.733 | 0.1372 | -0.210 | 0.811 |
| 0.1623 | -0.698 | 0.741 | 0.1623 | -0.240 | 0.806 |
| 0.1872 | -0.605 | 0.745 | 0.1872 | -0.262 | 0.804 |
| 0.2122 | -0.641 | 0.740 | 0.2122 | -0.274 | 0.803 |
| 0.2372 | -0.611 | 0.752 | 0.2372 | -0.271 | 0.803 |
| 0.2622 | -0.575 | 0.750 | 0.2622 | -0.264 | 0.804 |
| 0.2872 | -0.545 | 0.762 | 0.2872 | -0.253 | 0.806 |
| 0.3122 | -0.514 | 0.767 | 0.3122 | -0.246 | 0.807 |
| 0.3373 | -0.473 | 0.773 | 0.3373 | -0.223 | 0.810 |
| 0.3618 | -0.433 | 0.779 | 0.3618 | -0.199 | 0.814 |
| 0.3873 | -0.433 | 0.779 | 0.3873 | -0.217 | 0.811 |
| 0.4124 | -0.415 | 0.781 | 0.4124 | -0.215 | 0.811 |
| 0.4371 | -0.382 | 0.786 | 0.4371 | -0.196 | 0.814 |
| 0.4621 | -0.367 | 0.788 | 0.4621 | -0.195 | 0.814 |
| 0.4871 | -0.346 | 0.791 | 0.4871 | -0.184 | 0.816 |
| 0.5122 | -0.324 | 0.795 | 0.5122 | -0.176 | 0.817 |
| 0.5371 | -0.294 | 0.799 | 0.5371 | -0.159 | 0.820 |
| 0.5621 | -0.293 | 0.805 | 0.5621 | -0.130 | 0.824 |
| 0.5871 | -0.248 | 0.806 | 0.5871 | -0.135 | 0.823 |
| 0.6122 | -0.203 | 0.813 | 0.6122 | -0.099 | 0.828 |
| 0.6371 | -0.202 | 0.813 | 0.6371 | -0.105 | 0.827 |
| 0.6621 | -0.169 | 0.818 | 0.6621 | -0.082 | 0.831 |
| 0.6869 | -0.164 | 0.818 | 0.6869 | -0.085 | 0.830 |
| 0.7121 | -0.140 | 0.822 | 0.7121 | -0.073 | 0.832 |
| 0.7369 | -0.120 | 0.825 | 0.7369 | -0.058 | 0.834 |
| 0.7620 | -0.099 | 0.828 | 0.7620 | -0.047 | 0.836 |
| 0.7870 | -0.074 | 0.832 | 0.7870 | -0.031 | 0.839 |
| 0.8117 | -0.052 | 0.835 | 0.8117 | -0.017 | 0.841 |
| 0.8370 | -0.029 | 0.838 | 0.8370 | -0.001 | 0.843 |
| 0.8619 | -0.002 | 0.842 | 0.8619 | 0.019 | 0.846 |
| 0.8869 | 0.024 | 0.846 | 0.8869 | 0.038 | 0.849 |
| 0.9120 | 0.034 | 0.851 | 0.9120 | 0.060 | 0.852 |
| 0.9416 | 0.067 | 0.856 | 0.9416 | 0.085 | 0.856 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.530 ALPHA 3.00 REY 1.66e10

INTEGRATED FORCE COEFFICIENTS

CV = 0.3423 CM = 0.0052

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0119 | -0.860 | 0.714 | 0.0119 | 0.715 | 0.969 |
| 0.0371 | -1.300 | 0.694 | 0.0371 | 0.290 | 0.884 |
| 0.0623 | -1.123 | 0.677 | 0.0623 | 0.015 | 0.846 |
| 0.0872 | -0.966 | 0.700 | 0.0872 | -0.034 | 0.830 |
| 0.1122 | -0.899 | 0.710 | 0.1122 | -0.070 | 0.832 |
| 0.1372 | -0.837 | 0.719 | 0.1372 | -0.112 | 0.827 |
| 0.1623 | -0.807 | 0.723 | 0.1623 | -0.147 | 0.822 |
| 0.1872 | -0.772 | 0.720 | 0.1872 | -0.171 | 0.810 |
| 0.2122 | -0.740 | 0.733 | 0.2122 | -0.186 | 0.816 |
| 0.2372 | -0.701 | 0.739 | 0.2372 | -0.193 | 0.815 |
| 0.2620 | -0.654 | 0.746 | 0.2620 | -0.198 | 0.815 |
| 0.2872 | -0.614 | 0.752 | 0.2872 | -0.184 | 0.816 |
| 0.3122 | -0.579 | 0.757 | 0.3122 | -0.184 | 0.816 |
| 0.3375 | -0.533 | 0.764 | 0.3375 | -0.165 | 0.819 |
| 0.3610 | -0.490 | 0.770 | 0.3610 | -0.144 | 0.822 |
| 0.3873 | -0.406 | 0.771 | 0.3873 | -0.166 | 0.819 |
| 0.4124 | -0.459 | 0.775 | 0.4124 | -0.166 | 0.819 |
| 0.4371 | -0.423 | 0.780 | 0.4371 | -0.153 | 0.821 |
| 0.4621 | -0.402 | 0.783 | 0.4621 | -0.153 | 0.821 |
| 0.4871 | -0.370 | 0.787 | 0.4871 | -0.146 | 0.822 |
| 0.5120 | -0.335 | 0.790 | 0.5120 | -0.139 | 0.823 |
| 0.5371 | -0.327 | 0.794 | 0.5371 | -0.125 | 0.825 |
| 0.5621 | -0.285 | 0.800 | 0.5621 | -0.096 | 0.829 |
| 0.5870 | -0.278 | 0.802 | 0.5870 | -0.106 | 0.828 |
| 0.6122 | -0.232 | 0.800 | 0.6122 | -0.070 | 0.833 |
| 0.6371 | -0.224 | 0.809 | 0.6371 | -0.089 | 0.831 |
| 0.6661 | -0.192 | 0.814 | 0.6661 | -0.057 | 0.835 |
| 0.6869 | -0.164 | 0.815 | 0.6869 | -0.066 | 0.833 |
| 0.7121 | -0.157 | 0.819 | 0.7121 | -0.054 | 0.835 |
| 0.7369 | -0.153 | 0.823 | 0.7369 | -0.046 | 0.836 |
| 0.7620 | -0.111 | 0.826 | 0.7620 | -0.034 | 0.838 |
| 0.7870 | -0.002 | 0.830 | 0.7870 | -0.020 | 0.840 |
| 0.8117 | -0.359 | 0.834 | 0.8117 | -0.006 | 0.842 |
| 0.8370 | -0.033 | 0.830 | 0.8370 | 0.007 | 0.844 |
| 0.8619 | -0.003 | 0.842 | 0.8619 | 0.025 | 0.847 |
| 0.8869 | 0.326 | 0.846 | 0.8869 | 0.042 | 0.849 |
| 0.9120 | 0.856 | 0.851 | 0.9120 | 0.062 | 0.852 |
| 0.9416 | 0.809 | 0.856 | 0.9416 | 0.082 | 0.855 |

VACA-88:2 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.538 ALPHA 4.88 REY 1.67×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.4925$ $C_M = 0.0075$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | -1.242 | 0.052 | 0.0119 | 0.043 | 0.966 |
| 0.0371 | -1.517 | 0.048 | 0.0371 | 0.428 | 0.987 |
| 0.0623 | -1.378 | 0.048 | 0.0623 | 0.152 | 0.066 |
| 0.0872 | -1.155 | 0.072 | 0.0872 | 0.000 | 0.057 |
| 0.1122 | -1.092 | 0.067 | 0.1122 | 0.033 | 0.049 |
| 0.1372 | -0.960 | 0.703 | 0.1372 | -0.011 | 0.042 |
| 0.1623 | -0.921 | 0.707 | 0.1623 | -0.053 | 0.036 |
| 0.1872 | -0.874 | 0.714 | 0.1872 | -0.001 | 0.032 |
| 0.2122 | -0.827 | 0.723 | 0.2122 | -0.106 | 0.028 |
| 0.2372 | -0.779 | 0.728 | 0.2372 | -0.115 | 0.027 |
| 0.2623 | -0.727 | 0.735 | 0.2623 | -0.119 | 0.026 |
| 0.2872 | -0.681 | 0.740 | 0.2872 | -0.110 | 0.026 |
| 0.3122 | -0.641 | 0.740 | 0.3122 | -0.122 | 0.026 |
| 0.3375 | -0.589 | 0.756 | 0.3375 | -0.105 | 0.028 |
| 0.3618 | -0.543 | 0.763 | 0.3618 | -0.090 | 0.030 |
| 0.3873 | -0.530 | 0.764 | 0.3873 | -0.116 | 0.027 |
| 0.4124 | -0.502 | 0.763 | 0.4124 | -0.121 | 0.026 |
| 0.4371 | -0.482 | 0.774 | 0.4371 | -0.109 | 0.028 |
| 0.4621 | -0.439 | 0.778 | 0.4621 | -0.112 | 0.027 |
| 0.4871 | -0.413 | 0.782 | 0.4871 | -0.105 | 0.028 |
| 0.5120 | -0.383 | 0.786 | 0.5120 | -0.104 | 0.029 |
| 0.5371 | -0.353 | 0.793 | 0.5371 | -0.092 | 0.030 |
| 0.5621 | -0.312 | 0.797 | 0.5621 | -0.066 | 0.034 |
| 0.5870 | -0.300 | 0.798 | 0.5870 | -0.076 | 0.033 |
| 0.6122 | -0.251 | 0.806 | 0.6122 | -0.048 | 0.038 |
| 0.6371 | -0.243 | 0.807 | 0.6371 | -0.054 | 0.035 |
| 0.6601 | -0.210 | 0.812 | 0.6601 | -0.035 | 0.039 |
| 0.6869 | -0.200 | 0.813 | 0.6869 | -0.047 | 0.037 |
| 0.7121 | -0.172 | 0.817 | 0.7121 | -0.036 | 0.039 |
| 0.7369 | -0.145 | 0.821 | 0.7369 | -0.028 | 0.040 |
| 0.7623 | -0.119 | 0.825 | 0.7623 | -0.020 | 0.041 |
| 0.7870 | -0.097 | 0.833 | 0.7870 | -0.026 | 0.043 |
| 0.8117 | -0.081 | 0.834 | 0.8117 | 0.024 | 0.044 |
| 0.8373 | -0.034 | 0.838 | 0.8373 | 0.017 | 0.046 |
| 0.8619 | -0.005 | 0.842 | 0.8619 | 0.032 | 0.048 |
| 0.8869 | 0.023 | 0.846 | 0.8869 | 0.047 | 0.051 |
| 0.9120 | 0.053 | 0.850 | 0.9120 | 0.034 | 0.053 |
| 0.9416 | 0.086 | 0.855 | 0.9416 | 0.002 | 0.056 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.631 ALPHA 0.00 REYNOLDS 1.04E+10

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.6832$ $C_D = 0.0083$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | Cp | P/H | X/C | Cp | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | 0.132 | 0.609 | 0.0119 | 0.130 | 0.011 |
| 0.0371 | -0.269 | 0.729 | 0.0371 | -0.261 | 0.734 |
| 0.0623 | -0.530 | 0.676 | 0.0623 | -0.530 | 0.831 |
| 0.0872 | -0.507 | 0.602 | 0.0872 | -0.502 | 0.686 |
| 0.1122 | -0.507 | 0.602 | 0.1122 | -0.500 | 0.645 |
| 0.1372 | -0.501 | 0.603 | 0.1372 | -0.470 | 0.607 |
| 0.1623 | -0.513 | 0.601 | 0.1623 | -0.514 | 0.604 |
| 0.1872 | -0.516 | 0.600 | 0.1872 | -0.514 | 0.604 |
| 0.2122 | -0.500 | 0.602 | 0.2122 | -0.511 | 0.664 |
| 0.2372 | -0.469 | 0.606 | 0.2372 | -0.400 | 0.600 |
| 0.2620 | -0.466 | 0.600 | 0.2620 | -0.404 | 0.603 |
| 0.2872 | -0.441 | 0.605 | 0.2872 | -0.401 | 0.608 |
| 0.3122 | -0.422 | 0.609 | 0.3122 | -0.423 | 0.751 |
| 0.3375 | -0.300 | 0.706 | 0.3375 | -0.300 | 0.700 |
| 0.3610 | -0.304 | 0.712 | 0.3610 | -0.300 | 0.714 |
| 0.3873 | -0.305 | 0.710 | 0.3873 | -0.304 | 0.713 |
| 0.4124 | -0.349 | 0.713 | 0.4124 | -0.340 | 0.716 |
| 0.4371 | -0.324 | 0.710 | 0.4371 | -0.321 | 0.722 |
| 0.4621 | -0.311 | 0.721 | 0.4621 | -0.312 | 0.723 |
| 0.4871 | -0.293 | 0.725 | 0.4871 | -0.293 | 0.727 |
| 0.5120 | -0.277 | 0.720 | 0.5120 | -0.270 | 0.730 |
| 0.5371 | -0.251 | 0.733 | 0.5371 | -0.250 | 0.735 |
| 0.5621 | -0.215 | 0.740 | 0.5621 | -0.217 | 0.742 |
| 0.5870 | -0.212 | 0.741 | 0.5870 | -0.215 | 0.742 |
| 0.6122 | -0.170 | 0.749 | 0.6122 | -0.173 | 0.751 |
| 0.6371 | -0.173 | 0.740 | 0.6371 | -0.173 | 0.751 |
| 0.6661 | -0.140 | 0.755 | 0.6661 | -0.141 | 0.757 |
| 0.6869 | -0.130 | 0.755 | 0.6869 | -0.130 | 0.757 |
| 0.7121 | -0.110 | 0.759 | 0.7121 | -0.110 | 0.761 |
| 0.7369 | -0.090 | 0.763 | 0.7369 | -0.100 | 0.765 |
| 0.7620 | -0.079 | 0.767 | 0.7620 | -0.080 | 0.769 |
| 0.7873 | -0.096 | 0.772 | 0.7870 | -0.096 | 0.774 |
| 0.8117 | -0.095 | 0.776 | 0.8117 | -0.097 | 0.770 |
| 0.8370 | -0.015 | 0.780 | 0.8370 | -0.016 | 0.782 |
| 0.8619 | 0.111 | 0.785 | 0.8619 | 0.011 | 0.787 |
| 0.8869 | 0.035 | 0.790 | 0.8869 | 0.035 | 0.792 |
| 0.9120 | 0.005 | 0.794 | 0.9120 | 0.006 | 0.790 |
| 0.9416 | 0.003 | 0.001 | 0.9416 | 0.003 | 0.003 |

VACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.631 ALPHA. 1.00 REY 1.66-18

INTEGRATED FORCE COEFFICIENTS

CN = 0.1211 CM = 0.0029

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | -0.138 | 0.756 | 0.0119 | 0.365 | 0.056 |
| 0.0371 | -0.496 | 0.684 | 0.0371 | -0.055 | 0.773 |
| 0.0623 | -0.756 | 0.633 | 0.0623 | -0.336 | 0.710 |
| 0.0872 | -0.605 | 0.447 | 0.0872 | -0.337 | 0.717 |
| 0.1122 | -0.604 | 0.451 | 0.1122 | -0.357 | 0.714 |
| 0.1372 | -0.643 | 0.455 | 0.1372 | -0.368 | 0.711 |
| 0.1623 | -0.642 | 0.455 | 0.1623 | -0.392 | 0.727 |
| 0.1872 | -0.633 | 0.457 | 0.1872 | -0.404 | 0.724 |
| 0.2122 | -0.615 | 0.461 | 0.2122 | -0.406 | 0.784 |
| 0.2372 | -0.506 | 0.667 | 0.2372 | -0.395 | 0.726 |
| 0.2620 | -0.553 | 0.673 | 0.2620 | -0.379 | 0.789 |
| 0.2872 | -0.520 | 0.690 | 0.2872 | -0.363 | 0.712 |
| 0.3122 | -0.496 | 0.695 | 0.3122 | -0.351 | 0.715 |
| 0.3375 | -0.496 | 0.693 | 0.3375 | -0.323 | 0.720 |
| 0.3610 | -0.419 | 0.700 | 0.3610 | -0.295 | 0.726 |
| 0.3873 | -0.420 | 0.700 | 0.3873 | -0.309 | 0.723 |
| 0.4124 | -0.399 | 0.704 | 0.4124 | -0.298 | 0.725 |
| 0.4371 | -0.378 | 0.710 | 0.4371 | -0.277 | 0.729 |
| 0.4621 | -0.355 | 0.713 | 0.4621 | -0.269 | 0.731 |
| 0.4871 | -0.353 | 0.717 | 0.4871 | -0.251 | 0.714 |
| 0.5120 | -0.312 | 0.722 | 0.5120 | -0.239 | 0.737 |
| 0.5371 | -0.283 | 0.727 | 0.5371 | -0.210 | 0.741 |
| 0.5621 | -0.245 | 0.735 | 0.5621 | -0.186 | 0.747 |
| 0.5870 | -0.249 | 0.736 | 0.5870 | -0.187 | 0.747 |
| 0.6122 | -0.194 | 0.745 | 0.6122 | -0.144 | 0.756 |
| 0.6371 | -0.194 | 0.745 | 0.6371 | -0.149 | 0.755 |
| 0.6661 | -0.159 | 0.752 | 0.6661 | -0.119 | 0.751 |
| 0.6869 | -0.155 | 0.753 | 0.6869 | -0.119 | 0.760 |
| 0.7121 | -0.153 | 0.757 | 0.7121 | -0.100 | 0.764 |
| 0.7369 | -0.111 | 0.761 | 0.7369 | -0.062 | 0.760 |
| 0.7620 | -0.091 | 0.765 | 0.7620 | -0.065 | 0.771 |
| 0.7870 | -0.063 | 0.771 | 0.7870 | -0.044 | 0.775 |
| 0.8117 | -0.040 | 0.775 | 0.8117 | -0.027 | 0.779 |
| 0.8370 | -0.010 | 0.780 | 0.8370 | -0.000 | 0.762 |
| 0.8619 | 0.010 | 0.785 | 0.8619 | 0.016 | 0.767 |
| 0.8869 | 0.037 | 0.791 | 0.8869 | 0.039 | 0.792 |
| 0.9120 | 0.067 | 0.797 | 0.9120 | 0.065 | 0.797 |
| 0.9416 | 0.095 | 0.802 | 0.9416 | 0.090 | 0.802 |

NACA-0012 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

NACN NO. 0.638 ALPHA 2.00 REY 1.66*10⁶

INTEGRATED FORCE COEFFICIENTS

C_N = 0.2471 C_M = 0.8849

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | -0.408 | 0.693 | 0.0119 | 0.569 | 0.896 |
| 0.0371 | -0.758 | 0.632 | 0.0371 | 0.133 | 0.889 |
| 0.0623 | -1.313 | 0.563 | 0.0623 | -0.156 | 0.751 |
| 0.0872 | -0.808 | 0.608 | 0.0872 | -0.102 | 0.746 |
| 0.1122 | -0.836 | 0.619 | 0.1122 | -0.222 | 0.738 |
| 0.1372 | -0.792 | 0.627 | 0.1372 | -0.245 | 0.733 |
| 0.1623 | -0.768 | 0.630 | 0.1623 | -0.270 | 0.727 |
| 0.1872 | -0.752 | 0.634 | 0.1872 | -0.297 | 0.723 |
| 0.2122 | -0.726 | 0.640 | 0.2122 | -0.310 | 0.720 |
| 0.2372 | -0.685 | 0.648 | 0.2372 | -0.307 | 0.721 |
| 0.2622 | -0.645 | 0.656 | 0.2622 | -0.299 | 0.723 |
| 0.2872 | -0.607 | 0.664 | 0.2872 | -0.287 | 0.725 |
| 0.3122 | -0.573 | 0.671 | 0.3122 | -0.281 | 0.726 |
| 0.3375 | -0.528 | 0.680 | 0.3375 | -0.257 | 0.731 |
| 0.3618 | -0.484 | 0.688 | 0.3618 | -0.231 | 0.736 |
| 0.3873 | -0.440 | 0.689 | 0.3873 | -0.246 | 0.733 |
| 0.4124 | -0.405 | 0.694 | 0.4124 | -0.242 | 0.734 |
| 0.4371 | -0.423 | 0.703 | 0.4371 | -0.222 | 0.738 |
| 0.4621 | -0.402 | 0.704 | 0.4621 | -0.220 | 0.738 |
| 0.4871 | -0.375 | 0.710 | 0.4871 | -0.207 | 0.741 |
| 0.5120 | -0.351 | 0.714 | 0.5120 | -0.199 | 0.743 |
| 0.5371 | -0.328 | 0.720 | 0.5371 | -0.179 | 0.747 |
| 0.5621 | -0.268 | 0.720 | 0.5621 | -0.149 | 0.753 |
| 0.5870 | -0.272 | 0.730 | 0.5870 | -0.152 | 0.752 |
| 0.6122 | -0.224 | 0.739 | 0.6122 | -0.116 | 0.759 |
| 0.6371 | -0.218 | 0.741 | 0.6371 | -0.122 | 0.758 |
| 0.6611 | -0.181 | 0.748 | 0.6611 | -0.095 | 0.763 |
| 0.6869 | -0.175 | 0.749 | 0.6869 | -0.098 | 0.763 |
| 0.7121 | -0.150 | 0.754 | 0.7121 | -0.083 | 0.766 |
| 0.7369 | -0.124 | 0.759 | 0.7369 | -0.067 | 0.769 |
| 0.7620 | -0.101 | 0.764 | 0.7620 | -0.054 | 0.771 |
| 0.7870 | -0.073 | 0.769 | 0.7870 | -0.036 | 0.775 |
| 0.8117 | -0.050 | 0.774 | 0.8117 | -0.021 | 0.778 |
| 0.8370 | -0.026 | 0.779 | 0.8370 | -0.003 | 0.782 |
| 0.8619 | 0.005 | 0.785 | 0.8619 | 0.018 | 0.786 |
| 0.8869 | 0.034 | 0.791 | 0.8869 | 0.040 | 0.790 |
| 0.9120 | 0.305 | 0.797 | 0.9120 | 0.064 | 0.795 |
| 0.9416 | 0.895 | 0.803 | 0.9416 | 0.088 | 0.800 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.630 ALPHA 3.08 REYNOLDS 1.66E+10

INTEGRATED FORCE COEFFICIENTS

CN = 0.3713 CM = 0.0004

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0119 | -0.818 | 0.622 | 0.0119 | 0.728 | 0.927 |
| 0.0371 | -1.006 | 0.573 | 0.0371 | 0.294 | 0.841 |
| 0.0623 | -1.358 | 0.521 | 0.0623 | 0.007 | 0.764 |
| 0.0872 | -1.114 | 0.564 | 0.0872 | -0.043 | 0.774 |
| 0.1122 | -1.029 | 0.588 | 0.1122 | -0.093 | 0.764 |
| 0.1372 | -0.901 | 0.594 | 0.1372 | -0.128 | 0.757 |
| 0.1623 | -0.927 | 0.601 | 0.1623 | -0.168 | 0.749 |
| 0.1872 | -0.888 | 0.608 | 0.1872 | -0.195 | 0.743 |
| 0.2122 | -0.841 | 0.618 | 0.2122 | -0.213 | 0.742 |
| 0.2372 | -0.786 | 0.629 | 0.2372 | -0.218 | 0.739 |
| 0.2620 | -0.738 | 0.638 | 0.2620 | -0.217 | 0.739 |
| 0.2872 | -0.691 | 0.647 | 0.2872 | -0.218 | 0.740 |
| 0.3122 | -0.646 | 0.656 | 0.3122 | -0.212 | 0.748 |
| 0.3375 | -0.595 | 0.666 | 0.3375 | -0.194 | 0.744 |
| 0.3616 | -0.546 | 0.676 | 0.3616 | -0.172 | 0.746 |
| 0.3873 | -0.536 | 0.678 | 0.3873 | -0.194 | 0.744 |
| 0.4124 | -0.504 | 0.684 | 0.4124 | -0.194 | 0.744 |
| 0.4371 | -0.466 | 0.692 | 0.4371 | -0.178 | 0.747 |
| 0.4621 | -0.442 | 0.697 | 0.4621 | -0.176 | 0.747 |
| 0.4871 | -0.411 | 0.703 | 0.4871 | -0.167 | 0.749 |
| 0.5120 | -0.384 | 0.708 | 0.5120 | -0.161 | 0.750 |
| 0.5371 | -0.358 | 0.715 | 0.5371 | -0.144 | 0.753 |
| 0.5621 | -0.318 | 0.723 | 0.5621 | -0.116 | 0.759 |
| 0.5870 | -0.298 | 0.725 | 0.5870 | -0.123 | 0.758 |
| 0.6122 | -0.291 | 0.734 | 0.6122 | -0.085 | 0.765 |
| 0.6371 | -0.242 | 0.736 | 0.6371 | -0.095 | 0.763 |
| 0.6661 | -0.202 | 0.744 | 0.6661 | -0.072 | 0.768 |
| 0.6869 | -0.192 | 0.746 | 0.6869 | -0.078 | 0.767 |
| 0.7121 | -0.103 | 0.751 | 0.7121 | -0.063 | 0.769 |
| 0.7369 | -0.107 | 0.757 | 0.7369 | -0.050 | 0.772 |
| 0.7620 | -0.111 | 0.762 | 0.7620 | -0.039 | 0.774 |
| 0.7870 | -0.101 | 0.768 | 0.7870 | -0.023 | 0.777 |
| 0.8117 | -0.093 | 0.773 | 0.8117 | -0.009 | 0.782 |
| 0.8370 | -0.026 | 0.779 | 0.8370 | 0.005 | 0.783 |
| 0.8619 | 0.006 | 0.785 | 0.8619 | 0.019 | 0.786 |
| 0.8869 | 0.036 | 0.791 | 0.8869 | 0.048 | 0.792 |
| 0.9120 | 0.067 | 0.797 | 0.9120 | 0.063 | 0.794 |
| 0.9416 | 0.096 | 0.803 | 0.9416 | 0.083 | 0.798 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.649 ALPHA 0.00 REY 1.63×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.0009$ $C_M = 0.0002$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0119 | 0.155 | 0.707 | 0.0119 | 0.153 | 0.707 |
| 0.0371 | -0.200 | 0.695 | 0.0371 | -0.257 | 0.696 |
| 0.0623 | -0.550 | 0.629 | 0.0623 | -0.557 | 0.629 |
| 0.0872 | -0.535 | 0.634 | 0.0872 | -0.532 | 0.635 |
| 0.1122 | -0.542 | 0.632 | 0.1122 | -0.541 | 0.633 |
| 0.1372 | -0.530 | 0.633 | 0.1372 | -0.534 | 0.634 |
| 0.1623 | -0.554 | 0.630 | 0.1623 | -0.552 | 0.630 |
| 0.1872 | -0.559 | 0.628 | 0.1872 | -0.555 | 0.630 |
| 0.2122 | -0.551 | 0.633 | 0.2122 | -0.553 | 0.631 |
| 0.2372 | -0.529 | 0.635 | 0.2372 | -0.527 | 0.636 |
| 0.2620 | -0.505 | 0.640 | 0.2620 | -0.504 | 0.641 |
| 0.2872 | -0.477 | 0.646 | 0.2872 | -0.476 | 0.647 |
| 0.3122 | -0.457 | 0.651 | 0.3122 | -0.458 | 0.651 |
| 0.3375 | -0.421 | 0.659 | 0.3375 | -0.417 | 0.660 |
| 0.3618 | -0.385 | 0.667 | 0.3618 | -0.385 | 0.660 |
| 0.3873 | -0.391 | 0.666 | 0.3873 | -0.387 | 0.667 |
| 0.4124 | -0.374 | 0.669 | 0.4124 | -0.378 | 0.671 |
| 0.4371 | -0.346 | 0.676 | 0.4371 | -0.343 | 0.678 |
| 0.4621 | -0.332 | 0.679 | 0.4621 | -0.329 | 0.680 |
| 0.4871 | -0.311 | 0.683 | 0.4871 | -0.306 | 0.685 |
| 0.5120 | -0.292 | 0.688 | 0.5120 | -0.280 | 0.689 |
| 0.5371 | -0.267 | 0.693 | 0.5371 | -0.263 | 0.695 |
| 0.5621 | -0.241 | 0.701 | 0.5621 | -0.228 | 0.703 |
| 0.5870 | -0.227 | 0.702 | 0.5870 | -0.225 | 0.703 |
| 0.6122 | -0.184 | 0.712 | 0.6122 | -0.184 | 0.712 |
| 0.6371 | -0.181 | 0.712 | 0.6371 | -0.181 | 0.713 |
| 0.6661 | -0.148 | 0.720 | 0.6661 | -0.147 | 0.721 |
| 0.6869 | -0.145 | 0.723 | 0.6869 | -0.146 | 0.721 |
| 0.7121 | -0.122 | 0.725 | 0.7121 | -0.124 | 0.726 |
| 0.7369 | -0.102 | 0.730 | 0.7369 | -0.103 | 0.731 |
| 0.7620 | -0.082 | 0.735 | 0.7620 | -0.083 | 0.735 |
| 0.7870 | -0.057 | 0.740 | 0.7870 | -0.050 | 0.741 |
| 0.8117 | -0.035 | 0.745 | 0.8117 | -0.037 | 0.746 |
| 0.8370 | -0.013 | 0.750 | 0.8370 | -0.015 | 0.750 |
| 0.8619 | 0.015 | 0.756 | 0.8619 | 0.013 | 0.757 |
| 0.8869 | 0.041 | 0.762 | 0.8869 | 0.039 | 0.762 |
| 0.9120 | 0.072 | 0.769 | 0.9120 | 0.069 | 0.769 |
| 0.9416 | 0.101 | 0.775 | 0.9416 | 0.097 | 0.775 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.658 ALPHA 1.00 REYNOLDS 1.63E+6

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.1324$ $C_D = 0.0026$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0119 | -0.104 | 2.729 | 0.0119 | 0.367 | 0.648 |
| 0.0371 | -0.493 | 0.642 | 0.0371 | -0.044 | 0.744 |
| 0.0623 | -0.805 | 0.573 | 0.0623 | -0.341 | 0.670 |
| 0.0872 | -0.757 | 0.588 | 0.0872 | -0.350 | 0.676 |
| 0.1122 | -0.719 | 0.592 | 0.1122 | -0.377 | 0.678 |
| 0.1372 | -0.693 | 0.598 | 0.1372 | -0.388 | 0.668 |
| 0.1623 | -0.698 | 0.597 | 0.1623 | -0.415 | 0.662 |
| 0.1872 | -0.698 | 0.598 | 0.1872 | -0.438 | 0.658 |
| 0.2122 | -0.668 | 0.603 | 0.2122 | -0.432 | 0.657 |
| 0.2372 | -0.634 | 0.611 | 0.2372 | -0.422 | 0.660 |
| 0.2623 | -0.600 | 0.618 | 0.2623 | -0.408 | 0.663 |
| 0.2872 | -0.566 | 0.626 | 0.2872 | -0.398 | 0.667 |
| 0.3122 | -0.536 | 0.633 | 0.3122 | -0.377 | 0.678 |
| 0.3375 | -0.493 | 0.642 | 0.3375 | -0.347 | 0.676 |
| 0.3615 | -0.452 | 0.651 | 0.3615 | -0.316 | 0.683 |
| 0.3873 | -0.451 | 0.652 | 0.3873 | -0.327 | 0.681 |
| 0.4124 | -0.427 | 0.657 | 0.4124 | -0.315 | 0.684 |
| 0.4371 | -0.396 | 0.664 | 0.4371 | -0.292 | 0.689 |
| 0.4621 | -0.376 | 0.668 | 0.4621 | -0.282 | 0.691 |
| 0.4871 | -0.353 | 0.674 | 0.4871 | -0.266 | 0.694 |
| 0.5128 | -0.338 | 0.679 | 0.5128 | -0.251 | 0.698 |
| 0.5371 | -0.299 | 0.686 | 0.5371 | -0.227 | 0.703 |
| 0.5621 | -0.262 | 0.694 | 0.5621 | -0.195 | 0.710 |
| 0.5878 | -0.255 | 0.695 | 0.5878 | -0.194 | 0.710 |
| 0.6122 | -0.289 | 0.706 | 0.6122 | -0.152 | 0.720 |
| 0.6371 | -0.204 | 0.707 | 0.6371 | -0.154 | 0.719 |
| 0.6661 | -0.170 | 0.715 | 0.6661 | -0.124 | 0.726 |
| 0.6869 | -0.164 | 0.716 | 0.6869 | -0.124 | 0.726 |
| 0.7121 | -0.130 | 0.721 | 0.7121 | -0.104 | 0.731 |
| 0.7369 | -0.115 | 0.726 | 0.7369 | -0.086 | 0.735 |
| 0.7628 | -0.103 | 0.731 | 0.7628 | -0.067 | 0.739 |
| 0.7878 | -0.266 | 0.737 | 0.7878 | -0.044 | 0.744 |
| 0.8117 | -0.242 | 0.743 | 0.8117 | -0.025 | 0.748 |
| 0.8379 | -0.216 | 0.749 | 0.8379 | -0.005 | 0.753 |
| 0.8619 | 0.313 | 0.755 | 0.8619 | 0.028 | 0.758 |
| 0.8869 | 0.342 | 0.762 | 0.8869 | 0.043 | 0.763 |
| 0.9128 | 0.373 | 0.769 | 0.9128 | 0.071 | 0.769 |
| 0.9416 | 0.103 | 0.775 | 0.9416 | 0.096 | 0.775 |

NACA-8812 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.649 ALPHA 2.00 REV 1.63-18

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.2596$ $C_M = 0.0070$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | -0.398 | 0.665 | 0.0119 | 0.501 | 0.003 |
| 0.0371 | -0.745 | 0.587 | 0.0371 | 0.130 | 0.704 |
| 0.0623 | -1.116 | 0.505 | 0.0623 | -0.150 | 0.719 |
| 0.0872 | -0.963 | 0.539 | 0.0872 | -0.190 | 0.712 |
| 0.1122 | -0.915 | 0.550 | 0.1122 | -0.231 | 0.703 |
| 0.1372 | -0.803 | 0.561 | 0.1372 | -0.256 | 0.697 |
| 0.1623 | -0.551 | 0.564 | 0.1623 | -0.292 | 0.689 |
| 0.1872 | -0.820 | 0.569 | 0.1872 | -0.314 | 0.654 |
| 0.2122 | -0.791 | 0.577 | 0.2122 | -0.320 | 0.661 |
| 0.2372 | -0.742 | 0.580 | 0.2372 | -0.326 | 0.662 |
| 0.2620 | -0.694 | 0.599 | 0.2620 | -0.319 | 0.603 |
| 0.2872 | -0.650 | 0.609 | 0.2872 | -0.306 | 0.606 |
| 0.3122 | -0.612 | 0.617 | 0.3122 | -0.290 | 0.600 |
| 0.3375 | -0.594 | 0.620 | 0.3375 | -0.274 | 0.693 |
| 0.3610 | -0.516 | 0.630 | 0.3610 | -0.259 | 0.698 |
| 0.3873 | -0.500 | 0.640 | 0.3873 | -0.264 | 0.695 |
| 0.4124 | -0.400 | 0.646 | 0.4124 | -0.259 | 0.696 |
| 0.4371 | -0.443 | 0.655 | 0.4371 | -0.239 | 0.701 |
| 0.4621 | -0.419 | 0.660 | 0.4621 | -0.236 | 0.701 |
| 0.4871 | -0.390 | 0.666 | 0.4871 | -0.222 | 0.704 |
| 0.5120 | -0.304 | 0.672 | 0.5120 | -0.212 | 0.707 |
| 0.5371 | -0.331 | 0.679 | 0.5371 | -0.191 | 0.711 |
| 0.5621 | -0.292 | 0.680 | 0.5621 | -0.160 | 0.710 |
| 0.5870 | -0.240 | 0.691 | 0.5870 | -0.162 | 0.710 |
| 0.6122 | -0.243 | 0.701 | 0.6122 | -0.124 | 0.720 |
| 0.6371 | -0.222 | 0.704 | 0.6371 | -0.129 | 0.725 |
| 0.6661 | -0.104 | 0.712 | 0.6661 | -0.103 | 0.731 |
| 0.6869 | -0.176 | 0.714 | 0.6869 | -0.125 | 0.730 |
| 0.7121 | -0.190 | 0.720 | 0.7121 | -0.089 | 0.734 |
| 0.7369 | -0.145 | 0.725 | 0.7369 | -0.073 | 0.730 |
| 0.7620 | -0.101 | 0.731 | 0.7620 | -0.050 | 0.741 |
| 0.7870 | -0.272 | 0.737 | 0.7870 | -0.030 | 0.745 |
| 0.8117 | -0.246 | 0.743 | 0.8117 | -0.022 | 0.749 |
| 0.8370 | -0.200 | 0.749 | 0.8370 | -0.004 | 0.753 |
| 0.8619 | 0.313 | 0.756 | 0.8619 | 0.019 | 0.750 |
| 0.8869 | 0.342 | 0.762 | 0.8869 | 0.041 | 0.763 |
| 0.9120 | 0.873 | 0.769 | 0.9120 | 0.066 | 0.760 |
| 0.9416 | 0.103 | 0.776 | 0.9416 | 0.090 | 0.774 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.649 ALPHA 3.00 REYNOLDS 1.62*10⁶

INTEGRATED FORCE COEFFICIENTS

CV = 0.3968 CM = 0.0121

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0119 | -0.781 | 0.597 | 0.0119 | 0.736 | 0.917 |
| 0.0371 | -0.975 | 0.534 | 0.0371 | 0.381 | 0.821 |
| 0.0623 | -1.645 | 0.387 | 0.0623 | 0.007 | 0.755 |
| 0.0872 | -1.476 | 0.425 | 0.0872 | -0.044 | 0.744 |
| 0.1122 | -1.102 | 0.528 | 0.1122 | -0.098 | 0.732 |
| 0.1372 | -1.019 | 0.527 | 0.1372 | -0.134 | 0.724 |
| 0.1623 | -1.365 | 0.533 | 0.1623 | -0.178 | 0.714 |
| 0.1872 | -2.071 | 0.537 | 0.1872 | -0.207 | 0.708 |
| 0.2122 | -2.915 | 0.549 | 0.2122 | -0.227 | 0.703 |
| 0.2372 | -2.347 | 0.565 | 0.2372 | -0.238 | 0.703 |
| 0.2620 | -2.789 | 0.577 | 0.2620 | -0.231 | 0.703 |
| 0.2872 | -0.734 | 0.593 | 0.2872 | -0.225 | 0.704 |
| 0.3122 | -2.683 | 0.601 | 0.3122 | -0.224 | 0.704 |
| 0.3375 | -2.627 | 0.613 | 0.3375 | -0.204 | 0.709 |
| 0.3618 | -2.577 | 0.624 | 0.3618 | -0.184 | 0.713 |
| 0.3873 | -2.561 | 0.628 | 0.3873 | -0.203 | 0.709 |
| 0.4124 | -2.527 | 0.636 | 0.4124 | -0.202 | 0.709 |
| 0.4371 | -2.489 | 0.644 | 0.4371 | -0.187 | 0.712 |
| 0.4621 | -2.461 | 0.653 | 0.4621 | -0.187 | 0.712 |
| 0.4871 | -2.420 | 0.657 | 0.4871 | -0.176 | 0.715 |
| 0.5120 | -2.397 | 0.664 | 0.5120 | -0.178 | 0.716 |
| 0.5371 | -2.362 | 0.672 | 0.5371 | -0.151 | 0.728 |
| 0.5621 | -2.328 | 0.681 | 0.5621 | -0.125 | 0.726 |
| 0.5870 | -2.305 | 0.685 | 0.5870 | -0.129 | 0.725 |
| 0.6122 | -2.256 | 0.696 | 0.6122 | -0.093 | 0.733 |
| 0.6371 | -2.243 | 0.699 | 0.6371 | -0.102 | 0.731 |
| 0.6661 | -2.202 | 0.704 | 0.6661 | -0.078 | 0.737 |
| 0.6869 | -2.189 | 0.711 | 0.6869 | -0.081 | 0.736 |
| 0.7121 | -2.159 | 0.717 | 0.7121 | -0.169 | 0.739 |
| 0.7369 | -2.134 | 0.723 | 0.7369 | -0.055 | 0.742 |
| 0.7620 | -2.106 | 0.729 | 0.7620 | -0.043 | 0.744 |
| 0.7870 | -2.076 | 0.736 | 0.7870 | -0.026 | 0.748 |
| 0.8117 | -2.047 | 0.742 | 0.8117 | -0.011 | 0.751 |
| 0.8370 | -2.028 | 0.748 | 0.8370 | 0.005 | 0.755 |
| 0.8619 | 0.312 | 0.755 | 0.8619 | 0.022 | 0.759 |
| 0.8869 | 0.742 | 0.762 | 0.8869 | 0.043 | 0.763 |
| 0.9120 | 0.073 | 0.769 | 0.9120 | 0.064 | 0.768 |
| 0.9416 | 0.102 | 0.775 | 0.9416 | 0.085 | 0.773 |

NACA-8012 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.675 ALPHA 0.00 REY 1.63×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.8011$ $C_M = 0.8884$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | 0.171 | 0.777 | 0.0119 | 0.171 | 0.777 |
| 0.0371 | -0.252 | 0.677 | 0.0371 | -0.248 | 0.679 |
| 0.0623 | -0.572 | 0.602 | 0.0623 | -0.569 | 0.683 |
| 0.0872 | -0.551 | 0.607 | 0.0872 | -0.548 | 0.688 |
| 0.1122 | -0.501 | 0.684 | 0.1122 | -0.558 | 0.625 |
| 0.1372 | -0.501 | 0.684 | 0.1372 | -0.554 | 0.686 |
| 0.1623 | -0.503 | 0.599 | 0.1623 | -0.577 | 0.681 |
| 0.1872 | -0.591 | 0.597 | 0.1872 | -0.583 | 0.688 |
| 0.2122 | -0.594 | 0.599 | 0.2122 | -0.579 | 0.681 |
| 0.2372 | -0.558 | 0.605 | 0.2372 | -0.551 | 0.687 |
| 0.2620 | -0.553 | 0.611 | 0.2620 | -0.528 | 0.613 |
| 0.2872 | -0.503 | 0.618 | 0.2872 | -0.497 | 0.628 |
| 0.3122 | -0.479 | 0.623 | 0.3122 | -0.478 | 0.624 |
| 0.3375 | -0.448 | 0.632 | 0.3375 | -0.437 | 0.634 |
| 0.3618 | -0.404 | 0.641 | 0.3618 | -0.404 | 0.642 |
| 0.3873 | -0.409 | 0.648 | 0.3873 | -0.404 | 0.642 |
| 0.4124 | -0.387 | 0.645 | 0.4124 | -0.386 | 0.646 |
| 0.4371 | -0.357 | 0.652 | 0.4371 | -0.357 | 0.653 |
| 0.4621 | -0.342 | 0.655 | 0.4621 | -0.344 | 0.656 |
| 0.4871 | -0.328 | 0.661 | 0.4871 | -0.328 | 0.661 |
| 0.5120 | -0.308 | 0.665 | 0.5120 | -0.302 | 0.666 |
| 0.5371 | -0.273 | 0.672 | 0.5371 | -0.272 | 0.673 |
| 0.5621 | -0.247 | 0.680 | 0.5621 | -0.236 | 0.691 |
| 0.5870 | -0.242 | 0.681 | 0.5870 | -0.231 | 0.683 |
| 0.6122 | -0.188 | 0.692 | 0.6122 | -0.188 | 0.692 |
| 0.6371 | -0.185 | 0.692 | 0.6371 | -0.185 | 0.693 |
| 0.6661 | -0.152 | 0.703 | 0.6661 | -0.151 | 0.721 |
| 0.6869 | -0.148 | 0.701 | 0.6869 | -0.147 | 0.722 |
| 0.7121 | -0.145 | 0.707 | 0.7121 | -0.129 | 0.728 |
| 0.7369 | -0.104 | 0.712 | 0.7369 | -0.104 | 0.712 |
| 0.7620 | -0.093 | 0.717 | 0.7620 | -0.084 | 0.717 |
| 0.7870 | -0.057 | 0.723 | 0.7870 | -0.059 | 0.723 |
| 0.8117 | -0.034 | 0.728 | 0.8117 | -0.036 | 0.729 |
| 0.8370 | -0.012 | 0.733 | 0.8370 | -0.014 | 0.734 |
| 0.8619 | 0.017 | 0.740 | 0.8619 | 0.015 | 0.740 |
| 0.8869 | 0.043 | 0.746 | 0.8869 | 0.040 | 0.746 |
| 0.9120 | 0.075 | 0.754 | 0.9120 | 0.071 | 0.754 |
| 0.9416 | 0.104 | 0.761 | 0.9416 | 0.100 | 0.761 |

NACA-0012 293.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.674 ALPHA 1.80 REY 1.63×10^6

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.1344$ $C_M = 0.0039$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | -0.803 | 0.710 | 0.0119 | 0.393 | 0.830 |
| 0.0371 | -0.402 | 0.625 | 0.0371 | -0.044 | 0.727 |
| 0.0623 | -0.040 | 0.543 | 0.0623 | -0.353 | 0.634 |
| 0.0872 | -0.700 | 0.550 | 0.0872 | -0.365 | 0.552 |
| 0.1122 | -0.795 | 0.561 | 0.1122 | -0.394 | 0.645 |
| 0.1372 | -0.750 | 0.567 | 0.1372 | -0.407 | 0.642 |
| 0.1623 | -0.750 | 0.565 | 0.1623 | -0.439 | 0.634 |
| 0.1872 | -0.743 | 0.566 | 0.1872 | -0.456 | 0.630 |
| 0.2122 | -0.711 | 0.571 | 0.2122 | -0.460 | 0.629 |
| 0.2372 | -0.670 | 0.581 | 0.2372 | -0.447 | 0.632 |
| 0.2620 | -0.633 | 0.590 | 0.2620 | -0.432 | 0.636 |
| 0.2872 | -0.592 | 0.599 | 0.2872 | -0.411 | 0.641 |
| 0.3122 | -0.562 | 0.606 | 0.3122 | -0.390 | 0.644 |
| 0.3375 | -0.519 | 0.616 | 0.3375 | -0.366 | 0.651 |
| 0.3610 | -0.476 | 0.627 | 0.3610 | -0.334 | 0.659 |
| 0.3873 | -0.472 | 0.627 | 0.3873 | -0.344 | 0.656 |
| 0.4124 | -0.444 | 0.634 | 0.4124 | -0.330 | 0.660 |
| 0.4371 | -0.412 | 0.641 | 0.4371 | -0.305 | 0.665 |
| 0.4621 | -0.392 | 0.645 | 0.4621 | -0.296 | 0.667 |
| 0.4871 | -0.367 | 0.651 | 0.4871 | -0.270 | 0.671 |
| 0.5120 | -0.341 | 0.658 | 0.5120 | -0.263 | 0.675 |
| 0.5371 | -0.310 | 0.665 | 0.5371 | -0.237 | 0.681 |
| 0.5621 | -0.270 | 0.674 | 0.5621 | -0.203 | 0.689 |
| 0.5870 | -0.201 | 0.676 | 0.5870 | -0.203 | 0.689 |
| 0.6122 | -0.215 | 0.687 | 0.6122 | -0.160 | 0.699 |
| 0.6371 | -0.209 | 0.688 | 0.6371 | -0.163 | 0.699 |
| 0.6661 | -0.173 | 0.697 | 0.6661 | -0.132 | 0.706 |
| 0.6969 | -0.167 | 0.698 | 0.6969 | -0.132 | 0.706 |
| 0.7121 | -0.140 | 0.704 | 0.7121 | -0.112 | 0.710 |
| 0.7369 | -0.117 | 0.710 | 0.7369 | -0.092 | 0.715 |
| 0.7620 | -0.093 | 0.715 | 0.7620 | -0.074 | 0.719 |
| 0.7870 | -0.066 | 0.722 | 0.7870 | -0.051 | 0.725 |
| 0.8117 | -0.040 | 0.726 | 0.8117 | -0.031 | 0.729 |
| 0.8370 | -0.016 | 0.733 | 0.8370 | -0.010 | 0.734 |
| 0.8619 | 0.015 | 0.741 | 0.8619 | 0.017 | 0.740 |
| 0.8869 | 0.044 | 0.747 | 0.8869 | 0.041 | 0.746 |
| 0.9120 | 0.075 | 0.755 | 0.9120 | 0.070 | 0.753 |
| 0.9416 | 0.106 | 0.762 | 0.9416 | 0.097 | 0.760 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.674 ALPHA 2.00 REYNOLDS 1.63E6

INTEGRATED FORCE COEFFICIENTS

CL = 0.2747 CM = 0.0001

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | -0.354 | 0.654 | 0.0119 | 0.505 | 0.075 |
| 0.0371 | -0.700 | 0.570 | 0.0371 | 0.142 | 0.770 |
| 0.0623 | -1.279 | 0.436 | 0.0623 | -0.162 | 0.699 |
| 0.0872 | -3.999 | 0.502 | 0.0872 | -0.195 | 0.691 |
| 0.1122 | -3.909 | 0.504 | 0.1122 | -0.239 | 0.601 |
| 0.1372 | -0.926 | 0.519 | 0.1372 | -0.265 | 0.675 |
| 0.1623 | -0.917 | 0.521 | 0.1623 | -0.302 | 0.666 |
| 0.1872 | -3.980 | 0.525 | 0.1872 | -0.327 | 0.666 |
| 0.2122 | -3.053 | 0.536 | 0.2122 | -0.340 | 0.657 |
| 0.2372 | -3.700 | 0.551 | 0.2372 | -0.335 | 0.658 |
| 0.2620 | -3.737 | 0.563 | 0.2620 | -0.329 | 0.660 |
| 0.2872 | -0.686 | 0.575 | 0.2872 | -0.317 | 0.663 |
| 0.3122 | -0.645 | 0.585 | 0.3122 | -0.311 | 0.664 |
| 0.3375 | -0.593 | 0.597 | 0.3375 | -0.286 | 0.670 |
| 0.3619 | -0.544 | 0.609 | 0.3619 | -0.262 | 0.676 |
| 0.3873 | -0.531 | 0.612 | 0.3873 | -0.277 | 0.672 |
| 0.4124 | -0.499 | 0.619 | 0.4124 | -0.268 | 0.674 |
| 0.4371 | -3.462 | 0.620 | 0.4371 | -0.246 | 0.679 |
| 0.4621 | -3.437 | 0.633 | 0.4621 | -0.243 | 0.680 |
| 0.4871 | -3.404 | 0.641 | 0.4871 | -0.228 | 0.684 |
| 0.5120 | -3.376 | 0.648 | 0.5120 | -0.210 | 0.686 |
| 0.5371 | -0.342 | 0.656 | 0.5371 | -0.196 | 0.691 |
| 0.5621 | -0.301 | 0.666 | 0.5621 | -0.180 | 0.690 |
| 0.5870 | -0.208 | 0.669 | 0.5870 | -0.169 | 0.698 |
| 0.6122 | -0.241 | 0.680 | 0.6122 | -0.130 | 0.737 |
| 0.6371 | -3.230 | 0.682 | 0.6371 | -0.133 | 0.726 |
| 0.6619 | -3.192 | 0.691 | 0.6619 | -0.106 | 0.712 |
| 0.6869 | -3.152 | 0.694 | 0.6869 | -0.100 | 0.712 |
| 0.7121 | -3.153 | 0.700 | 0.7121 | -0.092 | 0.716 |
| 0.7369 | -3.125 | 0.707 | 0.7369 | -0.074 | 0.720 |
| 0.7620 | -0.100 | 0.713 | 0.7620 | -0.060 | 0.724 |
| 0.7870 | -0.070 | 0.720 | 0.7870 | -0.040 | 0.720 |
| 0.8117 | -0.042 | 0.726 | 0.8117 | -0.024 | 0.732 |
| 0.8370 | -0.015 | 0.733 | 0.8370 | -0.005 | 0.736 |
| 0.8619 | 0.017 | 0.740 | 0.8619 | 0.017 | 0.742 |
| 0.8869 | 0.046 | 0.747 | 0.8869 | 0.041 | 0.747 |
| 0.9120 | 0.070 | 0.754 | 0.9120 | 0.066 | 0.753 |
| 0.9416 | 0.106 | 0.761 | 0.9416 | 0.091 | 0.759 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.700 ALPHA 0.00 REYNOLDS 1.62x10⁶

INTEGRATED FORCE COEFFICIENTS

C_N = -0.0012 C_M = -0.0000

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | 0.109 | 0.767 | 0.0119 | 0.191 | 0.760 |
| 0.0371 | -0.240 | 0.661 | 0.0371 | -0.237 | 0.662 |
| 0.0623 | -0.500 | 0.577 | 0.0623 | -0.505 | 0.576 |
| 0.0872 | -0.505 | 0.501 | 0.0872 | -0.570 | 0.500 |
| 0.1122 | -0.503 | 0.577 | 0.1122 | -0.509 | 0.575 |
| 0.1372 | -0.504 | 0.576 | 0.1372 | -0.506 | 0.576 |
| 0.1623 | -0.610 | 0.570 | 0.1623 | -0.617 | 0.508 |
| 0.1872 | -0.623 | 0.567 | 0.1872 | -0.629 | 0.565 |
| 0.2122 | -0.617 | 0.569 | 0.2172 | -0.622 | 0.567 |
| 0.2372 | -0.508 | 0.576 | 0.2372 | -0.500 | 0.575 |
| 0.2620 | -0.501 | 0.502 | 0.2620 | -0.564 | 0.501 |
| 0.2872 | -0.526 | 0.591 | 0.2872 | -0.529 | 0.590 |
| 0.3122 | -0.501 | 0.597 | 0.3122 | -0.510 | 0.595 |
| 0.3375 | -0.403 | 0.607 | 0.3375 | -0.406 | 0.605 |
| 0.3618 | -0.426 | 0.616 | 0.3618 | -0.431 | 0.614 |
| 0.3873 | -0.427 | 0.615 | 0.3873 | -0.426 | 0.615 |
| 0.4124 | -0.406 | 0.621 | 0.4124 | -0.404 | 0.621 |
| 0.4371 | -0.374 | 0.628 | 0.4371 | -0.372 | 0.629 |
| 0.4621 | -0.300 | 0.632 | 0.4621 | -0.350 | 0.632 |
| 0.4871 | -0.334 | 0.639 | 0.4871 | -0.331 | 0.639 |
| 0.5120 | -0.313 | 0.644 | 0.5120 | -0.312 | 0.644 |
| 0.5371 | -0.294 | 0.651 | 0.5371 | -0.282 | 0.651 |
| 0.5621 | -0.247 | 0.660 | 0.5621 | -0.246 | 0.660 |
| 0.5870 | -0.240 | 0.662 | 0.5870 | -0.240 | 0.661 |
| 0.6122 | -0.196 | 0.673 | 0.6122 | -0.190 | 0.672 |
| 0.6371 | -0.194 | 0.673 | 0.6371 | -0.194 | 0.673 |
| 0.6661 | -0.150 | 0.682 | 0.6661 | -0.150 | 0.682 |
| 0.6869 | -0.154 | 0.683 | 0.6869 | -0.153 | 0.683 |
| 0.7121 | -0.130 | 0.689 | 0.7121 | -0.130 | 0.689 |
| 0.7369 | -0.107 | 0.695 | 0.7369 | -0.107 | 0.694 |
| 0.7620 | -0.005 | 0.700 | 0.7620 | -0.003 | 0.700 |
| 0.7870 | -0.050 | 0.707 | 0.7870 | -0.050 | 0.706 |
| 0.8117 | -0.034 | 0.713 | 0.8117 | -0.035 | 0.712 |
| 0.8370 | -0.012 | 0.710 | 0.8370 | -0.012 | 0.710 |
| 0.8619 | 0.017 | 0.726 | 0.8619 | 0.017 | 0.725 |
| 0.8869 | 0.045 | 0.732 | 0.8869 | 0.044 | 0.732 |
| 0.9120 | 0.076 | 0.743 | 0.9120 | 0.075 | 0.739 |
| 0.9410 | 0.106 | 0.740 | 0.9410 | 0.104 | 0.747 |

NACA-0012 293.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.728 ALPHA 1.00 REYNOLDS 1.63E+6

INTEGRATED FORCE COEFFICIENTS

C_D = 0.1424 C_M = 0.0040

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|----------------|-------|----------------------|----------------|-------|
| X/C | C _P | P/H | X/C | C _P | P/H |
| 0.0119 | -0.853 | 0.700 | 0.0119 | 0.405 | 0.021 |
| 0.0371 | -0.450 | 0.607 | 0.0371 | -0.035 | 0.712 |
| 0.0623 | -0.860 | 0.506 | 0.0623 | -0.350 | 0.032 |
| 0.0872 | -0.802 | 0.522 | 0.0872 | -0.373 | 0.020 |
| 0.1122 | -0.607 | 0.521 | 0.1122 | -0.405 | 0.020 |
| 0.1372 | -0.700 | 0.520 | 0.1372 | -0.423 | 0.016 |
| 0.1623 | -0.797 | 0.523 | 0.1623 | -0.450 | 0.007 |
| 0.1872 | -0.803 | 0.522 | 0.1872 | -0.479 | 0.002 |
| 0.2122 | -0.776 | 0.520 | 0.2122 | -0.404 | 0.001 |
| 0.2372 | -0.720 | 0.542 | 0.2372 | -0.469 | 0.004 |
| 0.2620 | -0.677 | 0.553 | 0.2620 | -0.454 | 0.000 |
| 0.2872 | -0.625 | 0.566 | 0.2872 | -0.429 | 0.014 |
| 0.3122 | -0.592 | 0.574 | 0.3122 | -0.416 | 0.010 |
| 0.3375 | -0.544 | 0.506 | 0.3375 | -0.305 | 0.025 |
| 0.3610 | -0.501 | 0.506 | 0.3610 | -0.353 | 0.033 |
| 0.3873 | -0.492 | 0.599 | 0.3873 | -0.363 | 0.031 |
| 0.4124 | -0.463 | 0.606 | 0.4124 | -0.345 | 0.035 |
| 0.4371 | -0.427 | 0.615 | 0.4371 | -0.310 | 0.042 |
| 0.4621 | -0.406 | 0.623 | 0.4621 | -0.309 | 0.044 |
| 0.4871 | -0.370 | 0.627 | 0.4871 | -0.209 | 0.049 |
| 0.5120 | -0.352 | 0.633 | 0.5120 | -0.273 | 0.053 |
| 0.5371 | -0.319 | 0.642 | 0.5371 | -0.246 | 0.060 |
| 0.5621 | -0.200 | 0.651 | 0.5621 | -0.211 | 0.060 |
| 0.5870 | -0.200 | 0.654 | 0.5870 | -0.209 | 0.069 |
| 0.6122 | -0.223 | 0.655 | 0.6122 | -0.167 | 0.079 |
| 0.6371 | -0.214 | 0.660 | 0.6371 | -0.160 | 0.079 |
| 0.6661 | -0.177 | 0.677 | 0.6661 | -0.135 | 0.067 |
| 0.6869 | -0.107 | 0.679 | 0.6869 | -0.135 | 0.067 |
| 0.7121 | -0.142 | 0.605 | 0.7121 | -0.114 | 0.092 |
| 0.7369 | -0.116 | 0.692 | 0.7369 | -0.095 | 0.097 |
| 0.7620 | -0.092 | 0.696 | 0.7620 | -0.075 | 0.702 |
| 0.7870 | -0.064 | 0.705 | 0.7870 | -0.051 | 0.700 |
| 0.8117 | -0.030 | 0.711 | 0.8117 | -0.030 | 0.713 |
| 0.8370 | -0.012 | 0.710 | 0.8370 | -0.009 | 0.710 |
| 0.8619 | 0.019 | 0.725 | 0.8619 | 0.010 | 0.725 |
| 0.8869 | 0.049 | 0.733 | 0.8869 | 0.043 | 0.731 |
| 0.9120 | 0.001 | 0.740 | 0.9120 | 0.072 | 0.730 |
| 0.9416 | 0.110 | 0.740 | 0.9416 | 0.099 | 0.745 |

NACA-0012 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.788 ALPHA 2.88 REYNOLDS 1.63x10⁶

INTEGRATED FORCE COEFFICIENTS

C_L = 0.2917 C_D = 0.0181

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0119 | -0.308 | 2.646 | 0.0119 | 0.587 | 0.866 |
| 0.0371 | -0.648 | 2.568 | 0.0371 | 0.145 | 0.757 |
| 0.0623 | -1.216 | 0.428 | 0.0623 | -0.163 | 0.681 |
| 0.0872 | -1.198 | 0.426 | 0.0872 | -0.282 | 0.672 |
| 0.1122 | -1.114 | 0.445 | 0.1122 | -0.247 | 0.661 |
| 0.1372 | -1.111 | 0.446 | 0.1372 | -0.275 | 0.654 |
| 0.1623 | -1.029 | 0.466 | 0.1623 | -0.318 | 0.643 |
| 0.1872 | -0.998 | 0.484 | 0.1872 | -0.345 | 0.636 |
| 0.2122 | -0.993 | 0.485 | 0.2122 | -0.350 | 0.633 |
| 0.2372 | -0.994 | 0.589 | 0.2372 | -0.355 | 0.634 |
| 0.2628 | -0.719 | 0.528 | 0.2628 | -0.349 | 0.636 |
| 0.2872 | -0.710 | 0.544 | 0.2872 | -0.334 | 0.639 |
| 0.3122 | -0.608 | 0.555 | 0.3122 | -0.329 | 0.640 |
| 0.3375 | -0.615 | 0.568 | 0.3375 | -0.382 | 0.647 |
| 0.3618 | -0.566 | 0.588 | 0.3618 | -0.276 | 0.653 |
| 0.3873 | -0.546 | 0.585 | 0.3873 | -0.290 | 0.652 |
| 0.4124 | -0.512 | 0.594 | 0.4124 | -0.281 | 0.652 |
| 0.4371 | -0.473 | 0.604 | 0.4371 | -0.268 | 0.657 |
| 0.4621 | -0.446 | 0.610 | 0.4621 | -0.255 | 0.659 |
| 0.4871 | -0.415 | 0.618 | 0.4871 | -0.239 | 0.663 |
| 0.5128 | -0.386 | 0.625 | 0.5128 | -0.228 | 0.665 |
| 0.5371 | -0.391 | 0.634 | 0.5371 | -0.285 | 0.671 |
| 0.5621 | -0.389 | 0.644 | 0.5621 | -0.175 | 0.678 |
| 0.5878 | -0.294 | 0.648 | 0.5878 | -0.176 | 0.678 |
| 0.6122 | -0.248 | 0.659 | 0.6122 | -0.137 | 0.688 |
| 0.6371 | -0.235 | 0.662 | 0.6371 | -0.148 | 0.687 |
| 0.6618 | -0.155 | 0.672 | 0.6618 | -0.111 | 0.694 |
| 0.6868 | -0.193 | 0.675 | 0.6868 | -0.111 | 0.694 |
| 0.7121 | -0.193 | 0.683 | 0.7121 | -0.895 | 0.698 |
| 0.7369 | -0.126 | 0.689 | 0.7369 | -0.777 | 0.723 |
| 0.7628 | -0.399 | 0.696 | 0.7628 | -0.862 | 0.785 |
| 0.7878 | -0.389 | 0.784 | 0.7878 | -0.848 | 0.712 |
| 0.8117 | -0.841 | 0.710 | 0.8117 | -0.824 | 0.716 |
| 0.8378 | -0.814 | 0.717 | 0.8378 | -0.884 | 0.721 |
| 0.8619 | 0.828 | 0.725 | 0.8619 | 0.819 | 0.726 |
| 0.8869 | 0.348 | 0.733 | 0.8869 | 0.842 | 0.732 |
| 0.9128 | 0.888 | 0.748 | 0.9128 | 0.867 | 0.738 |
| 0.9416 | 0.189 | 0.748 | 0.9416 | 0.893 | 0.745 |

NACA-0012 293.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.728 ALPHA 0.00 REYN 1.62×10^6

INTEGRATED FORCE COEFFICIENTS

$C_L = -0.0047$ $C_M = 0.0001$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | 0.205 | 0.760 | 0.0119 | 0.201 | 0.760 |
| 0.0371 | -0.226 | 0.649 | 0.0371 | -0.227 | 0.650 |
| 0.0623 | -0.507 | 0.557 | 0.0623 | -0.596 | 0.555 |
| 0.0872 | -0.579 | 0.559 | 0.0872 | -0.580 | 0.557 |
| 0.1122 | -0.605 | 0.552 | 0.1122 | -0.617 | 0.550 |
| 0.1372 | -0.610 | 0.551 | 0.1372 | -0.610 | 0.550 |
| 0.1623 | -0.644 | 0.542 | 0.1623 | -0.655 | 0.542 |
| 0.1872 | -0.664 | 0.537 | 0.1872 | -0.674 | 0.535 |
| 0.2122 | -0.660 | 0.530 | 0.2122 | -0.673 | 0.535 |
| 0.2372 | -0.625 | 0.547 | 0.2372 | -0.630 | 0.544 |
| 0.2620 | -0.594 | 0.555 | 0.2620 | -0.609 | 0.552 |
| 0.2872 | -0.554 | 0.565 | 0.2872 | -0.566 | 0.563 |
| 0.3122 | -0.526 | 0.572 | 0.3122 | -0.539 | 0.570 |
| 0.3375 | -0.485 | 0.583 | 0.3375 | -0.492 | 0.582 |
| 0.3610 | -0.444 | 0.593 | 0.3610 | -0.455 | 0.591 |
| 0.3873 | -0.443 | 0.594 | 0.3873 | -0.449 | 0.593 |
| 0.4124 | -0.420 | 0.599 | 0.4124 | -0.427 | 0.599 |
| 0.4371 | -0.387 | 0.600 | 0.4371 | -0.390 | 0.600 |
| 0.4621 | -0.372 | 0.612 | 0.4621 | -0.375 | 0.611 |
| 0.4871 | -0.345 | 0.615 | 0.4871 | -0.344 | 0.619 |
| 0.5120 | -0.322 | 0.625 | 0.5120 | -0.324 | 0.624 |
| 0.5371 | -0.293 | 0.632 | 0.5371 | -0.294 | 0.632 |
| 0.5621 | -0.295 | 0.542 | 0.5621 | -0.290 | 0.641 |
| 0.5870 | -0.248 | 0.644 | 0.5870 | -0.249 | 0.644 |
| 0.6122 | -0.202 | 0.656 | 0.6122 | -0.206 | 0.655 |
| 0.6371 | -0.198 | 0.657 | 0.6371 | -0.199 | 0.656 |
| 0.6661 | -0.163 | 0.666 | 0.6661 | -0.163 | 0.666 |
| 0.6869 | -0.157 | 0.667 | 0.6869 | -0.157 | 0.667 |
| 0.7121 | -0.142 | 0.674 | 0.7121 | -0.134 | 0.673 |
| 0.7369 | -0.109 | 0.682 | 0.7369 | -0.110 | 0.679 |
| 0.7620 | -0.086 | 0.686 | 0.7620 | -0.087 | 0.685 |
| 0.7870 | -0.060 | 0.692 | 0.7870 | -0.066 | 0.692 |
| 0.8117 | -0.036 | 0.698 | 0.8117 | -0.036 | 0.690 |
| 0.8370 | -0.011 | 0.705 | 0.8370 | -0.011 | 0.705 |
| 0.8619 | 0.019 | 0.715 | 0.8619 | 0.019 | 0.713 |
| 0.8869 | 0.047 | 0.720 | 0.8869 | 0.046 | 0.719 |
| 0.9120 | 0.079 | 0.726 | 0.9120 | 0.078 | 0.720 |
| 0.9416 | 0.107 | 0.735 | 0.9416 | 0.106 | 0.735 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.719 ALPHA 1.00 REYNOLDS 1.62E6

INTEGRATED FORCE COEFFICIENTS

CL = 0.1492 CM = 0.0953

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0119 | -0.822 | 0.703 | 0.0119 | 0.413 | 0.814 |
| 0.0371 | -0.828 | 0.799 | 0.0371 | -0.833 | 0.788 |
| 0.0623 | -0.924 | 0.472 | 0.0623 | -0.365 | 0.615 |
| 0.0872 | -0.889 | 0.581 | 0.0872 | -0.387 | 0.609 |
| 0.1122 | -0.856 | 0.489 | 0.1122 | -0.423 | 0.688 |
| 0.1372 | -0.825 | 0.497 | 0.1372 | -0.443 | 0.595 |
| 0.1623 | -0.846 | 0.492 | 0.1623 | -0.463 | 0.594 |
| 0.1872 | -0.866 | 0.482 | 0.1872 | -0.509 | 0.578 |
| 0.2122 | -0.891 | 0.488 | 0.2122 | -0.513 | 0.576 |
| 0.2372 | -0.784 | 0.588 | 0.2372 | -0.496 | 0.568 |
| 0.2620 | -0.718 | 0.525 | 0.2620 | -0.488 | 0.594 |
| 0.2872 | -0.662 | 0.539 | 0.2872 | -0.455 | 0.591 |
| 0.3122 | -0.622 | 0.549 | 0.3122 | -0.439 | 0.595 |
| 0.3375 | -0.578 | 0.562 | 0.3375 | -0.485 | 0.604 |
| 0.3618 | -0.523 | 0.575 | 0.3618 | -0.373 | 0.612 |
| 0.3873 | -0.518 | 0.578 | 0.3873 | -0.378 | 0.618 |
| 0.4124 | -0.488 | 0.586 | 0.4124 | -0.362 | 0.615 |
| 0.4371 | -0.443 | 0.595 | 0.4371 | -0.335 | 0.621 |
| 0.4621 | -0.419 | 0.601 | 0.4621 | -0.325 | 0.623 |
| 0.4871 | -0.398 | 0.609 | 0.4871 | -0.302 | 0.629 |
| 0.5120 | -0.363 | 0.616 | 0.5120 | -0.287 | 0.633 |
| 0.5371 | -0.338 | 0.624 | 0.5371 | -0.258 | 0.643 |
| 0.5621 | -0.291 | 0.634 | 0.5621 | -0.224 | 0.649 |
| 0.5870 | -0.278 | 0.637 | 0.5870 | -0.217 | 0.651 |
| 0.6122 | -0.252 | 0.649 | 0.6122 | -0.175 | 0.662 |
| 0.6371 | -0.223 | 0.652 | 0.6371 | -0.175 | 0.662 |
| 0.6661 | -0.183 | 0.662 | 0.6661 | -0.142 | 0.678 |
| 0.6869 | -0.173 | 0.664 | 0.6869 | -0.141 | 0.671 |
| 0.7121 | -0.145 | 0.671 | 0.7121 | -0.119 | 0.676 |
| 0.7369 | -0.128 | 0.678 | 0.7369 | -0.098 | 0.682 |
| 0.7620 | -0.094 | 0.684 | 0.7620 | -0.079 | 0.687 |
| 0.7870 | -0.065 | 0.692 | 0.7870 | -0.053 | 0.693 |
| 0.8117 | -0.038 | 0.699 | 0.8117 | -0.032 | 0.699 |
| 0.8370 | -0.011 | 0.705 | 0.8370 | -0.018 | 0.725 |
| 0.8619 | 0.021 | 0.714 | 0.8619 | 0.018 | 0.712 |
| 0.8869 | 0.050 | 0.721 | 0.8869 | 0.044 | 0.718 |
| 0.9120 | 0.082 | 0.738 | 0.9120 | 0.074 | 0.726 |
| 0.9416 | 0.113 | 0.737 | 0.9416 | 0.101 | 0.733 |

NACA-0012 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.721 ALPHA 2.00 REYNOLDS 1.63x10⁶

INTEGRATED FORCE COEFFICIENTS

C_D = 0.3288 C_L = 0.8111

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | -0.234 | 0.646 | 0.0119 | 0.591 | 0.868 |
| 0.0371 | -0.588 | 0.557 | 0.0371 | 0.145 | 0.745 |
| 0.0623 | -1.178 | 0.435 | 0.0623 | -0.169 | 0.664 |
| 0.0872 | -1.128 | 0.418 | 0.0872 | -0.211 | 0.653 |
| 0.1122 | -1.165 | 0.486 | 0.1122 | -0.259 | 0.641 |
| 0.1372 | -1.095 | 0.424 | 0.1372 | -0.292 | 0.633 |
| 0.1623 | -1.147 | 0.411 | 0.1623 | -0.336 | 0.621 |
| 0.1872 | -1.147 | 0.411 | 0.1872 | -0.367 | 0.613 |
| 0.2122 | -1.183 | 0.481 | 0.2122 | -0.383 | 0.600 |
| 0.2372 | -1.196 | 0.398 | 0.2372 | -0.377 | 0.611 |
| 0.2620 | -1.197 | 0.397 | 0.2620 | -0.371 | 0.612 |
| 0.2872 | -1.148 | 0.418 | 0.2872 | -0.355 | 0.616 |
| 0.3122 | -0.818 | 0.495 | 0.3122 | -0.344 | 0.619 |
| 0.3375 | -0.618 | 0.547 | 0.3375 | -0.321 | 0.625 |
| 0.3616 | -0.551 | 0.564 | 0.3616 | -0.295 | 0.632 |
| 0.3873 | -0.524 | 0.571 | 0.3873 | -0.339 | 0.628 |
| 0.4124 | -0.494 | 0.579 | 0.4124 | -0.298 | 0.631 |
| 0.4371 | -0.488 | 0.588 | 0.4371 | -0.271 | 0.637 |
| 0.4621 | -0.436 | 0.594 | 0.4621 | -0.271 | 0.637 |
| 0.4871 | -0.447 | 0.601 | 0.4871 | -0.254 | 0.642 |
| 0.5120 | -0.378 | 0.689 | 0.5120 | -0.241 | 0.645 |
| 0.5371 | -0.347 | 0.617 | 0.5371 | -0.218 | 0.651 |
| 0.5621 | -0.336 | 0.627 | 0.5621 | -0.187 | 0.659 |
| 0.5870 | -0.291 | 0.631 | 0.5870 | -0.187 | 0.659 |
| 0.6122 | -0.245 | 0.643 | 0.6122 | -0.146 | 0.678 |
| 0.6371 | -0.232 | 0.646 | 0.6371 | -0.149 | 0.669 |
| 0.6661 | -0.193 | 0.657 | 0.6661 | -0.119 | 0.677 |
| 0.6869 | -0.182 | 0.668 | 0.6869 | -0.128 | 0.676 |
| 0.7121 | -0.192 | 0.667 | 0.7121 | -0.191 | 0.681 |
| 0.7369 | -0.126 | 0.674 | 0.7369 | -0.083 | 0.686 |
| 0.7620 | -0.098 | 0.681 | 0.7620 | -0.065 | 0.691 |
| 0.7870 | -0.088 | 0.689 | 0.7870 | -0.044 | 0.696 |
| 0.8117 | -0.039 | 0.697 | 0.8117 | -0.029 | 0.701 |
| 0.8370 | -0.011 | 0.704 | 0.8370 | -0.006 | 0.706 |
| 0.8619 | 0.022 | 0.712 | 0.8619 | 0.019 | 0.712 |
| 0.8869 | 0.052 | 0.723 | 0.8869 | 0.043 | 0.718 |
| 0.9120 | 0.084 | 0.728 | 0.9120 | 0.071 | 0.726 |
| 0.9416 | 0.112 | 0.735 | 0.9416 | 0.097 | 0.732 |

VACA-0012 203.2 PM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.740 ALPHA 0.80 REY 1.62x10⁶

INTEGRATED FORCE COEFFICIENTS

CN = -0.8871 CM = 0.8881

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | 0.228 | 0.753 | 0.0119 | 0.223 | 0.754 |
| 0.0371 | -0.211 | 0.633 | 0.0371 | -0.209 | 0.639 |
| 0.0623 | -0.589 | 0.535 | 0.0623 | -0.598 | 0.535 |
| 0.0872 | -0.598 | 0.337 | 0.0872 | -0.600 | 0.534 |
| 0.1122 | -0.651 | 0.526 | 0.1122 | -0.645 | 0.522 |
| 0.1372 | -0.641 | 0.523 | 0.1372 | -0.634 | 0.520 |
| 0.1623 | -0.665 | 0.512 | 0.1623 | -0.781 | 0.587 |
| 0.1872 | -0.726 | 0.581 | 0.1872 | -0.748 | 0.495 |
| 0.2122 | -0.735 | 0.490 | 0.2122 | -0.777 | 0.485 |
| 0.2372 | -0.603 | 0.512 | 0.2372 | -0.715 | 0.502 |
| 0.2620 | -0.642 | 0.523 | 0.2620 | -0.668 | 0.517 |
| 0.2872 | -0.595 | 0.536 | 0.2872 | -0.611 | 0.530 |
| 0.3122 | -0.564 | 0.544 | 0.3122 | -0.581 | 0.538 |
| 0.3375 | -0.519 | 0.556 | 0.3375 | -0.531 | 0.552 |
| 0.3618 | -0.474 | 0.568 | 0.3618 | -0.488 | 0.563 |
| 0.3873 | -0.467 | 0.573 | 0.3873 | -0.477 | 0.566 |
| 0.4124 | -0.439 | 0.578 | 0.4124 | -0.449 | 0.573 |
| 0.4371 | -0.404 | 0.587 | 0.4371 | -0.412 | 0.583 |
| 0.4621 | -0.386 | 0.592 | 0.4621 | -0.393 | 0.587 |
| 0.4871 | -0.358 | 0.599 | 0.4871 | -0.363 | 0.595 |
| 0.5120 | -0.335 | 0.606 | 0.5120 | -0.338 | 0.602 |
| 0.5371 | -0.304 | 0.614 | 0.5371 | -0.306 | 0.611 |
| 0.5621 | -0.285 | 0.624 | 0.5621 | -0.267 | 0.621 |
| 0.5870 | -0.256 | 0.627 | 0.5870 | -0.256 | 0.624 |
| 0.6122 | -0.212 | 0.638 | 0.6122 | -0.212 | 0.636 |
| 0.6371 | -0.205 | 0.648 | 0.6371 | -0.204 | 0.648 |
| 0.6661 | -0.168 | 0.658 | 0.6661 | -0.168 | 0.648 |
| 0.6869 | -0.162 | 0.652 | 0.6869 | -0.161 | 0.658 |
| 0.7121 | -0.136 | 0.659 | 0.7121 | -0.135 | 0.656 |
| 0.7369 | -0.111 | 0.665 | 0.7369 | -0.112 | 0.663 |
| 0.7620 | -0.087 | 0.672 | 0.7620 | -0.087 | 0.669 |
| 0.7870 | -0.068 | 0.679 | 0.7870 | -0.061 | 0.677 |
| 0.8117 | -0.045 | 0.686 | 0.8117 | -0.035 | 0.683 |
| 0.8370 | -0.018 | 0.692 | 0.8370 | -0.018 | 0.698 |
| 0.8619 | 0.020 | 0.700 | 0.8619 | 0.020 | 0.698 |
| 0.8869 | 0.049 | 0.708 | 0.8869 | 0.048 | 0.706 |
| 0.9120 | 0.081 | 0.717 | 0.9120 | 0.081 | 0.715 |
| 0.9416 | 0.111 | 0.725 | 0.9416 | 0.109 | 0.722 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.740 ALPHA 1.00 REYNOLDS 1.62x10⁶

INTEGRATED FORCE COEFFICIENTS

C_D = 0.1663 C_M = 0.0057

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | 0.822 | 2.781 | 0.0119 | 0.418 | 0.687 |
| 0.0371 | -0.301 | 2.593 | 0.0371 | -0.028 | 0.688 |
| 0.0623 | -0.933 | 2.446 | 0.0623 | -0.367 | 0.590 |
| 0.0872 | -0.766 | 2.491 | 0.0872 | -0.392 | 0.591 |
| 0.1122 | -0.849 | 2.469 | 0.1122 | -0.436 | 0.579 |
| 0.1372 | -0.876 | 2.462 | 0.1372 | -0.459 | 0.573 |
| 0.1623 | -0.900 | 2.455 | 0.1623 | -0.503 | 0.561 |
| 0.1872 | -0.911 | 2.452 | 0.1872 | -0.534 | 0.553 |
| 0.2122 | -0.926 | 2.439 | 0.2122 | -0.544 | 0.551 |
| 0.2372 | -0.941 | 2.430 | 0.2372 | -0.524 | 0.556 |
| 0.2623 | -1.315 | 2.424 | 0.2623 | -0.500 | 0.560 |
| 0.2872 | -1.006 | 2.426 | 0.2872 | -0.481 | 0.567 |
| 0.3122 | -0.810 | 2.476 | 0.3122 | -0.464 | 0.572 |
| 0.3375 | -0.572 | 2.542 | 0.3375 | -0.428 | 0.581 |
| 0.3618 | -0.513 | 2.598 | 0.3618 | -0.394 | 0.593 |
| 0.3873 | -0.499 | 2.561 | 0.3873 | -0.396 | 0.593 |
| 0.4124 | -0.473 | 2.569 | 0.4124 | -0.378 | 0.595 |
| 0.4371 | -0.448 | 2.577 | 0.4371 | -0.349 | 0.602 |
| 0.4621 | -0.418 | 2.583 | 0.4621 | -0.336 | 0.606 |
| 0.4871 | -0.390 | 2.598 | 0.4871 | -0.314 | 0.612 |
| 0.5120 | -0.363 | 2.597 | 0.5120 | -0.297 | 0.616 |
| 0.5371 | -0.332 | 2.606 | 0.5371 | -0.269 | 0.624 |
| 0.5621 | -0.292 | 2.616 | 0.5621 | -0.234 | 0.633 |
| 0.5870 | -0.270 | 2.620 | 0.5870 | -0.228 | 0.635 |
| 0.6122 | -0.251 | 2.633 | 0.6122 | -0.185 | 0.646 |
| 0.6371 | -0.222 | 2.635 | 0.6371 | -0.182 | 0.647 |
| 0.6661 | -0.184 | 2.645 | 0.6661 | -0.149 | 0.656 |
| 0.6869 | -0.173 | 2.648 | 0.6869 | -0.145 | 0.657 |
| 0.7121 | -0.145 | 2.656 | 0.7121 | -0.122 | 0.663 |
| 0.7369 | -0.128 | 2.662 | 0.7369 | -0.101 | 0.669 |
| 0.7620 | -0.893 | 2.678 | 0.7620 | -0.888 | 0.674 |
| 0.7873 | -0.303 | 2.677 | 0.7873 | -0.855 | 0.681 |
| 0.8117 | -0.236 | 2.685 | 0.8117 | -0.834 | 0.686 |
| 0.8378 | -0.210 | 2.692 | 0.8378 | -0.810 | 0.693 |
| 0.8619 | 0.324 | 2.701 | 0.8619 | 0.818 | 0.700 |
| 0.8869 | 0.392 | 2.708 | 0.8869 | 0.845 | 0.707 |
| 0.9120 | 0.806 | 2.717 | 0.9120 | 0.876 | 0.716 |
| 0.9416 | 0.115 | 2.725 | 0.9416 | 0.104 | 0.723 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.739 ALPHA 2.00 REY 1.62×10^6

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.3337$ $C_M = 0.0007$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0119 | -0.176 | 0.649 | 0.0119 | 0.579 | 0.050 |
| 0.0371 | -0.521 | 0.557 | 0.0371 | 0.137 | 0.732 |
| 0.0623 | -1.103 | 0.402 | 0.0623 | -0.104 | 0.646 |
| 0.0872 | -1.056 | 0.414 | 0.0872 | -0.227 | 0.635 |
| 0.1122 | -1.112 | 0.399 | 0.1122 | -0.270 | 0.621 |
| 0.1372 | -1.066 | 0.411 | 0.1372 | -0.309 | 0.613 |
| 0.1623 | -1.105 | 0.421 | 0.1623 | -0.350 | 0.600 |
| 0.1872 | -1.123 | 0.397 | 0.1872 | -0.391 | 0.591 |
| 0.2122 | -1.101 | 0.307 | 0.2122 | -0.409 | 0.506 |
| 0.2372 | -1.101 | 0.301 | 0.2372 | -0.444 | 0.557 |
| 0.2620 | -1.205 | 0.375 | 0.2620 | -0.399 | 0.509 |
| 0.2872 | -1.196 | 0.378 | 0.2872 | -0.382 | 0.593 |
| 0.3122 | -1.215 | 0.372 | 0.3122 | -0.374 | 0.595 |
| 0.3375 | -1.194 | 0.370 | 0.3375 | -0.347 | 0.603 |
| 0.3610 | -0.999 | 0.430 | 0.3610 | -0.321 | 0.610 |
| 0.3873 | -0.678 | 0.517 | 0.3873 | -0.331 | 0.607 |
| 0.4124 | -0.547 | 0.550 | 0.4124 | -0.322 | 0.609 |
| 0.4371 | -0.401 | 0.560 | 0.4371 | -0.297 | 0.616 |
| 0.4621 | -0.431 | 0.581 | 0.4621 | -0.290 | 0.617 |
| 0.4871 | -0.395 | 0.593 | 0.4871 | -0.271 | 0.622 |
| 0.5120 | -0.303 | 0.600 | 0.5120 | -0.259 | 0.625 |
| 0.5371 | -0.329 | 0.600 | 0.5371 | -0.234 | 0.632 |
| 0.5621 | -0.293 | 0.610 | 0.5621 | -0.202 | 0.641 |
| 0.5870 | -0.276 | 0.622 | 0.5870 | -0.201 | 0.641 |
| 0.6122 | -0.234 | 0.633 | 0.6122 | -0.160 | 0.652 |
| 0.6371 | -0.223 | 0.637 | 0.6371 | -0.162 | 0.651 |
| 0.6601 | -0.104 | 0.647 | 0.6601 | -0.132 | 0.659 |
| 0.6869 | -0.172 | 0.653 | 0.6869 | -0.131 | 0.660 |
| 0.7121 | -0.144 | 0.657 | 0.7121 | -0.109 | 0.665 |
| 0.7369 | -0.110 | 0.664 | 0.7369 | -0.090 | 0.670 |
| 0.7620 | -0.091 | 0.671 | 0.7620 | -0.072 | 0.675 |
| 0.7870 | -0.063 | 0.679 | 0.7870 | -0.049 | 0.681 |
| 0.8117 | -0.036 | 0.686 | 0.8117 | -0.031 | 0.686 |
| 0.8370 | -0.010 | 0.693 | 0.8370 | -0.010 | 0.692 |
| 0.8619 | 0.023 | 0.702 | 0.8619 | 0.016 | 0.690 |
| 0.8869 | 0.050 | 0.705 | 0.8869 | 0.041 | 0.705 |
| 0.9120 | 0.003 | 0.710 | 0.9120 | 0.069 | 0.713 |
| 0.9416 | 0.110 | 0.725 | 0.9416 | 0.095 | 0.720 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.759 ALPHA 0.00 REY 1.59*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = -0.0045 CM = -0.0000

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | 0.243 | 0.749 | 0.0119 | 0.247 | 0.749 |
| 0.0371 | -0.100 | 0.631 | 0.0371 | -0.100 | 0.632 |
| 0.0623 | -0.503 | 0.522 | 0.0623 | -0.503 | 0.520 |
| 0.0872 | -0.506 | 0.521 | 0.0872 | -0.505 | 0.520 |
| 0.1122 | -0.642 | 0.506 | 0.1122 | -0.647 | 0.502 |
| 0.1372 | -0.654 | 0.502 | 0.1372 | -0.656 | 0.500 |
| 0.1623 | -0.696 | 0.491 | 0.1623 | -0.696 | 0.489 |
| 0.1872 | -0.752 | 0.476 | 0.1872 | -0.750 | 0.474 |
| 0.2122 | -0.811 | 0.459 | 0.2122 | -0.814 | 0.456 |
| 0.2372 | -0.843 | 0.451 | 0.2372 | -0.843 | 0.448 |
| 0.2623 | -0.861 | 0.446 | 0.2623 | -0.871 | 0.448 |
| 0.2872 | -0.810 | 0.457 | 0.2872 | -0.865 | 0.442 |
| 0.3122 | -0.622 | 0.511 | 0.3122 | -0.784 | 0.454 |
| 0.3375 | -0.521 | 0.539 | 0.3375 | -0.542 | 0.531 |
| 0.3618 | -0.405 | 0.549 | 0.3618 | -0.480 | 0.548 |
| 0.3873 | -0.400 | 0.553 | 0.3873 | -0.469 | 0.551 |
| 0.4124 | -0.454 | 0.559 | 0.4124 | -0.449 | 0.557 |
| 0.4371 | -0.422 | 0.566 | 0.4371 | -0.415 | 0.566 |
| 0.4621 | -0.402 | 0.571 | 0.4621 | -0.399 | 0.570 |
| 0.4871 | -0.372 | 0.579 | 0.4871 | -0.369 | 0.578 |
| 0.5120 | -0.348 | 0.586 | 0.5120 | -0.346 | 0.585 |
| 0.5371 | -0.317 | 0.595 | 0.5371 | -0.313 | 0.594 |
| 0.5621 | -0.277 | 0.606 | 0.5621 | -0.276 | 0.604 |
| 0.5873 | -0.265 | 0.609 | 0.5873 | -0.264 | 0.608 |
| 0.6122 | -0.219 | 0.622 | 0.6122 | -0.220 | 0.620 |
| 0.6371 | -0.211 | 0.624 | 0.6371 | -0.229 | 0.623 |
| 0.6661 | -0.173 | 0.634 | 0.6661 | -0.173 | 0.633 |
| 0.6869 | -0.166 | 0.636 | 0.6869 | -0.164 | 0.635 |
| 0.7121 | -0.148 | 0.643 | 0.7121 | -0.136 | 0.642 |
| 0.7369 | -0.116 | 0.650 | 0.7369 | -0.112 | 0.650 |
| 0.7620 | -0.298 | 0.657 | 0.7620 | -0.080 | 0.656 |
| 0.7873 | -0.302 | 0.665 | 0.7873 | -0.059 | 0.664 |
| 0.8117 | -0.336 | 0.672 | 0.8117 | -0.035 | 0.671 |
| 0.8373 | -0.310 | 0.679 | 0.8373 | -0.000 | 0.678 |
| 0.8619 | 0.022 | 0.688 | 0.8619 | 0.021 | 0.686 |
| 0.8869 | 0.051 | 0.696 | 0.8869 | 0.051 | 0.695 |
| 0.9120 | 0.003 | 0.705 | 0.9120 | 0.003 | 0.704 |
| 0.9416 | 0.112 | 0.713 | 0.9416 | 0.112 | 0.712 |

NACA-0012-203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.768 ALPHA 1.88 REY 1.63×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.1824$ $C_M = 0.0027$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 2.8119 | 0.863 | 2.699 | 0.8119 | 0.428 | 0.797 |
| 0.8371 | -0.332 | 2.598 | 0.8371 | -0.023 | 0.674 |
| 2.8623 | -0.688 | 2.438 | 0.8623 | -0.372 | 0.578 |
| 2.8872 | -0.746 | 2.475 | 0.8872 | -0.432 | 0.569 |
| 2.1122 | -0.798 | 2.463 | 0.1122 | -0.456 | 0.554 |
| 2.1372 | -0.837 | 2.459 | 0.1372 | -0.484 | 0.547 |
| 2.1623 | -0.878 | 2.439 | 0.1623 | -0.536 | 0.532 |
| 2.1872 | -0.908 | 2.431 | 0.1872 | -0.581 | 0.528 |
| 2.2122 | -0.957 | 2.417 | 0.2122 | -0.686 | 0.513 |
| 2.2372 | -0.983 | 2.410 | 0.2372 | -0.588 | 0.518 |
| 2.2623 | -1.013 | 2.432 | 0.2623 | -0.564 | 0.525 |
| 2.2872 | -1.025 | 2.398 | 0.2872 | -0.529 | 0.535 |
| 2.3122 | -1.044 | 2.393 | 0.3122 | -0.508 | 0.549 |
| 2.3375 | -1.033 | 2.396 | 0.3375 | -0.478 | 0.551 |
| 2.3618 | -1.036 | 2.395 | 0.3618 | -0.434 | 0.561 |
| 0.3873 | -0.921 | 2.427 | 0.3873 | -0.438 | 0.562 |
| 0.4124 | -0.591 | 2.518 | 0.4124 | -0.429 | 0.568 |
| 2.4371 | -0.464 | 2.553 | 0.4371 | -0.377 | 0.577 |
| 2.4621 | -0.406 | 2.569 | 0.4621 | -0.362 | 0.581 |
| 0.4871 | -0.369 | 2.583 | 0.4871 | -0.335 | 0.588 |
| 0.5128 | -0.337 | 2.588 | 0.5128 | -0.314 | 0.594 |
| 2.5371 | -0.309 | 2.596 | 0.5371 | -0.284 | 0.602 |
| 2.5621 | -0.275 | 2.635 | 0.5621 | -0.258 | 0.612 |
| 2.5873 | -0.262 | 2.639 | 0.5873 | -0.242 | 0.614 |
| 0.6122 | -0.222 | 2.624 | 0.6122 | -0.197 | 0.626 |
| 2.6371 | -0.209 | 2.624 | 0.6371 | -0.192 | 0.628 |
| 0.6661 | -0.174 | 2.633 | 0.6661 | -0.158 | 0.637 |
| 2.6869 | -0.164 | 2.636 | 0.6869 | -0.153 | 0.638 |
| 0.7121 | -0.137 | 2.644 | 0.7121 | -0.133 | 0.645 |
| 2.7369 | -0.113 | 2.657 | 0.7369 | -0.107 | 0.651 |
| 2.7623 | -0.087 | 2.657 | 0.7623 | -0.086 | 0.657 |
| 2.7878 | -0.059 | 2.665 | 0.7878 | -0.058 | 0.665 |
| 2.8117 | -0.301 | 2.673 | 0.8117 | -0.034 | 0.671 |
| 2.8378 | -0.007 | 2.689 | 0.8378 | -0.018 | 0.678 |
| 0.8619 | 0.825 | 2.683 | 0.8619 | 0.016 | 0.686 |
| 0.8869 | 0.754 | 2.696 | 0.8869 | 0.047 | 0.694 |
| 0.9128 | 0.887 | 2.785 | 0.9128 | 0.079 | 0.723 |
| 0.9416 | 0.113 | 0.713 | 0.9416 | 0.108 | 0.711 |

VACA-8812 283.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.759 ALPHA 2.88 REY 1.68-10⁶

INTEGRATED FORCE COEFFICIENTS

C_N = 0.3456 C_M = -0.8885

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | -0.103 | 3.654 | 0.0119 | 0.564 | 0.037 |
| 0.0371 | -0.491 | 3.598 | 0.0371 | 0.124 | 0.715 |
| 0.0623 | -1.817 | 3.482 | 0.0623 | -0.284 | 0.624 |
| 0.0872 | -0.976 | 3.413 | 0.0872 | -0.248 | 0.612 |
| 0.1122 | -1.037 | 3.396 | 0.1122 | -0.303 | 0.597 |
| 0.1372 | -1.311 | 3.404 | 0.1372 | -0.338 | 0.587 |
| 0.1623 | -1.742 | 3.395 | 0.1623 | -0.388 | 0.573 |
| 0.1872 | -1.863 | 3.389 | 0.1872 | -0.427 | 0.562 |
| 0.2122 | -1.103 | 3.378 | 0.2122 | -0.458 | 0.557 |
| 0.2372 | -1.127 | 3.372 | 0.2372 | -0.443 | 0.559 |
| 0.2620 | -1.157 | 3.363 | 0.2620 | -0.435 | 0.561 |
| 0.2872 | -1.151 | 3.365 | 0.2872 | -0.414 | 0.567 |
| 0.3122 | -1.176 | 3.358 | 0.3122 | -0.406 | 0.560 |
| 0.3375 | -1.164 | 3.361 | 0.3375 | -0.378 | 0.577 |
| 0.3618 | -1.173 | 3.359 | 0.3618 | -0.350 | 0.585 |
| 0.3873 | -1.169 | 3.355 | 0.3873 | -0.361 | 0.581 |
| 0.4124 | -1.359 | 3.391 | 0.4124 | -0.347 | 0.585 |
| 0.4371 | -0.709 | 3.407 | 0.4371 | -0.322 | 0.593 |
| 0.4621 | -0.572 | 3.525 | 0.4621 | -0.313 | 0.596 |
| 0.4871 | -0.509 | 3.542 | 0.4871 | -0.291 | 0.602 |
| 0.5123 | -0.456 | 3.557 | 0.5123 | -0.275 | 0.607 |
| 0.5371 | -0.403 | 3.572 | 0.5371 | -0.249 | 0.614 |
| 0.5621 | -0.362 | 3.583 | 0.5621 | -0.216 | 0.623 |
| 0.5873 | -0.322 | 3.594 | 0.5873 | -0.213 | 0.624 |
| 0.6122 | -0.275 | 3.607 | 0.6122 | -0.173 | 0.635 |
| 0.6371 | -0.248 | 3.617 | 0.6371 | -0.172 | 0.635 |
| 0.6661 | -0.199 | 3.628 | 0.6661 | -0.148 | 0.644 |
| 0.6869 | -0.178 | 3.634 | 0.6869 | -0.139 | 0.644 |
| 0.7121 | -0.143 | 3.643 | 0.7121 | -0.120 | 0.651 |
| 0.7369 | -0.117 | 3.651 | 0.7369 | -0.102 | 0.656 |
| 0.7623 | -0.305 | 3.663 | 0.7623 | -0.083 | 0.661 |
| 0.7878 | -0.358 | 3.667 | 0.7878 | -0.059 | 0.667 |
| 0.8117 | -0.833 | 3.675 | 0.8117 | -0.039 | 0.673 |
| 0.8373 | -0.306 | 3.681 | 0.8373 | -0.010 | 0.679 |
| 0.8619 | 0.325 | 3.693 | 0.8619 | 0.000 | 0.686 |
| 0.8869 | 0.351 | 3.697 | 0.8869 | 0.032 | 0.692 |
| 0.9123 | 0.381 | 3.705 | 0.9123 | 0.063 | 0.700 |
| 0.9416 | 0.103 | 3.711 | 0.9416 | 0.085 | 0.707 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.731 ALPHA 0.00 REY 1.57*10⁶

INTEGRATED FORCE COEFFICIENTS

C_x = 0.0016 C_y = -0.0003

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 2.0119 | 0.278 | 0.745 | 0.0119 | 0.267 | 0.745 |
| 0.0371 | -0.193 | 0.624 | 0.0371 | -0.156 | 0.624 |
| 0.0623 | -0.502 | 0.507 | 0.0623 | -0.559 | 0.510 |
| 0.0872 | -0.550 | 0.506 | 0.0872 | -0.561 | 0.509 |
| 0.1122 | -0.658 | 0.468 | 0.1122 | -0.634 | 0.490 |
| 0.1372 | -0.693 | 0.401 | 0.1372 | -0.650 | 0.401 |
| 0.1623 | -0.676 | 0.475 | 0.1623 | -0.670 | 0.476 |
| 0.1872 | -0.724 | 0.461 | 0.1872 | -0.720 | 0.462 |
| 0.2122 | -0.793 | 0.441 | 0.2122 | -0.797 | 0.442 |
| 0.2372 | -0.827 | 0.431 | 0.2372 | -0.833 | 0.431 |
| 0.2620 | -0.865 | 0.421 | 0.2620 | -0.868 | 0.422 |
| 0.2872 | -0.874 | 0.410 | 0.2872 | -0.870 | 0.410 |
| 0.3122 | -0.896 | 0.412 | 0.3122 | -0.900 | 0.412 |
| 0.3375 | -0.804 | 0.415 | 0.3375 | -0.800 | 0.416 |
| 0.3610 | -0.891 | 0.413 | 0.3610 | -0.896 | 0.414 |
| 0.3873 | -0.808 | 0.414 | 0.3873 | -0.868 | 0.422 |
| 0.4124 | -0.643 | 0.404 | 0.4124 | -0.555 | 0.511 |
| 0.4371 | -0.442 | 0.542 | 0.4371 | -0.413 | 0.551 |
| 0.4621 | -0.379 | 0.559 | 0.4621 | -0.370 | 0.564 |
| 0.4871 | -0.343 | 0.573 | 0.4871 | -0.342 | 0.571 |
| 0.5120 | -0.317 | 0.577 | 0.5120 | -0.320 | 0.570 |
| 0.5371 | -0.292 | 0.584 | 0.5371 | -0.295 | 0.585 |
| 0.5621 | -0.262 | 0.593 | 0.5621 | -0.263 | 0.594 |
| 0.5870 | -0.250 | 0.596 | 0.5870 | -0.252 | 0.597 |
| 0.6122 | -0.211 | 0.607 | 0.6122 | -0.213 | 0.608 |
| 0.6371 | -0.202 | 0.610 | 0.6371 | -0.203 | 0.611 |
| 0.6661 | -0.160 | 0.623 | 0.6661 | -0.167 | 0.621 |
| 0.6869 | -0.159 | 0.622 | 0.6869 | -0.159 | 0.624 |
| 0.7121 | -0.133 | 0.630 | 0.7121 | -0.133 | 0.631 |
| 0.7369 | -0.109 | 0.636 | 0.7369 | -0.109 | 0.639 |
| 0.7620 | -0.084 | 0.644 | 0.7620 | -0.085 | 0.645 |
| 0.7870 | -0.057 | 0.651 | 0.7870 | -0.050 | 0.653 |
| 0.8117 | -0.030 | 0.659 | 0.8117 | -0.031 | 0.660 |
| 0.8370 | -0.006 | 0.666 | 0.8370 | -0.007 | 0.667 |
| 0.8619 | 0.024 | 0.674 | 0.8619 | 0.024 | 0.676 |
| 0.8869 | 0.053 | 0.683 | 0.8869 | 0.050 | 0.683 |
| 0.9120 | 0.086 | 0.692 | 0.9120 | 0.084 | 0.693 |
| 0.9416 | 0.113 | 0.700 | 0.9416 | 0.111 | 0.701 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.779 ALPHA 0.58 REY $1.57 \cdot 10^6$

INTEGRATED FORCE COEFFICIENTS

CV = 0.1819 CM = -0.0016

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0119 | 0.109 | 0.723 | 0.0119 | 0.349 | 0.758 |
| 0.0371 | -0.223 | 0.686 | 0.0371 | -0.284 | 0.645 |
| 0.0623 | -0.720 | 0.465 | 0.0623 | -0.462 | 0.537 |
| 0.0872 | -0.623 | 0.492 | 0.0872 | -0.462 | 0.531 |
| 0.1122 | -0.709 | 0.468 | 0.1122 | -0.546 | 0.513 |
| 0.1372 | -0.748 | 0.459 | 0.1372 | -0.565 | 0.507 |
| 0.1623 | -0.778 | 0.448 | 0.1623 | -0.616 | 0.493 |
| 0.1872 | -0.800 | 0.442 | 0.1872 | -0.672 | 0.477 |
| 0.2122 | -0.827 | 0.425 | 0.2122 | -0.737 | 0.458 |
| 0.2372 | -0.856 | 0.417 | 0.2372 | -0.768 | 0.449 |
| 0.2622 | -0.927 | 0.405 | 0.2622 | -0.808 | 0.443 |
| 0.2872 | -0.935 | 0.403 | 0.2872 | -0.802 | 0.443 |
| 0.3122 | -0.961 | 0.396 | 0.3122 | -0.813 | 0.437 |
| 0.3375 | -0.953 | 0.398 | 0.3375 | -0.735 | 0.459 |
| 0.3618 | -0.964 | 0.395 | 0.3618 | -0.513 | 0.522 |
| 0.3873 | -0.969 | 0.388 | 0.3873 | -0.447 | 0.541 |
| 0.4124 | -1.003 | 0.384 | 0.4124 | -0.425 | 0.547 |
| 0.4371 | -0.713 | 0.466 | 0.4371 | -0.397 | 0.555 |
| 0.4621 | -0.477 | 0.533 | 0.4621 | -0.383 | 0.553 |
| 0.4871 | -0.342 | 0.557 | 0.4871 | -0.357 | 0.567 |
| 0.5120 | -0.338 | 0.573 | 0.5120 | -0.335 | 0.573 |
| 0.5371 | -0.308 | 0.584 | 0.5371 | -0.386 | 0.562 |
| 0.5621 | -0.265 | 0.594 | 0.5621 | -0.278 | 0.59 |
| 0.5870 | -0.246 | 0.599 | 0.5870 | -0.259 | 0.595 |
| 0.6122 | -0.218 | 0.609 | 0.6122 | -0.216 | 0.627 |
| 0.6371 | -0.196 | 0.613 | 0.6371 | -0.207 | 0.618 |
| 0.6661 | -0.163 | 0.623 | 0.6661 | -0.171 | 0.620 |
| 0.6869 | -0.153 | 0.626 | 0.6869 | -0.164 | 0.622 |
| 0.7121 | -0.126 | 0.633 | 0.7121 | -0.138 | 0.638 |
| 0.7369 | -0.105 | 0.639 | 0.7369 | -0.114 | 0.637 |
| 0.7620 | -0.079 | 0.647 | 0.7620 | -0.098 | 0.643 |
| 0.7878 | -0.053 | 0.654 | 0.7878 | -0.062 | 0.652 |
| 0.8117 | -0.026 | 0.662 | 0.8117 | -0.036 | 0.659 |
| 0.8370 | -0.003 | 0.668 | 0.8370 | -0.012 | 0.666 |
| 0.8619 | 0.026 | 0.677 | 0.8619 | 0.019 | 0.674 |
| 0.8869 | 0.055 | 0.685 | 0.8869 | 0.047 | 0.682 |
| 0.9120 | 0.089 | 0.695 | 0.9120 | 0.081 | 0.692 |
| 0.9416 | 0.113 | 0.701 | 0.9416 | 0.108 | 0.703 |

NACA-2012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.731 ALPHA 1.00 REY 1.50×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.1844$ $C_M = -0.0049$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0119 | 0.120 | 0.782 | 0.0119 | 0.421 | 0.760 |
| 0.0371 | -0.273 | 0.590 | 0.0371 | -0.020 | 0.662 |
| 0.0623 | -0.806 | 0.438 | 0.0623 | -0.379 | 0.560 |
| 0.0872 | -0.736 | 0.458 | 0.0872 | -0.412 | 0.550 |
| 0.1122 | -0.726 | 0.460 | 0.1122 | -0.474 | 0.532 |
| 0.1372 | -0.773 | 0.447 | 0.1372 | -0.501 | 0.525 |
| 0.1623 | -0.826 | 0.432 | 0.1623 | -0.560 | 0.508 |
| 0.1872 | -0.853 | 0.424 | 0.1872 | -0.619 | 0.491 |
| 0.2122 | -0.917 | 0.406 | 0.2122 | -0.662 | 0.473 |
| 0.2372 | -0.940 | 0.400 | 0.2372 | -0.710 | 0.465 |
| 0.2620 | -0.960 | 0.380 | 0.2620 | -0.737 | 0.457 |
| 0.2872 | -0.986 | 0.386 | 0.2872 | -0.750 | 0.450 |
| 0.3122 | -1.016 | 0.370 | 0.3122 | -0.680 | 0.471 |
| 0.3375 | -1.006 | 0.381 | 0.3375 | -0.516 | 0.520 |
| 0.3618 | -1.023 | 0.376 | 0.3618 | -0.462 | 0.530 |
| 0.3873 | -1.045 | 0.369 | 0.3873 | -0.460 | 0.536 |
| 0.4124 | -1.074 | 0.361 | 0.4124 | -0.439 | 0.542 |
| 0.4371 | -1.050 | 0.366 | 0.4371 | -0.404 | 0.552 |
| 0.4621 | -0.861 | 0.416 | 0.4621 | -0.380 | 0.557 |
| 0.4871 | -0.564 | 0.507 | 0.4871 | -0.361 | 0.565 |
| 0.5120 | -0.453 | 0.539 | 0.5120 | -0.339 | 0.571 |
| 0.5371 | -0.393 | 0.556 | 0.5371 | -0.300 | 0.580 |
| 0.5621 | -0.345 | 0.570 | 0.5621 | -0.271 | 0.590 |
| 0.5870 | -0.299 | 0.583 | 0.5870 | -0.261 | 0.593 |
| 0.6122 | -0.253 | 0.596 | 0.6122 | -0.210 | 0.605 |
| 0.6371 | -0.221 | 0.605 | 0.6371 | -0.211 | 0.608 |
| 0.6601 | -0.179 | 0.617 | 0.6601 | -0.174 | 0.610 |
| 0.6869 | -0.159 | 0.622 | 0.6869 | -0.160 | 0.620 |
| 0.7121 | -0.129 | 0.631 | 0.7121 | -0.143 | 0.627 |
| 0.7369 | -0.103 | 0.639 | 0.7369 | -0.119 | 0.634 |
| 0.7620 | -0.075 | 0.646 | 0.7620 | -0.096 | 0.640 |
| 0.7870 | -0.049 | 0.654 | 0.7870 | -0.078 | 0.648 |
| 0.8117 | -0.021 | 0.662 | 0.8117 | -0.046 | 0.655 |
| 0.8370 | 0.003 | 0.669 | 0.8370 | -0.022 | 0.662 |
| 0.8619 | 0.031 | 0.677 | 0.8619 | 0.009 | 0.670 |
| 0.8869 | 0.059 | 0.685 | 0.8869 | 0.036 | 0.678 |
| 0.9120 | 0.089 | 0.693 | 0.9120 | 0.069 | 0.688 |
| 0.9416 | 0.110 | 0.699 | 0.9416 | 0.095 | 0.695 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.021 ALPHA 0.00 REY 1.60E10

INTEGRATED FORCE COEFFICIENTS

C_x = -0.0061 C_m = 0.0015

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | C _p | P/H | X/C | C _p | P/H |
|--------|----------------|-------|--------|----------------|-------|
| 0.0119 | 0.293 | 0.742 | 0.0119 | 0.294 | 0.742 |
| 0.0371 | -0.122 | 0.620 | 0.0371 | -0.122 | 0.620 |
| 0.0623 | -0.540 | 0.497 | 0.0623 | -0.538 | 0.498 |
| 0.0872 | -2.525 | 0.501 | 0.0872 | -0.526 | 0.502 |
| 0.1122 | -2.605 | 0.478 | 0.1122 | -0.607 | 0.478 |
| 0.1372 | -2.635 | 0.469 | 0.1372 | -0.630 | 0.469 |
| 0.1623 | -2.674 | 0.457 | 0.1623 | -0.676 | 0.457 |
| 0.1872 | -2.700 | 0.450 | 0.1872 | -0.704 | 0.449 |
| 0.2122 | -2.765 | 0.431 | 0.2122 | -0.766 | 0.441 |
| 0.2372 | -2.802 | 0.420 | 0.2372 | -0.805 | 0.419 |
| 0.2620 | -2.842 | 0.409 | 0.2620 | -0.844 | 0.400 |
| 0.2872 | -2.852 | 0.405 | 0.2872 | -0.857 | 0.404 |
| 0.3122 | -2.803 | 0.396 | 0.3122 | -0.888 | 0.395 |
| 0.3375 | -2.876 | 0.390 | 0.3375 | -0.882 | 0.397 |
| 0.3610 | -2.892 | 0.394 | 0.3610 | -0.890 | 0.392 |
| 0.3873 | -2.918 | 0.386 | 0.3873 | -0.925 | 0.384 |
| 0.4124 | -2.946 | 0.378 | 0.4124 | -0.956 | 0.375 |
| 0.4371 | -2.930 | 0.380 | 0.4371 | -0.947 | 0.370 |
| 0.4621 | -2.969 | 0.371 | 0.4621 | -0.970 | 0.369 |
| 0.4871 | -2.956 | 0.393 | 0.4871 | -0.941 | 0.350 |
| 0.5120 | -2.908 | 0.403 | 0.5120 | -0.640 | 0.468 |
| 0.5371 | -2.414 | 0.534 | 0.5371 | -2.420 | 0.530 |
| 0.5621 | -0.335 | 0.557 | 0.5621 | -0.317 | 0.554 |
| 0.5870 | -2.207 | 0.570 | 0.5870 | -0.297 | 0.569 |
| 0.6122 | -2.242 | 0.585 | 0.6122 | -0.250 | 0.593 |
| 0.6371 | -2.210 | 0.594 | 0.6371 | -0.216 | 0.593 |
| 0.6601 | -0.171 | 0.606 | 0.6601 | -0.176 | 0.604 |
| 0.6869 | -2.153 | 0.611 | 0.6869 | -0.157 | 0.610 |
| 0.7121 | -0.124 | 0.619 | 0.7121 | -0.120 | 0.610 |
| 0.7369 | -0.101 | 0.626 | 0.7369 | -0.103 | 0.626 |
| 0.7620 | -0.075 | 0.634 | 0.7620 | -0.070 | 0.634 |
| 0.7870 | -2.050 | 0.641 | 0.7870 | -0.050 | 0.641 |
| 0.8117 | -2.024 | 0.649 | 0.8117 | -0.025 | 0.649 |
| 0.8370 | -2.002 | 0.655 | 0.8370 | -0.001 | 0.655 |
| 0.8619 | 0.027 | 0.664 | 0.8619 | 0.026 | 0.663 |
| 0.8869 | 0.053 | 0.671 | 0.8869 | 0.053 | 0.671 |
| 0.9120 | 0.085 | 0.681 | 0.9120 | 0.083 | 0.680 |
| 0.9416 | 0.107 | 0.687 | 0.9416 | 0.105 | 0.657 |

NACA-0012 293.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.839 ALPHA 0.50 REYNOLDS 1.69*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.8644 CM = -0.0221

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0119 | 0.237 | 2.725 | 0.0119 | 0.354 | 0.759 |
| 0.0371 | -0.100 | 2.686 | 0.0371 | -0.070 | 0.634 |
| 0.0623 | -0.667 | 2.459 | 0.0623 | -0.461 | 0.519 |
| 0.0872 | -0.563 | 2.498 | 0.0872 | -0.474 | 0.515 |
| 0.1122 | -0.674 | 2.465 | 0.1122 | -0.547 | 0.493 |
| 0.1372 | -0.689 | 2.453 | 0.1372 | -0.573 | 0.485 |
| 0.1623 | -0.731 | 2.440 | 0.1623 | -0.605 | 0.476 |
| 0.1872 | -0.757 | 2.433 | 0.1872 | -0.657 | 0.461 |
| 0.2122 | -0.815 | 2.416 | 0.2122 | -0.727 | 0.440 |
| 0.2372 | -0.844 | 2.407 | 0.2372 | -0.762 | 0.438 |
| 0.2623 | -0.883 | 2.396 | 0.2623 | -0.802 | 0.416 |
| 0.2872 | -0.944 | 2.393 | 0.2872 | -0.815 | 0.414 |
| 0.3122 | -0.924 | 2.384 | 0.3122 | -0.843 | 0.436 |
| 0.3375 | -0.916 | 2.386 | 0.3375 | -0.835 | 0.420 |
| 0.3619 | -0.934 | 2.381 | 0.3619 | -0.849 | 0.404 |
| 0.3873 | -0.958 | 2.374 | 0.3873 | -0.875 | 0.396 |
| 0.4124 | -0.941 | 2.364 | 0.4124 | -0.906 | 0.388 |
| 0.4371 | -0.901 | 2.367 | 0.4371 | -0.893 | 0.391 |
| 0.4621 | -1.214 | 2.357 | 0.4621 | -0.920 | 0.383 |
| 0.4871 | -1.268 | 2.361 | 0.4871 | -0.849 | 0.424 |
| 0.5120 | -0.759 | 2.432 | 0.5120 | -0.526 | 0.499 |
| 0.5371 | -0.446 | 2.510 | 0.5371 | -0.373 | 0.545 |
| 0.5621 | -0.400 | 2.530 | 0.5621 | -0.311 | 0.563 |
| 0.5870 | -0.349 | 2.553 | 0.5870 | -0.271 | 0.575 |
| 0.6122 | -0.302 | 2.567 | 0.6122 | -0.229 | 0.587 |
| 0.6371 | -0.260 | 2.579 | 0.6371 | -0.205 | 0.594 |
| 0.6661 | -0.213 | 2.593 | 0.6661 | -0.171 | 0.604 |
| 0.6869 | -0.186 | 2.601 | 0.6869 | -0.161 | 0.607 |
| 0.7121 | -0.171 | 2.611 | 0.7121 | -0.138 | 0.614 |
| 0.7369 | -0.119 | 2.620 | 0.7369 | -0.116 | 0.620 |
| 0.7620 | -0.087 | 2.630 | 0.7620 | -0.095 | 0.627 |
| 0.7870 | -0.050 | 2.638 | 0.7870 | -0.072 | 0.633 |
| 0.8117 | -0.027 | 2.647 | 0.8117 | -0.049 | 0.640 |
| 0.8370 | -0.003 | 2.655 | 0.8370 | -0.027 | 0.646 |
| 0.8619 | 0.027 | 2.663 | 0.8619 | 0.002 | 0.655 |
| 0.8869 | 0.055 | 2.672 | 0.8869 | 0.027 | 0.663 |
| 0.9120 | 0.085 | 2.681 | 0.9120 | 0.058 | 0.672 |
| 0.9416 | 0.105 | 2.687 | 0.9416 | 0.081 | 0.670 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.800 ALPHA 1.00 REYNOLDS 1.50E+6

INTEGRATED FORCE COEFFICIENTS

C_L = 0.8933 C_M = -0.0002

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|----------------|-------|----------------------|----------------|-------|
| X/C | C _P | P/H | X/C | C _P | P/H |
| 0.0119 | 0.188 | 0.718 | 0.0119 | 0.485 | 0.774 |
| 0.0371 | -0.206 | 0.594 | 0.0371 | -0.829 | 0.647 |
| 0.0623 | -0.729 | 0.441 | 0.0623 | -0.483 | 0.536 |
| 0.0872 | -0.626 | 0.471 | 0.0872 | -0.428 | 0.529 |
| 0.1122 | -0.672 | 0.457 | 0.1122 | -0.498 | 0.508 |
| 0.1372 | -0.711 | 0.446 | 0.1372 | -0.522 | 0.501 |
| 0.1623 | -0.767 | 0.429 | 0.1623 | -0.568 | 0.488 |
| 0.1872 | -0.796 | 0.421 | 0.1872 | -0.626 | 0.471 |
| 0.2122 | -0.851 | 0.404 | 0.2122 | -0.695 | 0.458 |
| 0.2372 | -0.888 | 0.395 | 0.2372 | -0.732 | 0.439 |
| 0.2528 | -0.919 | 0.384 | 0.2528 | -0.771 | 0.420 |
| 0.2672 | -0.928 | 0.351 | 0.2672 | -0.731 | 0.425 |
| 0.3122 | -0.959 | 0.372 | 0.3122 | -0.789 | 0.416 |
| 0.3375 | -0.958 | 0.375 | 0.3375 | -0.800 | 0.419 |
| 0.3618 | -0.969 | 0.369 | 0.3618 | -0.813 | 0.415 |
| 0.3873 | -0.993 | 0.362 | 0.3873 | -0.838 | 0.408 |
| 0.4124 | -1.025 | 0.353 | 0.4124 | -0.864 | 0.400 |
| 0.4371 | -1.014 | 0.356 | 0.4371 | -0.853 | 0.404 |
| 0.4621 | -1.039 | 0.349 | 0.4621 | -0.888 | 0.395 |
| 0.4871 | -0.822 | 0.413 | 0.4871 | -0.824 | 0.412 |
| 0.5120 | -0.542 | 0.495 | 0.5120 | -0.688 | 0.478 |
| 0.5371 | -0.438 | 0.526 | 0.5371 | -0.398 | 0.548 |
| 0.5621 | -0.396 | 0.538 | 0.5621 | -0.313 | 0.563 |
| 0.5870 | -0.364 | 0.548 | 0.5870 | -0.252 | 0.572 |
| 0.6122 | -0.333 | 0.557 | 0.6122 | -0.243 | 0.583 |
| 0.6371 | -0.306 | 0.565 | 0.6371 | -0.228 | 0.588 |
| 0.6661 | -0.272 | 0.575 | 0.6661 | -0.202 | 0.595 |
| 0.6869 | -0.249 | 0.591 | 0.6869 | -0.195 | 0.597 |
| 0.7121 | -0.219 | 0.593 | 0.7121 | -0.175 | 0.603 |
| 0.7369 | -0.198 | 0.599 | 0.7369 | -0.157 | 0.608 |
| 0.7620 | -0.178 | 0.608 | 0.7620 | -0.138 | 0.614 |
| 0.7870 | -0.128 | 0.617 | 0.7870 | -0.118 | 0.620 |
| 0.8117 | -0.093 | 0.627 | 0.8117 | -0.094 | 0.626 |
| 0.8370 | -0.063 | 0.636 | 0.8370 | -0.079 | 0.631 |
| 0.8619 | -0.032 | 0.645 | 0.8619 | -0.055 | 0.638 |
| 0.8869 | -0.004 | 0.654 | 0.8869 | -0.033 | 0.645 |
| 0.9120 | 0.026 | 0.662 | 0.9120 | -0.006 | 0.653 |
| 0.9416 | 0.046 | 0.668 | 0.9416 | 0.015 | 0.659 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

NACA NO. 0.828 ALPHA 0.80 REY 1.42⁶10

INTEGRATED FORCE COEFFICIENTS

CN = 0.9374 CN = -0.8151

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | 0.321 | 0.748 | 0.0119 | 0.319 | 0.748 |
| 0.0371 | -0.298 | 0.615 | 0.0371 | -0.298 | 0.616 |
| 0.0623 | -0.513 | 0.487 | 0.0623 | -0.507 | 0.498 |
| 0.0872 | -0.484 | 0.496 | 0.0872 | -0.485 | 0.497 |
| 0.1122 | -0.569 | 0.470 | 0.1122 | -0.565 | 0.473 |
| 0.1372 | -0.657 | 0.458 | 0.1372 | -0.667 | 0.468 |
| 0.1623 | -0.645 | 0.447 | 0.1623 | -0.652 | 0.447 |
| 0.1872 | -0.678 | 0.437 | 0.1872 | -0.680 | 0.438 |
| 0.2122 | -0.736 | 0.419 | 0.2122 | -0.739 | 0.422 |
| 0.2372 | -0.771 | 0.488 | 0.2372 | -0.773 | 0.418 |
| 0.2623 | -0.809 | 0.397 | 0.2623 | -0.814 | 0.398 |
| 0.2872 | -0.826 | 0.392 | 0.2872 | -0.828 | 0.393 |
| 0.3122 | -0.892 | 0.384 | 0.3122 | -0.897 | 0.365 |
| 0.3375 | -0.894 | 0.383 | 0.3375 | -0.896 | 0.365 |
| 0.3618 | -0.869 | 0.379 | 0.3618 | -0.875 | 0.379 |
| 0.3873 | -0.899 | 0.373 | 0.3873 | -0.901 | 0.371 |
| 0.4124 | -0.927 | 0.361 | 0.4124 | -0.933 | 0.361 |
| 0.4371 | -0.926 | 0.361 | 0.4371 | -0.928 | 0.363 |
| 0.4621 | -0.955 | 0.353 | 0.4621 | -0.959 | 0.354 |
| 0.4871 | -0.968 | 0.351 | 0.4871 | -0.931 | 0.362 |
| 0.5128 | -0.973 | 0.347 | 0.5128 | -0.844 | 0.388 |
| 0.5371 | -0.954 | 0.353 | 0.5371 | -0.742 | 0.419 |
| 0.5621 | -0.895 | 0.371 | 0.5621 | -0.657 | 0.445 |
| 0.5878 | -0.792 | 0.402 | 0.5878 | -0.581 | 0.468 |
| 0.6122 | -0.679 | 0.436 | 0.6122 | -0.516 | 0.488 |
| 0.6371 | -0.548 | 0.476 | 0.6371 | -0.464 | 0.523 |
| 0.6661 | -0.469 | 0.520 | 0.6661 | -0.422 | 0.516 |
| 0.6869 | -0.429 | 0.512 | 0.6869 | -0.396 | 0.524 |
| 0.7121 | -0.373 | 0.529 | 0.7121 | -0.363 | 0.534 |
| 0.7369 | -0.351 | 0.536 | 0.7369 | -0.334 | 0.543 |
| 0.7628 | -0.348 | 0.539 | 0.7628 | -0.312 | 0.549 |
| 0.7878 | -0.331 | 0.542 | 0.7878 | -0.295 | 0.554 |
| 0.8117 | -0.314 | 0.547 | 0.8117 | -0.288 | 0.559 |
| 0.8378 | -0.304 | 0.553 | 0.8378 | -0.262 | 0.564 |
| 0.8619 | -0.291 | 0.554 | 0.8619 | -0.249 | 0.568 |
| 0.8869 | -0.276 | 0.558 | 0.8869 | -0.234 | 0.573 |
| 0.9128 | -0.264 | 0.562 | 0.9128 | -0.219 | 0.577 |
| 0.9416 | -0.241 | 0.569 | 0.9416 | -0.193 | 0.585 |

NACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.820 ALPHA 0.50 REV 1.42+10⁶

INTEGRATED FORCE COEFFICIENTS

C_N = -0.0363 C_M = 0.0207

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0119 | 0.274 | 0.726 | 0.0119 | 0.362 | 0.752 |
| 0.0371 | -0.129 | 0.604 | 0.0371 | -0.054 | 0.625 |
| 0.0623 | -0.603 | 0.460 | 0.0623 | -0.439 | 0.509 |
| 0.0872 | -0.515 | 0.467 | 0.0872 | -0.453 | 0.505 |
| 0.1122 | -0.603 | 0.460 | 0.1122 | -0.532 | 0.481 |
| 0.1372 | -0.645 | 0.447 | 0.1372 | -0.560 | 0.469 |
| 0.1623 | -0.688 | 0.434 | 0.1623 | -0.606 | 0.450 |
| 0.1872 | -0.719 | 0.425 | 0.1872 | -0.639 | 0.440 |
| 0.2122 | -0.773 | 0.409 | 0.2122 | -0.700 | 0.429 |
| 0.2372 | -0.806 | 0.399 | 0.2372 | -0.744 | 0.416 |
| 0.2620 | -0.841 | 0.380 | 0.2620 | -0.781 | 0.405 |
| 0.2872 | -0.857 | 0.363 | 0.2872 | -0.799 | 0.400 |
| 0.3122 | -0.883 | 0.375 | 0.3122 | -0.824 | 0.392 |
| 0.3375 | -0.884 | 0.375 | 0.3375 | -0.829 | 0.391 |
| 0.3618 | -0.909 | 0.370 | 0.3618 | -0.843 | 0.386 |
| 0.3873 | -0.920 | 0.362 | 0.3873 | -0.873 | 0.377 |
| 0.4124 | -0.951 | 0.355 | 0.4124 | -0.901 | 0.368 |
| 0.4371 | -0.954 | 0.354 | 0.4371 | -0.899 | 0.369 |
| 0.4621 | -0.981 | 0.345 | 0.4621 | -0.929 | 0.368 |
| 0.4871 | -0.975 | 0.347 | 0.4871 | -0.935 | 0.350 |
| 0.5120 | -0.910 | 0.364 | 0.5120 | -0.931 | 0.353 |
| 0.5371 | -0.798 | 0.401 | 0.5371 | -0.949 | 0.354 |
| 0.5621 | -0.648 | 0.446 | 0.5621 | -0.944 | 0.356 |
| 0.5870 | -0.555 | 0.474 | 0.5870 | -0.961 | 0.350 |
| 0.6122 | -0.454 | 0.505 | 0.6122 | -0.951 | 0.353 |
| 0.6371 | -0.396 | 0.523 | 0.6371 | -0.938 | 0.357 |
| 0.6661 | -0.309 | 0.531 | 0.6661 | -0.710 | 0.427 |
| 0.6869 | -0.300 | 0.534 | 0.6869 | -0.532 | 0.481 |
| 0.7121 | -0.346 | 0.537 | 0.7121 | -0.441 | 0.500 |
| 0.7369 | -0.335 | 0.541 | 0.7369 | -0.380 | 0.524 |
| 0.7623 | -0.330 | 0.542 | 0.7623 | -0.362 | 0.532 |
| 0.7870 | -0.316 | 0.546 | 0.7870 | -0.344 | 0.537 |
| 0.8117 | -0.311 | 0.540 | 0.8117 | -0.330 | 0.542 |
| 0.8370 | -0.298 | 0.552 | 0.8370 | -0.315 | 0.546 |
| 0.8619 | -0.206 | 0.556 | 0.8619 | -0.306 | 0.549 |
| 0.8859 | -0.200 | 0.561 | 0.8859 | -0.293 | 0.553 |
| 0.9120 | -0.257 | 0.564 | 0.9120 | -0.200 | 0.557 |
| 0.9416 | -0.240 | 0.570 | 0.9416 | -0.259 | 0.563 |

VACA-0012 203.2 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.821 ALPHA 1.80 REYNOLDS 1.42E+10

INTEGRATED FORCE COEFFICIENTS

CM = -0.0605 CH = 0.0493

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0119 | 0.234 | 0.713 | 0.0119 | 0.484 | 0.766 |
| 0.0371 | -0.199 | 0.594 | 0.0371 | -0.022 | 0.637 |
| 0.0623 | -0.601 | 0.442 | 0.0623 | -0.389 | 0.526 |
| 0.0872 | -0.553 | 0.475 | 0.0872 | -0.419 | 0.517 |
| 0.1122 | -0.610 | 0.455 | 0.1122 | -0.496 | 0.493 |
| 0.1372 | -0.664 | 0.441 | 0.1372 | -0.528 | 0.484 |
| 0.1623 | -0.714 | 0.426 | 0.1623 | -0.561 | 0.474 |
| 0.1872 | -0.748 | 0.416 | 0.1872 | -0.612 | 0.458 |
| 0.2122 | -0.803 | 0.399 | 0.2122 | -0.681 | 0.437 |
| 0.2372 | -0.834 | 0.390 | 0.2372 | -0.720 | 0.425 |
| 0.2628 | -0.878 | 0.379 | 0.2628 | -0.759 | 0.414 |
| 0.2872 | -0.884 | 0.375 | 0.2872 | -0.777 | 0.408 |
| 0.3122 | -0.900 | 0.368 | 0.3122 | -0.805 | 0.400 |
| 0.3375 | -0.910 | 0.367 | 0.3375 | -0.805 | 0.400 |
| 0.3619 | -0.926 | 0.362 | 0.3619 | -0.828 | 0.395 |
| 0.3873 | -0.952 | 0.354 | 0.3873 | -0.849 | 0.386 |
| 0.4124 | -0.980 | 0.346 | 0.4124 | -0.879 | 0.378 |
| 0.4371 | -0.978 | 0.349 | 0.4371 | -0.875 | 0.379 |
| 0.4621 | -0.934 | 0.368 | 0.4621 | -0.905 | 0.369 |
| 0.4871 | -0.794 | 0.402 | 0.4871 | -0.989 | 0.360 |
| 0.5123 | -0.663 | 0.442 | 0.5123 | -0.926 | 0.363 |
| 0.5371 | -0.569 | 0.478 | 0.5371 | -0.922 | 0.354 |
| 0.5621 | -0.493 | 0.493 | 0.5621 | -0.916 | 0.356 |
| 0.5870 | -0.426 | 0.514 | 0.5870 | -0.933 | 0.351 |
| 0.6122 | -0.390 | 0.524 | 0.6122 | -0.923 | 0.364 |
| 0.6371 | -0.376 | 0.529 | 0.6371 | -0.935 | 0.363 |
| 0.6661 | -0.350 | 0.536 | 0.6661 | -0.905 | 0.369 |
| 0.6869 | -0.343 | 0.539 | 0.6869 | -0.861 | 0.353 |
| 0.7121 | -0.333 | 0.542 | 0.7121 | -0.789 | 0.428 |
| 0.7369 | -0.325 | 0.544 | 0.7369 | -0.518 | 0.486 |
| 0.7628 | -0.314 | 0.547 | 0.7628 | -0.429 | 0.513 |
| 0.7878 | -0.301 | 0.551 | 0.7878 | -0.371 | 0.531 |
| 0.8117 | -0.294 | 0.553 | 0.8117 | -0.338 | 0.541 |
| 0.8378 | -0.277 | 0.559 | 0.8378 | -0.312 | 0.548 |
| 0.8619 | -0.264 | 0.563 | 0.8619 | -0.292 | 0.554 |
| 0.8869 | -0.258 | 0.567 | 0.8869 | -0.281 | 0.558 |
| 0.9120 | -0.250 | 0.570 | 0.9120 | -0.264 | 0.563 |
| 0.9416 | -0.217 | 0.577 | 0.9416 | -0.235 | 0.572 |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.500 ALPHA 0.00 REYNOLDS 0.93x10⁶

INTEGRATED FORCE COEFFICIENTS

C_L = 0.0252 C_D = 0.0034

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0287 | -0.208 | 0.012 | 0.0757 | -0.465 | 0.774 |
| 0.0707 | -0.508 | 0.760 | 0.1294 | -0.500 | 0.769 |
| 0.1299 | -0.489 | 0.771 | 0.1838 | -0.465 | 0.774 |
| 0.1813 | -0.477 | 0.772 | 0.2368 | -0.441 | 0.778 |
| 0.2387 | -0.450 | 0.776 | 0.2864 | -0.403 | 0.783 |
| 0.2836 | -0.408 | 0.783 | 0.3383 | -0.361 | 0.790 |
| 0.3337 | -0.378 | 0.787 | 0.3904 | -0.333 | 0.794 |
| 0.3858 | -0.348 | 0.793 | 0.4376 | -0.294 | 0.800 |
| 0.4359 | -0.307 | 0.798 | 0.4870 | -0.265 | 0.824 |
| 0.4862 | -0.275 | 0.802 | 0.5371 | -0.217 | 0.811 |
| 0.5388 | -0.227 | 0.809 | 0.5879 | -0.197 | 0.814 |
| 0.5882 | -0.188 | 0.815 | 0.6371 | -0.162 | 0.819 |
| 0.6395 | -0.164 | 0.819 | 0.6874 | -0.141 | 0.822 |
| 0.6894 | -0.129 | 0.824 | 0.7359 | -0.118 | 0.827 |
| 0.7443 | -0.094 | 0.829 | | | |
| 0.7954 | -0.055 | 0.835 | | | |
| 0.8414 | -0.019 | 0.848 | | | |
| 0.8891 | 0.026 | 0.847 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.499 ALPHA 1.00 REYNOLDS 0.63x10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1091 CM = 0.0012

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0207 | -0.493 | 0.771 | 0.0757 | -0.386 | 0.790 |
| 0.0707 | -0.477 | 0.744 | 0.1294 | -0.392 | 0.787 |
| 0.1299 | -0.615 | 0.753 | 0.1830 | -0.371 | 0.759 |
| 0.1813 | -0.575 | 0.759 | 0.2360 | -0.364 | 0.790 |
| 0.2307 | -0.531 | 0.765 | 0.2864 | -0.339 | 0.793 |
| 0.2836 | -0.476 | 0.773 | 0.3363 | -0.300 | 0.790 |
| 0.3337 | -0.437 | 0.779 | 0.3924 | -0.287 | 0.821 |
| 0.3858 | -0.393 | 0.785 | 0.4376 | -0.254 | 0.826 |
| 0.4350 | -0.352 | 0.791 | 0.4870 | -0.230 | 0.809 |
| 0.4860 | -0.312 | 0.797 | 0.5371 | -0.189 | 0.815 |
| 0.5380 | -0.270 | 0.805 | 0.5879 | -0.174 | 0.810 |
| 0.5882 | -0.215 | 0.812 | 0.6371 | -0.144 | 0.822 |
| 0.6395 | -0.184 | 0.816 | 0.6874 | -0.127 | 0.825 |
| 0.6894 | -0.144 | 0.822 | 0.7359 | -0.096 | 0.829 |
| 0.7443 | -0.105 | 0.820 | | | |
| 0.7954 | -0.062 | 0.834 | | | |
| 0.8414 | -0.024 | 0.840 | | | |
| 0.8891 | 0.022 | 0.846 | | | |

NA-2-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.500 ALPHA 2.00 REY 0.03+10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2247 CH = 0.0024

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0227 | -0.022 | 0.722 | 0.0757 | -0.146 | 0.021 |
| 0.0797 | -0.005 | 0.715 | 0.1294 | -0.200 | 0.004 |
| 0.1299 | -0.751 | 0.732 | 0.1830 | -0.273 | 0.002 |
| 0.1813 | -0.601 | 0.742 | 0.2360 | -0.283 | 0.001 |
| 0.2307 | -0.610 | 0.752 | 0.2864 | -0.271 | 0.003 |
| 0.2836 | -0.547 | 0.762 | 0.3383 | -0.240 | 0.006 |
| 0.3337 | -0.495 | 0.770 | 0.3904 | -0.230 | 0.000 |
| 0.3850 | -0.442 | 0.770 | 0.4376 | -0.211 | 0.012 |
| 0.4350 | -0.393 | 0.785 | 0.4870 | -0.195 | 0.014 |
| 0.4860 | -0.340 | 0.791 | 0.5371 | -0.157 | 0.020 |
| 0.5380 | -0.291 | 0.800 | 0.5879 | -0.145 | 0.021 |
| 0.5902 | -0.241 | 0.807 | 0.6371 | -0.117 | 0.026 |
| 0.6395 | -0.205 | 0.813 | 0.6874 | -0.105 | 0.027 |
| 0.6894 | -0.102 | 0.819 | 0.7359 | -0.079 | 0.031 |
| 0.7443 | -0.119 | 0.825 | | | |
| 0.7954 | -0.071 | 0.832 | | | |
| 0.8414 | -0.020 | 0.839 | | | |
| 0.8891 | 0.023 | 0.846 | | | |

NACA-BB12 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.520 ALPHA 3.00 REYNOLDS 0.84e10

INTEGRATED FORCE COEFFICIENTS

CN = 0.3361 CM = 0.0040

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0207 | -1.101 | 0.660 | 0.0777 | -0.005 | 0.042 |
| 0.0787 | -1.206 | 0.667 | 0.1796 | -0.140 | 0.021 |
| 0.1299 | -0.894 | 0.711 | 0.2700 | -0.101 | 0.016 |
| 0.1813 | -0.794 | 0.726 | 0.3600 | -0.206 | 0.012 |
| 0.2327 | -0.710 | 0.738 | 0.4500 | -0.203 | 0.012 |
| 0.2836 | -0.629 | 0.751 | 0.5400 | -0.193 | 0.014 |
| 0.3337 | -0.526 | 0.761 | 0.6300 | -0.100 | 0.015 |
| 0.3850 | -0.402 | 0.770 | 0.7200 | -0.167 | 0.010 |
| 0.4358 | -0.435 | 0.778 | 0.8100 | -0.155 | 0.020 |
| 0.4868 | -0.303 | 0.786 | 0.9000 | -0.123 | 0.024 |
| 0.5382 | -0.317 | 0.796 | 0.9900 | -0.117 | 0.025 |
| 0.5882 | -0.264 | 0.804 | 0.0000 | -0.097 | 0.020 |
| 0.6395 | -0.224 | 0.812 | 0.0000 | -0.090 | 0.029 |
| 0.6894 | -0.179 | 0.816 | 0.0000 | -0.060 | 0.032 |
| 0.7443 | -0.100 | 0.823 | | | |
| 0.7954 | -0.277 | 0.831 | | | |
| 0.8414 | -0.232 | 0.830 | | | |
| 0.8891 | 0.021 | 0.846 | | | |

NACA-8912 181.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.531 ALPHA 4.80 REY 0.83*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.4451 CM = 0.0073

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0207 | -1.501 | 0.688 | 0.0757 | 0.121 | 0.868 |
| 0.0787 | -1.250 | 0.657 | 0.1294 | -0.045 | 0.836 |
| 0.1299 | -1.026 | 0.698 | 0.1830 | -0.091 | 0.829 |
| 0.1813 | -0.897 | 0.709 | 0.2360 | -0.131 | 0.823 |
| 0.2307 | -0.794 | 0.72 | 0.2864 | -0.137 | 0.822 |
| 0.2836 | -0.694 | 0.740 | 0.3383 | -0.135 | 0.823 |
| 0.3337 | -0.612 | 0.751 | 0.3904 | -0.136 | 0.822 |
| 0.3858 | -0.541 | 0.762 | 0.4376 | -0.125 | 0.824 |
| 0.4359 | -0.478 | 0.771 | 0.4870 | -0.119 | 0.825 |
| 0.4868 | -0.418 | 0.780 | 0.5371 | -0.093 | 0.829 |
| 0.5388 | -0.346 | 0.791 | 0.5879 | -0.098 | 0.829 |
| 0.5882 | -0.287 | 0.803 | 0.6371 | -0.071 | 0.832 |
| 0.6395 | -0.238 | 0.807 | 0.6874 | -0.066 | 0.833 |
| 0.6894 | -0.188 | 0.814 | 0.7359 | -0.048 | 0.835 |
| 0.7443 | -0.132 | 0.823 | | | |
| 0.7954 | -0.080 | 0.830 | | | |
| 0.8414 | -0.033 | 0.837 | | | |
| 0.8891 | 0.010 | 0.845 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.630 ALPHA 8.00 REYNOLDS 0.82x10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.8211 CM = 0.0848

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.8287 | -0.108 | 0.747 | 0.0757 | -0.525 | 0.685 |
| 2.8797 | -0.557 | 0.677 | 0.1294 | -0.549 | 0.676 |
| 0.1299 | -0.550 | 0.679 | 0.1830 | -0.511 | 0.684 |
| 0.1813 | -0.523 | 0.680 | 0.2368 | -0.484 | 0.689 |
| 0.2327 | -0.492 | 0.686 | 0.2864 | -0.445 | 0.697 |
| 0.2836 | -0.445 | 0.696 | 0.3383 | -0.401 | 0.705 |
| 0.3337 | -0.409 | 0.703 | 0.3904 | -0.369 | 0.712 |
| 0.3850 | -0.369 | 0.711 | 0.4376 | -0.324 | 0.720 |
| 0.4358 | -0.331 | 0.719 | 0.4870 | -0.291 | 0.727 |
| 0.4860 | -0.296 | 0.726 | 0.5371 | -0.243 | 0.736 |
| 0.5368 | -0.243 | 0.736 | 0.5879 | -0.213 | 0.742 |
| 0.5882 | -0.200 | 0.745 | 0.6371 | -0.176 | 0.750 |
| 0.6395 | -0.170 | 0.751 | 0.6874 | -0.149 | 0.755 |
| 0.6894 | -0.134 | 0.759 | 0.7359 | -0.112 | 0.762 |
| 0.7443 | -0.097 | 0.765 | | | |
| 0.7954 | -0.056 | 0.773 | | | |
| 0.8414 | -0.016 | 0.781 | | | |
| 0.8891 | 0.029 | 0.790 | | | |

NACA-0012 1/1.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.598 ALPHA 1.00 REY 0.82×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.1194$ $C_M = 0.0023$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0227 | -0.498 | 0.689 | 0.0757 | -0.322 | 0.722 |
| 0.0787 | -0.739 | 0.648 | 0.1294 | -0.410 | 0.785 |
| 0.1299 | -0.688 | 0.658 | 0.1838 | -0.404 | 0.786 |
| 0.1813 | -0.646 | 0.658 | 0.2368 | -0.397 | 0.788 |
| 0.2387 | -0.542 | 0.669 | 0.2864 | -0.373 | 0.712 |
| 0.2836 | -0.528 | 0.682 | 0.3383 | -0.339 | 0.719 |
| 0.3337 | -0.479 | 0.691 | 0.3904 | -0.317 | 0.723 |
| 0.3858 | -0.429 | 0.701 | 0.4376 | -0.282 | 0.738 |
| 0.4358 | -0.384 | 0.710 | 0.4878 | -0.255 | 0.735 |
| 0.4868 | -0.348 | 0.719 | 0.5371 | -0.212 | 0.744 |
| 0.5368 | -0.278 | 0.731 | 0.5879 | -0.189 | 0.748 |
| 0.5852 | -0.230 | 0.748 | 0.6371 | -0.154 | 0.755 |
| 0.6395 | -0.193 | 0.748 | 0.6874 | -0.134 | 0.759 |
| 0.6894 | -0.151 | 0.756 | 0.7359 | -0.103 | 0.765 |
| 0.7443 | -0.108 | 0.764 | | | |
| 0.7954 | -0.064 | 0.773 | | | |
| 0.8414 | -0.023 | 0.782 | | | |
| 0.8891 | 0.029 | 0.791 | | | |

AD-A076 131

AERONAUTICAL RESEARCH LABS MELBOURNE (AUSTRALIA)

F/G 20/4

TRANSONIC WIND TUNNEL TESTS ON A SERIES OF TWO-DIMENSIONAL AERO--ETC(U)

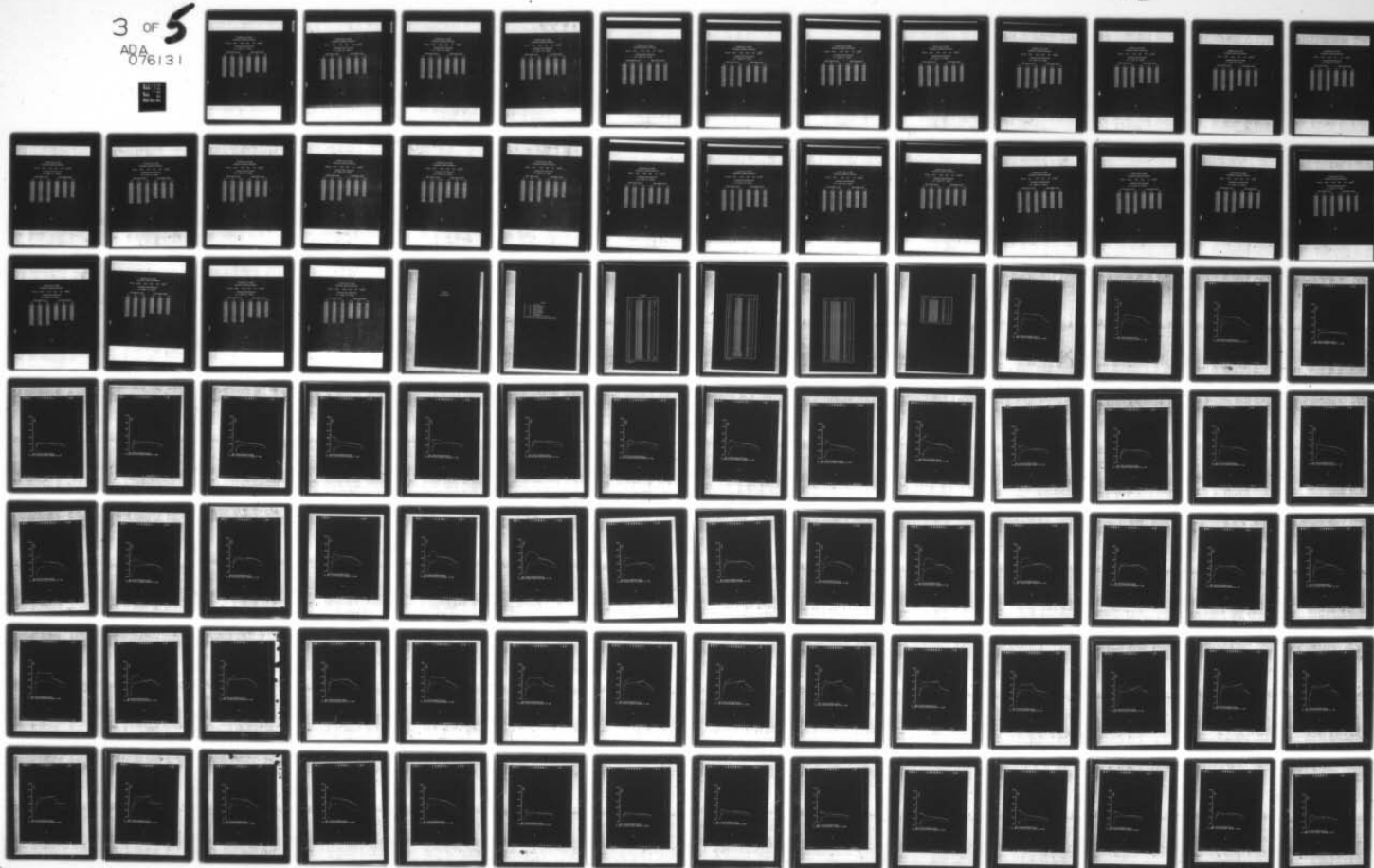
JAN 79 B D FAIRLIE , N POLLOCK

UNCLASSIFIED

ARL/AERO NOTE-384

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NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.598 ALPHA 2.08 REY 8.82×10^6

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.2397$ $C_M = 0.0447$

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 2.8287 | -0.815 | 2.624 | 0.0797 | -0.161 | 0.753 |
| 2.8787 | -0.952 | 2.597 | 0.1294 | -0.235 | 0.728 |
| 2.1299 | -0.843 | 2.619 | 0.1838 | -0.391 | 0.725 |
| 2.1813 | -2.762 | 2.635 | 0.2368 | -0.313 | 0.723 |
| 2.2387 | -2.686 | 2.650 | 0.2864 | -0.299 | 0.726 |
| 2.2336 | -0.686 | 2.665 | 0.3383 | -0.276 | 0.730 |
| 2.3337 | -0.542 | 2.678 | 0.3984 | -0.263 | 0.733 |
| 2.3858 | -0.488 | 2.698 | 0.4376 | -0.237 | 0.738 |
| 2.4359 | -0.422 | 2.782 | 0.4878 | -0.217 | 0.742 |
| 2.4968 | -0.374 | 2.711 | 0.5371 | -0.179 | 0.749 |
| 2.5383 | -0.388 | 2.724 | 0.5879 | -0.168 | 0.753 |
| 2.5882 | -0.255 | 2.734 | 0.6371 | -0.133 | 0.758 |
| 2.6375 | -0.212 | 2.743 | 0.6874 | -0.115 | 0.762 |
| 2.6894 | -0.169 | 2.751 | 0.7359 | -0.080 | 0.767 |
| 2.7443 | -0.121 | 2.761 | | | |
| 2.7954 | -0.071 | 2.771 | | | |
| 2.8414 | -0.024 | 2.783 | | | |
| 2.8891 | 0.228 | 2.793 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.620 ALPHA 3.00 REY 0.63×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.3591$ $C_M = 0.0067$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.0207 | -1.106 | 0.549 | 0.0757 | -0.011 | 0.781 |
| 2.0707 | -1.195 | 0.547 | 0.1294 | -0.165 | 0.751 |
| 2.1299 | -1.298 | 0.584 | 0.1830 | -0.281 | 0.744 |
| 2.1813 | -0.809 | 0.607 | 0.2368 | -0.238 | 0.738 |
| 2.2387 | -0.706 | 0.628 | 0.2864 | -0.228 | 0.738 |
| 2.2836 | -0.606 | 0.648 | 0.3353 | -0.217 | 0.740 |
| 2.3337 | -0.611 | 0.662 | 0.3904 | -0.209 | 0.742 |
| 2.3850 | -0.536 | 0.677 | 0.4376 | -0.189 | 0.746 |
| 2.4358 | -0.467 | 0.691 | 0.4870 | -0.175 | 0.749 |
| 2.4860 | -0.400 | 0.703 | 0.5371 | -0.143 | 0.755 |
| 2.5368 | -0.365 | 0.717 | 0.5879 | -0.133 | 0.757 |
| 2.5882 | -0.276 | 0.729 | 0.6371 | -0.112 | 0.761 |
| 2.6395 | -0.226 | 0.739 | 0.6874 | -0.099 | 0.764 |
| 2.6894 | -0.177 | 0.748 | 0.7359 | -0.075 | 0.769 |
| 2.7443 | -0.124 | 0.759 | | | |
| 2.7954 | -0.074 | 0.769 | | | |
| 2.8414 | -0.024 | 0.779 | | | |
| 2.8891 | 0.028 | 0.789 | | | |

VACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.650 ALPHA 0.00 REY 0.02+10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.0271 CM = 0.0041

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0287 | -0.178 | 0.712 | 0.0757 | -0.532 | 0.633 |
| 0.0787 | -0.578 | 0.625 | 0.1294 | -0.587 | 0.621 |
| 0.1299 | -0.576 | 0.624 | 0.1830 | -0.553 | 0.628 |
| 0.1813 | -0.571 | 0.625 | 0.2366 | -0.523 | 0.635 |
| 0.2337 | -0.537 | 0.632 | 0.2864 | -0.481 | 0.644 |
| 0.2936 | -0.486 | 0.644 | 0.3383 | -0.432 | 0.655 |
| 0.3337 | -0.446 | 0.652 | 0.3904 | -0.394 | 0.664 |
| 0.3858 | -0.408 | 0.663 | 0.4376 | -0.348 | 0.674 |
| 0.4358 | -0.358 | 0.672 | 0.4878 | -0.329 | 0.683 |
| 0.4868 | -0.318 | 0.681 | 0.5371 | -0.258 | 0.694 |
| 0.5389 | -0.282 | 0.693 | 0.5879 | -0.227 | 0.701 |
| 0.5882 | -0.218 | 0.703 | 0.6371 | -0.186 | 0.718 |
| 0.6395 | -0.182 | 0.711 | 0.6874 | -0.156 | 0.717 |
| 0.6894 | -0.144 | 0.728 | 0.7359 | -0.118 | 0.726 |
| 0.7443 | -0.101 | 0.729 | | | |
| 0.7954 | -0.058 | 0.739 | | | |
| 0.8414 | -0.016 | 0.748 | | | |
| 0.8891 | 0.033 | 0.759 | | | |

NACA-0012 181.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.658 ALPHA 1.32 REYNOLDS 8.82x10⁶

INTEGRATED FORCE COEFFICIENTS

C_L = 0.1271 C_D = 0.0038

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0287 | -0.465 | 8.648 | 0.0757 | -0.348 | 8.675 |
| 0.0787 | -0.798 | 8.576 | 0.1294 | -0.448 | 8.653 |
| 0.1299 | -0.743 | 8.586 | 0.1830 | -0.436 | 8.654 |
| 0.1813 | -0.784 | 8.595 | 0.2360 | -0.431 | 8.655 |
| 0.2327 | -0.642 | 8.689 | 0.2864 | -0.482 | 8.662 |
| 0.2836 | -0.569 | 8.625 | 0.3383 | -0.365 | 8.670 |
| 0.3337 | -0.515 | 8.637 | 0.3904 | -0.339 | 8.676 |
| 0.3858 | -0.458 | 8.658 | 0.4376 | -0.388 | 8.684 |
| 0.4358 | -0.485 | 8.661 | 0.4878 | -0.278 | 8.691 |
| 0.4868 | -0.357 | 8.672 | 0.5571 | -0.229 | 8.701 |
| 0.5383 | -0.293 | 8.686 | 0.5879 | -0.199 | 8.786 |
| 0.5582 | -0.243 | 8.698 | 0.6371 | -0.164 | 8.714 |
| 0.6395 | -0.282 | 8.797 | 0.6874 | -0.141 | 8.719 |
| 0.6894 | -0.168 | 8.716 | 0.7359 | -0.108 | 8.727 |
| 0.7443 | -0.112 | 8.727 | | | |
| 0.7954 | -0.204 | 8.737 | | | |
| 0.8414 | -0.016 | 8.748 | | | |
| 0.8891 | 0.032 | 8.759 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.649 ALPHA 2.00 REY 0.02018⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2521 CM = 0.0073

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0207 | -0.790 | 0.579 | 0.0757 | -0.166 | 0.716 |
| 0.0707 | -1.048 | 0.519 | 0.1294 | -0.290 | 0.686 |
| 0.1299 | -0.921 | 0.540 | 0.1830 | -0.320 | 0.682 |
| 0.1813 | -0.835 | 0.567 | 0.2368 | -0.335 | 0.678 |
| 0.2307 | -0.740 | 0.586 | 0.2864 | -0.321 | 0.681 |
| 0.2836 | -0.645 | 0.609 | 0.3383 | -0.298 | 0.687 |
| 0.3337 | -0.576 | 0.625 | 0.3904 | -0.263 | 0.690 |
| 0.3858 | -0.508 | 0.640 | 0.4376 | -0.252 | 0.697 |
| 0.4350 | -0.446 | 0.654 | 0.4870 | -0.230 | 0.702 |
| 0.4860 | -0.388 | 0.666 | 0.5371 | -0.191 | 0.718 |
| 0.5380 | -0.317 | 0.682 | 0.5879 | -0.173 | 0.714 |
| 0.5882 | -0.263 | 0.694 | 0.6371 | -0.143 | 0.721 |
| 0.6395 | -0.217 | 0.704 | 0.6874 | -0.123 | 0.726 |
| 0.6894 | -0.169 | 0.715 | 0.7359 | -0.093 | 0.732 |
| 0.7443 | -0.117 | 0.727 | | | |
| 0.7954 | -0.066 | 0.738 | | | |
| 0.8414 | -0.017 | 0.749 | | | |
| 0.8891 | 0.035 | 0.761 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.649 ALPHA 3.89 REYNOLDS 0.62-10⁶

INTEGRATED FORCE COEFFICIENTS

C_N = 0.3920 C_M = 0.0148

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|----------------|-------|----------------------|----------------|-------|
| X/C | C _P | P/H | X/C | C _P | P/H |
| 0.0207 | -1.119 | 2.524 | 0.0757 | -0.018 | 0.749 |
| 0.0787 | -1.513 | 2.416 | 0.1294 | -0.175 | 0.714 |
| 0.1299 | -1.218 | 2.484 | 0.1830 | -0.217 | 0.735 |
| 0.1913 | -2.941 | 2.543 | 0.2368 | -0.246 | 0.698 |
| 0.2327 | -2.834 | 2.557 | 0.2864 | -0.244 | 0.698 |
| 0.2836 | -3.724 | 2.592 | 0.3303 | -0.233 | 0.701 |
| 0.3337 | -2.638 | 2.611 | 0.3904 | -0.227 | 0.702 |
| 2.3858 | -2.557 | 2.629 | 0.4376 | -0.294 | 0.707 |
| 2.4358 | -2.484 | 2.645 | 0.4870 | -0.189 | 0.711 |
| 2.4860 | -2.419 | 2.659 | 0.5371 | -0.156 | 0.718 |
| 2.5360 | -2.343 | 2.677 | 0.5879 | -0.144 | 0.721 |
| 2.5892 | -2.281 | 2.698 | 0.6371 | -0.118 | 0.726 |
| 2.6395 | -2.230 | 2.702 | 0.6874 | -0.105 | 0.729 |
| 2.6894 | -2.177 | 2.713 | 0.7359 | -0.081 | 0.735 |
| 2.7443 | -2.121 | 2.726 | | | |
| 2.7954 | -2.068 | 2.738 | | | |
| 2.8414 | -2.017 | 2.749 | | | |
| 2.8891 | -2.066 | 2.761 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.674 ALPHA 0.00 REV 0.02-10⁶

INTEGRATED FORCE COEFFICIENTS

C_x = 0.0263 C_m = 0.0045

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0207 | -0.161 | 0.699 | 0.0757 | -0.549 | 0.688 |
| 0.0787 | -0.501 | 0.681 | 0.1294 | -0.608 | 0.594 |
| 0.1299 | -0.592 | 0.598 | 0.1830 | -0.579 | 0.601 |
| 0.1813 | -0.591 | 0.598 | 0.2360 | -0.540 | 0.628 |
| 0.2387 | -0.595 | 0.607 | 0.2864 | -0.503 | 0.619 |
| 0.2836 | -0.502 | 0.619 | 0.3383 | -0.450 | 0.631 |
| 0.3337 | -0.459 | 0.629 | 0.3904 | -0.400 | 0.641 |
| 0.3856 | -0.413 | 0.643 | 0.4376 | -0.350 | 0.653 |
| 0.4358 | -0.369 | 0.650 | 0.4870 | -0.310 | 0.662 |
| 0.4863 | -0.323 | 0.660 | 0.5371 | -0.266 | 0.675 |
| 0.5363 | -0.268 | 0.674 | 0.5879 | -0.232 | 0.682 |
| 0.5882 | -0.223 | 0.685 | 0.6371 | -0.191 | 0.692 |
| 0.6395 | -0.186 | 0.693 | 0.6874 | -0.159 | 0.700 |
| 0.6894 | -0.146 | 0.703 | 0.7359 | -0.128 | 0.709 |
| 0.7443 | -0.102 | 0.713 | | | |
| 0.7954 | -0.059 | 0.723 | | | |
| 0.8414 | -0.015 | 0.733 | | | |
| 0.8891 | 0.034 | 0.745 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.073 ALPHA 1.00 REYNOLDS 0.82-10⁶

INTEGRATED FORCE COEFFICIENTS

C_L = 0.1319 C_D = 0.0044

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | C _P | P/H | X/C | C _P | P/H |
|--------|----------------|-------|--------|----------------|-------|
| 0.0207 | -0.444 | 0.634 | 0.0757 | -0.350 | 0.656 |
| 0.0707 | -0.015 | 0.547 | 0.1294 | -0.453 | 0.632 |
| 0.1299 | -0.775 | 0.557 | 0.1830 | -0.451 | 0.632 |
| 0.1813 | -0.785 | 0.566 | 0.2368 | -0.446 | 0.633 |
| 0.2307 | -0.669 | 0.581 | 0.2864 | -0.416 | 0.649 |
| 0.2836 | -0.592 | 0.599 | 0.3383 | -0.378 | 0.649 |
| 0.3337 | -0.532 | 0.614 | 0.3904 | -0.352 | 0.655 |
| 0.3853 | -0.473 | 0.627 | 0.4376 | -0.313 | 0.664 |
| 0.4358 | -0.416 | 0.641 | 0.4870 | -0.282 | 0.672 |
| 0.4868 | -0.366 | 0.652 | 0.5371 | -0.236 | 0.683 |
| 0.5380 | -0.301 | 0.668 | 0.5879 | -0.207 | 0.688 |
| 0.5892 | -0.249 | 0.680 | 0.6371 | -0.172 | 0.697 |
| 0.6395 | -0.205 | 0.690 | 0.6874 | -0.145 | 0.703 |
| 0.6894 | -0.161 | 0.700 | 0.7359 | -0.110 | 0.711 |
| 0.7443 | -0.111 | 0.712 | | | |
| 0.7954 | -0.092 | 0.723 | | | |
| 0.8414 | -0.015 | 0.734 | | | |
| 0.8891 | 0.004 | 0.746 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.674 ALPHA 2.00 REY 0.02010⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2773 CN = 0.0121

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0207 | -0.748 | 0.563 | 0.0797 | -0.172 | 0.697 |
| 0.0707 | -1.173 | 0.462 | 0.1294 | -0.318 | 0.664 |
| 0.1299 | -3.975 | 0.580 | 0.1830 | -0.334 | 0.659 |
| 0.1813 | -0.803 | 0.530 | 0.2368 | -0.348 | 0.655 |
| 0.2307 | -2.775 | 0.555 | 0.2864 | -0.333 | 0.659 |
| 0.2836 | -0.671 | 0.580 | 0.3383 | -0.307 | 0.665 |
| 0.3337 | -3.590 | 0.599 | 0.3904 | -0.291 | 0.669 |
| 0.3858 | -0.520 | 0.615 | 0.4376 | -0.261 | 0.676 |
| 0.4350 | -2.425 | 0.630 | 0.4870 | -0.235 | 0.682 |
| 0.4860 | -0.398 | 0.644 | 0.5371 | -0.199 | 0.690 |
| 0.5380 | -3.324 | 0.661 | 0.5879 | -0.179 | 0.695 |
| 0.5902 | -3.267 | 0.674 | 0.6371 | -0.148 | 0.702 |
| 0.6395 | -0.210 | 0.686 | 0.6874 | -0.126 | 0.708 |
| 0.6894 | -0.109 | 0.697 | 0.7359 | -0.097 | 0.714 |
| 0.7443 | -0.115 | 0.710 | | | |
| 0.7954 | -0.064 | 0.722 | | | |
| 0.8414 | -3.012 | 0.734 | | | |
| 0.8891 | 0.859 | 0.746 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.699 ALPHA 0.00 REY 0.82⁶10

INTEGRATED FORCE COEFFICIENTS

CN = 0.3283 CM = 0.0052

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0207 | -0.143 | 0.686 | 0.0757 | -0.562 | 0.583 |
| 0.0707 | -0.593 | 0.575 | 0.1294 | -0.631 | 0.566 |
| 0.1299 | -0.613 | 0.570 | 0.1830 | -0.606 | 0.572 |
| 0.1813 | -0.619 | 0.569 | 0.2360 | -0.575 | 0.588 |
| 0.2307 | -0.584 | 0.577 | 0.2864 | -0.526 | 0.592 |
| 0.2836 | -0.527 | 0.591 | 0.3383 | -0.470 | 0.626 |
| 0.3337 | -0.482 | 0.603 | 0.3904 | -0.426 | 0.617 |
| 0.3858 | -0.430 | 0.615 | 0.4376 | -0.374 | 0.629 |
| 0.4358 | -0.383 | 0.627 | 0.4870 | -0.320 | 0.641 |
| 0.4860 | -0.330 | 0.638 | 0.5371 | -0.275 | 0.654 |
| 0.5360 | -0.276 | 0.654 | 0.5879 | -0.239 | 0.663 |
| 0.5862 | -0.228 | 0.665 | 0.6371 | -0.196 | 0.673 |
| 0.6395 | -0.199 | 0.675 | 0.6874 | -0.162 | 0.682 |
| 0.6894 | -0.148 | 0.685 | 0.7359 | -0.123 | 0.691 |
| 0.7443 | -0.104 | 0.696 | | | |
| 0.7954 | -0.059 | 0.707 | | | |
| 0.8414 | -0.012 | 0.719 | | | |
| 0.8891 | 0.047 | 0.731 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.738 ALPHA 1.00 REY 0.62⁶15

INTEGRATED FORCE COEFFICIENTS

CV = 0.1428 CM = 0.0067

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0287 | -0.413 | 0.619 | 0.0757 | -0.358 | 0.635 |
| 0.0787 | -0.850 | 0.518 | 0.1294 | -0.463 | 0.687 |
| 0.1299 | -0.813 | 0.528 | 0.1838 | -0.478 | 0.625 |
| 0.1813 | -0.783 | 0.527 | 0.2368 | -0.466 | 0.686 |
| 0.2387 | -0.746 | 0.546 | 0.2864 | -0.435 | 0.614 |
| 0.2836 | -0.619 | 0.568 | 0.3383 | -0.395 | 0.624 |
| 0.3337 | -0.549 | 0.585 | 0.3984 | -0.365 | 0.631 |
| 0.3858 | -0.485 | 0.681 | 0.4376 | -0.324 | 0.641 |
| 0.4358 | -0.426 | 0.616 | 0.4872 | -0.289 | 0.652 |
| 0.4860 | -0.372 | 0.629 | 0.5371 | -0.243 | 0.661 |
| 0.5388 | -0.382 | 0.646 | 0.5879 | -0.213 | 0.669 |
| 0.5892 | -0.249 | 0.668 | 0.6371 | -0.176 | 0.678 |
| 0.6395 | -0.204 | 0.671 | 0.6874 | -0.147 | 0.685 |
| 0.6894 | -0.158 | 0.682 | 0.7359 | -0.112 | 0.694 |
| 0.7443 | -0.188 | 0.694 | | | |
| 0.7954 | -0.899 | 0.787 | | | |
| 0.8414 | -0.811 | 0.718 | | | |
| 0.8991 | 0.841 | 0.731 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.730 ALPHA 2.00 REY 0.02+10⁶

INTEGRATED FORCE COEFFICIENTS

C_x = 0.3232 C_y = 0.0203

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0207 | -0.602 | 3.552 | 0.0757 | -0.175 | 3.679 |
| 0.0707 | -1.221 | 3.419 | 0.1294 | -0.310 | 3.644 |
| 0.1299 | -1.124 | 3.443 | 0.1830 | -0.345 | 3.637 |
| 0.1813 | -0.934 | 3.493 | 0.2368 | -0.359 | 3.634 |
| 0.2387 | -0.805 | 3.522 | 0.2864 | -0.343 | 3.638 |
| 0.2836 | -0.694 | 3.550 | 0.3383 | -0.321 | 3.643 |
| 0.3337 | -0.607 | 3.571 | 0.3924 | -0.301 | 3.648 |
| 0.3858 | -0.551 | 3.598 | 0.4376 | -0.270 | 3.656 |
| 0.4358 | -0.463 | 3.600 | 0.4870 | -0.243 | 3.662 |
| 0.4863 | -0.402 | 3.623 | 0.5371 | -0.205 | 3.672 |
| 0.5380 | -0.327 | 3.641 | 0.5879 | -0.183 | 3.677 |
| 0.5902 | -0.269 | 3.656 | 0.6371 | -0.153 | 3.685 |
| 0.6395 | -0.217 | 3.668 | 0.6674 | -0.130 | 3.678 |
| 0.6894 | -0.168 | 3.680 | 0.7359 | -0.108 | 3.698 |
| 2.7443 | -0.113 | 3.694 | | | |
| 2.7954 | -0.061 | 3.707 | | | |
| 2.8414 | -0.009 | 3.720 | | | |
| 2.8891 | 0.042 | 3.732 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.710 ALPHA 0.00 REY 0.02⁶10

INTEGRATED FORCE COEFFICIENTS

CN = 0.0297 CM = 0.0054

UPPER SURFACE VALUES

| X/C | CP | P/H |
|--------|--------|-------|
| 0.0207 | -0.127 | 3.677 |
| 0.0787 | -0.610 | 3.593 |
| 0.1299 | -0.643 | 3.545 |
| 0.1813 | -0.661 | 3.543 |
| 0.2307 | -0.610 | 3.551 |
| 0.2836 | -0.595 | 3.567 |
| 0.3337 | -0.582 | 3.581 |
| 0.3850 | -0.447 | 3.595 |
| 0.4358 | -0.396 | 3.620 |
| 0.4960 | -0.349 | 3.620 |
| 0.5380 | -0.283 | 3.637 |
| 0.5882 | -0.233 | 3.550 |
| 0.6395 | -0.174 | 3.663 |
| 0.6894 | -0.191 | 3.671 |
| 0.7443 | -0.105 | 3.682 |
| 0.7954 | -0.098 | 3.694 |
| 0.8414 | -0.012 | 3.706 |
| 0.8891 | 0.030 | 3.719 |

LOWER SURFACE VALUES

| X/C | CP | P/H |
|--------|--------|-------|
| 0.0757 | -0.584 | 0.560 |
| 0.1294 | -0.662 | 0.540 |
| 0.1830 | -0.648 | 0.543 |
| 0.2360 | -0.609 | 0.553 |
| 0.2864 | -0.555 | 0.567 |
| 0.3303 | -0.493 | 0.583 |
| 0.3904 | -0.444 | 0.596 |
| 0.4376 | -0.387 | 0.610 |
| 0.4870 | -0.339 | 0.622 |
| 0.5371 | -0.284 | 0.637 |
| 0.5879 | -0.244 | 0.647 |
| 0.6371 | -0.200 | 0.650 |
| 0.6874 | -0.163 | 0.660 |
| 0.7359 | -0.123 | 0.670 |

NACA-8512 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.723 ALPHA 1.00 REY 0.82e10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1719 CM = 0.0113

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0207 | -0.300 | 0.610 | 0.0757 | -0.363 | 0.614 |
| 0.0787 | -0.021 | 0.471 | 0.1294 | -0.407 | 0.502 |
| 0.1299 | -0.071 | 0.404 | 0.1830 | -0.497 | 0.500 |
| 0.1813 | -0.097 | 0.470 | 0.2366 | -0.491 | 0.501 |
| 0.2307 | -0.777 | 0.500 | 0.2864 | -0.460 | 0.509 |
| 0.2836 | -0.660 | 0.530 | 0.3383 | -0.410 | 0.600 |
| 0.3337 | -0.577 | 0.560 | 0.3904 | -0.302 | 0.609 |
| 0.3858 | -0.507 | 0.576 | 0.4376 | -0.339 | 0.620 |
| 0.4358 | -0.441 | 0.594 | 0.4870 | -0.301 | 0.630 |
| 0.4868 | -0.304 | 0.609 | 0.5371 | -0.252 | 0.643 |
| 0.5380 | -0.311 | 0.620 | 0.5879 | -0.220 | 0.651 |
| 0.5882 | -0.255 | 0.642 | 0.6371 | -0.102 | 0.661 |
| 0.6395 | -0.200 | 0.654 | 0.6874 | -0.151 | 0.660 |
| 0.6804 | -0.101 | 0.666 | 0.7359 | -0.115 | 0.670 |
| 0.7443 | -0.109 | 0.680 | | | |
| 0.7954 | -0.059 | 0.692 | | | |
| 0.8414 | -0.009 | 0.705 | | | |
| 0.8891 | 0.042 | 0.719 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.721 ALPHA 2.00 REY 0.82*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.3118 CM = 0.0166

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.8287 | -0.617 | 0.548 | 0.0757 | -0.184 | 0.658 |
| 2.8787 | -1.168 | 0.488 | 0.1294 | -0.328 | 0.623 |
| 2.1259 | -1.203 | 0.398 | 0.1830 | -0.359 | 0.615 |
| 2.1813 | -1.197 | 0.399 | 0.2368 | -0.378 | 0.618 |
| 2.2387 | -1.186 | 0.482 | 0.2864 | -0.365 | 0.613 |
| 2.2836 | -0.778 | 0.589 | 0.3383 | -0.338 | 0.628 |
| 2.3337 | -0.598 | 0.555 | 0.3904 | -0.318 | 0.625 |
| 2.3858 | -0.518 | 0.574 | 0.4376 | -0.286 | 0.634 |
| 2.4359 | -0.495 | 0.598 | 0.4878 | -0.257 | 0.641 |
| 2.4850 | -0.396 | 0.685 | 0.5371 | -0.215 | 0.652 |
| 2.5380 | -0.322 | 0.624 | 0.5879 | -0.192 | 0.658 |
| 2.5882 | -0.253 | 0.639 | 0.6371 | -0.159 | 0.667 |
| 2.6395 | -0.213 | 0.652 | 0.6874 | -0.134 | 0.673 |
| 2.6894 | -0.184 | 0.665 | 0.7359 | -0.103 | 0.681 |
| 2.7443 | -0.188 | 0.679 | | | |
| 2.7954 | -0.256 | 0.693 | | | |
| 2.8414 | -0.285 | 0.706 | | | |
| 2.8891 | 0.046 | 0.719 | | | |

NACA-8812 181.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.739 ALPHA 0.00 REYNOLDS 8,820,180

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.8318$ $C_D = 0.0059$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0287 | -0.187 | 0.667 | 0.8757 | -0.595 | 0.537 |
| 0.0787 | -0.619 | 0.531 | 0.1294 | -0.693 | 0.511 |
| 0.1299 | -0.678 | 0.517 | 0.1838 | -0.697 | 0.510 |
| 0.1813 | -0.714 | 0.506 | 0.2368 | -0.629 | 0.528 |
| 0.2387 | -0.664 | 0.519 | 0.2864 | -0.589 | 0.539 |
| 0.2836 | -0.592 | 0.538 | 0.3383 | -0.518 | 0.558 |
| 0.3337 | -0.529 | 0.555 | 0.3984 | -0.444 | 0.572 |
| 0.3858 | -0.467 | 0.571 | 0.4376 | -0.482 | 0.589 |
| 0.4358 | -0.412 | 0.586 | 0.4878 | -0.351 | 0.672 |
| 0.4858 | -0.368 | 0.603 | 0.5371 | -0.292 | 0.618 |
| 0.5389 | -0.292 | 0.618 | 0.5879 | -0.251 | 0.629 |
| 0.5882 | -0.248 | 0.632 | 0.6371 | -0.205 | 0.641 |
| 0.6395 | -0.196 | 0.644 | 0.6874 | -0.166 | 0.651 |
| 0.6894 | -0.192 | 0.659 | 0.7359 | -0.124 | 0.663 |
| 0.7443 | -0.184 | 0.668 | | | |
| 0.7954 | -0.186 | 0.681 | | | |
| 0.8414 | -0.189 | 0.693 | | | |
| 0.8891 | 0.041 | 0.707 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.741 ALPHA 1.00 REV 0.02+10

INTEGRATED FORCE COEFFICIENTS

CN = 0.1989 CM = 0.0143

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.8287 | -3.336 | 0.685 | 0.0757 | -0.771 | 0.595 |
| 2.8787 | -3.937 | 0.444 | 0.1294 | -0.583 | 0.560 |
| 2.1299 | -0.647 | 0.468 | 0.1838 | -0.523 | 0.555 |
| 0.1813 | -0.926 | 0.447 | 0.2368 | -0.497 | 0.562 |
| 0.2387 | -0.997 | 0.428 | 0.2864 | -0.486 | 0.565 |
| 0.2836 | -0.883 | 0.472 | 0.3383 | -0.441 | 0.577 |
| 0.3337 | -0.571 | 0.542 | 0.3984 | -0.484 | 0.587 |
| 0.3858 | -0.585 | 0.568 | 0.4376 | -0.353 | 0.608 |
| 0.4358 | -3.441 | 0.577 | 0.4778 | -0.311 | 0.611 |
| 0.4868 | -0.345 | 0.592 | 0.5371 | -0.262 | 0.625 |
| 0.5388 | -0.311 | 0.611 | 0.5879 | -0.227 | 0.634 |
| 0.5882 | -0.256 | 0.626 | 0.6371 | -0.196 | 0.645 |
| 0.6395 | -0.206 | 0.639 | 0.6874 | -0.156 | 0.653 |
| 0.6894 | -0.159 | 0.652 | 0.7359 | -0.118 | 0.663 |
| 0.7443 | -0.106 | 0.666 | | | |
| 0.7954 | -0.055 | 0.688 | | | |
| 0.8414 | -0.005 | 0.693 | | | |
| 0.8891 | 0.046 | 0.787 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.741 ALPHA 2.00 REY 0.02-10

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.3346$ $C_M = 0.0134$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0287 | -0.543 | 0.340 | 0.0757 | -0.104 | 0.641 |
| 0.0787 | -1.004 | 0.404 | 0.1294 | -0.346 | 0.638 |
| 0.1299 | -1.146 | 0.388 | 0.1838 | -0.383 | 0.590 |
| 0.1813 | -1.107 | 0.382 | 0.2368 | -0.388 | 0.591 |
| 0.2337 | -1.213 | 0.370 | 0.2864 | -0.348 | 0.589 |
| 0.2736 | -1.192 | 0.376 | 0.3383 | -0.362 | 0.596 |
| 0.3337 | -1.126 | 0.393 | 0.3904 | -0.339 | 0.682 |
| 0.3858 | -0.591 | 0.536 | 0.4376 | -0.302 | 0.612 |
| 0.4358 | -0.451 | 0.573 | 0.4870 | -0.272 | 0.620 |
| 0.4868 | -0.371 | 0.593 | 0.5371 | -0.227 | 0.632 |
| 0.5380 | -0.304 | 0.612 | 0.5879 | -0.199 | 0.639 |
| 0.5882 | -0.249 | 0.627 | 0.6371 | -0.163 | 0.649 |
| 0.6395 | -0.208 | 0.648 | 0.6874 | -0.130 | 0.656 |
| 0.6894 | -0.152 | 0.653 | 0.7359 | -0.106 | 0.664 |
| 0.7443 | -0.099 | 0.667 | | | |
| 0.7954 | -0.049 | 0.688 | | | |
| 0.8414 | 0.000 | 0.694 | | | |
| 0.8891 | 0.050 | 0.707 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

NACH NO. 0.758 ALPHA 0.00 REY 0.01×10^6

INTEGRATED FORCE COEFFICIENTS

CN = 0.0200 CM = 0.0042

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0287 | -0.303 | 0.668 | 0.0757 | -0.689 | 0.516 |
| 0.0787 | -0.621 | 0.512 | 0.1294 | -0.714 | 0.486 |
| 0.1299 | -0.664 | 0.495 | 0.1830 | -0.789 | 0.465 |
| 0.1813 | -0.789 | 0.469 | 0.2360 | -0.754 | 0.475 |
| 0.2387 | -0.828 | 0.455 | 0.2864 | -0.642 | 0.506 |
| 0.2836 | -0.646 | 0.585 | 0.3383 | -0.554 | 0.538 |
| 0.3337 | -0.557 | 0.538 | 0.3984 | -0.488 | 0.548 |
| 0.3853 | -0.487 | 0.549 | 0.4376 | -0.428 | 0.567 |
| 0.4356 | -0.426 | 0.566 | 0.4878 | -0.363 | 0.592 |
| 0.4860 | -0.369 | 0.581 | 0.5371 | -0.332 | 0.599 |
| 0.5380 | -0.299 | 0.680 | 0.5879 | -0.257 | 0.611 |
| 0.5882 | -0.244 | 0.616 | 0.6371 | -0.289 | 0.625 |
| 0.6395 | -0.201 | 0.628 | 0.6874 | -0.169 | 0.636 |
| 0.6894 | -0.155 | 0.648 | 0.7359 | -0.125 | 0.648 |
| 0.7443 | -0.105 | 0.654 | | | |
| 0.7954 | -0.055 | 0.668 | | | |
| 0.8414 | -0.007 | 0.681 | | | |
| 0.8991 | 0.044 | 0.695 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.759 ALPHA 1.00 REV 0.01-10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.2825 CN = 0.8135

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0287 | -0.291 | 0.603 | 0.0757 | -0.378 | 0.576 |
| 0.0787 | -0.099 | 0.435 | 0.1294 | -0.521 | 0.539 |
| 0.1299 | -0.075 | 0.442 | 0.1830 | -0.555 | 0.533 |
| 0.1813 | -0.060 | 0.433 | 0.2368 | -0.532 | 0.536 |
| 0.2307 | -0.066 | 0.417 | 0.2864 | -0.528 | 0.539 |
| 0.2836 | -0.090 | 0.400 | 0.3383 | -0.460 | 0.554 |
| 0.3337 | -0.099 | 0.400 | 0.3924 | -0.425 | 0.566 |
| 0.3850 | -0.526 | 0.530 | 0.4376 | -0.371 | 0.580 |
| 0.4356 | -0.421 | 0.567 | 0.4878 | -0.326 | 0.593 |
| 0.4858 | -0.365 | 0.582 | 0.5371 | -0.272 | 0.600 |
| 0.5380 | -0.299 | 0.600 | 0.5879 | -0.234 | 0.610 |
| 0.5882 | -0.240 | 0.614 | 0.6371 | -0.192 | 0.630 |
| 0.6395 | -0.199 | 0.628 | 0.6874 | -0.159 | 0.639 |
| 0.6894 | -0.153 | 0.640 | 0.7359 | -0.120 | 0.650 |
| 0.7443 | -0.101 | 0.655 | | | |
| 0.7954 | -0.051 | 0.669 | | | |
| 0.8414 | -0.001 | 0.682 | | | |
| 0.8891 | 0.050 | 0.696 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.759 ALPHA 2.00 REY 8.01×10^6

INTEGRATED FORCE COEFFICIENTS

$C_N = 0.3417$ $C_M = 0.0006$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0297 | -0.478 | 0.553 | 0.0757 | -0.209 | 0.624 |
| 0.0787 | -1.212 | 0.484 | 0.1294 | -0.348 | 0.581 |
| 0.1299 | -1.862 | 0.384 | 0.1838 | -0.414 | 0.568 |
| 0.1813 | -1.115 | 0.375 | 0.2368 | -0.413 | 0.568 |
| 0.2387 | -1.173 | 0.359 | 0.2864 | -0.428 | 0.566 |
| 0.2836 | -1.168 | 0.361 | 0.3383 | -0.398 | 0.575 |
| 0.3327 | -1.201 | 0.351 | 0.3904 | -0.344 | 0.582 |
| 0.3858 | -1.111 | 0.376 | 0.4376 | -0.324 | 0.593 |
| 0.4358 | -0.597 | 0.518 | 0.4878 | -0.289 | 0.602 |
| 0.4863 | -0.442 | 0.561 | 0.5371 | -0.243 | 0.615 |
| 0.5388 | -0.348 | 0.589 | 0.5879 | -0.214 | 0.623 |
| 0.5882 | -0.263 | 0.618 | 0.6371 | -0.176 | 0.633 |
| 0.6395 | -0.197 | 0.629 | 0.6874 | -0.149 | 0.641 |
| 0.6894 | -0.143 | 0.643 | 0.7359 | -0.138 | 0.650 |
| 0.7443 | -0.085 | 0.659 | | | |
| 0.7954 | -0.038 | 0.672 | | | |
| 0.8414 | 0.018 | 0.685 | | | |
| 0.8891 | 0.054 | 0.697 | | | |

NACA-8012 181.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.779 ALPHA 0.88 REYNOLDS 6.79e10

INTEGRATED FORCE COEFFICIENTS

$C_L = 0.8139$ $C_M = 0.8826$

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0207 | -0.852 | 0.654 | 0.0757 | -0.597 | 0.498 |
| 0.0787 | -0.602 | 0.497 | 0.1294 | -0.683 | 0.473 |
| 0.1299 | -0.656 | 0.482 | 0.1838 | -0.781 | 0.445 |
| 0.1813 | -0.758 | 0.453 | 0.2368 | -0.793 | 0.442 |
| 0.2387 | -0.841 | 0.429 | 0.2864 | -0.868 | 0.423 |
| 0.2836 | -0.882 | 0.423 | 0.3383 | -0.856 | 0.424 |
| 0.3337 | -0.895 | 0.414 | 0.3984 | -0.547 | 0.512 |
| 0.3858 | -0.598 | 0.581 | 0.4376 | -0.399 | 0.554 |
| 0.4358 | -0.481 | 0.554 | 0.4877 | -0.346 | 0.578 |
| 0.4868 | -0.346 | 0.573 | 0.5371 | -0.291 | 0.585 |
| 0.5388 | -0.284 | 0.587 | 0.5879 | -0.248 | 0.597 |
| 0.5882 | -0.245 | 0.602 | 0.6371 | -0.201 | 0.611 |
| 0.6395 | -0.192 | 0.614 | 0.6874 | -0.161 | 0.622 |
| 0.6894 | -0.147 | 0.627 | 0.7359 | -0.118 | 0.634 |
| 0.7443 | -0.097 | 0.641 | | | |
| 0.7954 | -0.049 | 0.655 | | | |
| 0.8414 | -0.001 | 0.668 | | | |
| 0.8891 | 0.049 | 0.682 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.791 ALPHA 0.58 REYNOLDS 0.79-10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.8988 CM = 0.8889

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0287 | -0.143 | 0.620 | 0.0757 | -0.404 | 0.531 |
| 0.0787 | -0.736 | 0.458 | 0.1294 | -0.620 | 0.492 |
| 0.1299 | -0.717 | 0.464 | 0.1830 | -0.705 | 0.468 |
| 0.1813 | -0.810 | 0.438 | 0.2368 | -0.787 | 0.467 |
| 0.2387 | -0.896 | 0.413 | 0.2864 | -0.732 | 0.468 |
| 0.2836 | -0.927 | 0.404 | 0.3383 | -0.543 | 0.514 |
| 0.3337 | -0.973 | 0.391 | 0.3904 | -0.476 | 0.533 |
| 0.3858 | -0.927 | 0.404 | 0.4376 | -0.409 | 0.552 |
| 0.4358 | -0.476 | 0.533 | 0.4878 | -0.353 | 0.560 |
| 0.4868 | -0.351 | 0.569 | 0.5371 | -0.292 | 0.556 |
| 0.5388 | -0.279 | 0.589 | 0.5879 | -0.249 | 0.598 |
| 0.5882 | -0.229 | 0.624 | 0.6371 | -0.203 | 0.611 |
| 0.6395 | -0.185 | 0.616 | 0.6874 | -0.164 | 0.622 |
| 0.6894 | -0.142 | 0.628 | 0.7359 | -0.122 | 0.634 |
| 0.7443 | -0.092 | 0.643 | | | |
| 0.7954 | -0.045 | 0.656 | | | |
| 0.8414 | 0.003 | 0.670 | | | |
| 0.8891 | 0.053 | 0.684 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.750 ALPHA 1.00 REYNOLDS 2.79x10⁶

INTEGRATED FORCE COEFFICIENTS

C_L = 0.1918 C_D = 0.0049

| UPPER SURFACE VALUES | | | LOWER SURFACE VALUES | | |
|----------------------|--------|-------|----------------------|--------|-------|
| X/C | CP | P/H | X/C | CP | P/H |
| 0.0207 | -0.230 | 0.603 | 0.0757 | -0.391 | 0.556 |
| 0.0707 | -0.034 | 0.430 | 0.1294 | -0.540 | 0.511 |
| 0.1299 | -0.054 | 0.425 | 0.1830 | -0.620 | 0.459 |
| 0.1813 | -0.000 | 0.415 | 0.2360 | -0.606 | 0.495 |
| 0.2307 | -0.049 | 0.397 | 0.2864 | -0.623 | 0.490 |
| 0.2836 | -0.071 | 0.391 | 0.3303 | -0.520 | 0.517 |
| 0.3337 | -1.020 | 0.375 | 0.3904 | -0.409 | 0.534 |
| 0.3850 | -1.039 | 0.372 | 0.4376 | -0.403 | 0.553 |
| 0.4350 | -0.040 | 0.426 | 0.4870 | -0.350 | 0.560 |
| 0.4860 | -0.433 | 0.545 | 0.5371 | -0.271 | 0.555 |
| 0.5300 | -0.311 | 0.579 | 0.5879 | -0.240 | 0.597 |
| 0.5802 | -0.230 | 0.600 | 0.6371 | -0.203 | 0.609 |
| 0.6395 | -0.103 | 0.616 | 0.6874 | -0.166 | 0.620 |
| 0.6894 | -0.136 | 0.629 | 0.7359 | -0.126 | 0.632 |
| 0.7442 | -0.002 | 0.645 | | | |
| 0.7954 | -0.036 | 0.650 | | | |
| 0.8414 | 0.011 | 0.671 | | | |
| 0.8891 | 0.050 | 0.685 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.030 ALPHA 0.00 REY 0.00⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.0076 CM = 0.0019

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0287 | -0.018 | 0.650 | 0.0757 | -0.501 | 0.405 |
| 0.0707 | -0.576 | 0.406 | 0.1294 | -0.646 | 0.466 |
| 0.1299 | -0.619 | 0.473 | 0.1830 | -0.751 | 0.435 |
| 0.1913 | -0.722 | 0.443 | 0.2368 | -0.773 | 0.428 |
| 0.2387 | -0.809 | 0.410 | 0.2864 | -0.852 | 0.405 |
| 0.2936 | -0.842 | 0.400 | 0.3303 | -0.869 | 0.420 |
| 0.3337 | -0.893 | 0.393 | 0.3904 | -0.900 | 0.358 |
| 0.3850 | -0.902 | 0.390 | 0.4376 | -0.883 | 0.396 |
| 0.4358 | -0.907 | 0.389 | 0.4870 | -0.447 | 0.524 |
| 0.4860 | -0.455 | 0.525 | 0.5371 | -0.295 | 0.569 |
| 0.5300 | -0.286 | 0.571 | 0.5879 | -0.220 | 0.509 |
| 0.5802 | -0.219 | 0.591 | 0.6371 | -0.181 | 0.602 |
| 0.6395 | -0.169 | 0.606 | 0.6974 | -0.141 | 0.614 |
| 0.6894 | -0.127 | 0.610 | 0.7359 | -0.101 | 0.626 |
| 0.7443 | -0.080 | 0.632 | | | |
| 0.7954 | -0.035 | 0.645 | | | |
| 0.8414 | 0.011 | 0.659 | | | |
| 0.8891 | 0.020 | 0.673 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.820 ALPHA 0.50 REY 0.80010⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1172 CM = 0.0841

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 2.2207 | -0.895 | 0.627 | 0.0757 | -0.474 | 0.516 |
| 2.0707 | -0.699 | 0.450 | 0.1294 | -0.580 | 0.479 |
| 2.1299 | -0.663 | 0.460 | 0.1830 | -0.690 | 0.491 |
| 2.1813 | -0.763 | 0.431 | 0.2360 | -0.718 | 0.444 |
| 2.2307 | -0.849 | 0.406 | 0.2864 | -0.792 | 0.422 |
| 2.2836 | -0.888 | 0.394 | 0.3383 | -0.889 | 0.417 |
| 2.3337 | -0.943 | 0.376 | 0.3904 | -0.840 | 0.400 |
| 2.3858 | -0.953 | 0.375 | 0.4376 | -0.671 | 0.450 |
| 2.4358 | -0.901 | 0.357 | 0.4870 | -0.354 | 0.551 |
| 2.4868 | -0.664 | 0.460 | 0.5371 | -0.277 | 0.574 |
| 2.5380 | -0.361 | 0.549 | 0.5879 | -0.233 | 0.587 |
| 2.5882 | -0.259 | 0.579 | 0.6371 | -0.193 | 0.599 |
| 2.6395 | -0.184 | 0.601 | 0.6874 | -0.156 | 0.610 |
| 2.6894 | -0.130 | 0.617 | 0.7359 | -0.117 | 0.621 |
| 2.7443 | -0.073 | 0.634 | | | |
| 2.7954 | -0.030 | 0.647 | | | |
| 2.8414 | 0.018 | 0.661 | | | |
| 2.8891 | 0.063 | 0.674 | | | |

VACA-8012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.831 ALPHA 1.00 REY 2.88*10⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.1685 CM = -0.0028

UPPER SURFACE VALUES

| X/C | CP | P/H |
|--------|--------|-------|
| 0.0287 | -0.162 | 2.698 |
| 0.0787 | -0.762 | 0.431 |
| 0.1299 | -0.788 | 0.423 |
| 0.1913 | -0.638 | 0.411 |
| 0.2387 | -0.896 | 0.391 |
| 0.2836 | -0.923 | 0.383 |
| 0.3337 | -0.981 | 0.366 |
| 0.3858 | -0.998 | 0.361 |
| 0.4358 | -1.035 | 0.350 |
| 0.4868 | -0.729 | 0.441 |
| 0.5380 | -0.402 | 0.537 |
| 0.5882 | -0.301 | 0.567 |
| 0.6395 | -0.218 | 0.591 |
| 0.6894 | -0.154 | 0.610 |
| 0.7443 | -0.088 | 0.629 |
| 0.7954 | -0.033 | 0.645 |
| 0.8414 | 0.016 | 0.668 |
| 0.8891 | 0.059 | 0.672 |

LOWER SURFACE VALUES

| X/C | CP | P/H |
|--------|--------|-------|
| 0.0757 | -0.397 | 0.538 |
| 0.1294 | -0.550 | 0.493 |
| 0.1830 | -0.641 | 0.466 |
| 0.2368 | -0.661 | 0.461 |
| 0.2864 | -0.725 | 0.442 |
| 0.3383 | -0.738 | 0.438 |
| 0.3904 | -0.721 | 0.443 |
| 0.4376 | -0.414 | 0.533 |
| 0.4878 | -0.342 | 0.555 |
| 0.5371 | -0.299 | 0.572 |
| 0.5879 | -0.251 | 0.581 |
| 0.6371 | -0.209 | 0.594 |
| 0.6874 | -0.172 | 0.605 |
| 0.7359 | -0.133 | 0.616 |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.028 ALPHA 0.00 REYNOLDS 0.01010⁶

INTEGRATED FORCE COEFFICIENTS

CN = 0.0027 CM = 0.0020

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | CP | P/H | X/C | CP | P/H |
|--------|--------|-------|--------|--------|-------|
| 0.0287 | 0.817 | 0.648 | 0.0757 | -0.568 | 0.474 |
| 0.0787 | -0.549 | 0.477 | 0.1294 | -0.686 | 0.459 |
| 0.1299 | -0.568 | 0.467 | 0.1830 | -0.716 | 0.426 |
| 0.1813 | -0.682 | 0.436 | 0.2368 | -0.743 | 0.418 |
| 0.2387 | -0.771 | 0.418 | 0.2864 | -0.824 | 0.394 |
| 0.2836 | -0.889 | 0.398 | 0.3383 | -0.847 | 0.387 |
| 0.3337 | -0.864 | 0.381 | 0.3904 | -0.891 | 0.373 |
| 0.3858 | -0.878 | 0.377 | 0.4376 | -0.982 | 0.370 |
| 0.4358 | -0.915 | 0.366 | 0.4870 | -0.926 | 0.363 |
| 0.4860 | -0.942 | 0.358 | 0.5371 | -0.785 | 0.438 |
| 0.5383 | -0.660 | 0.443 | 0.5879 | -0.350 | 0.537 |
| 0.5882 | -0.339 | 0.540 | 0.6371 | -0.245 | 0.569 |
| 0.6395 | -0.232 | 0.573 | 0.6874 | -0.168 | 0.592 |
| 0.6894 | -0.162 | 0.594 | 0.7359 | -0.111 | 0.610 |
| 0.7443 | -0.093 | 0.615 | | | |
| 0.7954 | -0.042 | 0.638 | | | |
| 0.8414 | 0.007 | 0.645 | | | |
| 0.8891 | 0.051 | 0.658 | | | |

NACA-0012 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.820 ALPHA 0.50 REYNOLDS 0.81x10⁶

INTEGRATED FORCE COEFFICIENTS

C_D = 0.0570 C_M = 0.0076

UPPER SURFACE VALUES

| X/C | C _P | P/H |
|--------|----------------|-------|
| 0.0287 | -0.844 | 0.638 |
| 0.0787 | -0.641 | 0.449 |
| 0.1299 | -0.621 | 0.455 |
| 0.1813 | -0.718 | 0.426 |
| 0.2327 | -0.798 | 0.421 |
| 0.2836 | -0.841 | 0.388 |
| 0.3337 | -0.898 | 0.371 |
| 0.3850 | -0.914 | 0.366 |
| 0.4358 | -0.947 | 0.357 |
| 0.4868 | -0.964 | 0.351 |
| 0.5398 | -0.914 | 0.407 |
| 0.5882 | -0.825 | 0.545 |
| 0.6395 | -0.727 | 0.571 |
| 0.6894 | -0.612 | 0.591 |
| 0.7443 | -0.483 | 0.612 |
| 0.7954 | -0.348 | 0.628 |
| 0.8414 | 0.303 | 0.644 |
| 0.8891 | 0.846 | 0.657 |

LOWER SURFACE VALUES

| X/C | C _P | P/H |
|--------|----------------|-------|
| 0.0757 | -0.474 | 0.528 |
| 0.1294 | -0.576 | 0.469 |
| 0.1838 | -0.677 | 0.439 |
| 0.2368 | -0.784 | 0.431 |
| 0.2864 | -0.784 | 0.426 |
| 0.3383 | -0.888 | 0.399 |
| 0.3984 | -0.851 | 0.386 |
| 0.4376 | -0.863 | 0.383 |
| 0.4878 | -0.886 | 0.375 |
| 0.5371 | -0.782 | 0.431 |
| 0.5879 | -0.331 | 0.544 |
| 0.6371 | -0.235 | 0.573 |
| 0.6874 | -0.167 | 0.593 |
| 0.7359 | -0.128 | 0.625 |

NACA-3312 101.6 MM CHORD
EXPERIMENTAL PRESSURE DISTRIBUTION

MACH NO. 0.819 ALPHA 1.00 REV 8.01-10

INTEGRATED FORCE COEFFICIENTS

C_x = 0.0699 C_y = 0.0160

UPPER SURFACE VALUES

LOWER SURFACE VALUES

| X/C | C _p | P/H | X/C | C _p | P/H |
|--------|----------------|-------|--------|----------------|-------|
| 0.0207 | -0.896 | 0.615 | 0.0757 | -0.486 | 0.528 |
| 0.0757 | -0.692 | 0.435 | 0.1294 | -0.540 | 0.488 |
| 0.1294 | -0.719 | 0.426 | 0.1830 | -0.639 | 0.450 |
| 0.1830 | -0.767 | 0.412 | 0.2368 | -0.667 | 0.442 |
| 0.2368 | -0.837 | 0.391 | 0.2864 | -0.740 | 0.410 |
| 0.2864 | -0.866 | 0.382 | 0.3383 | -0.770 | 0.412 |
| 0.3383 | -0.920 | 0.363 | 0.3904 | -0.815 | 0.397 |
| 0.3904 | -0.946 | 0.358 | 0.4376 | -0.820 | 0.393 |
| 0.4376 | -0.986 | 0.345 | 0.4870 | -0.851 | 0.386 |
| 0.4870 | -0.765 | 0.412 | 0.5371 | -0.884 | 0.420 |
| 0.5368 | -0.387 | 0.526 | 0.5879 | -0.422 | 0.515 |
| 0.5882 | -0.385 | 0.551 | 0.6371 | -0.259 | 0.565 |
| 0.6395 | -0.250 | 0.568 | 0.6874 | -0.191 | 0.586 |
| 0.6894 | -0.198 | 0.584 | 0.7359 | -0.145 | 0.599 |
| 0.7443 | -0.144 | 0.480 | | | |
| 0.7954 | -0.095 | 0.615 | | | |
| 0.8414 | -0.045 | 0.630 | | | |
| 0.8891 | -0.005 | 0.642 | | | |

APPENDIX B
PLOTTED RESULTS

NOTATION

| | | |
|----|--------------|-----------------------------------------|
| AL | (α) | Angle of incidence |
| CM | (C_m) | Pitching moment coefficient |
| CN | (C_N) | Normal force coefficient |
| H | (H) | Free stream total pressure |
| M | (M) | Free stream Mach number |
| P | (p) | Local static pressure |
| R | (R) | Reynolds number based on aerofoil chord |

Note: Symbols in brackets are corresponding symbols used in main part of report.

CONTENTS

| No. | Model | M | α |
|-----|-----------------|-------|----------|
| 1 | 203-2 mm BGK-1* | 0.750 | 0.55 |
| 2 | 203-2 mm BGK-1* | 0.749 | 0.55 |
| 3 | 203-2 mm BGK-1* | 0.749 | 0.55 |
| 4 | 203-2 mm BGK-1 | 0.50 | -2.5 |
| 5 | 203-2 mm BGK-1 | 0.50 | 0 |
| 6 | 203-2 mm BGK-1 | 0.50 | 1 |
| 7 | 203-2 mm BGK-1 | 0.50 | 2 |
| 8 | 203-2 mm BGK-1 | 0.50 | 3 |
| 9 | 203-2 mm BGK-1 | 0.50 | 4 |
| 10 | 203-2 mm BGK-1 | 0.60 | -2.5 |
| 11 | 203-2 mm BGK-1 | 0.60 | 0 |
| 12 | 203-2 mm BGK-1 | 0.60 | 1 |
| 13 | 203-2 mm BGK-1 | 0.60 | 2 |
| 14 | 203-2 mm BGK-1 | 0.60 | 3 |
| 15 | 203-2 mm BGK-1 | 0.60 | 4 |
| 16 | 203-2 mm BGK-1 | 0.65 | -2.4 |
| 17 | 203-2 mm BGK-1 | 0.65 | 0 |
| 18 | 203-2 mm BGK-1 | 0.65 | 1 |
| 19 | 203-2 mm BGK-1 | 0.65 | 2 |
| 20 | 203-2 mm BGK-1 | 0.65 | 3 |
| 21 | 203-2 mm BGK-1 | 0.70 | 2.5 |
| 22 | 203-2 mm BGK-1 | 0.70 | 0 |
| 23 | 203-2 mm BGK-1 | 0.70 | 1 |
| 24 | 203-2 mm BGK-1 | 0.70 | 2 |
| 25 | 203-2 mm BGK-1 | 0.70 | 3 |
| 26 | 203-2 mm BGK-1 | 0.72 | -2.5 |
| 27 | 203-2 mm BGK-1 | 0.72 | 0 |
| 28 | 203-2 mm BGK-1 | 0.72 | 1 |
| 29 | 203-2 mm BGK-1 | 0.72 | 2 |
| 30 | 203-2 mm BGK-1 | 0.74 | -2.5 |
| 31 | 203-2 mm BGK-1 | 0.74 | -1 |
| 32 | 203-2 mm BGK-1 | 0.74 | 0 |
| 33 | 203-2 mm BGK-1 | 0.74 | 1 |
| 34 | 203-2 mm BGK-1 | 0.74 | 2 |
| 35 | 203-2 mm BGK-1 | 0.76 | -2.5 |
| 36 | 203-2 mm BGK-1 | 0.76 | 1 |
| 37 | 203-2 mm BGK-1 | 0.76 | 0 |
| 38 | 203-2 mm BGK-1 | 0.76 | 1 |
| 39 | 203-2 mm BGK-1 | 0.76 | 2 |
| 40 | 203-2 mm BGK-1 | 0.78 | -2.5 |
| 41 | 203-2 mm BGK-1 | 0.78 | 1 |
| 42 | 203-2 mm BGK-1 | 0.78 | 0 |
| 43 | 203-2 mm BGK-1 | 0.78 | 1 |
| 44 | 203-2 mm BGK-1 | 0.78 | 2 |
| 45 | 203-2 mm BGK-1 | 0.80 | -2.5 |
| 46 | 203-2 mm BGK-1 | 0.80 | 1 |
| 47 | 203-2 mm BGK-1 | 0.80 | 0 |
| 48 | 203-2 mm BGK-1 | 0.80 | 1 |
| 49 | 203-2 mm BGK-1 | 0.80 | 2 |
| 50 | 101-6 mm BGK-1* | 0.753 | 0.60 |
| 51 | 101-6 mm BGK-1* | 0.752 | 0.60 |

* Free transition.

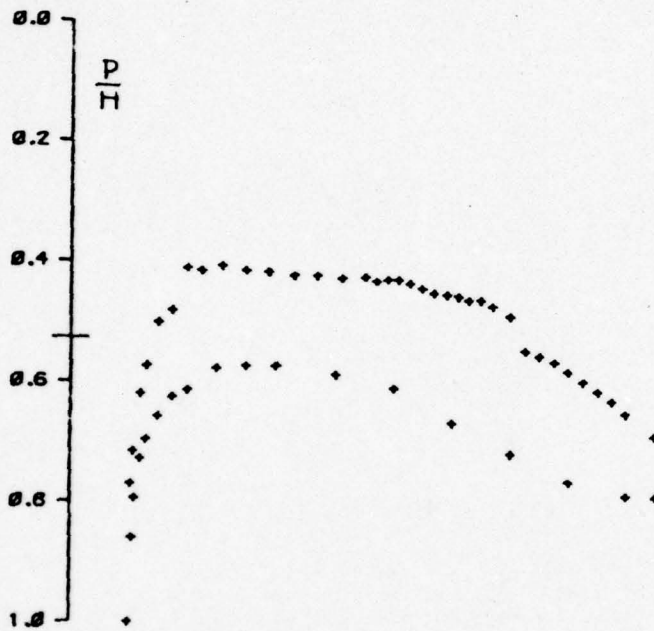
| No. | Model | M | α |
|-----|--------------------|-------|----------|
| 52 | 001 6 mm BGK-1* | 0.755 | 0.60 |
| 53 | 001 6 mm BGK-1 | 0.50 | -2.5 |
| 54 | 001 6 mm BGK-1 | 0.50 | 0 |
| 55 | 001 6 mm BGK-1 | 0.50 | 1 |
| 56 | 001 6 mm BGK-1 | 0.50 | 2 |
| 57 | 001 6 mm BGK-1 | 0.50 | 3 |
| 58 | 001 6 mm BGK-1 | 0.50 | 4 |
| 59 | 001 6 mm BGK-1 | 0.660 | -2.5 |
| 60 | 001 6 mm BGK-1 | 0.50 | 0 |
| 61 | 001 6 mm BGK-1 | 0.60 | 1 |
| 62 | 001 6 mm BGK-1 | 0.60 | 2 |
| 63 | 001 6 mm BGK-1 | 0.60 | 3 |
| 64 | 001 6 mm BGK-1 | 0.60 | 4 |
| 65 | 001 6 mm BGK-1 | 0.65 | -2.5 |
| 66 | 001 6 mm EGK-1 | 0.65 | 0 |
| 67 | 001 6 mm BGK-1 | 0.65 | 1 |
| 68 | 001 6 mm BGK-1 | 0.65 | 2 |
| 69 | 001 6 mm BGK-1 | 0.65 | 3 |
| 70 | 001 6 mm BGK-1 | 0.70 | -2.5 |
| 71 | 001 6 mm BGK-1 | 0.70 | 0 |
| 72 | 001 6 mm BGK-1 | 0.70 | 1 |
| 73 | 001 6 mm BGK-1 | 0.70 | 2 |
| 74 | 001 6 mm BGK-1 | 0.70 | 3 |
| 75 | 001 6 mm BGK-1 | 0.72 | -2.5 |
| 76 | 001 6 mm BGK-1 | 0.72 | 0 |
| 77 | 001 6 mm BGK-1 | 0.72 | 1 |
| 78 | 001 6 mm BGK-1 | 0.72 | 2 |
| 79 | 001 6 mm BGK-1 | 0.74 | -2.5 |
| 80 | 001 6 mm BGK-1 | 0.74 | -1 |
| 81 | 001 6 mm BGK-1 | 0.74 | 0 |
| 82 | 001 6 mm BGK-1 | 0.74 | 1 |
| 83 | 001 6 mm BGK-1 | 0.74 | 2 |
| 84 | 001 6 mm BGK-1 | 0.76 | -2.5 |
| 85 | 001 6 mm BGK-1 | 0.76 | -1 |
| 86 | 001 6 mm BGK-1 | 0.76 | 0 |
| 87 | 001 6 mm BGK-1 | 0.76 | 1 |
| 88 | 001 6 mm BGK-1 | 0.76 | 2 |
| 89 | 001 6 mm BGK-1 | 0.78 | -2.5 |
| 90 | 001 6 mm BGK-1 | 0.78 | -1 |
| 91 | 001 6 mm BGK-1 | 0.78 | 0 |
| 92 | 001 6 mm BGK-1 | 0.78 | 1 |
| 93 | 001 6 mm BGK-1 | 0.78 | 2 |
| 94 | 001 6 mm BGK-1 | 0.80 | -2.5 |
| 95 | 001 6 mm BGK-1 | 0.80 | -1 |
| 96 | 001 6 mm BGK-1 | 0.80 | 0 |
| 97 | 001 6 mm BGK-1 | 0.80 | 1 |
| 98 | 001 6 mm BGK-1 | 0.80 | 2 |
| 99 | 200 2 mm NACA-0012 | 0.50 | 0 |
| 100 | 200 2 mm NACA-0012 | 0.50 | 1 |
| 101 | 200 2 mm NACA-0012 | 0.50 | 2 |
| 102 | 200 2 mm NACA-0012 | 0.50 | 3 |

* Free transition.

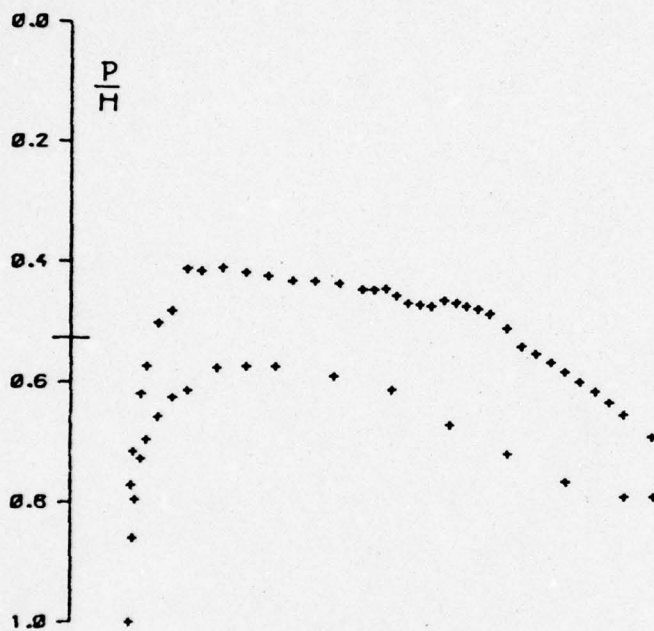
| No. | Model | M | z |
|-----|--------------------|-------|-----|
| 103 | 203 2 mm NACA-0012 | 0.50 | 4 |
| 104 | 203 2 mm NACA-0012 | 0.60 | 0 |
| 105 | 203 2 mm NACA-0012 | 0.60 | 1 |
| 106 | 203 2 mm NACA-0012 | 0.60 | 2 |
| 107 | 203 2 mm NACA-0012 | 0.60 | 3 |
| 108 | 203 2 mm NACA-0012 | 0.65 | 0 |
| 109 | 203 2 mm NACA-0012 | 0.65 | 1 |
| 110 | 203 2 mm NACA-0012 | 0.65 | 2 |
| 111 | 203 2 mm NACA-0012 | 0.65 | 3 |
| 112 | 203 2 mm NACA-0012 | 0.675 | 0 |
| 113 | 203 2 mm NACA-0012 | 0.675 | 1 |
| 114 | 203 2 mm NACA-0012 | 0.675 | 2 |
| 115 | 203 2 mm NACA-0012 | 0.70 | 0 |
| 116 | 203 2 mm NACA-0012 | 0.70 | 1 |
| 117 | 203 2 mm NACA-0012 | 0.70 | 2 |
| 118 | 203 2 mm NACA-0012 | 0.72 | 0 |
| 119 | 203 2 mm NACA-0012 | 0.72 | 1 |
| 120 | 203 2 mm NACA-0012 | 0.72 | 2 |
| 121 | 203 2 mm NACA-0012 | 0.74 | 0 |
| 122 | 203 2 mm NACA-0012 | 0.74 | 1 |
| 123 | 203 2 mm NACA-0012 | 0.74 | 2 |
| 124 | 203 2 mm NACA-0012 | 0.76 | 0 |
| 125 | 203 2 mm NACA-0012 | 0.76 | 1 |
| 126 | 203 2 mm NACA-0012 | 0.76 | 2 |
| 127 | 203 2 mm NACA-0012 | 0.78 | 0 |
| 128 | 203 2 mm NACA-0012 | 0.78 | 0.5 |
| 129 | 203 2 mm NACA-0012 | 0.78 | 1 |
| 130 | 203 2 mm NACA-0012 | 0.80 | 0 |
| 131 | 203 2 mm NACA-0012 | 0.80 | 0.5 |
| 132 | 203 2 mm NACA-0012 | 0.80 | 1 |
| 133 | 203 2 mm NACA-0012 | 0.82 | 0 |
| 134 | 203 2 mm NACA-0012 | 0.82 | 0.5 |
| 135 | 203 2 mm NACA-0012 | 0.82 | 1.0 |
| 136 | 101 6 mm NACA-0012 | 0.50 | 0 |
| 137 | 101 6 mm NACA-0012 | 0.50 | 1 |
| 138 | 101 6 mm NACA-0012 | 0.50 | 2 |
| 139 | 101 6 mm NACA-0012 | 0.50 | 3 |
| 140 | 101 6 mm NACA-0012 | 0.50 | 4 |
| 141 | 101 6 mm NACA-0012 | 0.60 | 0 |
| 142 | 101 6 mm NACA-0012 | 0.60 | 1 |
| 143 | 101 6 mm NACA-0012 | 0.60 | 2 |
| 144 | 101 6 mm NACA-0012 | 0.60 | 3 |
| 145 | 101 6 mm NACA-0012 | 0.65 | 0 |
| 146 | 101 6 mm NACA-0012 | 0.65 | 1 |
| 147 | 101 6 mm NACA-0012 | 0.65 | 2 |
| 148 | 101 6 mm NACA-0012 | 0.65 | 3 |
| 149 | 101 6 mm NACA-0012 | 0.675 | 0 |
| 150 | 101 6 mm NACA-0012 | 0.675 | 1 |
| 151 | 101 6 mm NACA-0012 | 0.675 | 2 |
| 152 | 101 6 mm NACA-0012 | 0.70 | 0 |
| 153 | 101 6 mm NACA-0012 | 0.70 | 1 |

h

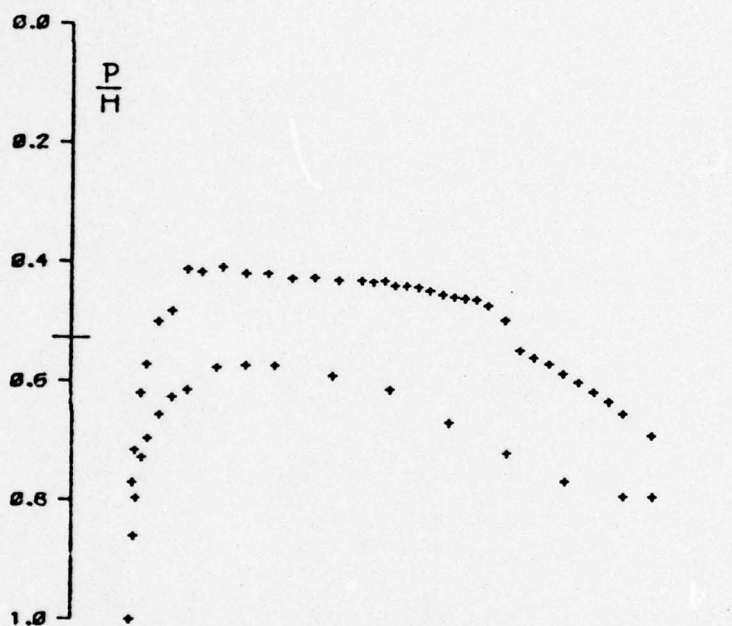
| No. | Model | M | π |
|-----|--------------------|------|-------|
| 154 | 101-6 mm NACA-0012 | 0.70 | 2 |
| 155 | 101-6 mm NACA-0012 | 0.72 | 0 |
| 156 | 101-6 mm NACA-0012 | 0.72 | 1 |
| 157 | 101-6 mm NACA-0012 | 0.72 | 2 |
| 158 | 101-6 mm NACA-0012 | 0.74 | 0 |
| 159 | 101-6 mm NACA-0012 | 0.74 | 1 |
| 160 | 101-6 mm NACA-0012 | 0.74 | 2 |
| 161 | 101-6 mm NACA-0012 | 0.76 | 0 |
| 162 | 101-6 mm NACA-0012 | 0.76 | 1 |
| 163 | 101-6 mm NACA-0012 | 0.76 | 2 |
| 164 | 101-6 mm NACA-0012 | 0.78 | 0 |
| 165 | 101-6 mm NACA-0012 | 0.78 | 0.5 |
| 166 | 101-6 mm NACA-0012 | 0.78 | 1 |
| 167 | 101-6 mm NACA-0012 | 0.80 | 0 |
| 168 | 101-6 mm NACA-0012 | 0.80 | 0.5 |
| 169 | 101-6 mm NACA-0012 | 0.80 | 1 |
| 170 | 101-6 mm NACA-0012 | 0.82 | 0 |
| 171 | 101-6 mm NACA-0012 | 0.82 | 0.5 |
| 172 | 101-6 mm NACA-0012 | 0.82 | 1 |



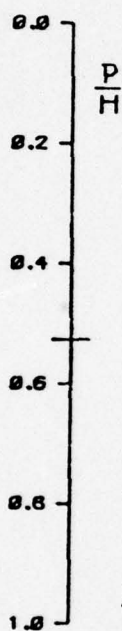
+ BGK-1 203.2 MM CHORD SOLID WALLS
 $M = .750$ $AL = 0.55$ $CL = 0.565$ $CM = 0.130$ $R = 1.656$



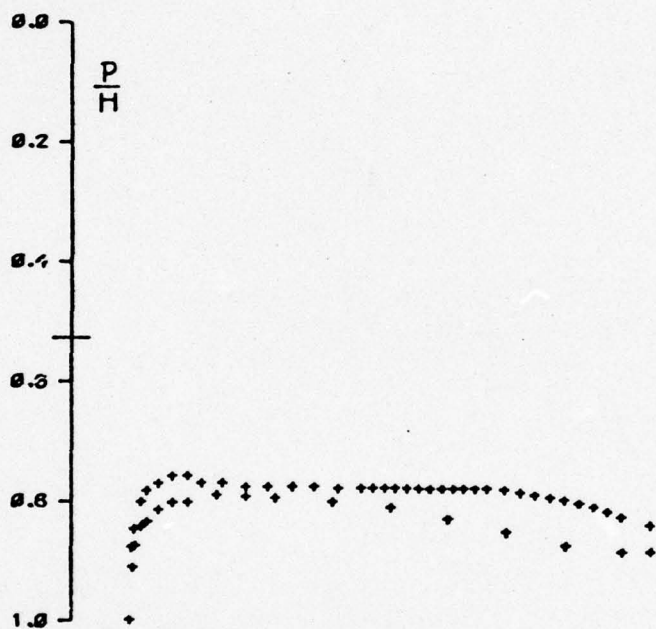
+ BGK-1 203.2 MM CHORD SOLID WALLS
 M= .749 AL= 0.55 CN= 0.566 CM= 0.126 R= 1.658



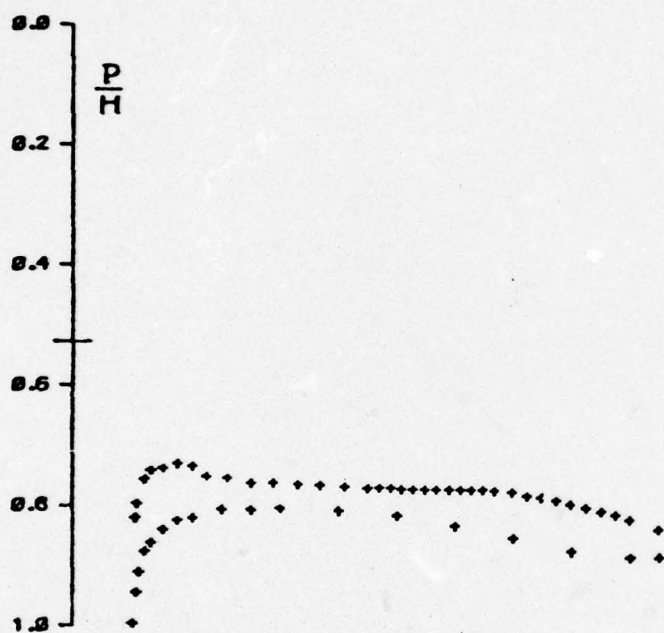
+ BGK-1 203.2 MM CHORD SOLID WALLS
 ME .749 AL= 0.55 CE= 0.586 CE=0.130 R= 1.658



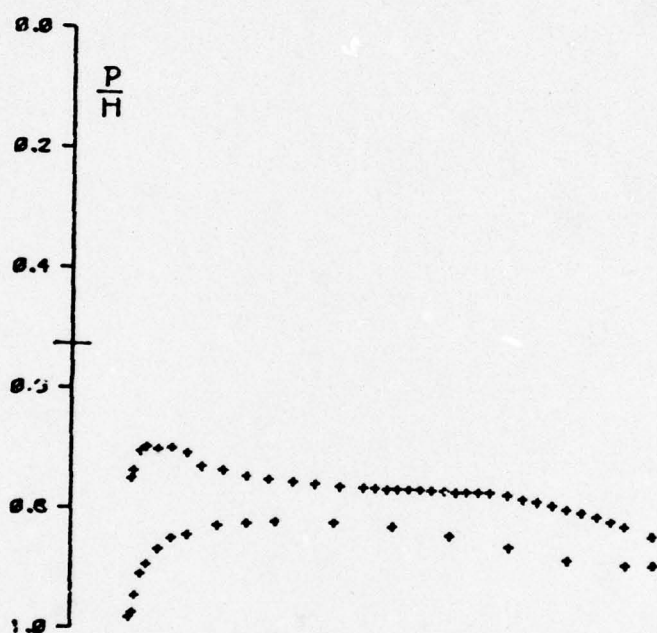
+ BK-1 203.2 MM CHORD SOLID WALLS
 M= .500 AL=2.50 CM=0.031 CM=0.033 R= 1.660



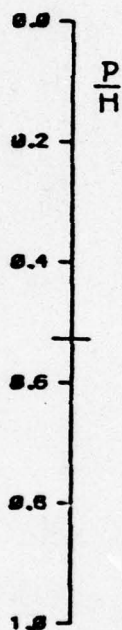
+ BOK-1 203.2 MM CHORD SOLID WALLS
 ME .501 AL= 0.00 CM= 0.276 CM= 0.053 R= 1.051



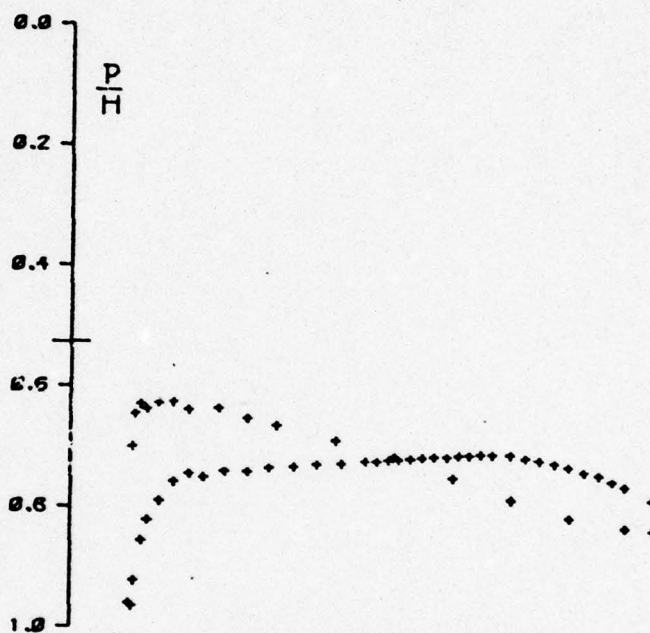
+ BGK-1 283.2 MM CHORD SOLID WALLS
 M= .489 AL= 1.00 CM= 0.394 CM= 0.063 R= 1.656



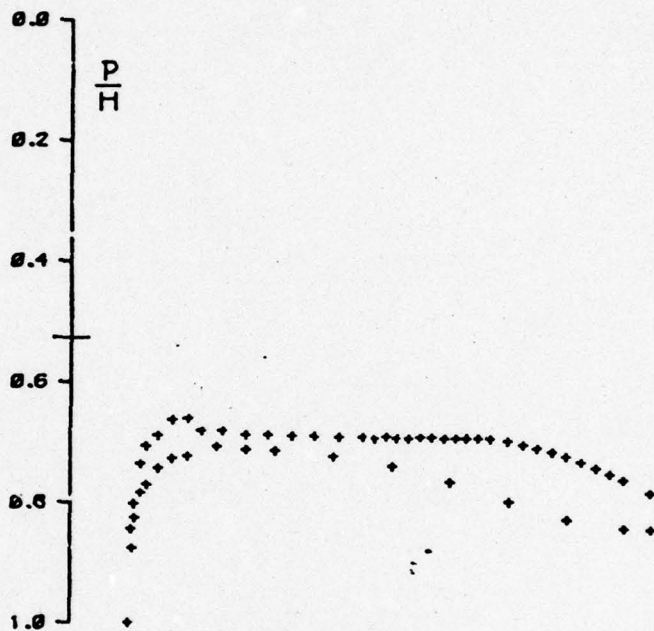
+ BGK-1 253.2 MM CHORD SOLID WALLS
 M= .468 N= 2.00 O= 0.531 O2= 0.003 R= 1.656



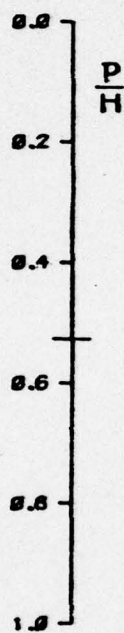
+ BGK-1. 283.2 MM CHORD SOLID WALLS
 ME .489 AL= 4.08 CN= 0.753 CM= 0.877 R= 1.656



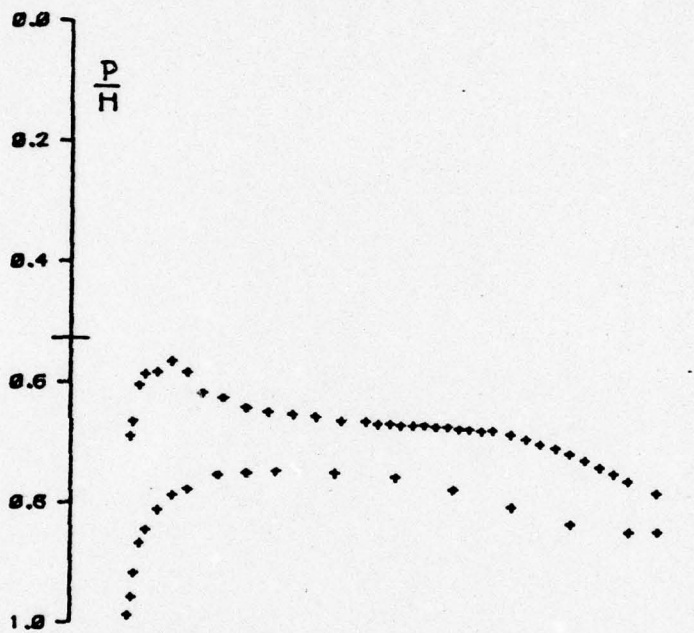
+ BGK-1 203.2 MM CHORD SOLID WALLS
 $M = .586$ $AL = 2.50$ $CH = 0.053$ $DE = 0.006$ $R = 1.643$



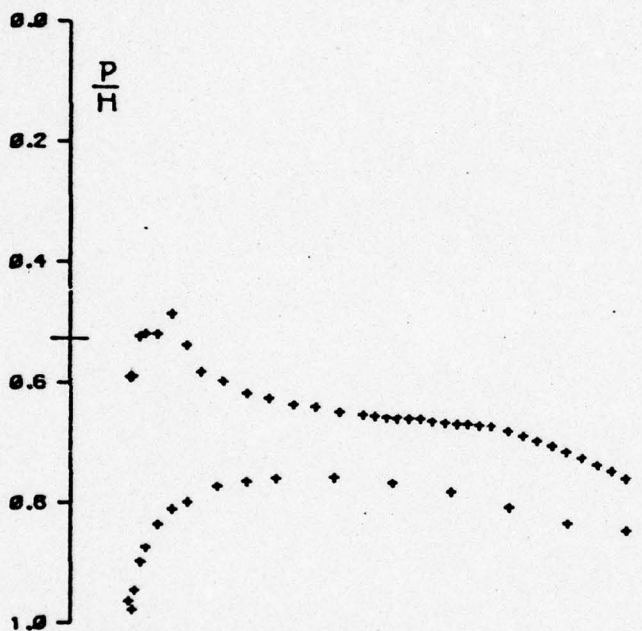
+ B&K-1 203.2 MM CHORD SOLID WALLS
 $M = .598$ $AL = 0.00$ $CM = 0.289$ $CM = 0.059$ $R = 1.643$



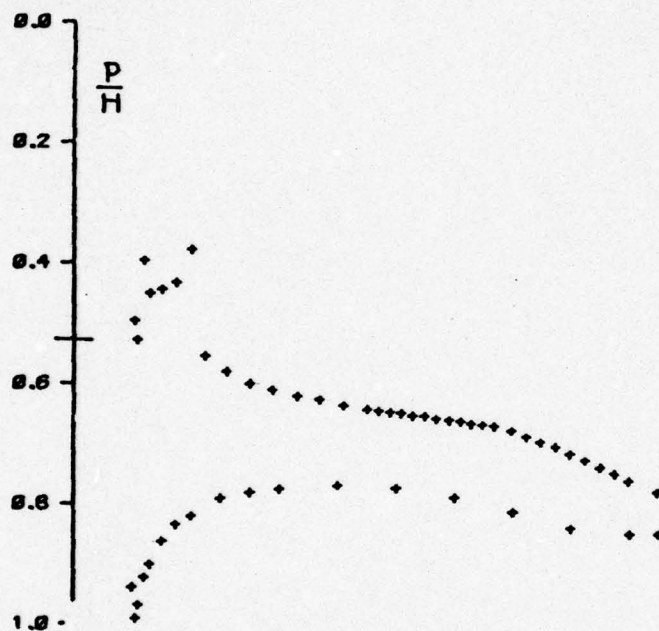
+ BGK-1 223.2 MM CHORD SOLID WALLS
 $M = .691$ $AL = 1.00$ $CM = 0.420$ $CM = 0.086$ $R = 1.651$



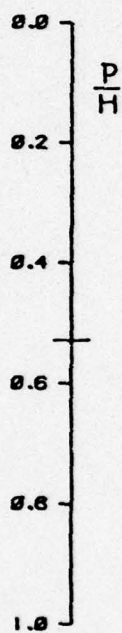
+ BOK-1 253.2 MM CHORD SOLID WALLS
 M= .588 AL= 2.00 CM= 0.518 CM= 0.835 R= 1.645



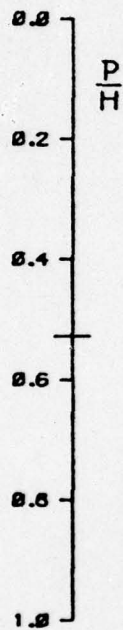
+ BGK-1 283.2 MM CHORD SOLID WALLS
 $\mu = .600$ $AL = 3.00$ $CE = 0.671$ $CE = 0.051$ $R = 1.646$



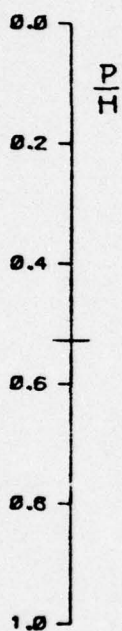
+ BQK-1 203.2 MM CHORD SOLID WALLS
 M= .600 AL= 4.00 CN= 0.023 CM=0.025 R= 1.648



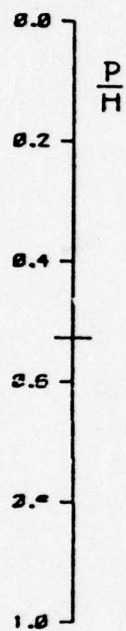
+ 5GK-1 203.2 MM CHORD SOLID WALLS
 M= .651 AL=2.50 CM=0.063 CM=0.081 R= 1.633



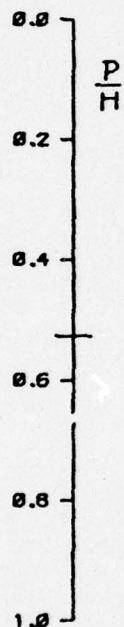
+ BGK-1 203.2 MM CHORD SOLID WALLS
 $M = .652$ $AL = 0.00$ $CN = 0.296$ $CM = 0.082$ $R = 1.640$



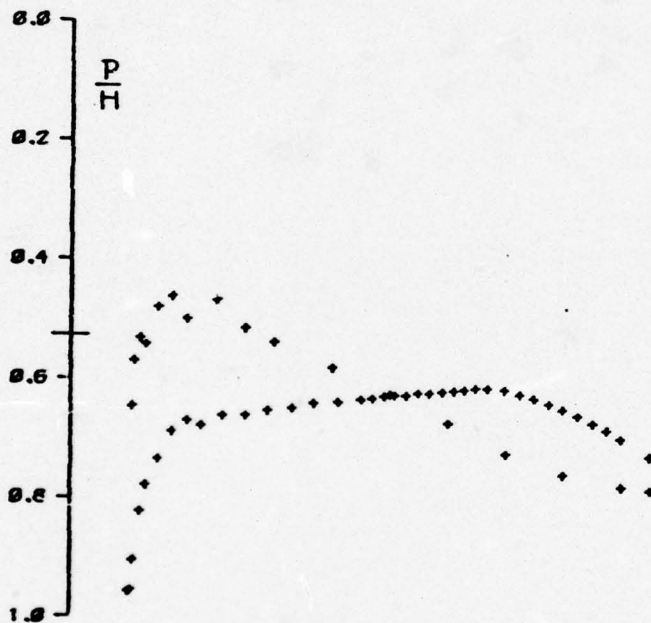
* BGR-1 283.2 MM CHORD SOLID WALLS
 ME .649 AL= 1.00 CF= 0.436 CM= 0.090 R= 1.632



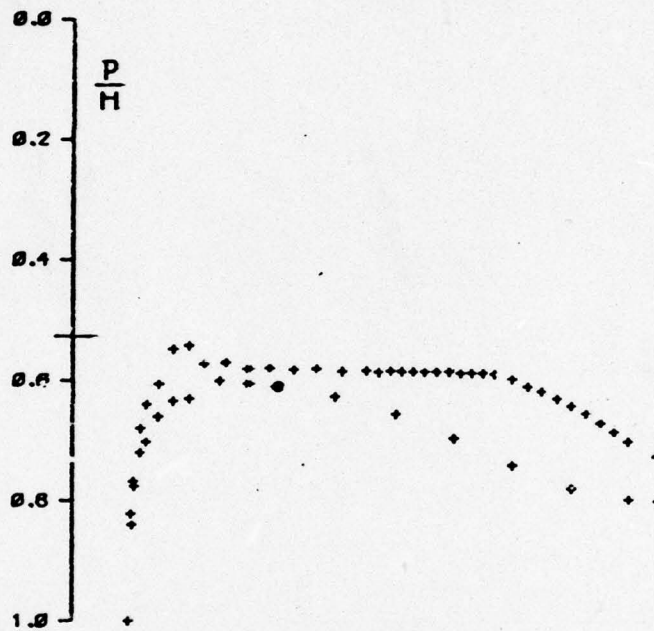
+ BGK-1 203.2 MM CHORD SOLID WALLS
 ME .649 AL= 2.00 CWF 2.568 CWF=0.086 R= 1.532



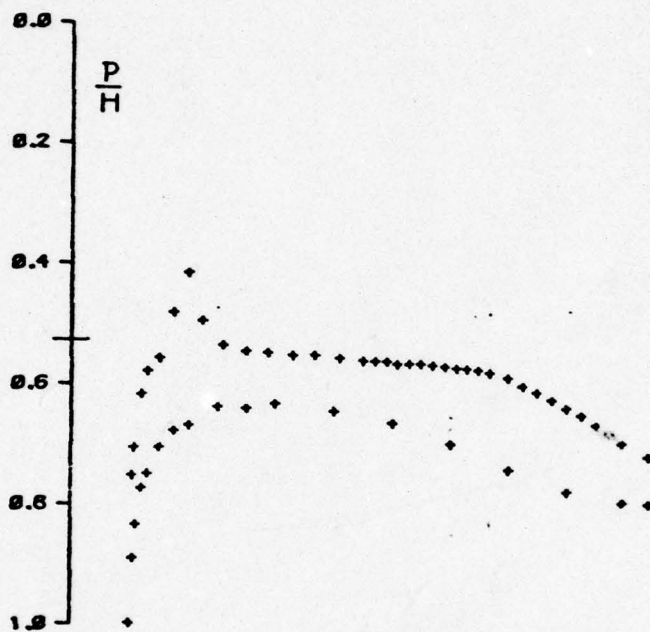
BGK-1 203.2 MM CHORD SOLID WALLS
 ME .649 AL= 3.00 ONE 0.719 ONE=0.081 R= 1.635



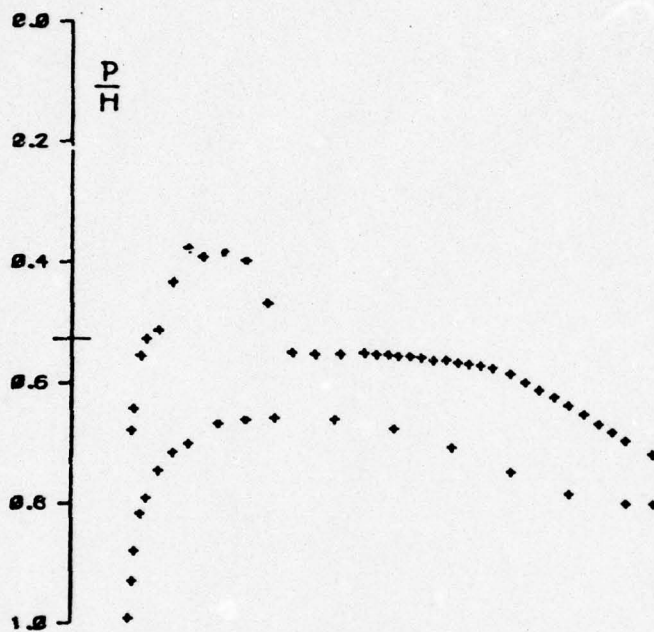
+ BGK-1 283.2 MM CHORD SOLID WALLS
 M= .688 AL=2.50 ON=0.082 OM=0.085 R= 1.625



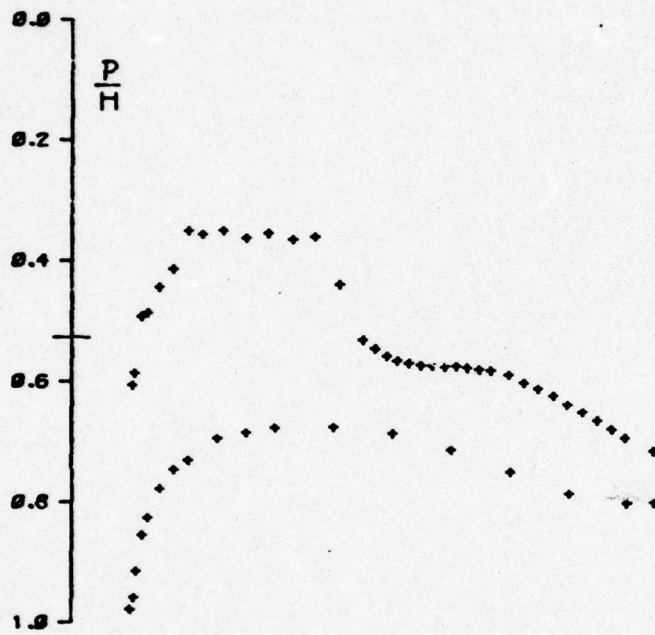
+ BGR-1 203.2 MM CHORD SOLID WALLS
 M= .781 AL= 0.00 CM= 0.302 CM= 0.055 R= 1.632



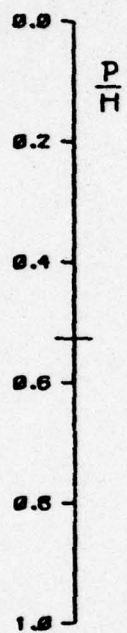
+ BOK-1 203.2 MM CHORD SOLID WALLS
 ME 701 AL= 1.00 CH= 0.452 CM= 0.082 R= 1.632



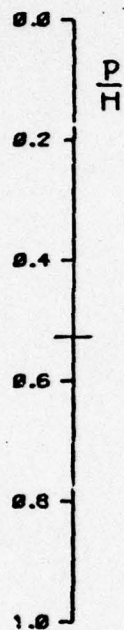
+ BGR-1 203.2 MM CHORD SOLID WALLS
 ME .700 AL= 2.00 CE 0.619 CM=0.007 RE 1.632



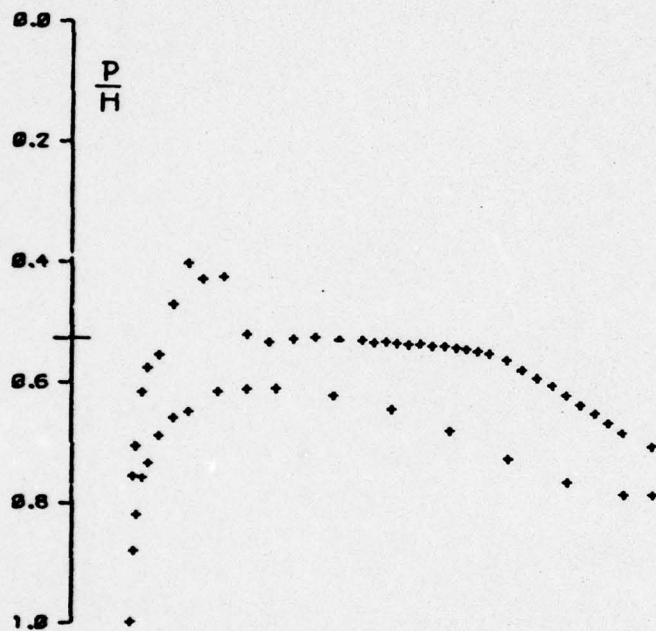
+ BOK-1 203.2 MM CHORD SOLID WALLS
 ME .689 AL 3.00 CM 0.797 CM 1.085 RE 1.633



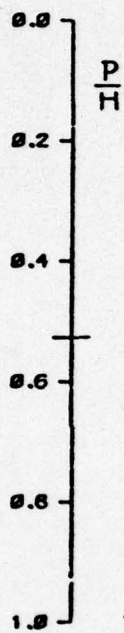
+ BCK-1 203.2 MM CHORD SOLID WALLS
 ME = .720 AL = 2.50 CN = 0.112 CM = 0.098 R = 1.645



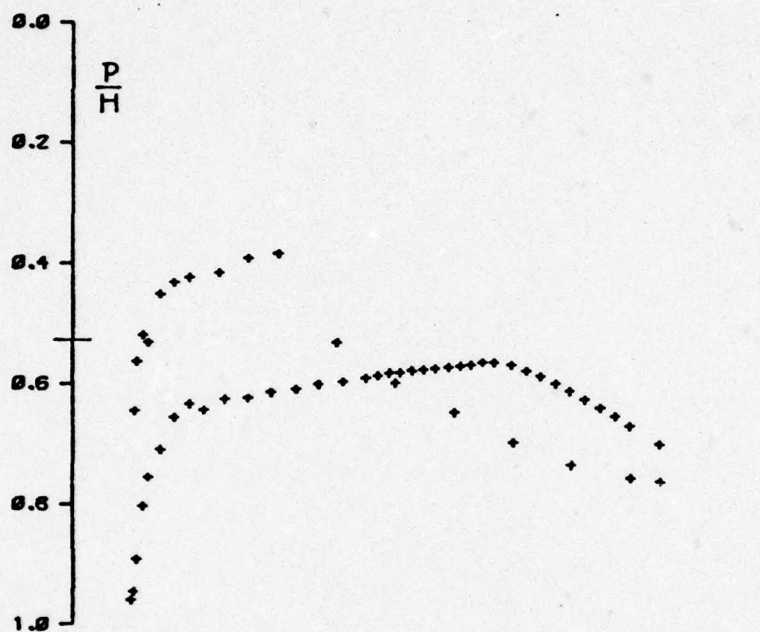
+ BOK-1 203.2 MM CHORD SOLID WALLS
 $M = .729$ $A_L = 0.00$ $C_N = 0.306$ $C_D = 0.087$ $R = 1.646$



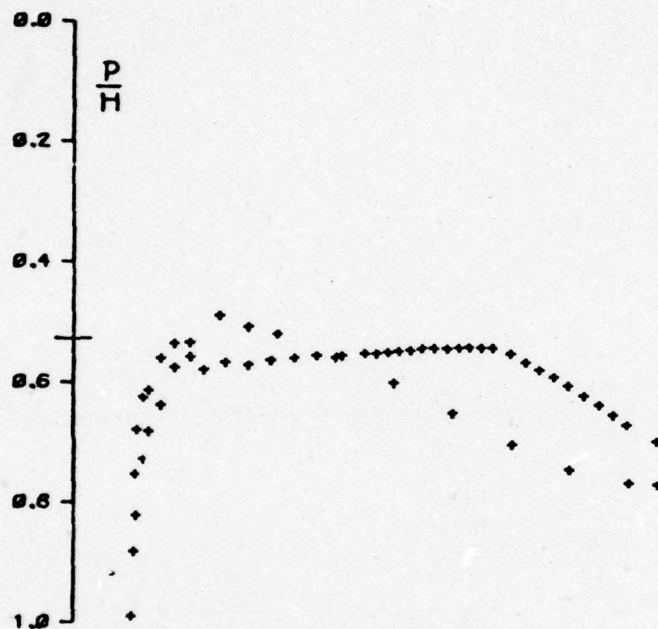
+ BGK-1 203.2 MM CHORD SOLID WALLS
 $M = .721$ $AL = 1.00$ $CM = 0.467$ $CM = 0.063$ $R = 1.646$



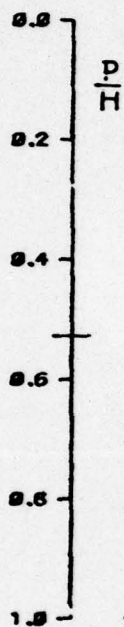
+ BOK-1 203.2 MM CHORD SOLID WALLS
 ME .719 AL= 2.00 CM= 0.652 CM= 0.080 R= 1.646



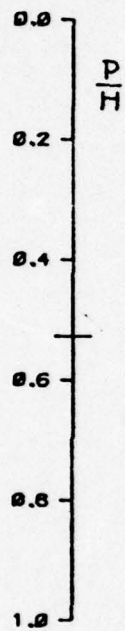
+ BGK-1 223.2 MM CHORD SOLID WALLS
 M= .740 AL=2.50 CB=0.113 CB=0.101 R= 1.643



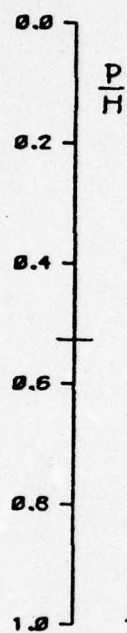
+ BGK-1 223.2 MM CHORD SOLID WALLS
 ME .749 AL=1.00 CM=0.145 CM=0.039 R=1.645



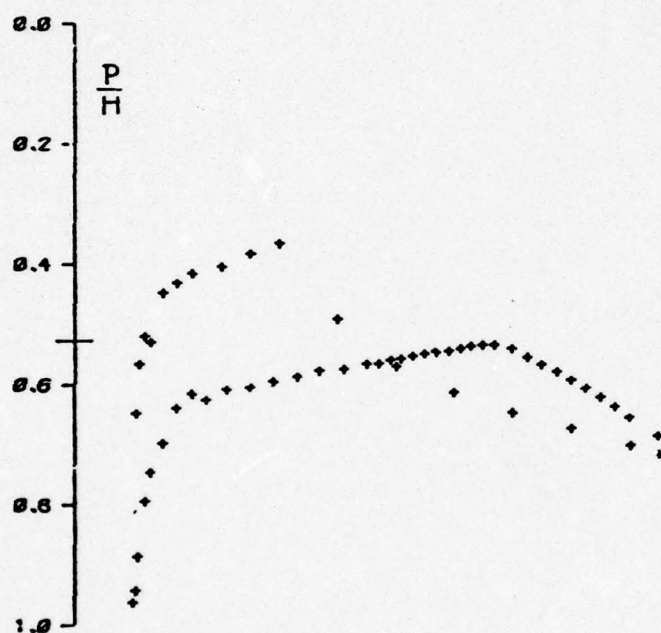
+ BGK-1 203.2 MM CHORD SOLID WALLS
 ME .741 AL= 0.00 ON= 0.306 ON=0.088 R= 1.646



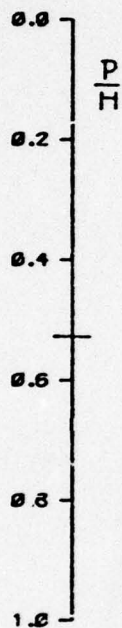
+ BGK-1 203.2 MM CHORD SOLID WALLS
 ME .740 AL= 1.00 CM= 0.491 CM=0.034 R= 1.643



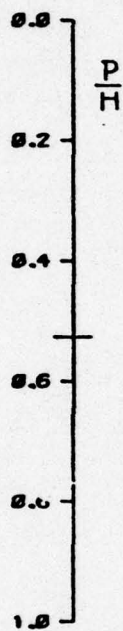
+ BGK-1 203.2 MM CHORD SOLID WALLS
 ME .740 AL= 2.00 CN= 0.667 CM=0.888 R= 1.643



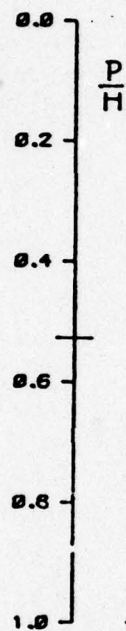
+ BGK-1 283.2 MM CHORD SOLID WALLS
 M= .759 AL=2.50 CN=0.152 CN=0.073 R= 1.628



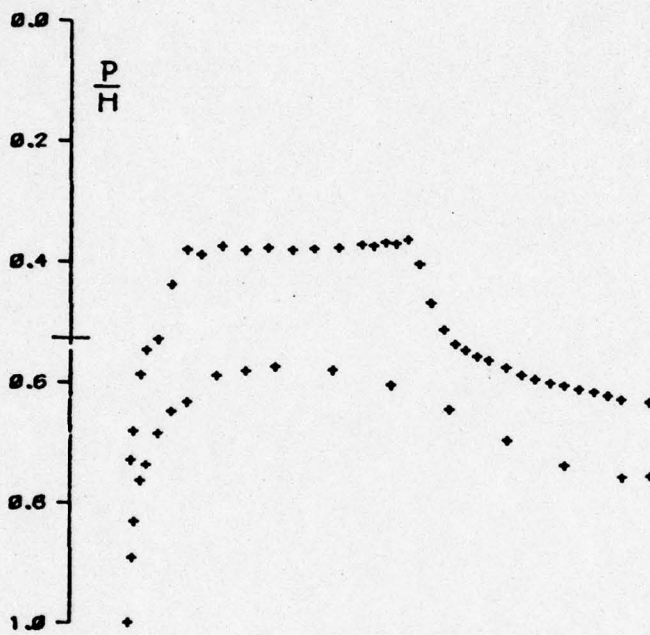
+ BGK-1 203.2 MM CHORD SOLID WALLS
 ME .758 AL>1.00 OE=0.134 OM=0.103 RE 1.628



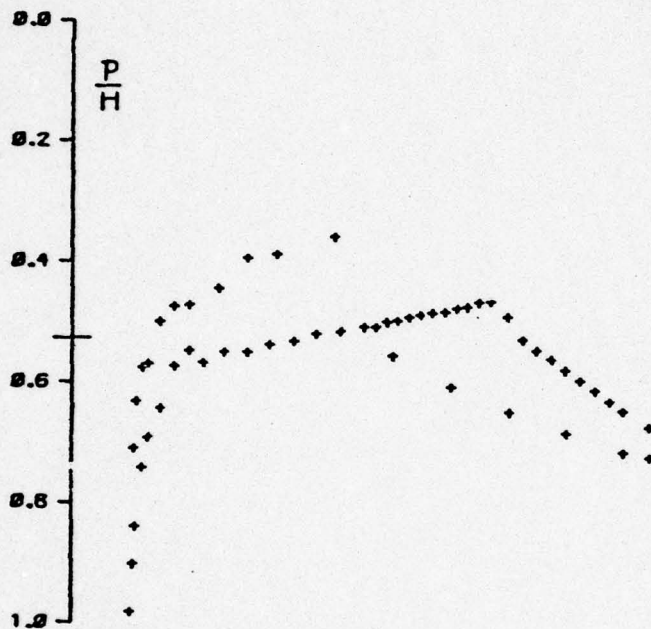
+ 3GK-1 203.2 MM CHORD SOLID WALLS
 M= .760 AL= 0.00 CN= 0.306 CM= 0.100 R= 1.630



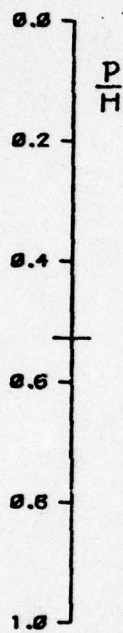
+ BGK-1 203.2 MM CHORD SOLID WALLS
 ME .759 AL= 1.00 CM= 0.498 CM= 0.097 R= 1.626



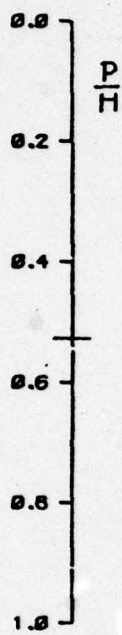
+ BGK-1 203.2 MM CHORD SOLID WALLS
 M= .780 AL= 2.20 CT= 0.576 CM= 0.087 R= 1.633



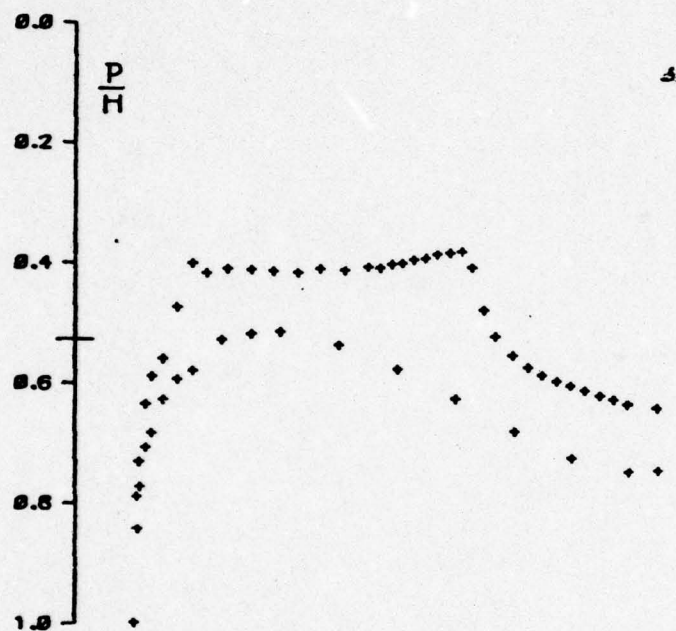
+ 20K-1 203.2 MM CHORD SOLID WALLS
 ME .762 AL-2.50 CM-0.013 CM-0.062 R= 1.607



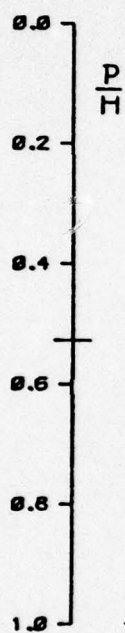
+ BGK-1 203.2 MM CHORD SOLID WALLS
 ME .780 AL=1.00 CM=0.099 CM=0.101 R=1.602



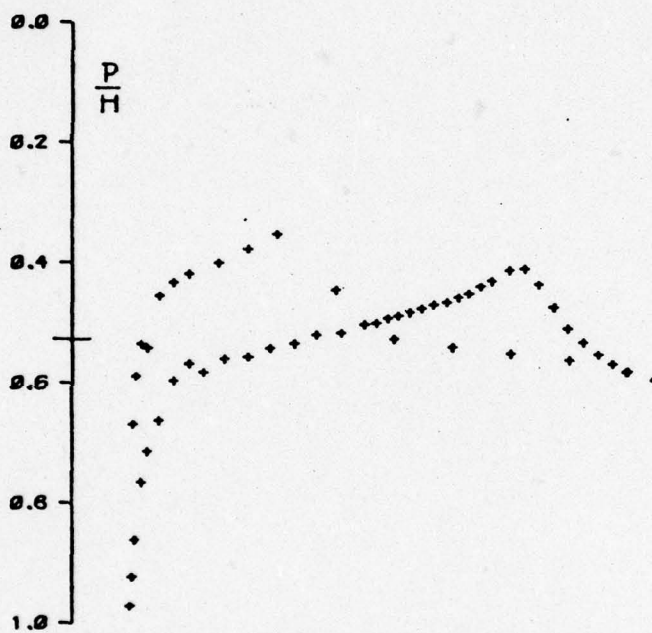
+ BGK-1 203.2 MM CHORD SOLID WALLS
 M= .776 AL= 0.00 CM= 0.301 CM= 0.106 R= 1.589



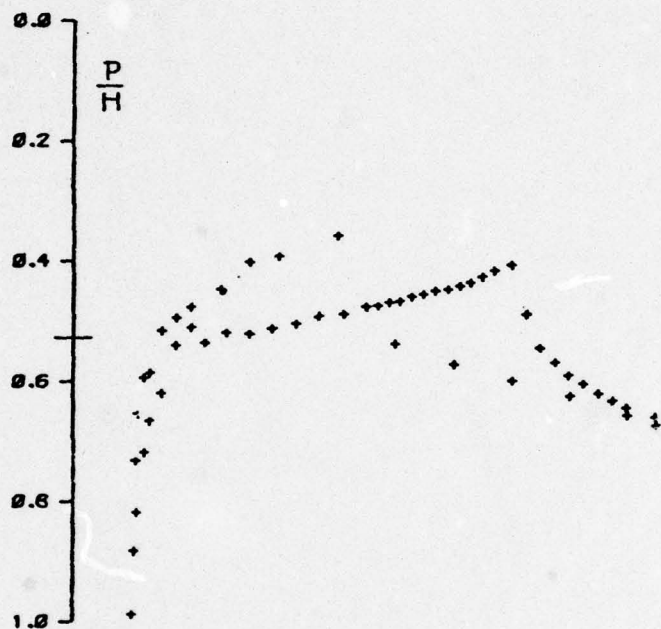
+ BOK-1 203.2 MM CHORD SOLID WALLS
 $M = .789$ $AL = 1.00$ $CM = 0.438$ $CM = 0.101$ $R = 1.600$



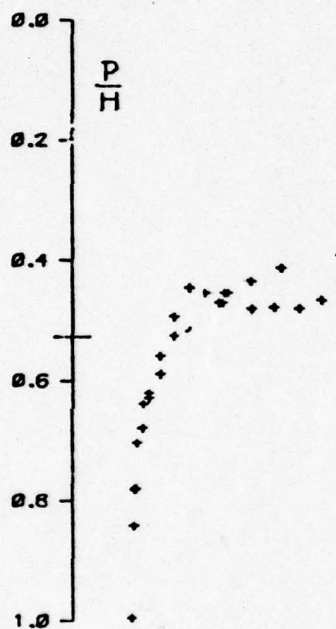
+ BGK-1 223.2 MM CHORD SOLID WALLS
 M= .781 AL= 2.00 CM= 0.476 CM= 0.085 R= 1.612



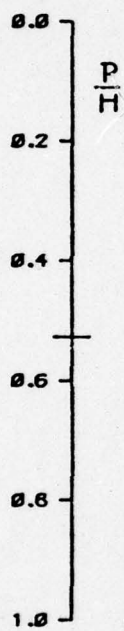
+ BGK-1 203.2 MM CHORD SOLID WALLS
 $M = .620$ $A_L = 2.50$ $C_N = 0.101$ $C_M = 0.055$ $R = 1.637$



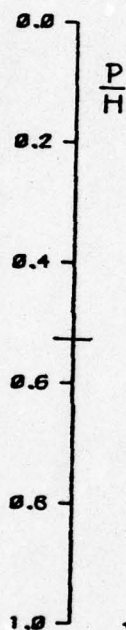
+ BGK-1 203.2 MM CHORD SOLID WALLS
 $M = .799$ $AL = 1.00$ $CL = 0.024$ $CL = 0.053$ $R = 1.630$



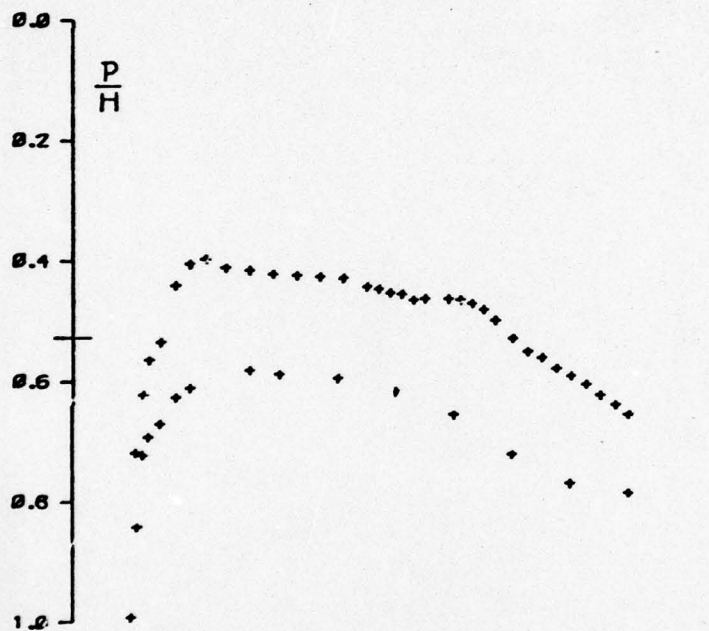
+ BGK-1 203.2 mm CHORD SOLID WALLS
 M= .799 AL= 0.00 CN= 0.169 CM=0.091 R= 1.632



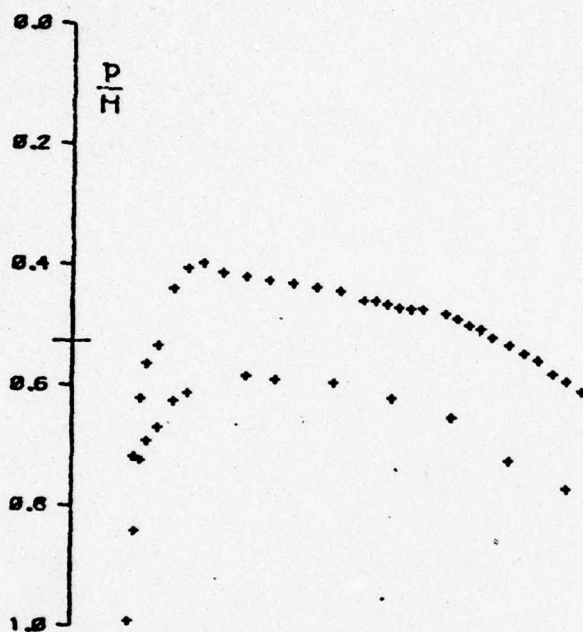
+ BGK-1 203.2 MM CHORD SOLID WALLS
 ME .001 AL= 1.00 CM= 0.325 CM= 0.089 R= 1.637



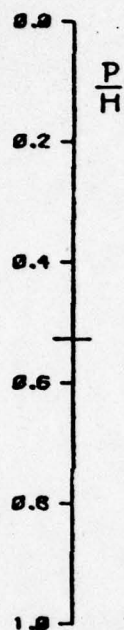
+ B&K-1 223.2 MM CHORD SOLID WALLS
 ME .798 AL= 2.00 CM= 0.400 CM= 0.098 R= 1.633



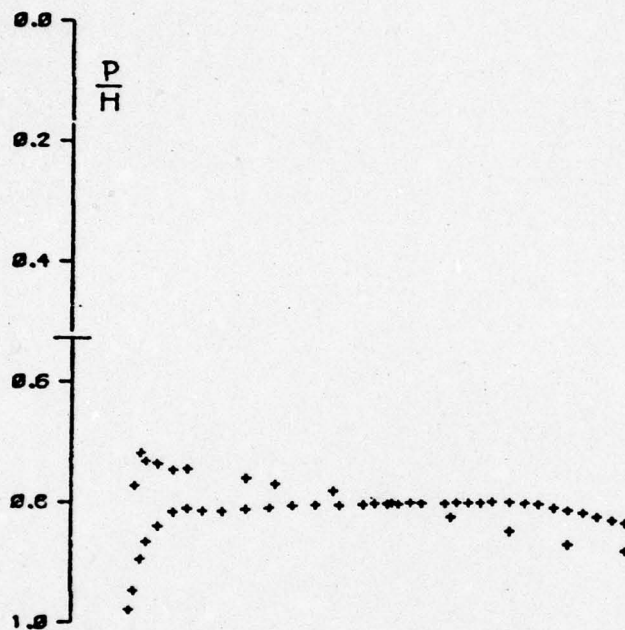
+ BGK-1 101.6 MM CHORD SOLID WALLS
 M = .753 AL = 0.60 ON = 0.563 OM = 0.116 R = 0.629



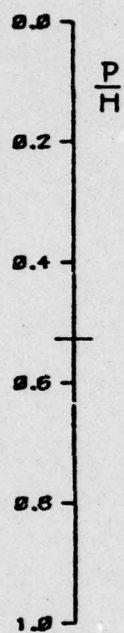
+ NGR-1 101.6 MM CHORD SOLID WALLS
 $M = 0.752$ $A_L = 0.68$ $C_{D1} = 0.546$ $C_{D2} = 0.115$ $R = 0.629$



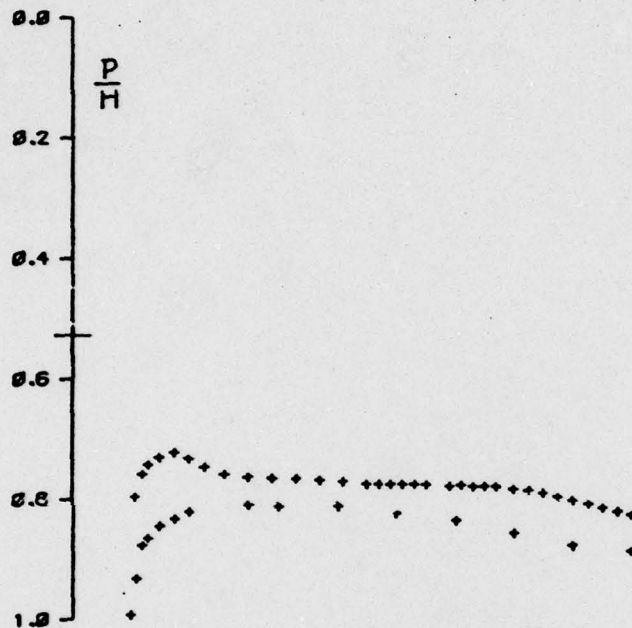
+ BGK-1 101.6 MM CHORD SOLID WALLS
 M= 754 AL= 0.60 CL= 0.548 CM= 0.115 R= 1.400



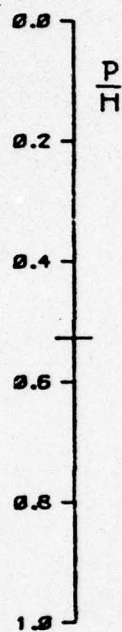
+ BGK-1 101.6 MM CHORD SOLID WALLS
 M= .502 AL=2.50 CN=0.034 CM=0.075 R= 0.630



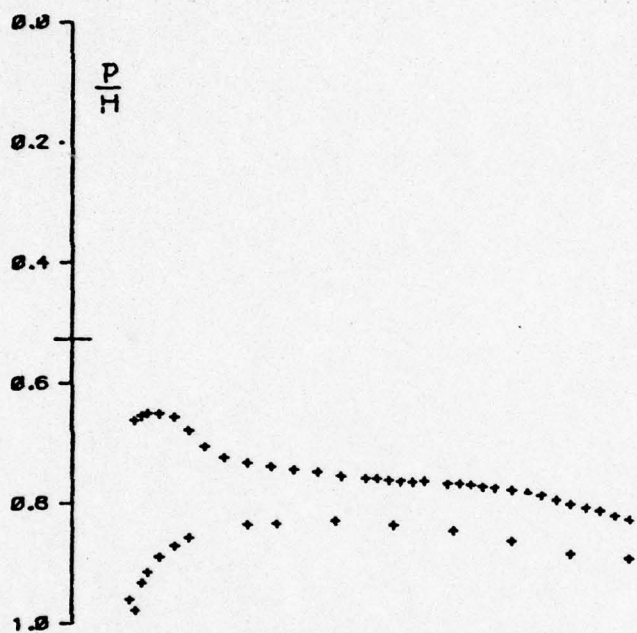
+ BGK-1 101.6 MM CHORD SOLID WALLS
M= .500 AL= 0.00 CL= 0.275 CM= 0.077 R= 0.529



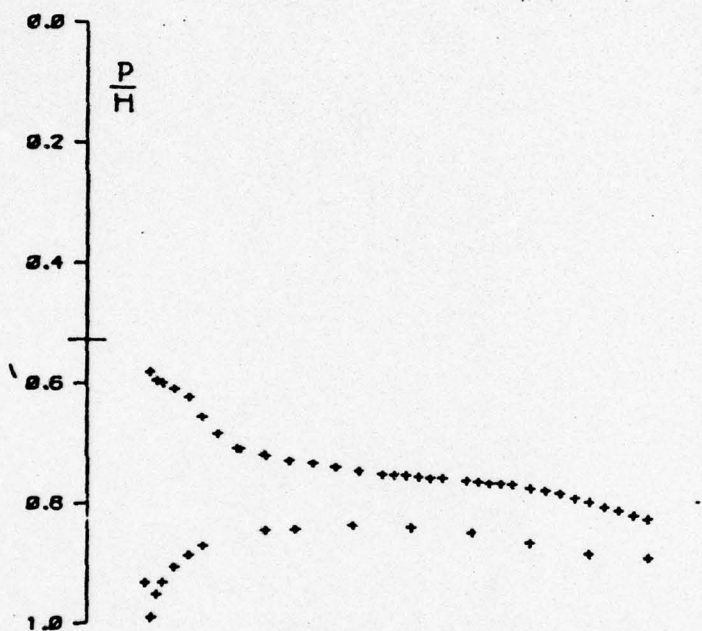
+ BGK-1 101.6 MM CHORD SOLID WALLS
 M= .500 AL= 1.00 ON= 0.391 OM= 0.076 R= 0.629



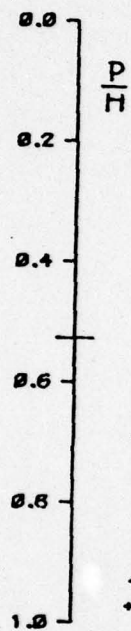
+ BGK-1 101.6 MM CHORD SOLID WALLS
 $M = .500$ $AL = 2.00$ $CL = 0.504$ $CL = 0.075$ $R = 0.629$



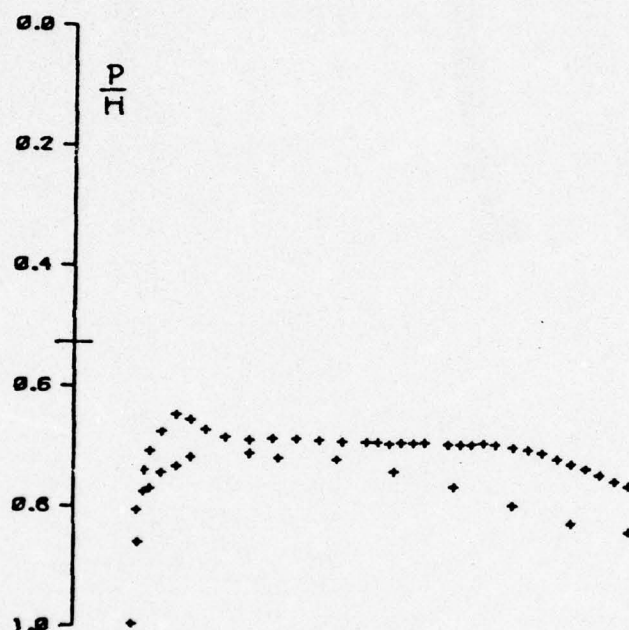
+ BGK-1 101.6 MM CHORD SOLID WALLS
 M= .502 AL= 3.00 CM=0.615 CM=0.073 R= 0.637



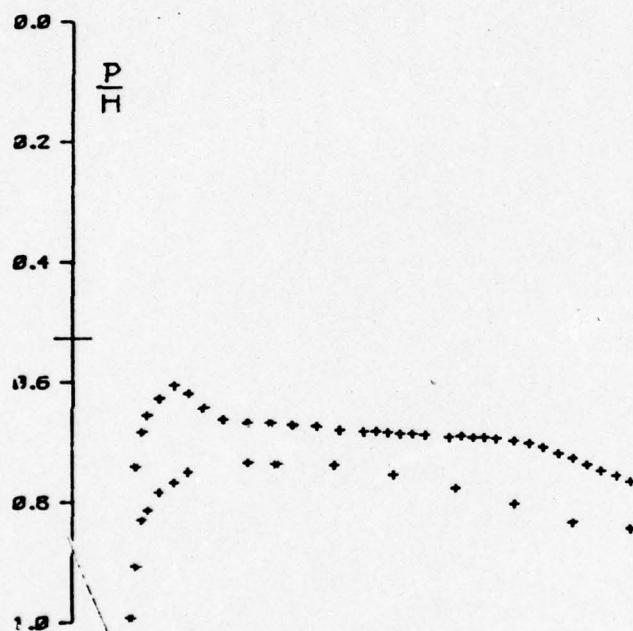
+ BGK-1 101.6 MM CHORD SOLID WALLS
 $M = .499$ $AL = 4.00$ $Q = 2.724$ $Q_1 = 0.071$ $R = 0.629$



+ BGK-1 101.6 MM CHORD SOLID WALLS
 ME .601 AL 2.50 CE 0.051 CM 0.002 R 0.014



+ BGK-1 101.6 MM CHORD SOLID WALLS
 M= .620 AL= 0.00 CN= 0.285 CM= 0.031 R= 0.814



+ BGC-1 101.6 MM O.D. SOLID WALLS
 ME = .000 AL = 1.00 CE = 0.411 CM = 0.000 R = 0.814

AD-A076 131

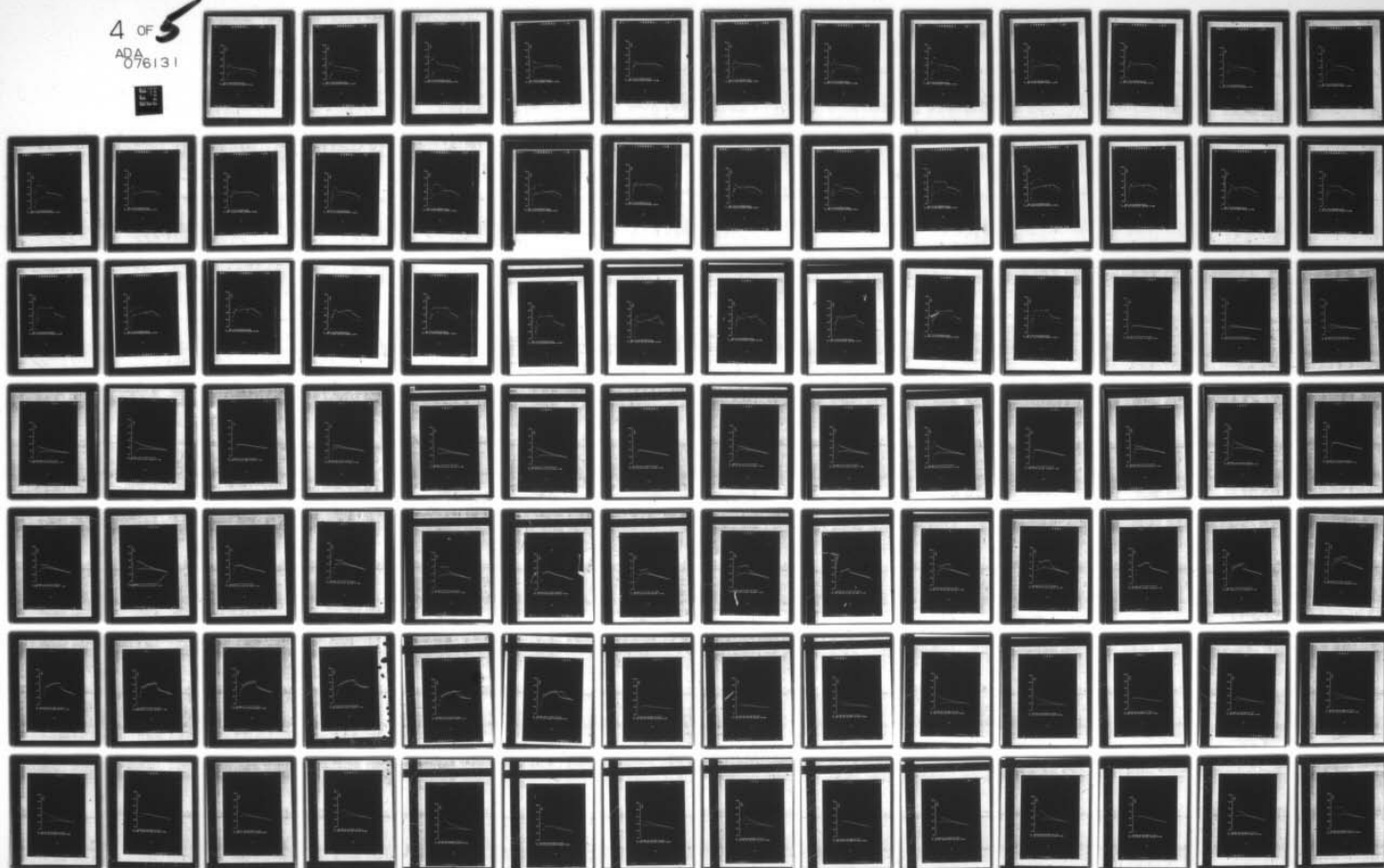
AERONAUTICAL RESEARCH LABS MELBOURNE (AUSTRALIA)
TRANSONIC WIND TUNNEL TESTS ON A SERIES OF TWO-DIMENSIONAL AERO--ETC(U)
JAN 79 B D FAIRLIE , N POLLOCK
ARL/AERO NOTE-384

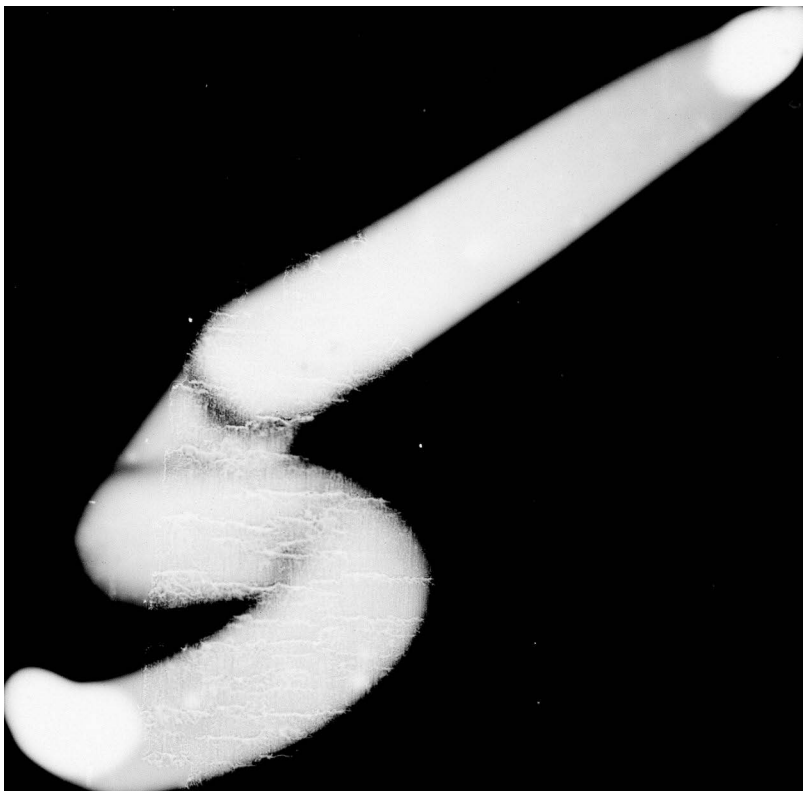
F/G 20/4

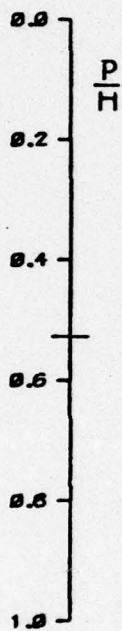
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NL

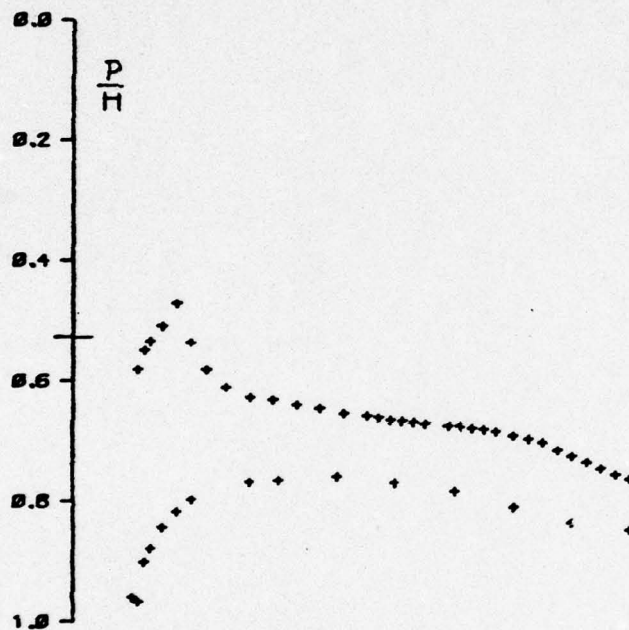
4 OF 5
AD-A076131



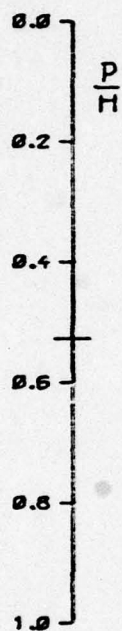




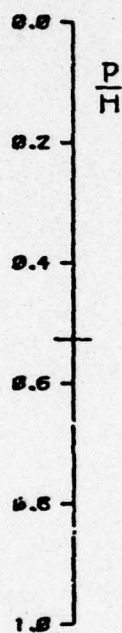
+ BGK-1 101.6 MM CHORD SOLID WALLS
 ME .600 AL 2.00 CM 0.534 CM 0.078 R 0.814



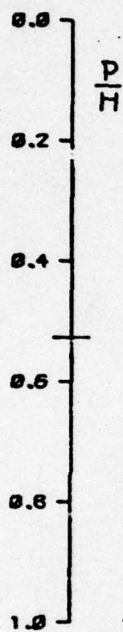
+ DGR-1 101.6 MM CHORD SOLID WALLS
 M= .601 AL= 3.00 CM= 0.646 CM= 0.074 R= 0.617



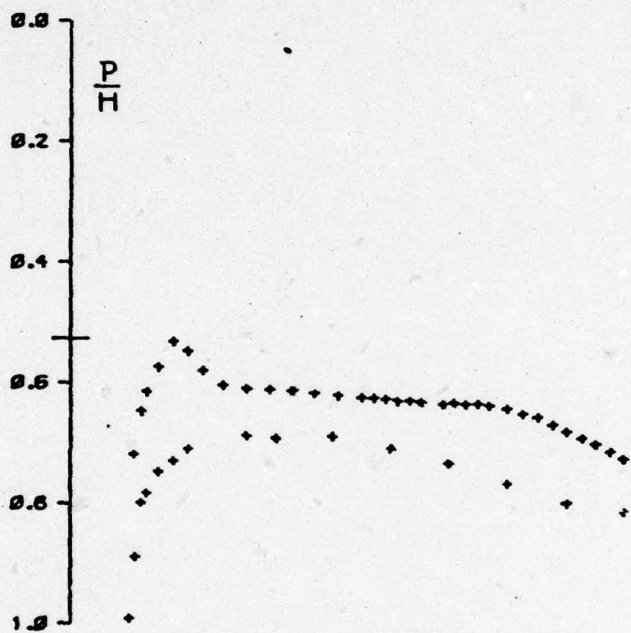
+ BGR-1 101.6 MM CHORD SOLID WALLS
 ME SSS AL= 4.02 CM= 0.765 CM= 0.088 R= 0.612



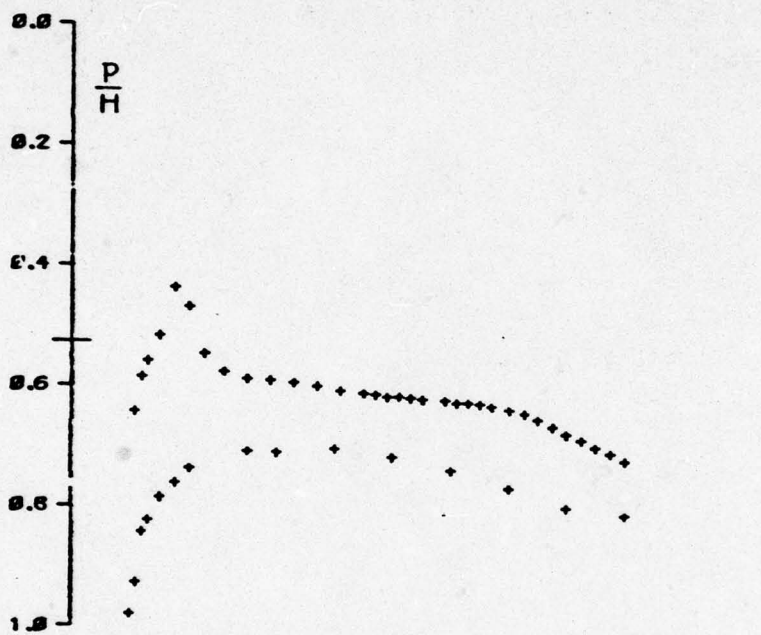
+ BOK-1 101.6 MM CHORD SOLID WALLS
 ME .649 AL-2.50 CN-0.066 CM-0.065 R= 0.039



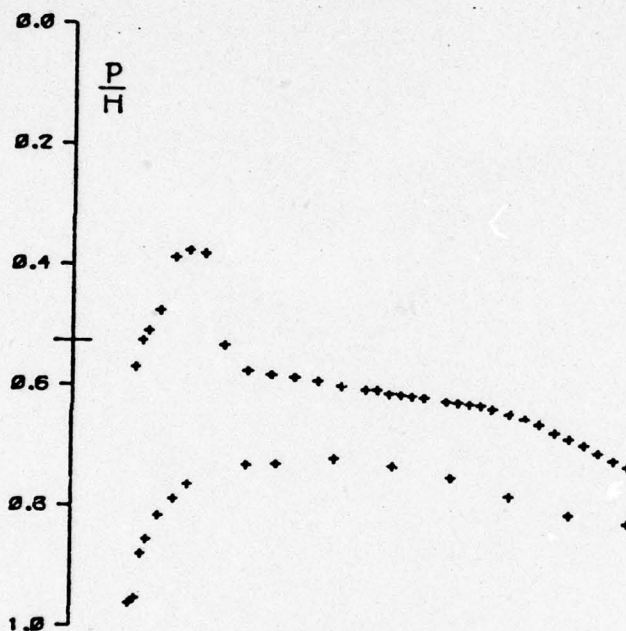
+ BGK-1 101.6 MM CHORD SOLID WALLS
 ME .649 AL= 0.00 DE 0.250 CM=0.054 R= 0.020



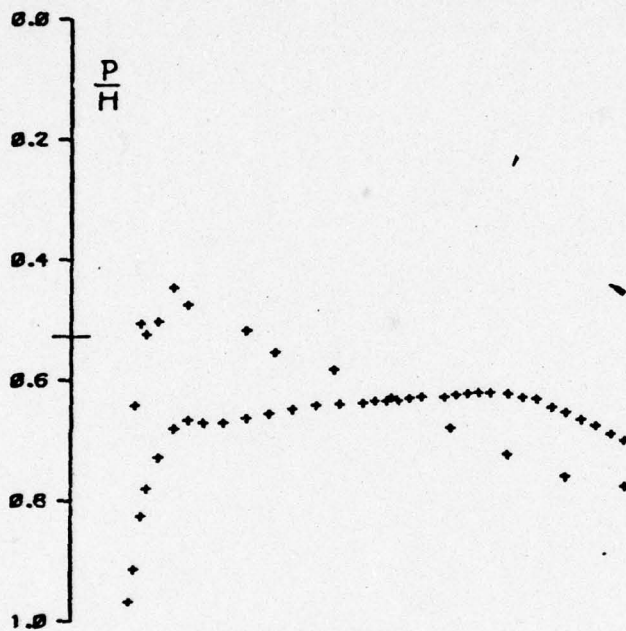
+ BGK-1 101.6 MM CHORD SOLID WALLS
 $M = .652$ $AL = 1.00$ $CR = 0.419$ $CM = 0.002$ $R = 0.011$



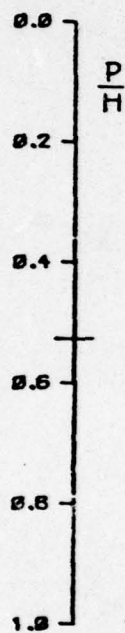
+ BOK-1 101.6 MM CHORD SOLID WALLS
 ME .648 AL= 2.00 CM= 0.547 CM=0.078 R= 0.000



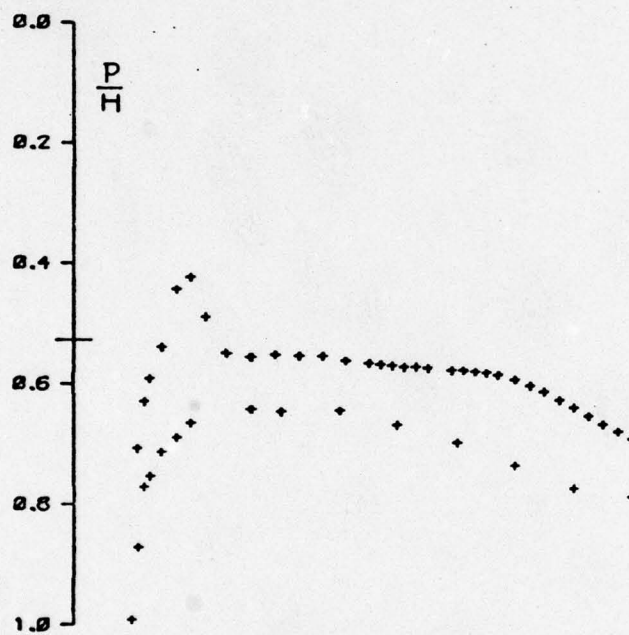
+ BGK-1 101.6 MM CHORD SOLID WALLS
 M= .650 AL= 3.00 CF= 0.674 CM= 0.873 R= 0.812



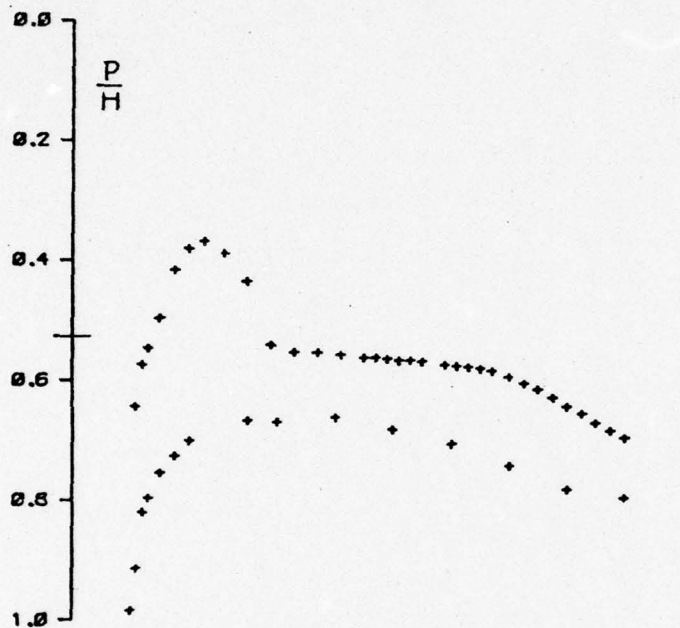
+ BGK-1 101.6 MM CHORD SOLID WALLS
 ME .700 AL=2.50 CB=0.039 CB=0.031 R=0.631



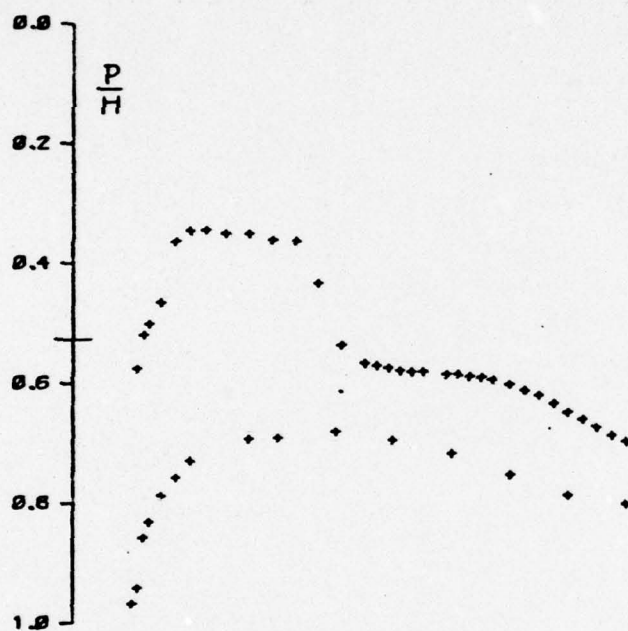
+ BGK-1 101.6 MM CHORD SOLID WALLS
 M= .700 AL= 0.00 CL= 0.294 CM= 0.005 R= 0.611



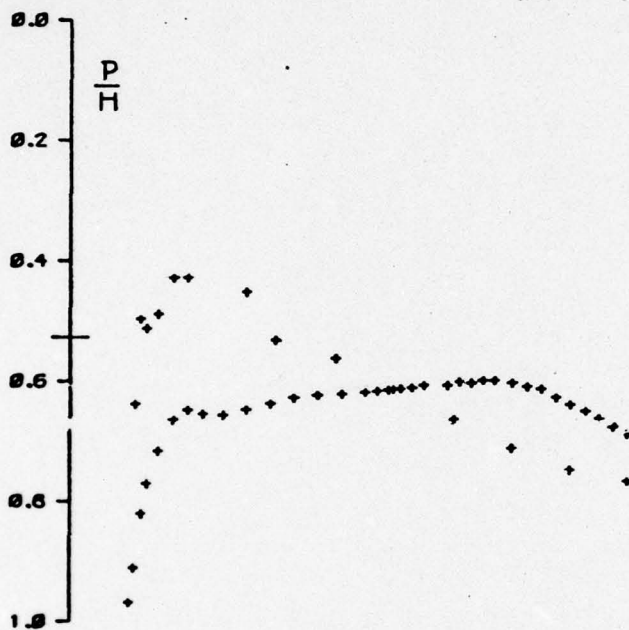
+ BGK-1 101.6 MM CHORD SOLID WALLS
 N= .701 AL= 1.00 CF= 0.442 CM= 0.084 R= 0.611



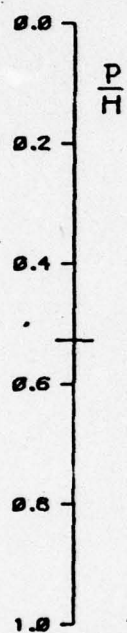
+ BGK-1 101.6 MM CHORD SOLID WALLS
 ME .689 AL= 2.00 CN= 0.583 CM=0.079 R= 0.809



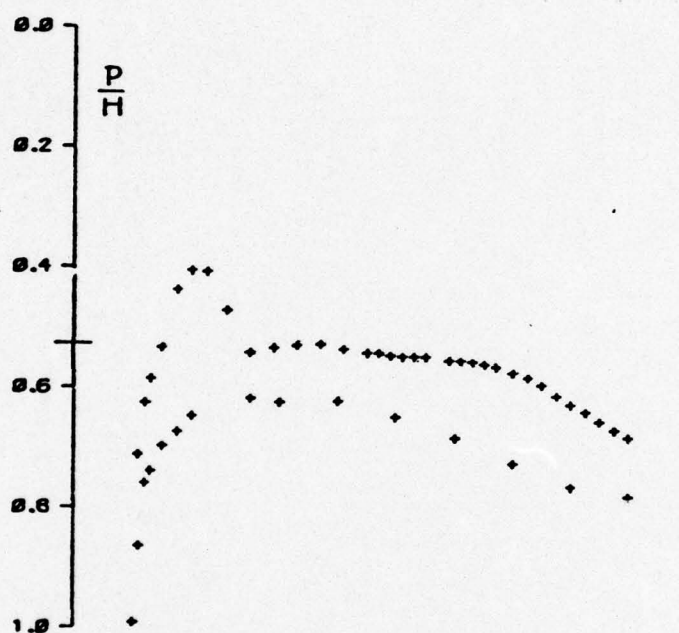
+ BGK-1 101.6 MM CHORD SOLID WALLS
 ME 700 AL= 3.00 CE= 0.756 CF= 0.074 R= 0.811



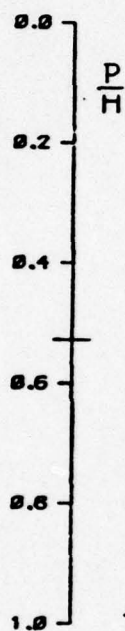
+ BGK-1 101.6 MM CHORD SOLID WALLS
 ME .722 AL-2.50 CM-0.100 CM-0.083 R=0.811



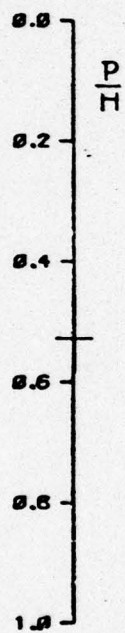
+ BGK-1 101.6 MM CHORD SOLID WALLS
 M= .722 AL= 0.00 CL= 0.294 CM= 0.009 R= 0.009



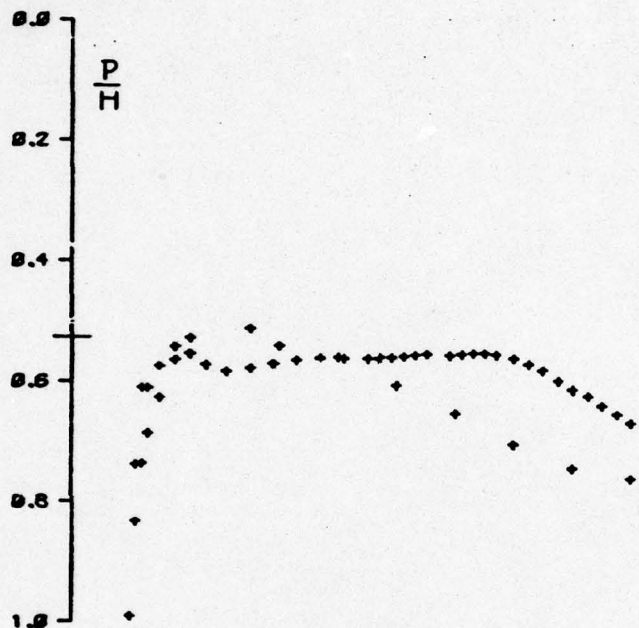
+ BOK-1 101.6 M⁴ CHORD SOLID WALLS
 $M = .721$ $AL = 1.20$ $CE = 0.449$ $CM = 0.284$ $R = 0.607$



+ BGK-1 121.6 MM CHORD SOLID WALLS
 M= .719 AL= 2.00 CM= 0.617 CM= 0.078 R= 0.337

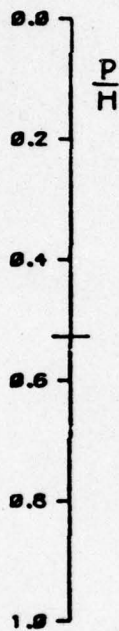


+ BOK-1 101.6 MM CHORD SOLID WALLS
 ME 739 AL=2.50 CN=0.129 CM=0.085 R=0.811

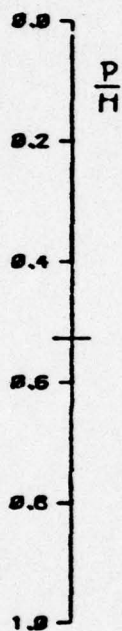


+ BOK-1 101.6 MM CHORD SOLID WALLS
 N= 740 AL=1.00 DT=0.140 CM=0.252 R=0.611

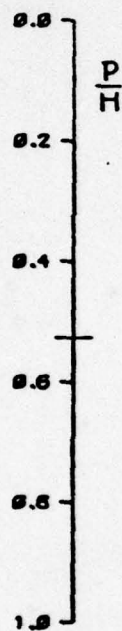
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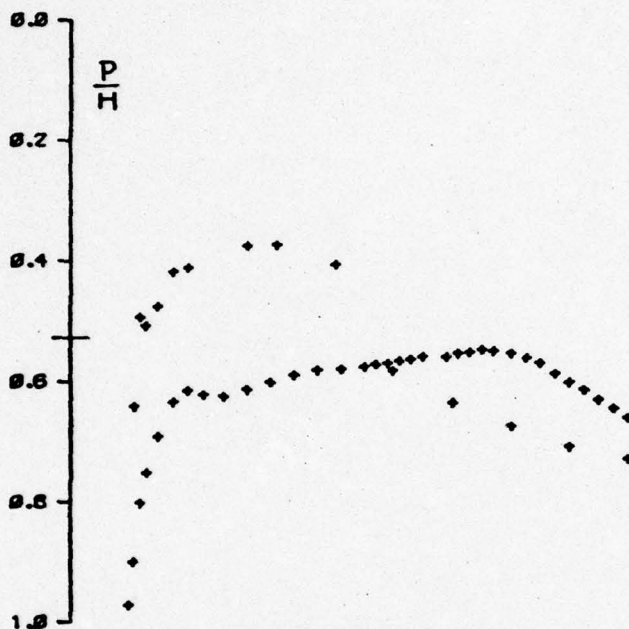
+ BGK-1 101.6 MM CHORD SOLID WALLS
 M= 740 AL= 0.00 ON= 0.300 OM= 0.600 R= 0.611



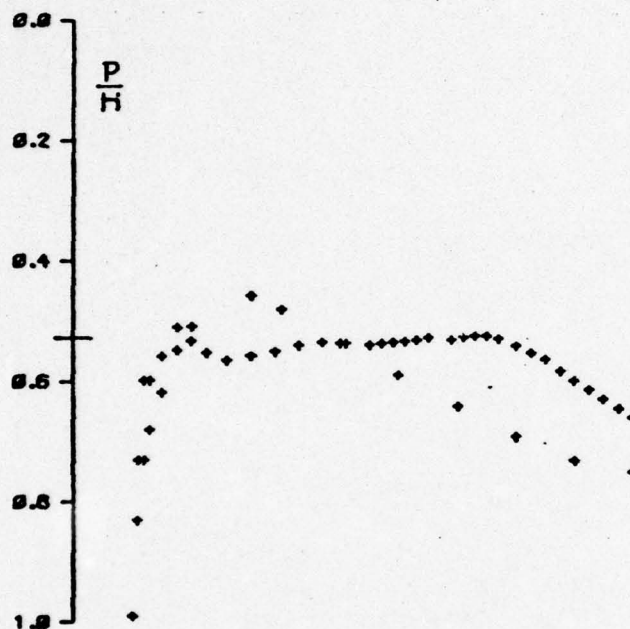
+ BGK-1 101.6 MM CHORD SOLID WALLS
 ME 738 AL= 1.00 CM= 0.463 CM= 0.055 R= 0.611



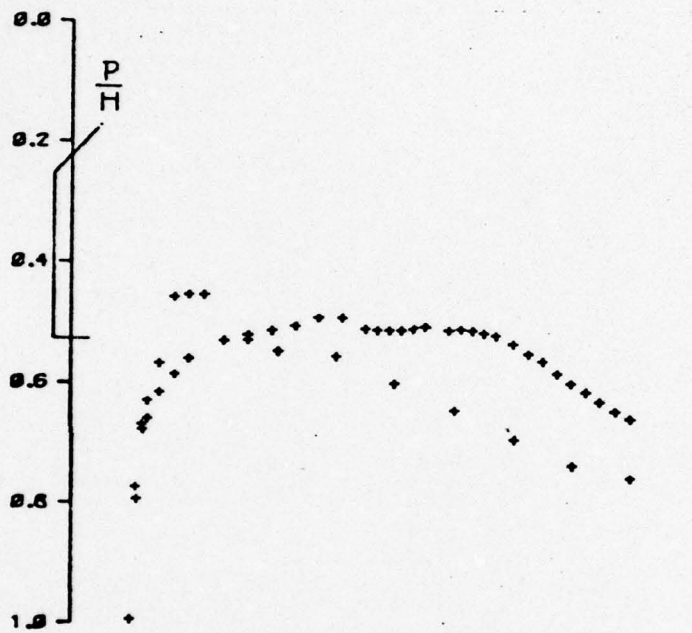
+ DEK-1 101.6 MM CHORD SOLID WALLS
 M= 0.739 AL= 2.00 CL= 0.641 CR= 0.953 R= 0.611



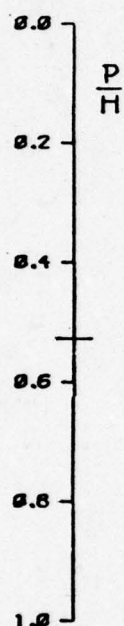
+ BOK-1 101.6 MM CHORD SOLID WALLS
 M= .761 AL=2.50 CN=0.16Z CM=0.034 R= 0.020



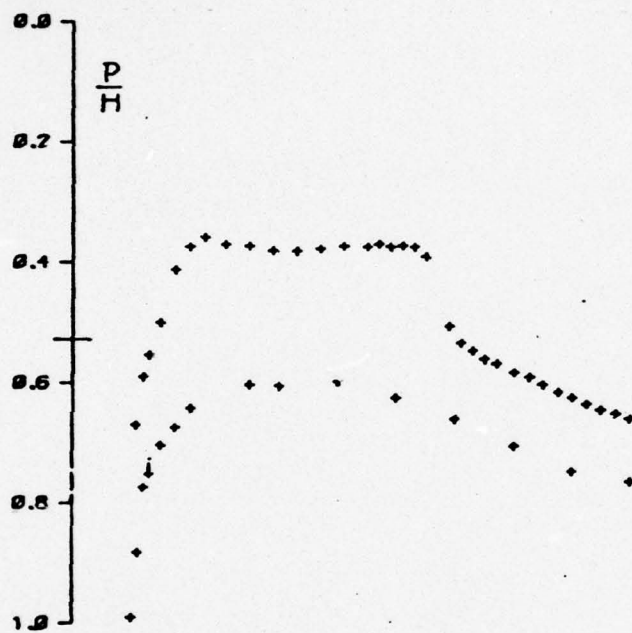
+ BGK-1 101.6 MM CHORD SOLID WALLS
 ME 762 AL-1.00 CM=0.124 CM=0.055 R=0.000



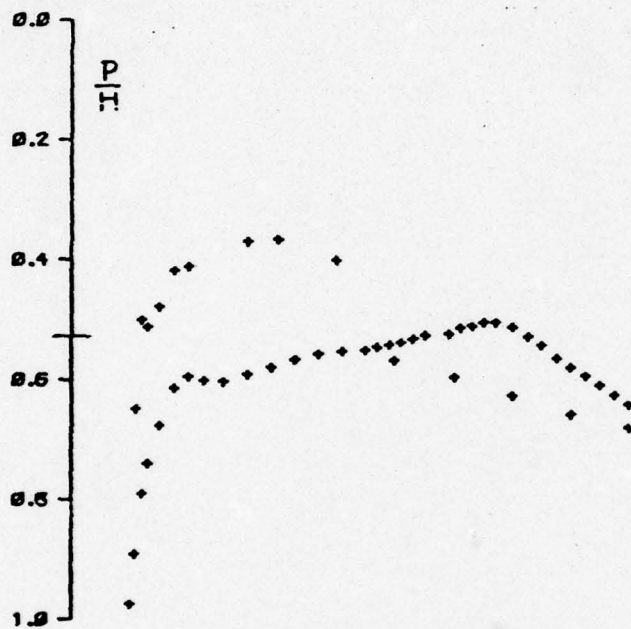
+ BGK-1 101.6 MM CHORD SOLID WALLS
 $\mu = .760$ $AL = 0.00$ $CL = 0.300$ $CM = 0.001$ $R = 0.006$



+ BGR-1 101.6 MM CHORD SOLID WALLS
 M= .761 AL= 1.00 CN= 0.486 CM= 0.008 R= 0.009



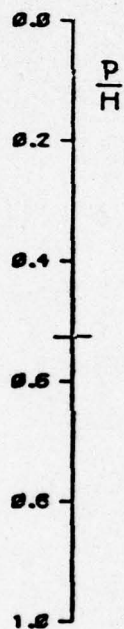
+ BGK-1 121.6 MM CHURD SOLID WALLS
 M= .758 AL= 2.00 CM= P. 513 CM=0.007 R= 0.007



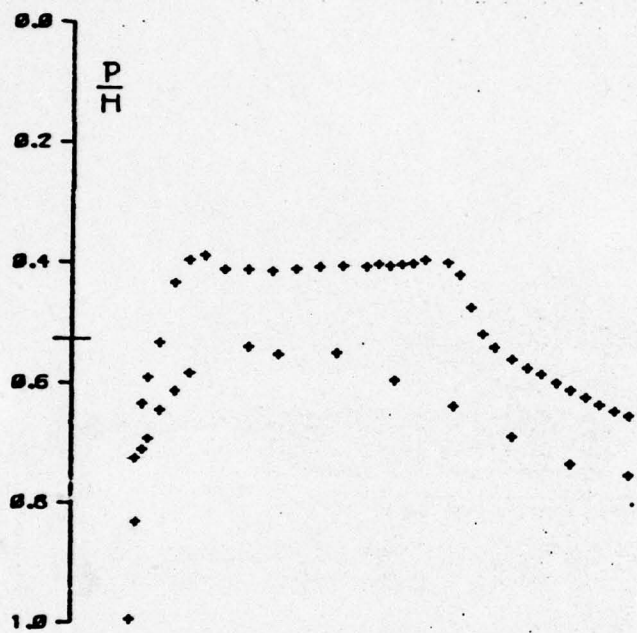
+ BOK-1 101.6 MM CHORD SOLID WALLS
 ME .779 AL-2.50 CN-0.150 CM-0.065 RE 0.765



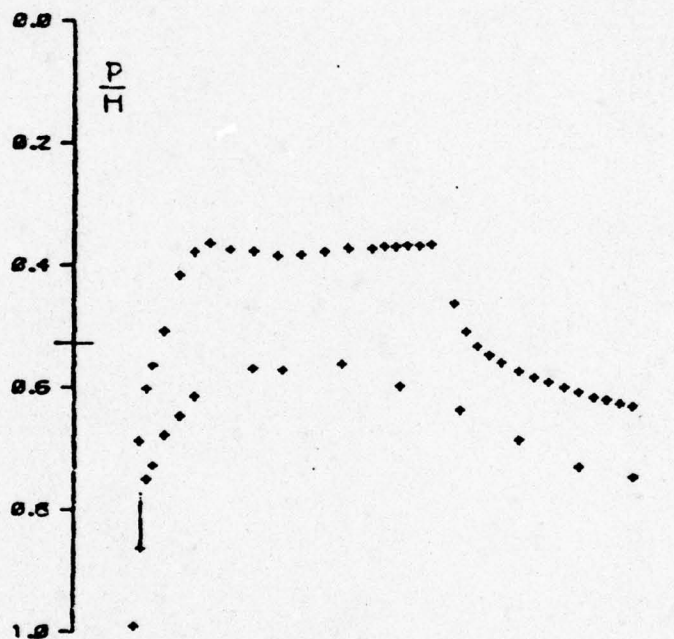
+ BQK-1 101.6 MM CHORD SOLID WALLS
 M= 783 AL=1.00 CM= 0.105 CM=0.085 R= 0.788



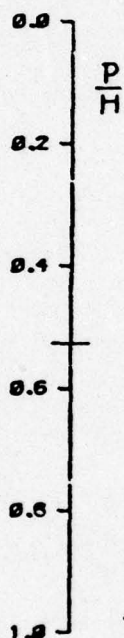
+ AGE-: 101.6 MM CHORD SOLID WALLS
 M= 732 AL= 0.00 CN= 0.293 CM= 0.084 R= 0.781



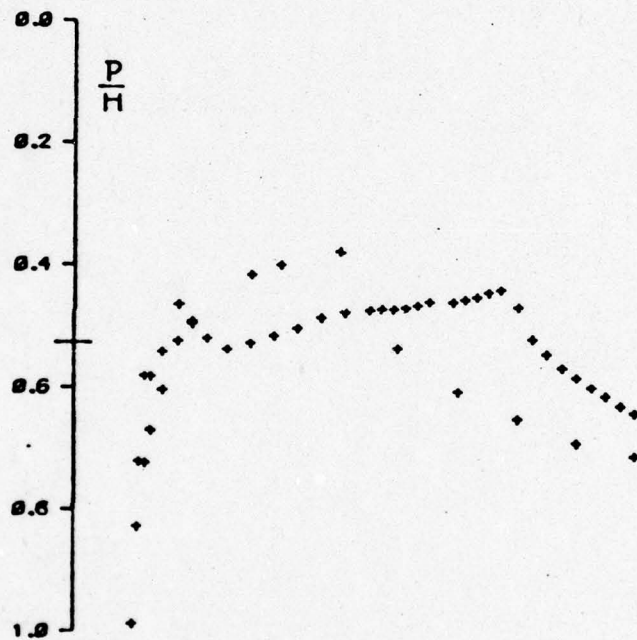
+ BOK-1 101.6 MM CHORD SOLID WALLS
 ME .776 AL= 1.00 CN= 0.166 CM= 0.082 R= 0.763



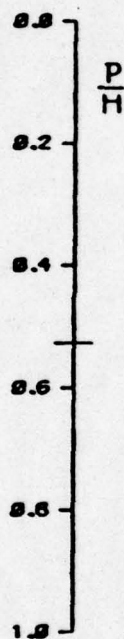
+ BOK-1 101.6 MM CHORD SOLID WALLS
 ME .778 AL= 2.00 CM= 0.536 CM= 0.086 RE= 0.786



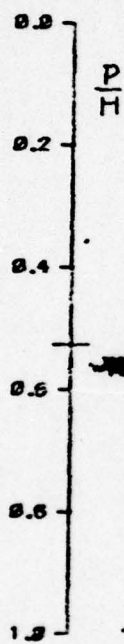
+ BGK-1 101.6 MM CHORD SOLID WALLS
 ME .000 AL-2.50 ON-0.100 OB-0.000 R= 0.764



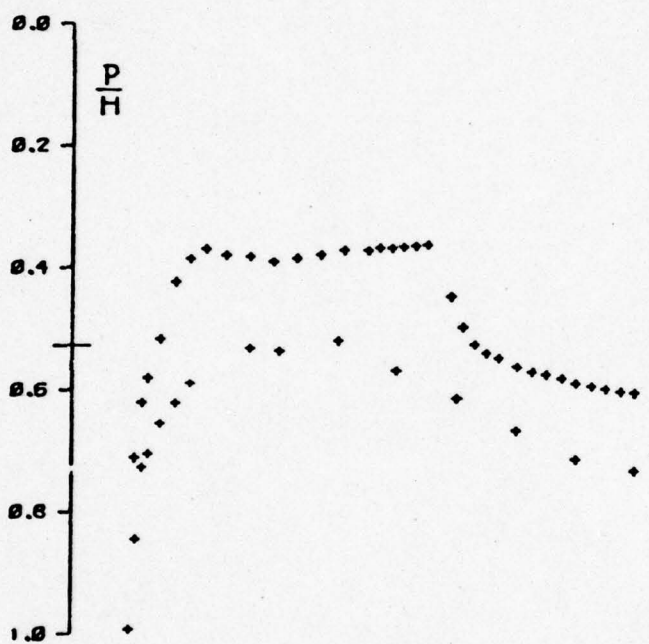
+ BGR-1 101.6 MM CHORD SOLID WALLS
 M = .031 AL = 1.00 CL = 0.072 CM = 0.034 R = 0.706



+ BGR-1 101.6 MM CHORD SOLID WALLS
 ME .789 AL= 0.00 CL= 0.245 CM= 0.005 R= 0.784

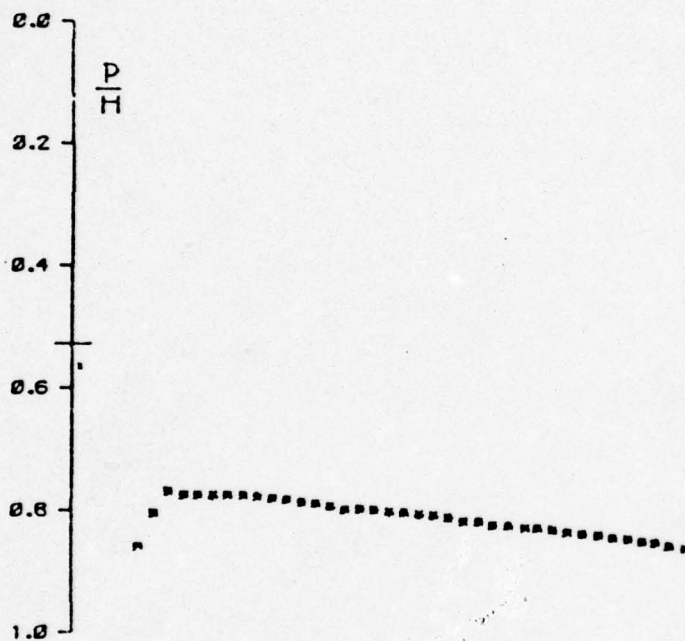


+ BGK-1 101.5 mm CHORD SOLID WALLS
 $M = 0.799$ $A_L = 1.00$ $C_H = 0.373$ $C_M = 0.069$ $R = 0.768$



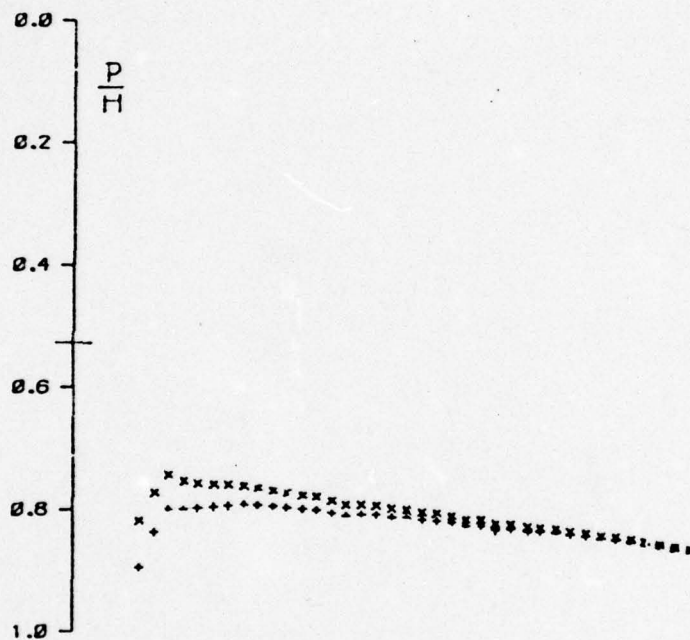
+ BOK-1 101.6 MM CHORD SOLID WALLS
 ME .001 AL= 2.00 CE= 0.450 CM=0.084 R= 0.766

94



x NACA 0012 203.2 MM CHORD SOLID BALLS
 M= 5.02 AL= 2.00 CH=0.001 CM= 2.000 R= 1.669

yy



+ x NACA 0012 203+2 MM CHORD SOLID BALLS
 $M = .500$ $AL = 1.00$ $C = 0.115$ $CM = 0.002$ $R = 1.661$

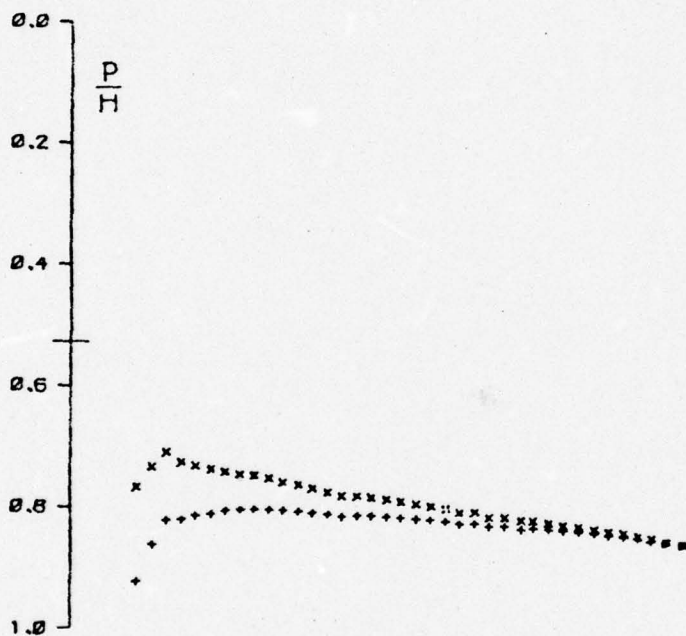
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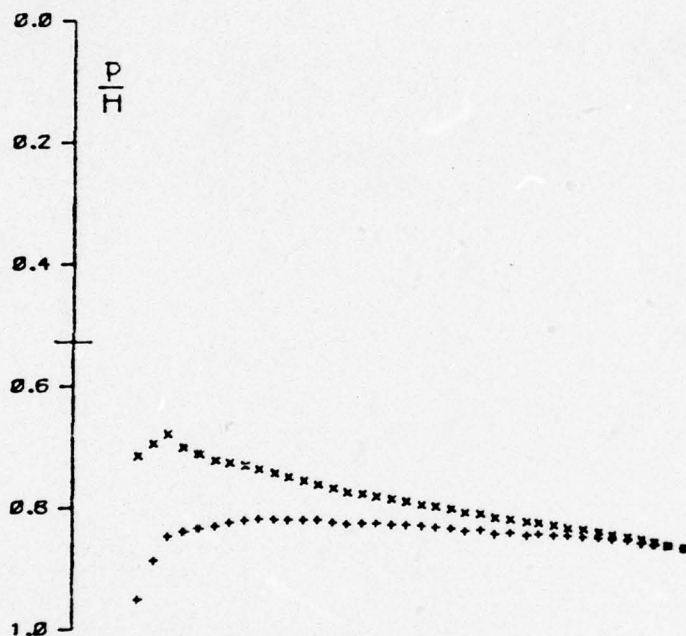
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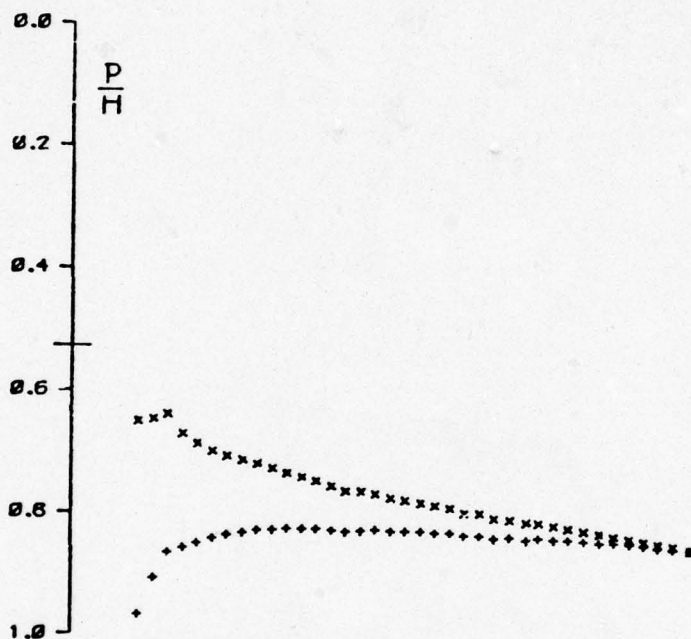
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+ x NACA 0012 203.2 MM CHORD SOLID WINGS
 M= .499 AL= 2.00 Cx= 0.230 Cy= 0.003 R= 1.661



+ x NACA 0012 203.2 MM CHORD SOLID WALLS
 M= .500 AL= 3.00 OF= 0.342 OM= 0.005 R= 1.661



+ x NACA 0012 203.2 MM CHORD SOLID BALLS
 M= .500 AL= 4.00 CM= 0.453 CM= 0.008 R= 1.665

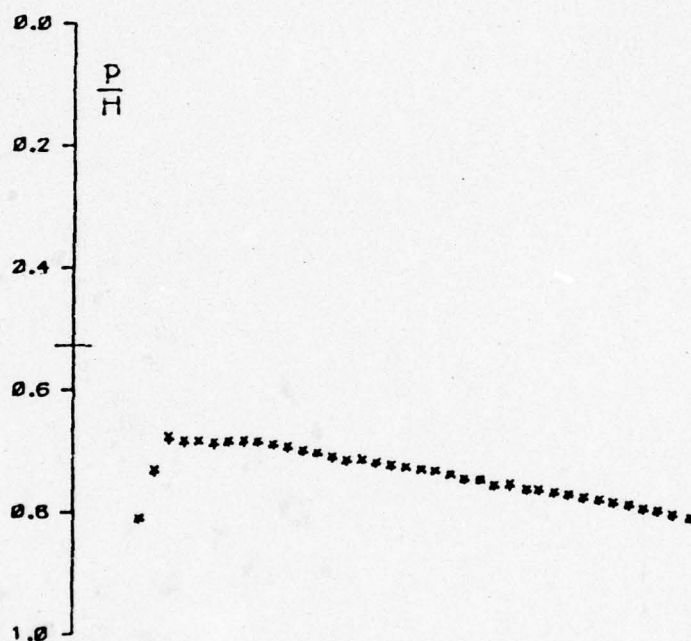
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5570

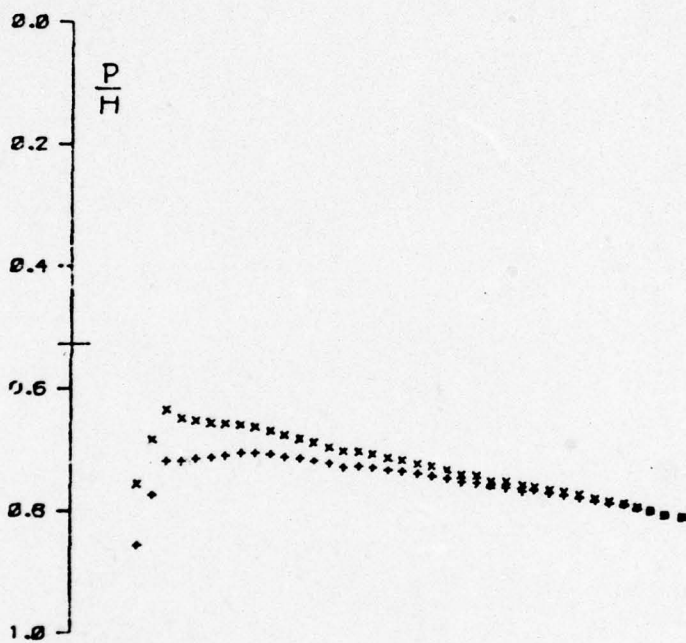
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0/70

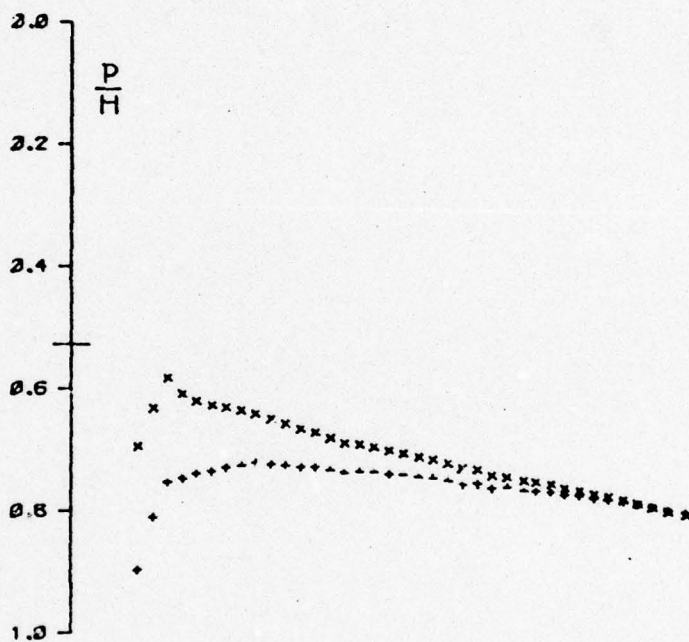
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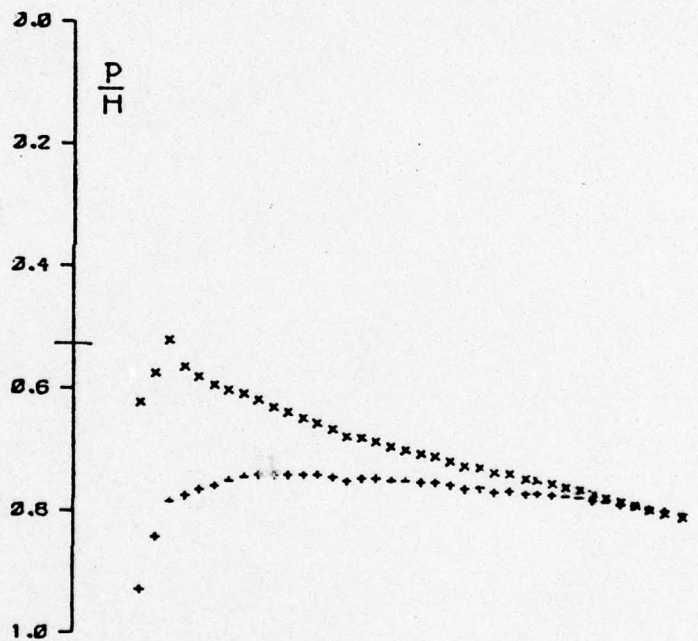
+ x NACA 0012 203.2 mm CHORD SOLID WALLS
 $M_\infty = .601$ $AL = 2.00$ $CN = 0.000$ $CM = 0.000$ $R = 1.656$



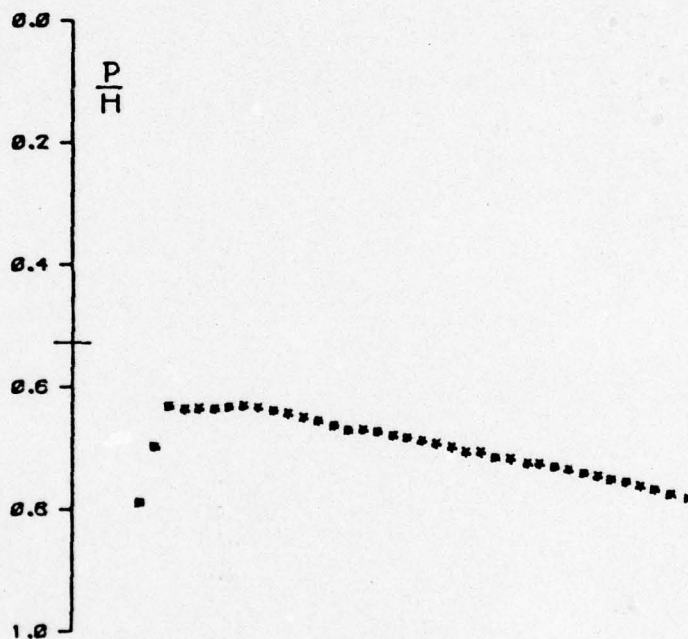
+ x NACA 0012 203.2 MM CHORD SOLID WALLS
 $M = .621$ $AL = 1.00$ $CF = 0.121$ $CF = 2.003$ $R = 1.656$



+ x NACA 0012 203.2 MM CHORD SOLID BALLS
 $M = .600$ $AL = 2.00$ $CH = 2.247$ $CH = 2.025$ $R = 1.658$



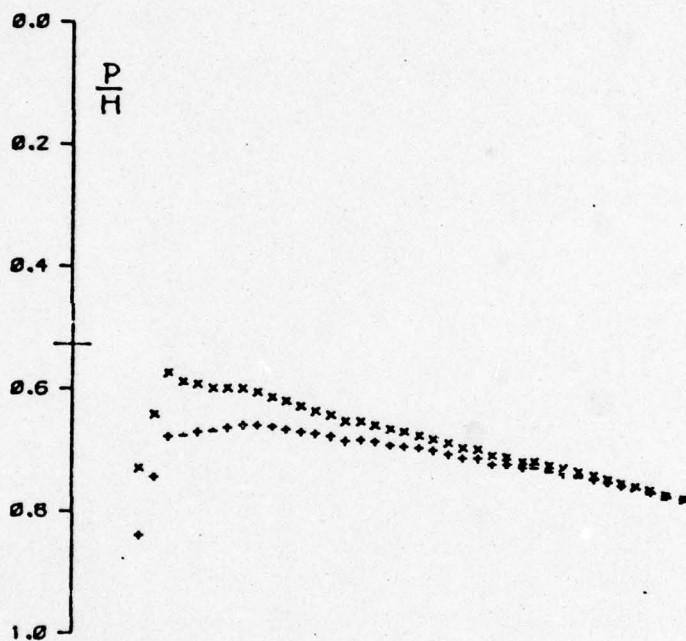
+ x NACA 0012 203.2 MM CHORD SOLID WALLS
 $M = .600$ $AL = 3.00$ $CN = 2.371$ $CM = 2.008$ $R = 1.658$



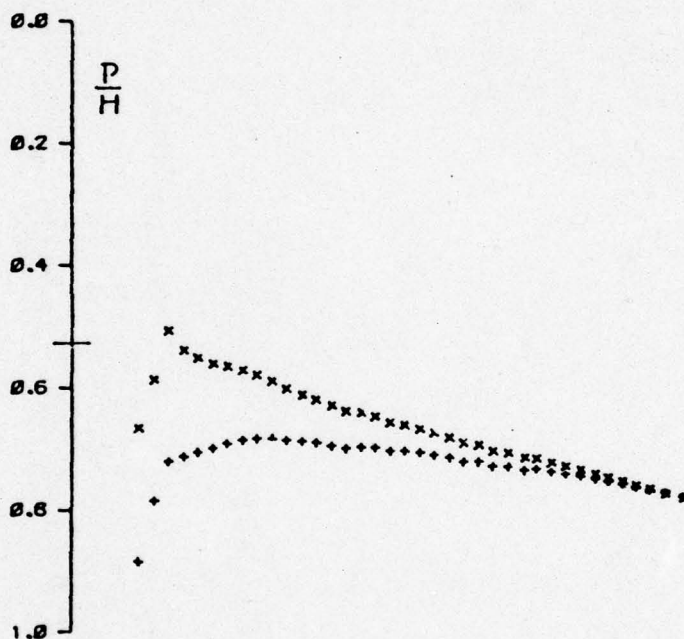
+ X NACA 0012 203.2 MM CHORD SOLID BALLS
 M= .648 AL= 2.00 CN= 0.001 CM= 2.000 R= 1.625

10M

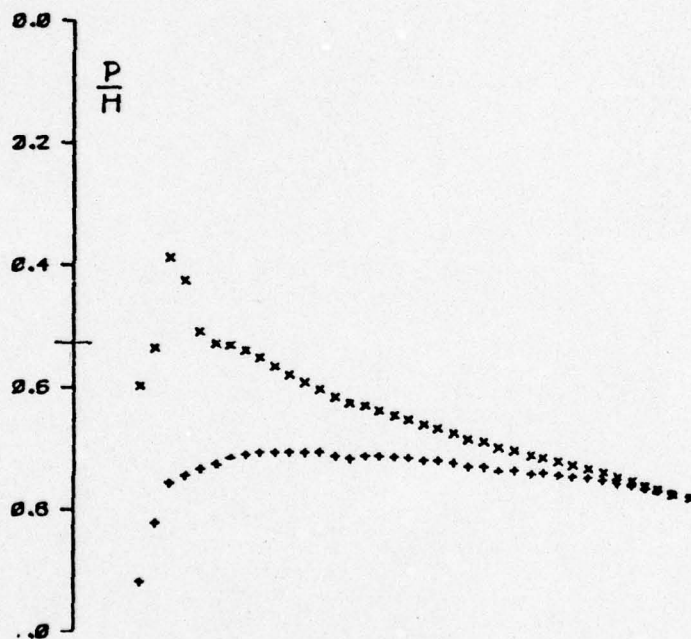
103 1000



+ x NACA 0012 2x3.2 mm CHORD SOLID WALLS
 ME .650 AL= 1.00 CE 2.132 CM 0.003 RE 1.632

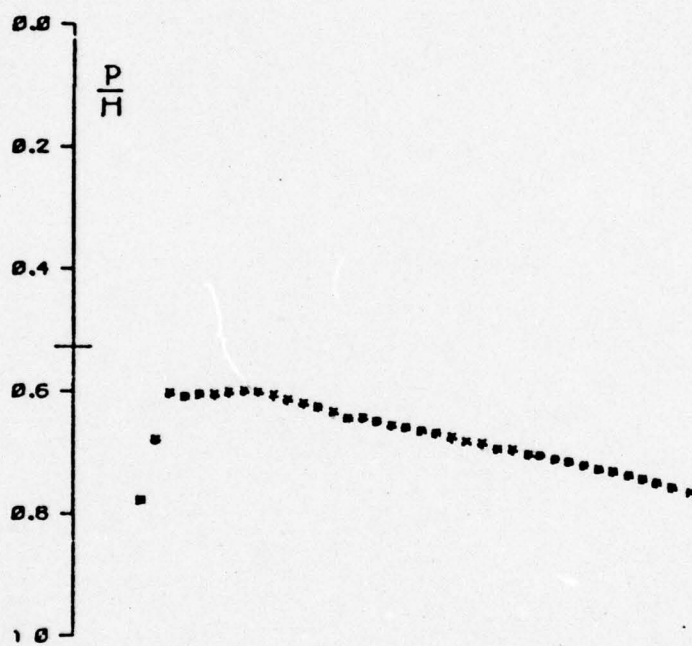


+ x NACA 0012 203.2 mm CHORD SOLID WINGS
 $M = 0.649$ $AL = 2.00$ $CF = 0.280$ $CF = 0.007$ $R = 1.628$

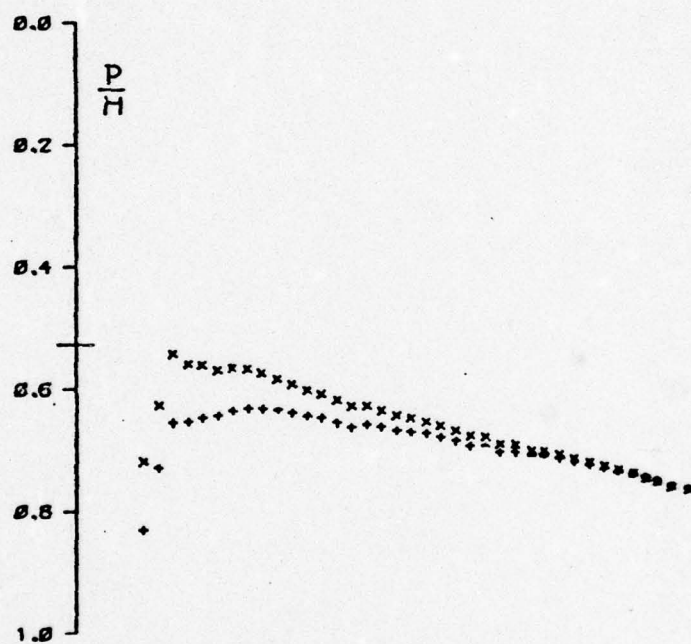


+ x NACA 0012 203-2 MM CHORD SOLID WALLS
 $M = .648$ $AL = 3.00$ $C_{L} = 0.397$ $C_{D} = 0.012$ $R = 1.628$

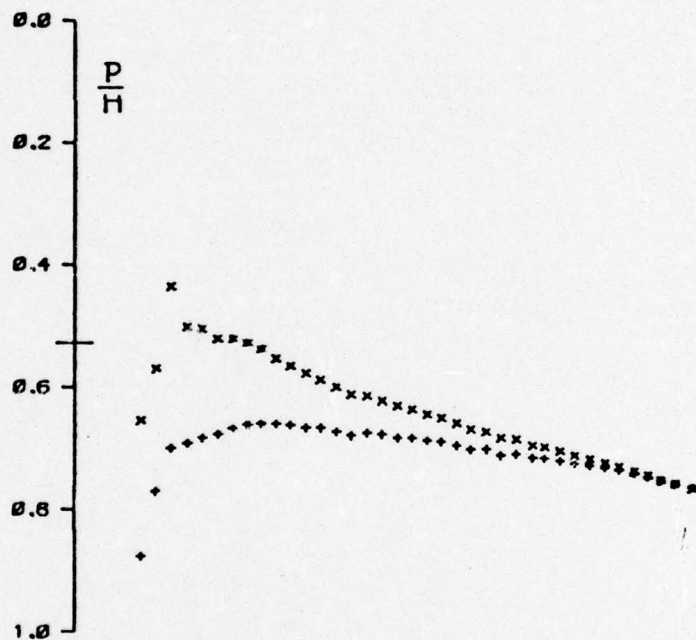
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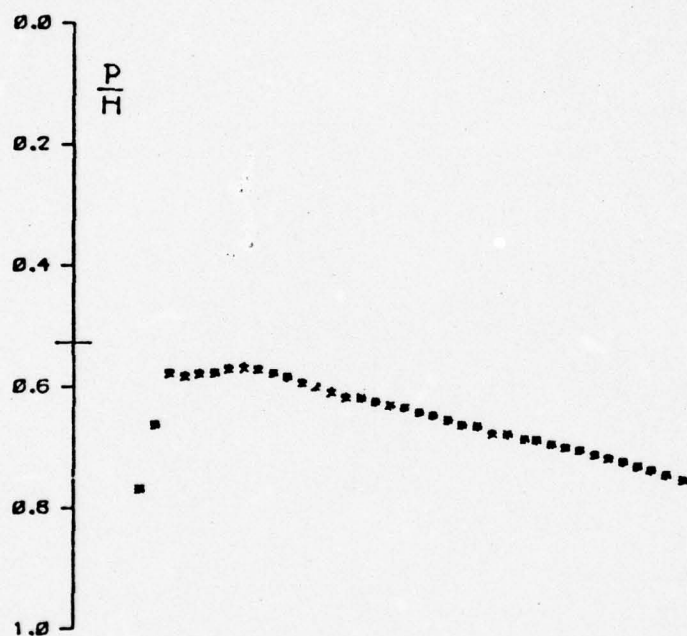
+ x NACA 0012 203.2 MM CHORD SOLID WALLS
 $M = .675$ $AL = 0.00$ $CL = 0.001$ $CM = 0.000$ $R = 1.530$



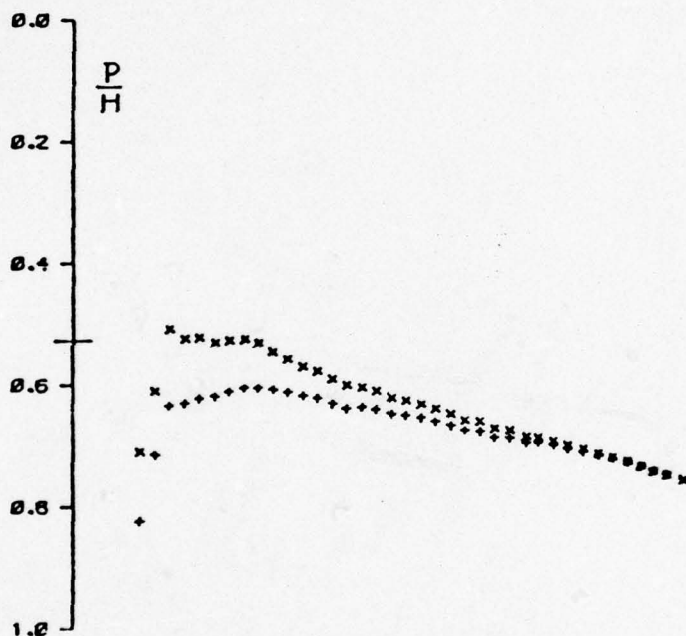
+ x NACA 0012 203.2 MM CHORD SOLID WALLS
 ME .674 AL= 1.00 CME 0.134 CME 0.004 R= 1.632



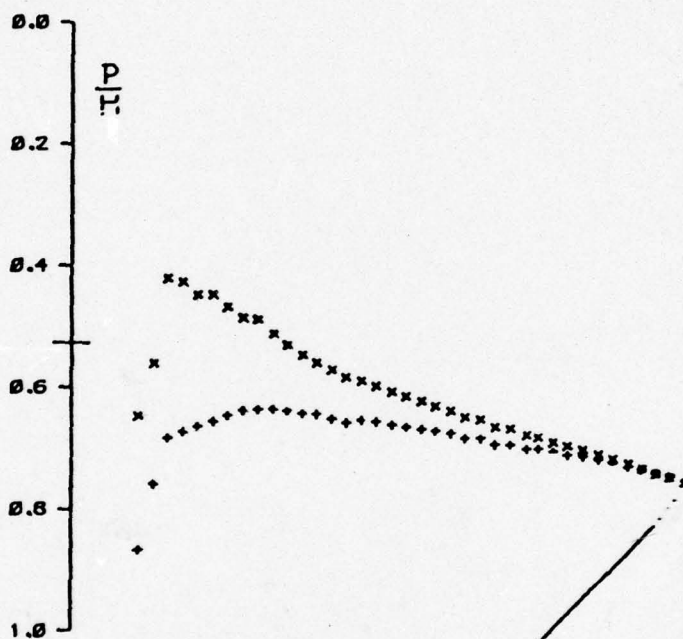
+ x NACA 0012 223-2 MM CHORD SOLID WALLS
 ME .674 AL= 2.00 CE 0.275 CM= 0.008 R= 1.632



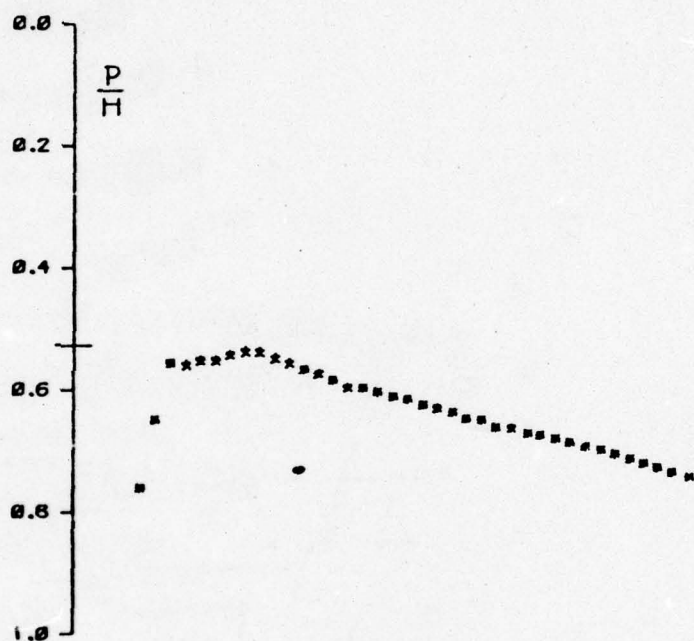
+ X NACA 0012 203.2 MM CHORD SOLID WALLS
 ME .700 AL= 0.00 CM=0.001 CM= 0.000 R= 1.624



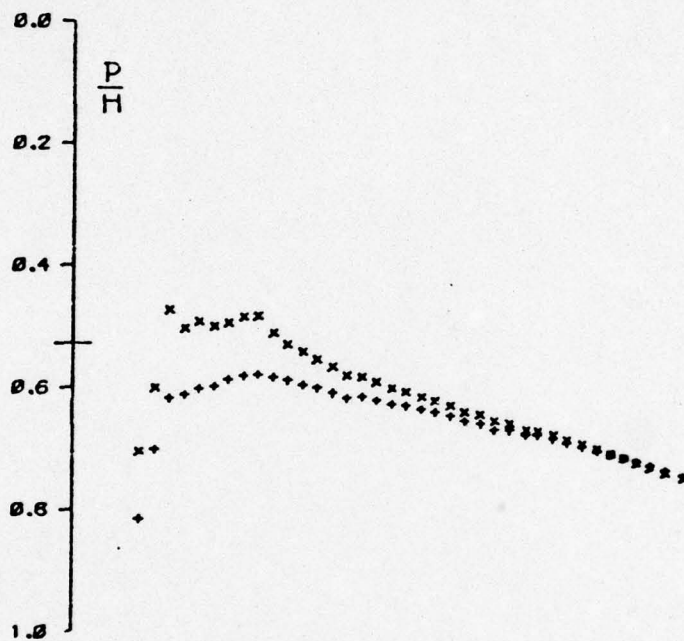
+ x NACA 0012 203.2 MM CHORD SOLID WALL:
 M_∞ .700 AL= 1.00 CF= 0.142 CM= 0.025 P. 1.627



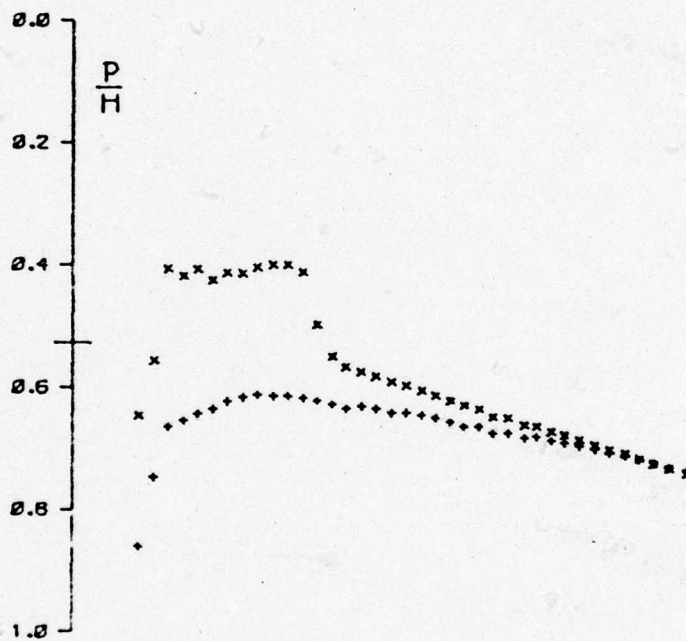
X NACA 0012 203.2 MM CHORD SOLID WALLS
 M= .700 AL= 2.00 CM= 2.292 CM= 0.010 R= 1.627



+ x NACA 0012 203.2 MM CHORD SOLID WALLS
 M_∞ = .720 A₁ = 0.00 C_f = 0.005 C_f = 0.000 R = 1.619



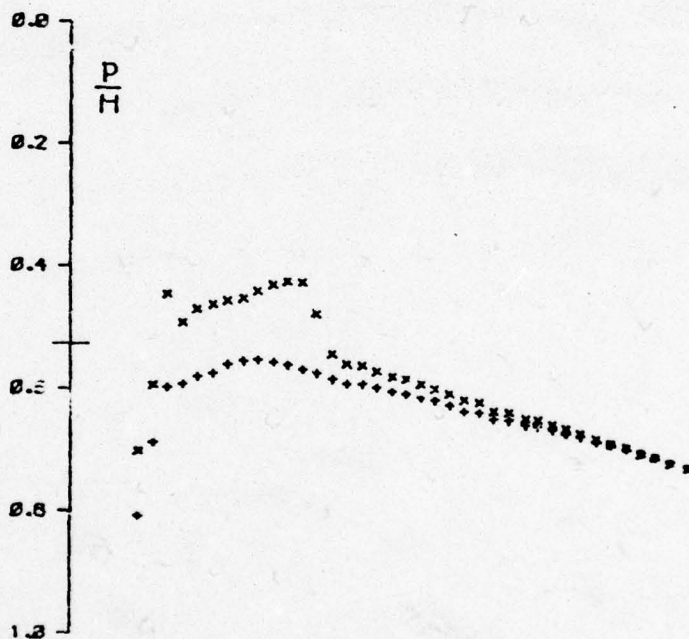
+ x NACA 0012 203.2 MM CHORD SOLID WALLS
 ME .719 AL= 1.00 CME 0.149 CME 0.005 R= 1.620



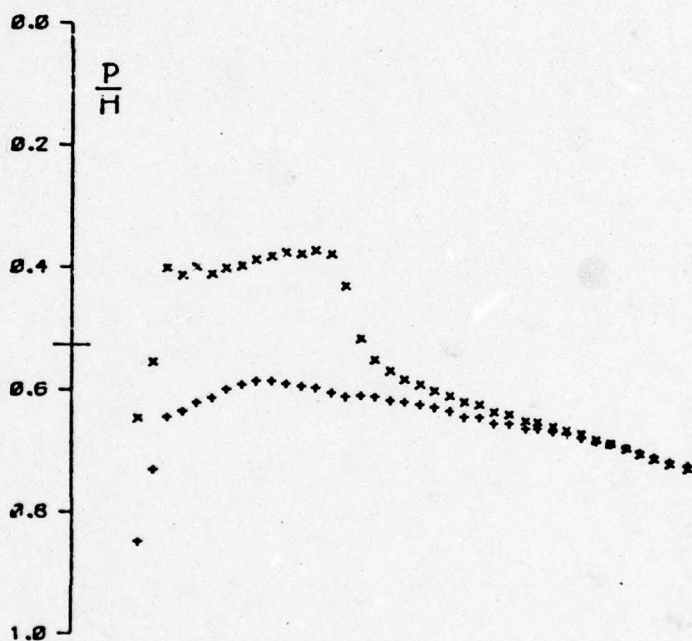
+ x NACA 0012 203.2 MM CHORD SOLID BALLS
 ME .721 AL= 2.00 CN= 0.321 CM= 0.211 R= 1.625



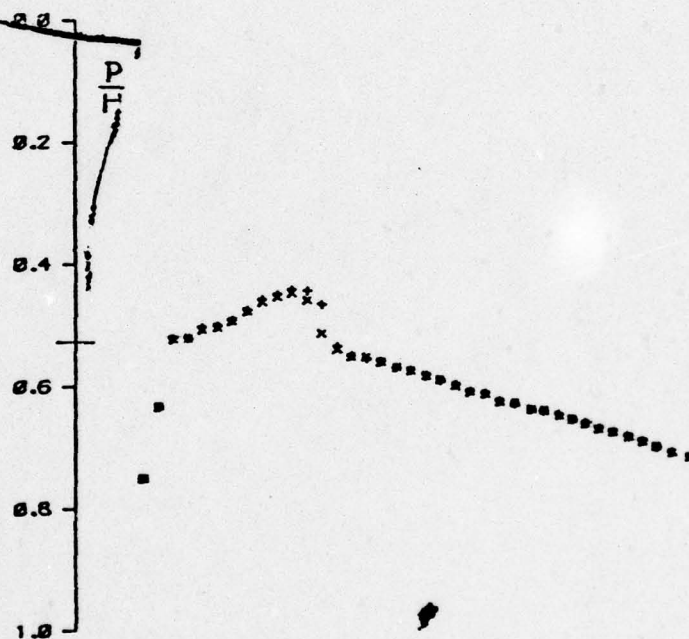
+ X NACA 0012 203.2 MM CHORD SOLID BALLS
 ME .740 AL= 0.00 CM=0.007 CM= 0.000 R= 1.618



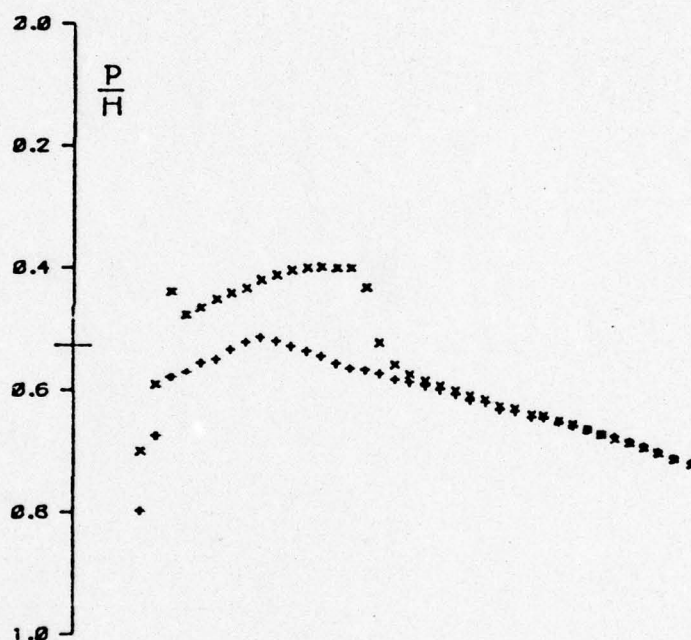
+ x NACA 0012 203.2 MM CHORD SOLID WALLS
 $M = 0.740$ $AL = 1.00$ $CN = 0.166$ $CM = 0.006$ $R = 1.622$



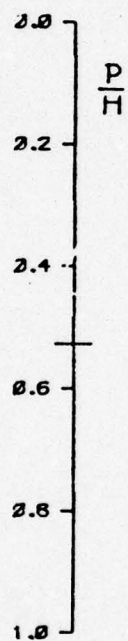
+ x NACA 0012 283.2 MM CHORD SOLID WALLS
 $M = .739$ $\alpha = 2.00$ $C_{Df} = 0.334$ $C_{Df} = 0.009$ $R = 1.619$



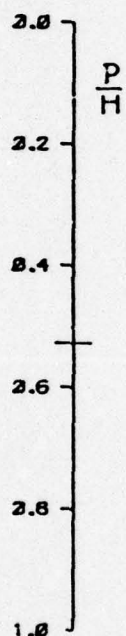
+ x NACA 0012 203.2 MM CHORD SOLID WALLS
 $M = .758$ $AL = 0.00$ $CM = -0.021$ $CM = 0.000$ $R = 1.594$



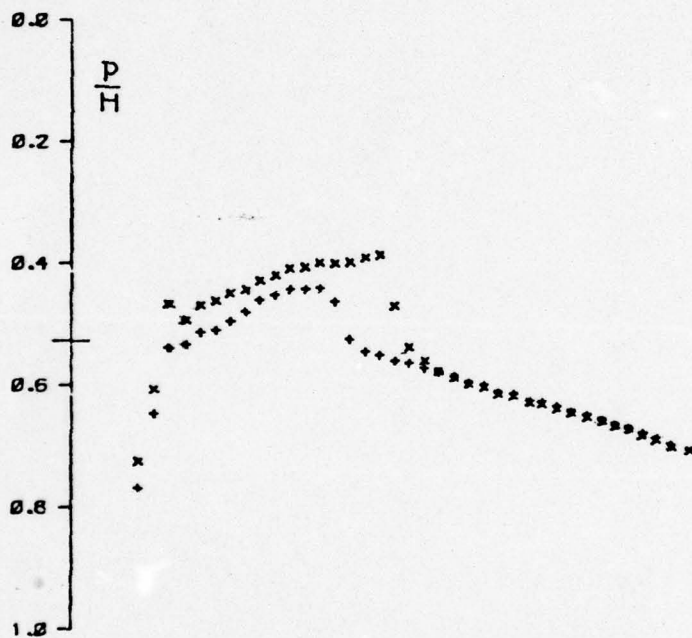
+ x NACA 0012 243.2 MM CHORD SOLID WALLS
 M= .760 AL= 1.00 CN= 0.160 CM= 0.003 R= 1.596



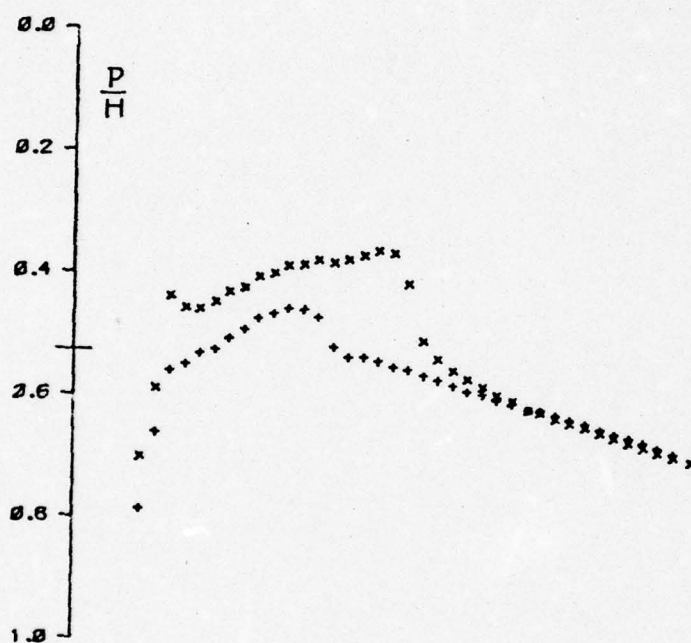
+ x NACA 0012 203.2 MM CHORD SOLID WALLS
 $M = .759$ $AL = 2.00$ $UN = 2.346$ $CM = 0.001$ $P = 1.587$



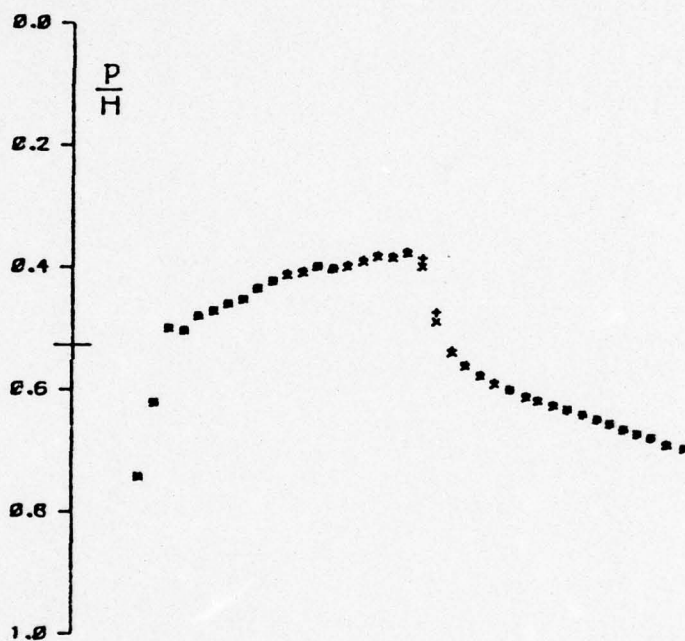
+ x NACA 0012 203.2 MM CHORD SOLID WALLS
 $M = .781$ $AL = 2.00$ $CM = 0.002$ $CM = -0.000$ $R = 1.571$



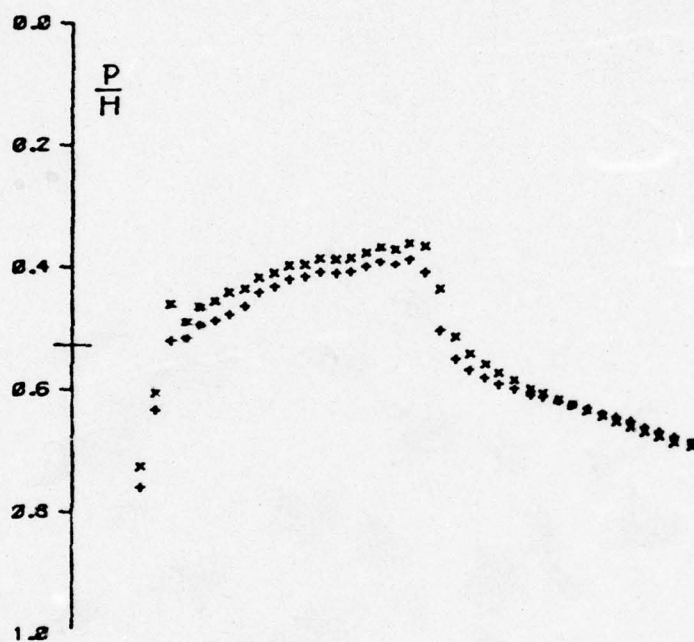
+ x NACA 0012 243.2 MM CHORD SOLID BALLS
 $M = .779$ $A = 2.50$ $C_F = 0.102$ $C_M = -0.002$ $R = 1.571$



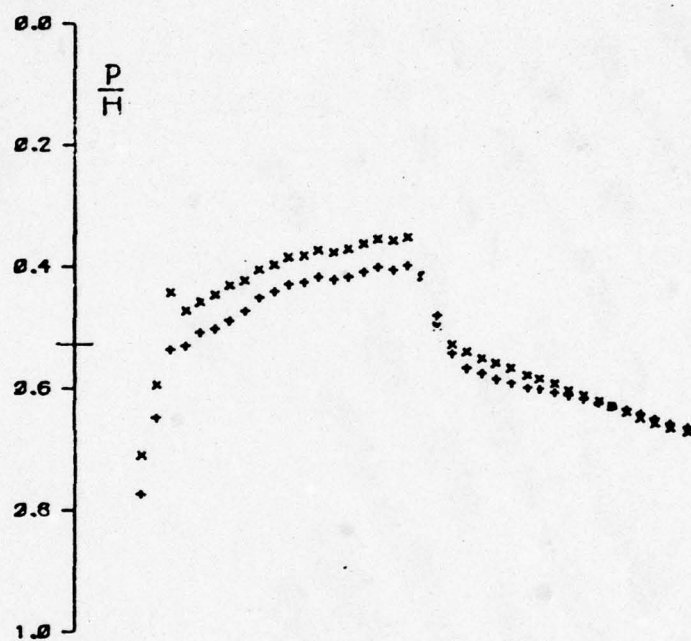
+ x NACA 0012 203.2 MM CHORD SOLID WINGS
 NE 781 AL=1.00 CM=0.184 CM=0.005 R=1.578



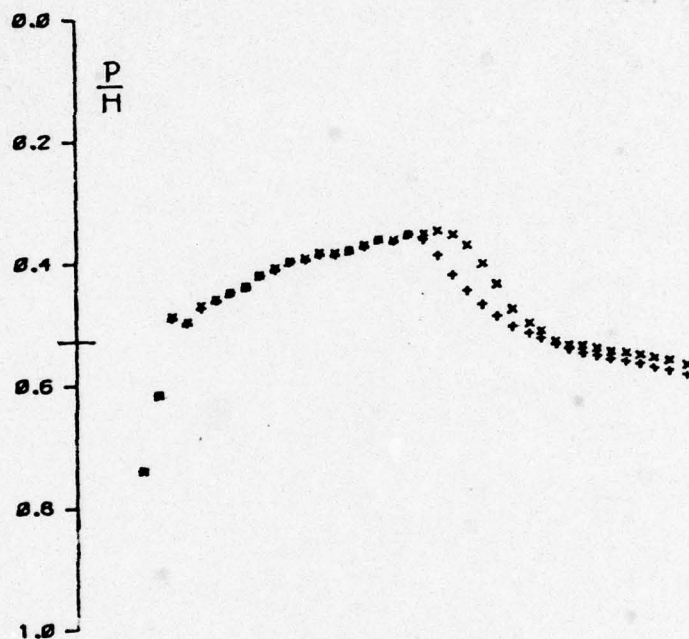
+ x NACA 0012 203.2 MM CHORD SOLID WALLS
 M = .801 AL = 2.00 CN = 0.006 CM = 0.001 R = 1.601



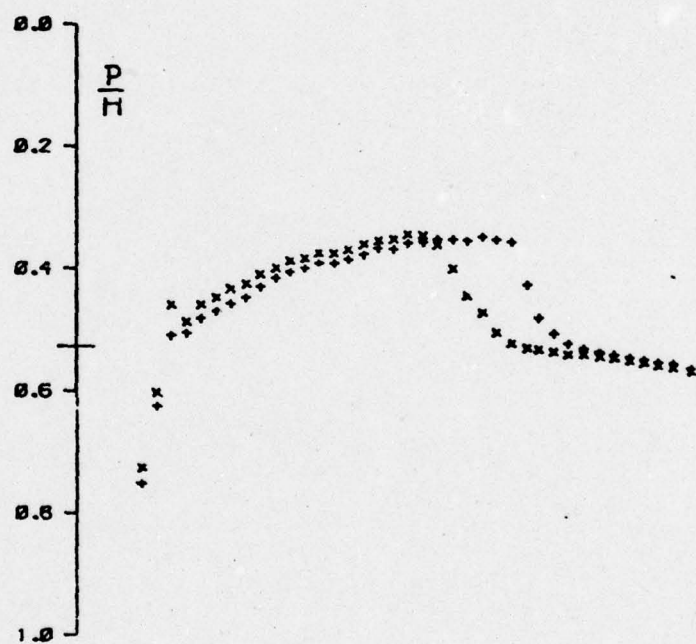
x NACA 0012 203.2 MM CHORD SOLID WALLS
 Re 800 AL=0.50 UN=0.064 CM=0.002 F=1.587



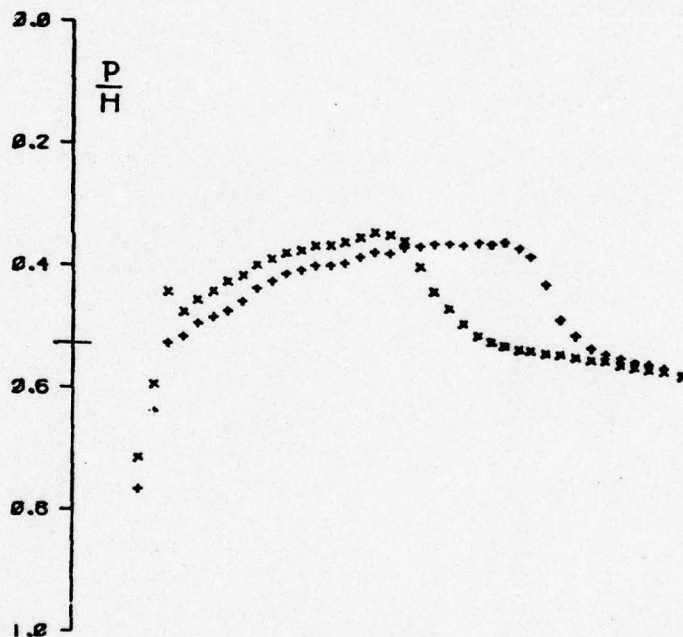
+ x NACA 0012 243.2 MM CHORD SOLID WALLS
 ME .600 AL 1.00 CN 0.093 CM 0.000 Z 1.001



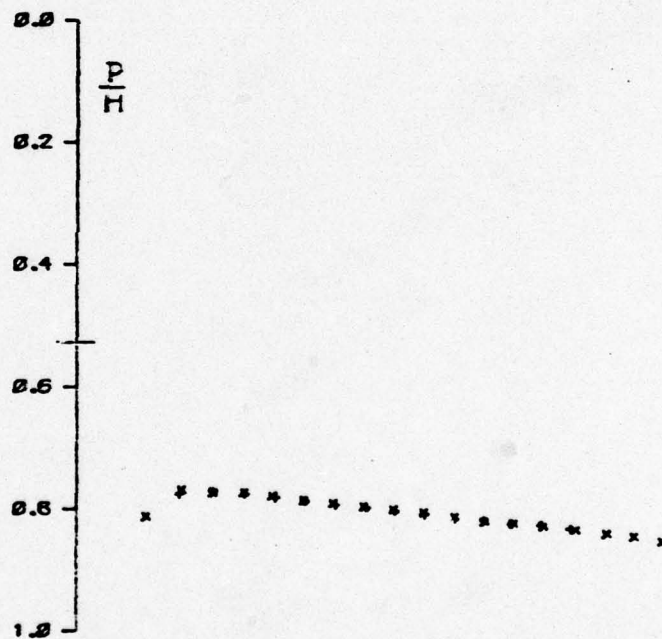
* x NACA 0012 203.2 MM CHORD SOLID WALLS
 M=0.62 AL=0.00 CM=0.237 CM=0.015 R=1.415



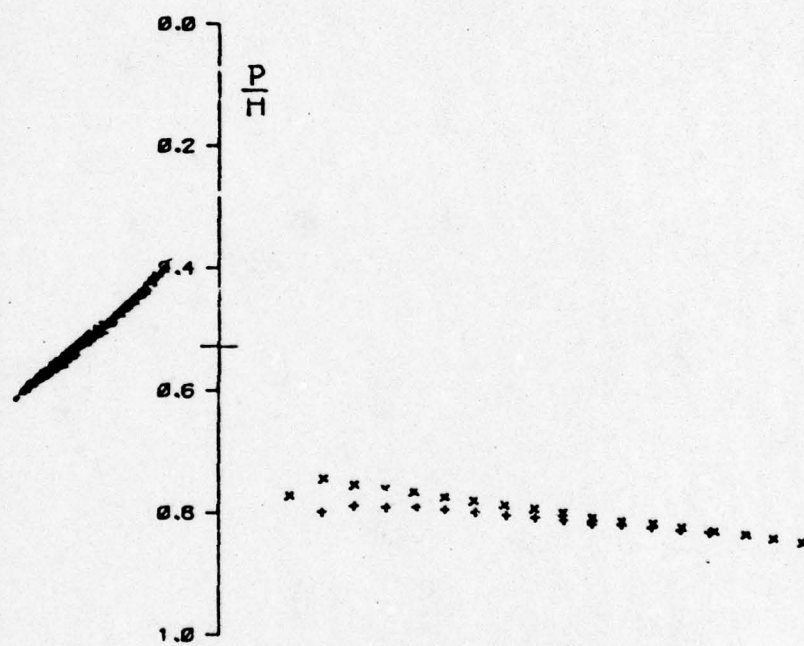
+ x NACA 0012 200.2 MM CHORD SOLID WALLS
 M= 0.20 AL= 2.50 CL= 0.336 CM= 0.029 R= 1.415



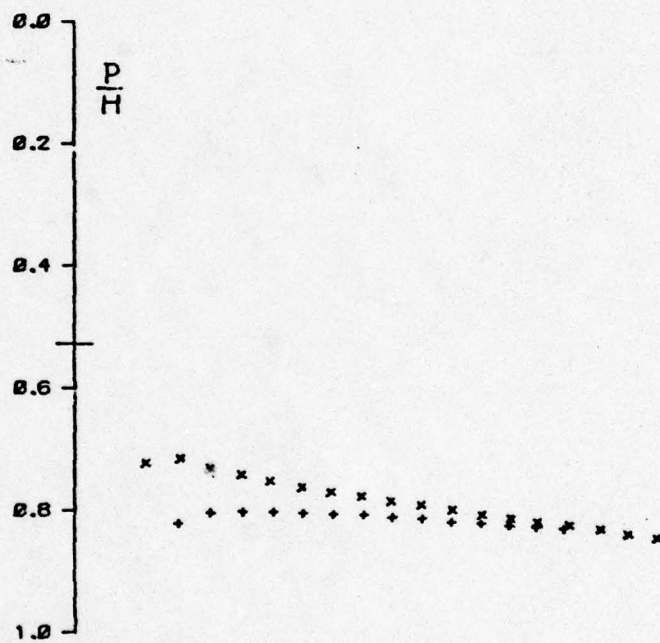
+ x NACA 0012 203.2 MM CHORD SOLID WALLS
 $M = 0.621$ $AL = 1.00$ $CH = 0.061$ $CH = 2.048$ $Re = 1.419$



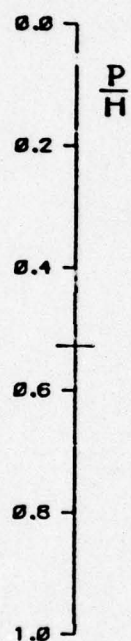
+ x NACA 0012 101.6 MM CHORD SOLID WALLS
 M= 0.500 AL= 0.00 CL= 0.025 CM= 0.003 R= 0.632



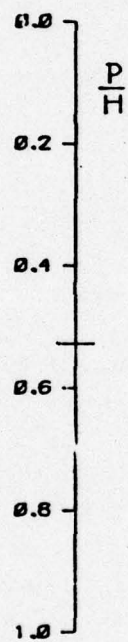
+ x NACA 0012 101.6 MM CHORD SOLID WALLS
 ME .499 AL= 1.00 CM= 2.109 CM= 2.321 R= 0.629



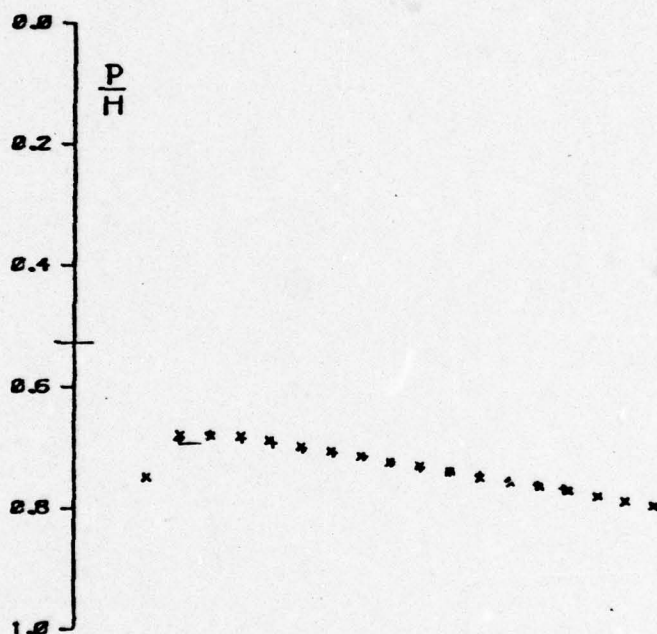
+ x NACA 0012 101.6 MM CHORD SOLID WALLS
 M= .500 AL= 2.00 CM= 0.225 CM= 0.002 R= 0.632



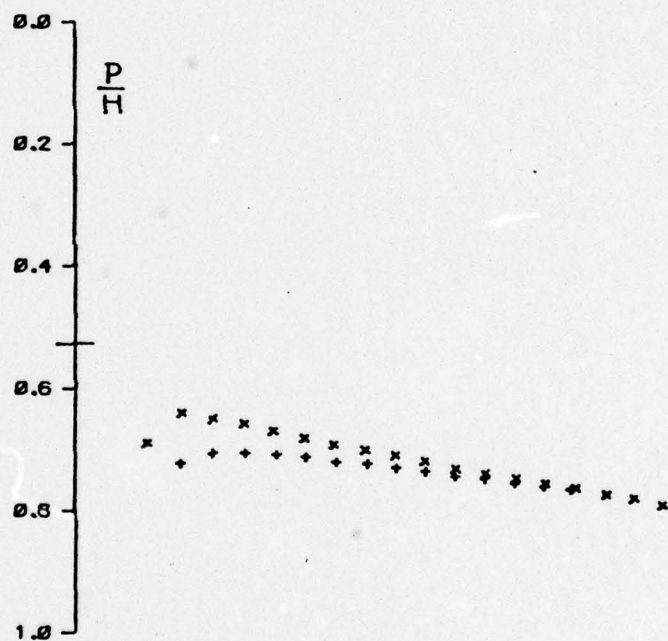
+ x NACA 0012 101.6 MM CHORD SOLID BALLS
 M= .500 AL= 3.00 CF= 0.336 CM= 0.002 R= 0.835



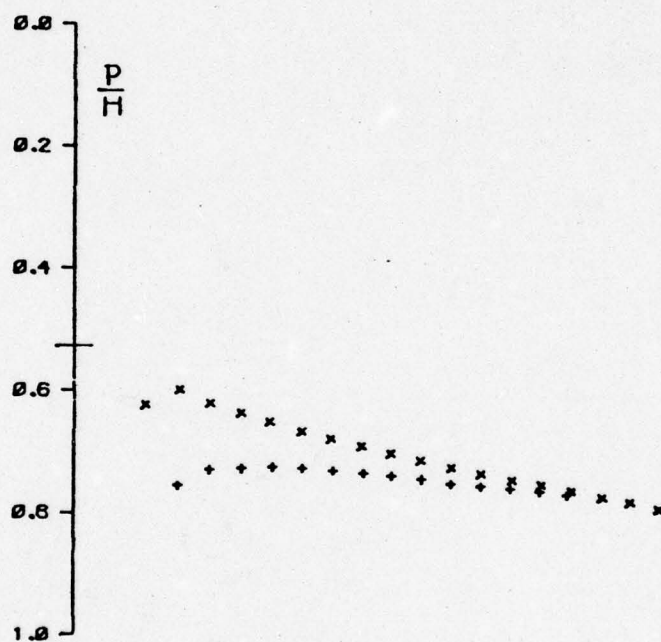
+ x NACA 0012 101.6 MM CHORD SOLID WINGS
 $M = .501$ $AL = 4.00$ $CL = 0.445$ $CM = 0.007$ $R = 0.632$



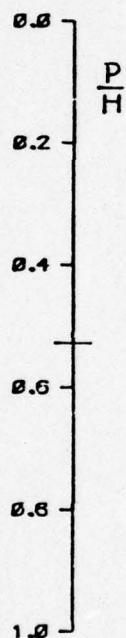
+ x NACA 0012 101.6 MM CHORD SOLID BALLS
 ME .600 AL= 0.00 CM= 0.021 CM= 0.004 RE 0.024



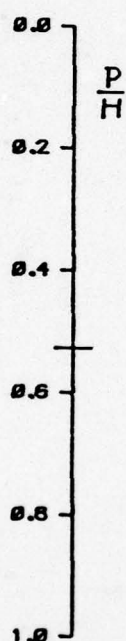
+ x NACA 0012 101.6 MM CHORD SOLID WALLS
 M= .588 AL= 1.00 CN= 0.119 CN= 0.002 R= 0.624



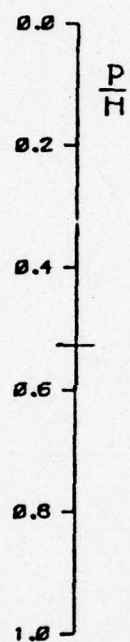
+ x NACA 0012 101.6 MM CHORD SOLID WALLS
 $M = .598$ $AL = 2.00$ $Cl = 0.239$ $Cl = 0.005$ $R = 0.024$



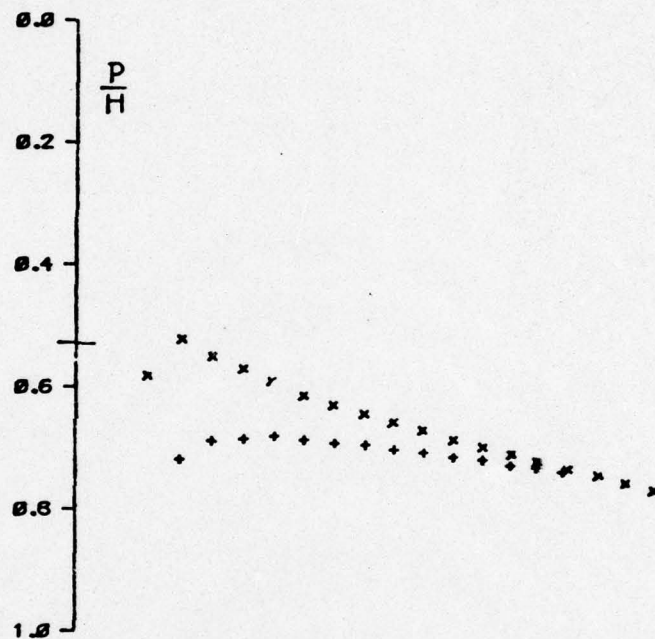
+ x NACA 0012 101.6 MM CHORD SOLID BALLS
 M = .600 AL = 3.00 CN = 0.359 CM = 0.009 R = 0.027



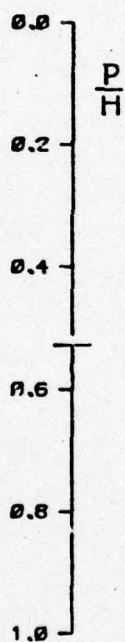
+ x NACA 0012 101.6 MM CHORD SOLID WALLS
 M = .650 AL = 0.00 CN = 0.027 CM = 0.004 R = 0.021



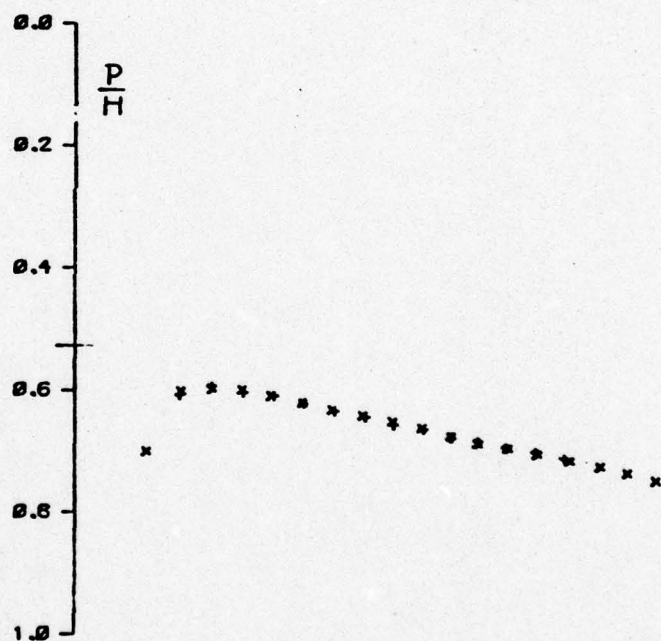
+ x NACA 251.1 101.6 MM CHORD SOLID WALLS
 ME .650 AL .00 CN .0127 CN .0023 R .0021



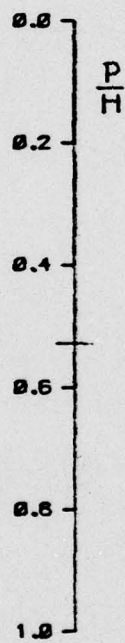
+ x NACA 0012 101.6 MM CHORD SOLID BALLS
 ME .649 AL= 2.00 CM= 0.252 CM= 0.007 R= 0.019



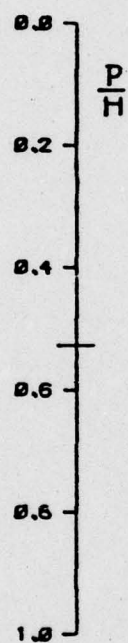
+ x NACA 0012 101.6 MM CHORD SOLID WALLS
 M= .643 AL= 3.00 CM= 0.392 CM= 0.015 R= 0.819



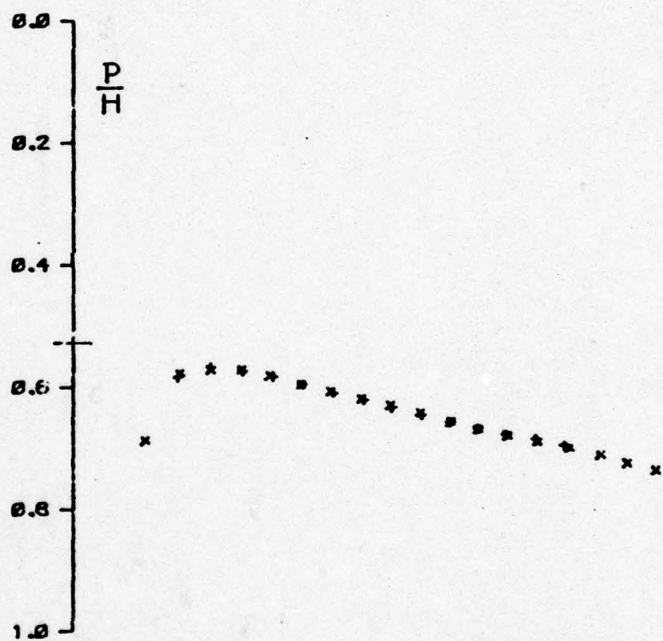
+ x NACA 0012 101.6 MM CHORD SOLID WALLS
 $M = .674$ $AL = 0.00$ $CM = 0.026$ $CM = 0.004$ $R = 0.022$



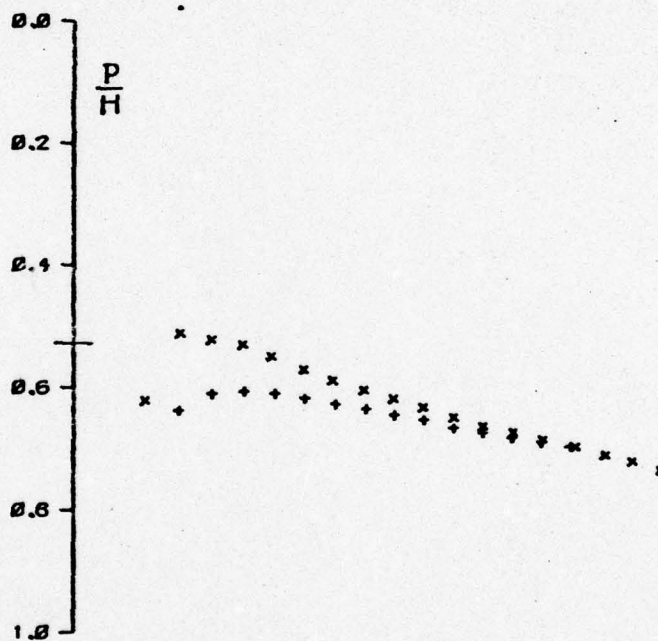
+ x NACA 0012 101.6 MM CHORD SOLID WALLS
 $M = .673$ $AL = 1.00$ $CL = 0.132$ $CM = 0.004$ $R = 0.622$



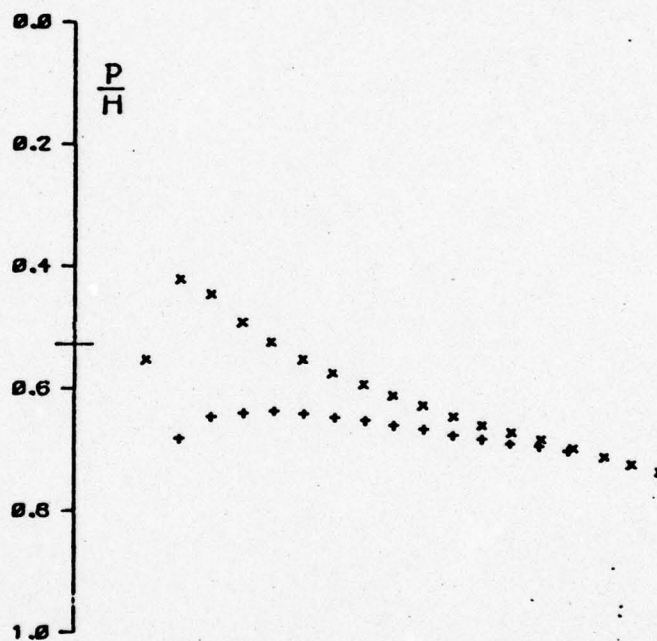
+ x NACA 0012 101.6 MM CHORD SOLID BALLS
 $M = .574$ $\alpha = 2.00$ $C_D = 0.277$ $C_L = 0.012$ $R = 0.022$

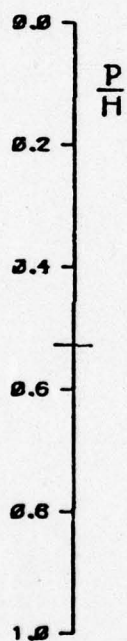


+ x NACA 6312 10% .6 MM CHORD SOLID WALLS
 $M = .659$ $AL = 0.10$ $Q = 0.028$ $Q = 0.025$ $R = 0.621$

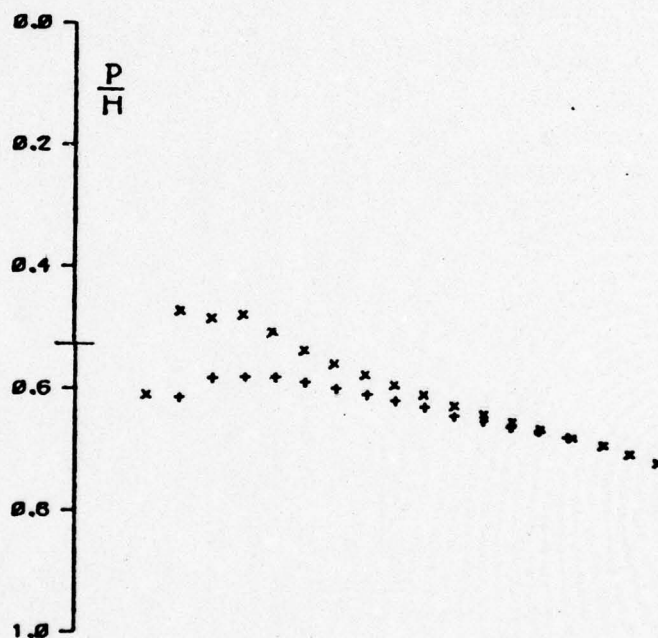


+ x NACA 0012 101.6 MM CHORD SOLID WALLS
 M= 0.75 AL= 1.00 CN= 0.143 CM= 0.007 R= 0.021

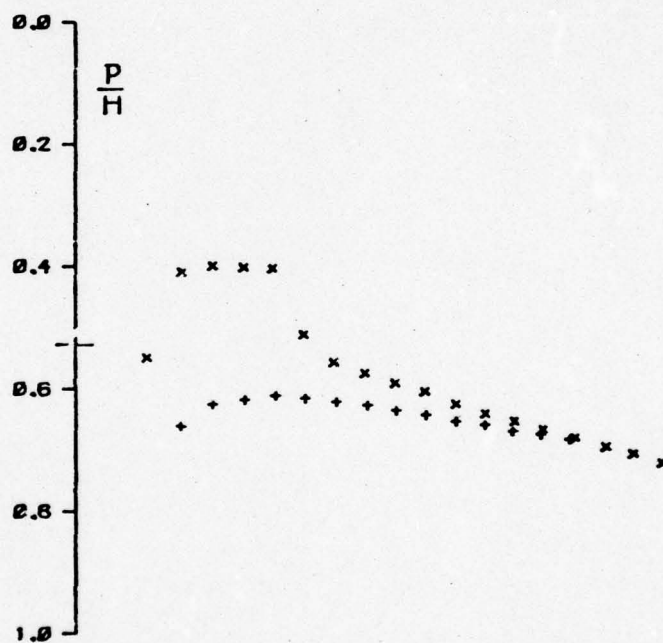




+ x NACA 0012 101.6 MM CHORD SOLID WALLS
 M= .718 AL= 0.00 CM= 0.030 CM= 0.025 R= 0.817



+ x NACA 0012 101.6 MM CHORD SOLID WINGS
 M= 0.72 AL= 1.00 CM= 0.172 CM= 0.011 R= 0.519



+ x NACA 0012 101.6 MM CHORD SOLID WINGS
 M= 0.71 AL= 2.00 CM= 0.311 CM= 0.017 R= 0.822

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AERONAUTICAL RESEARCH LABS MELBOURNE (AUSTRALIA)
TRANSONIC WIND TUNNEL TESTS ON A SERIES OF TWO-DIMENSIONAL AERO--ETC(U)
JAN 79 B D FAIRLIE , N POLLOCK
ARL/AERO NOTE-384

F/G 20/4

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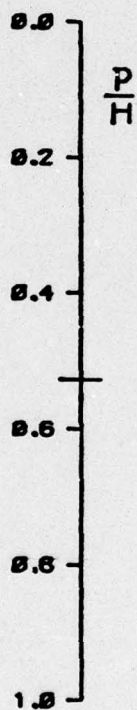


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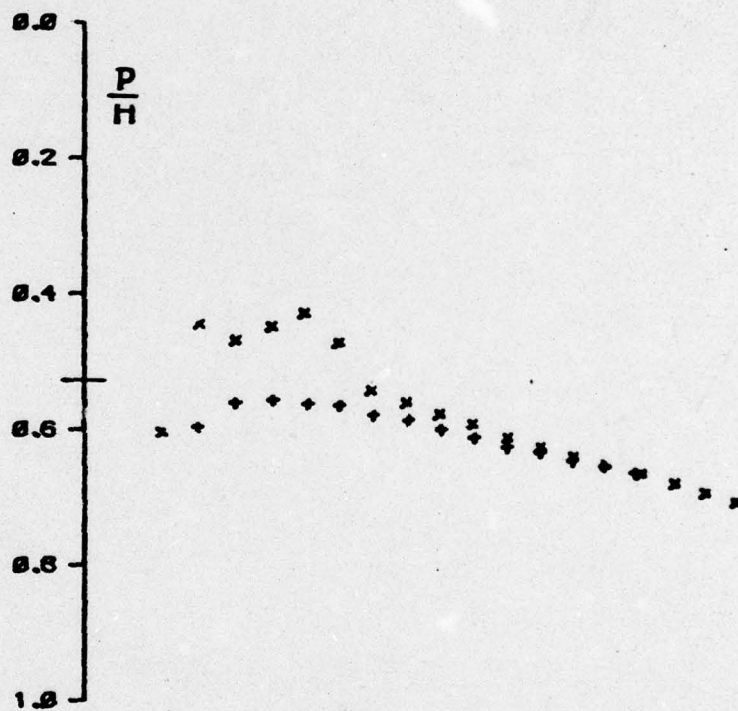
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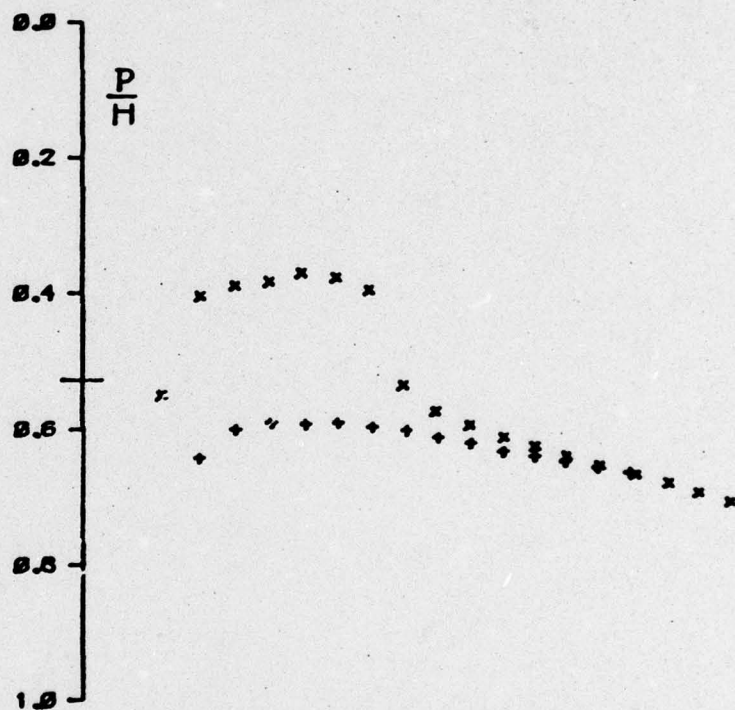
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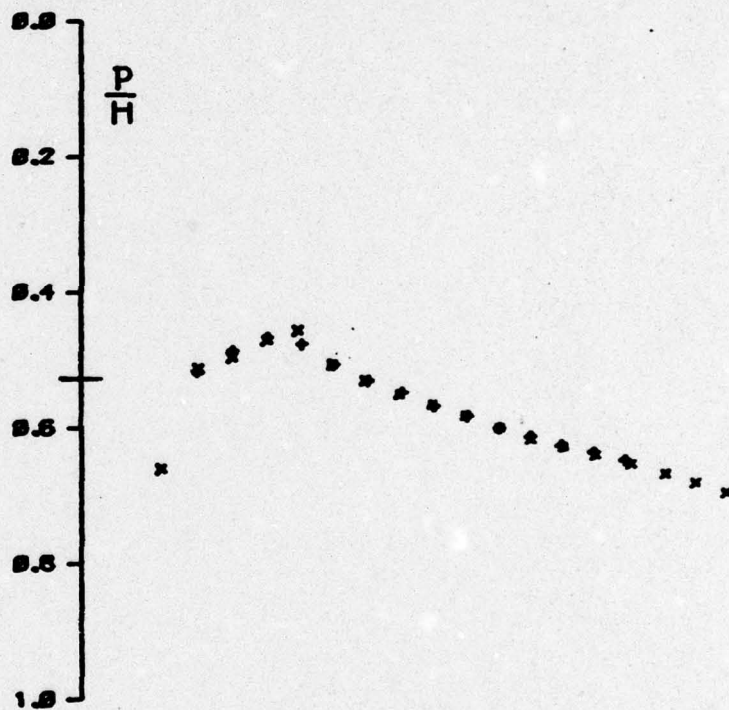
+ x NACA 0012 101.6 MM CHORD SOLID BALLS
 $M = .739$ $AL = 0.00$ $CL = 0.032$ $CM = 0.006$ $R = 0.022$



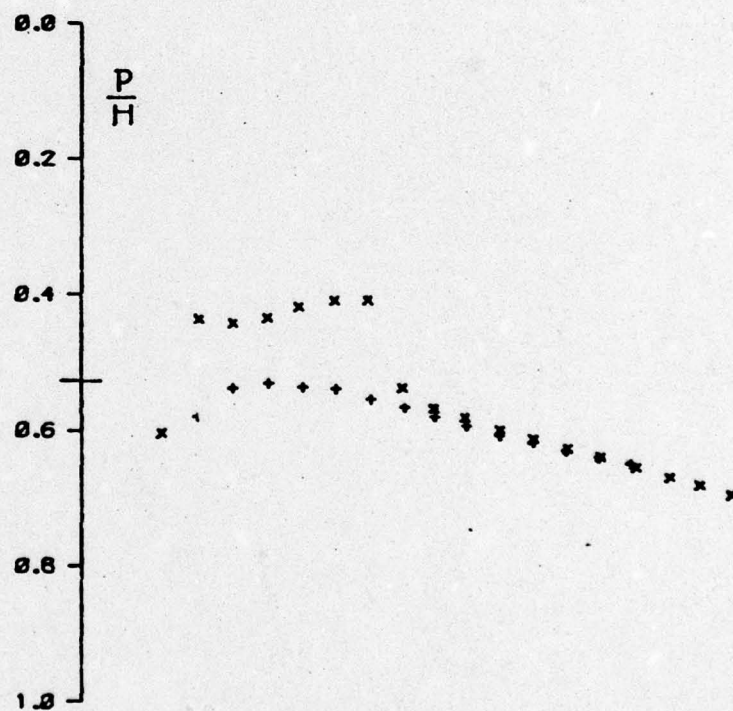
+ x NACA 2312 101.6 MM CHORD SOLID WALLS
 M = 741 AL = 1.00 CL = 0.191 CM = 0.014 R = 0.021



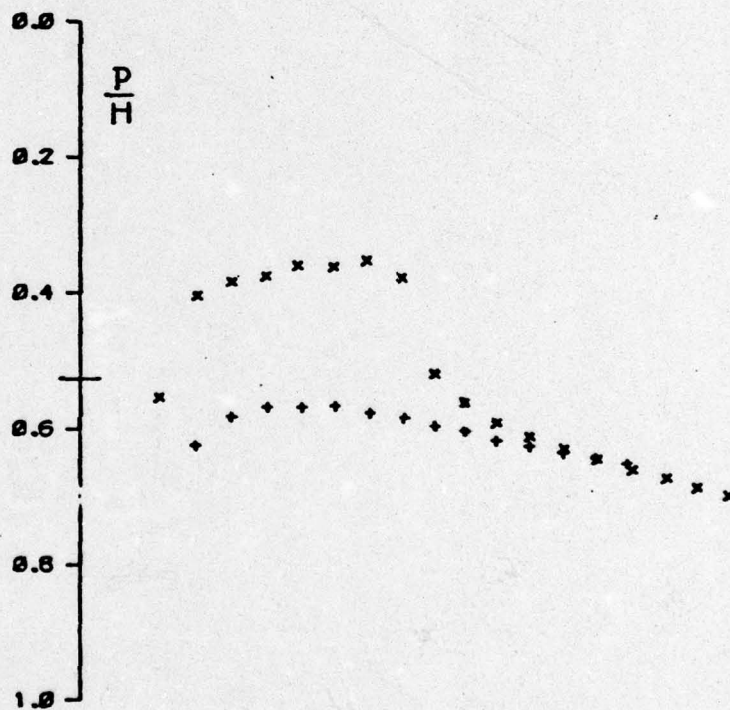
+ x NACA 2312 101.6 MM CHORD SOLID WALLS
 M= 0.71 AL= 2.00 CN= 1.335 CM= 0.013 R= 0.021



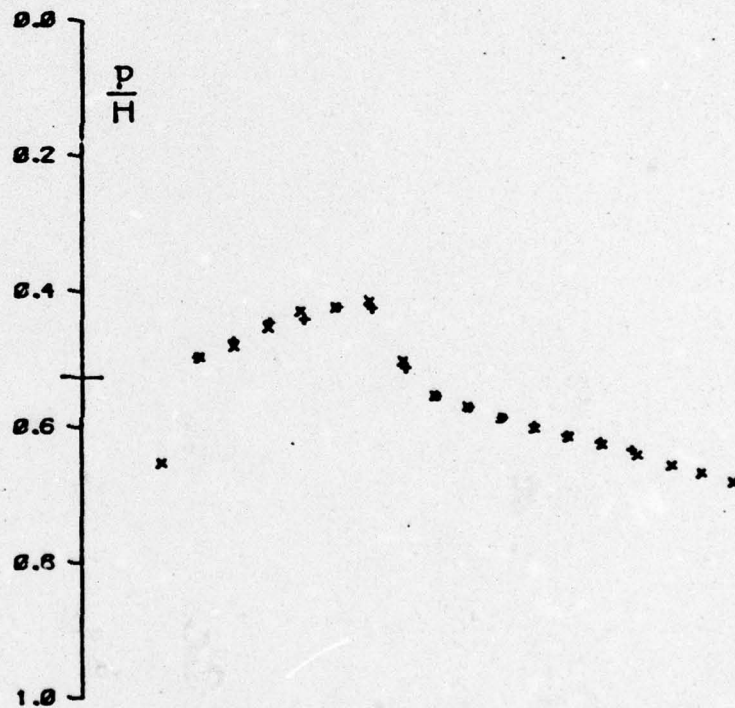
+ x NACA 0012 101.6 MM CHORD SOLID WALLS
 M= .758 AL= 0.00 CN= 0.021 CM= 0.004 R= 0.009



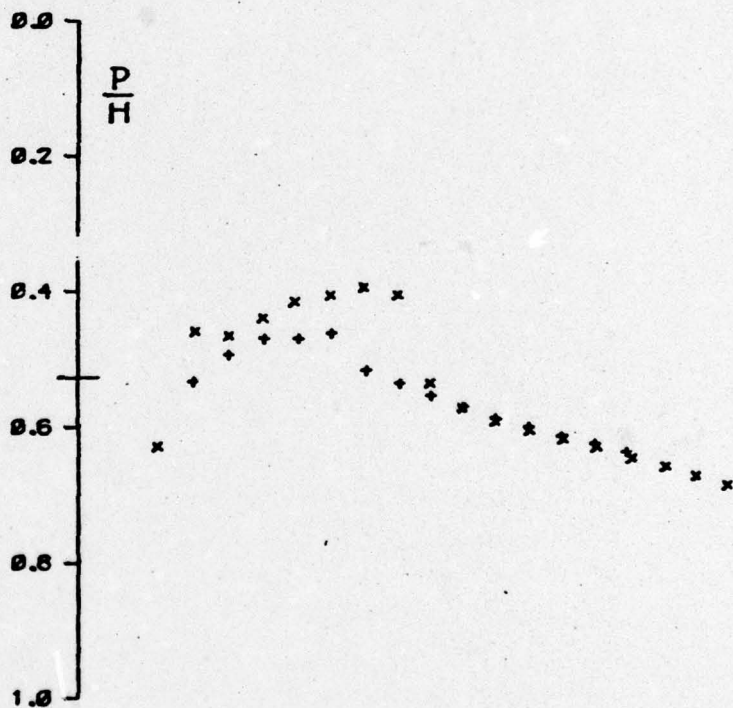
+ x NACA 0012 101.6 MM CHORD SLEED WALLS
 M = .759 AL = 1.00 CN = 0.203 CM = 0.014 R = 0.809



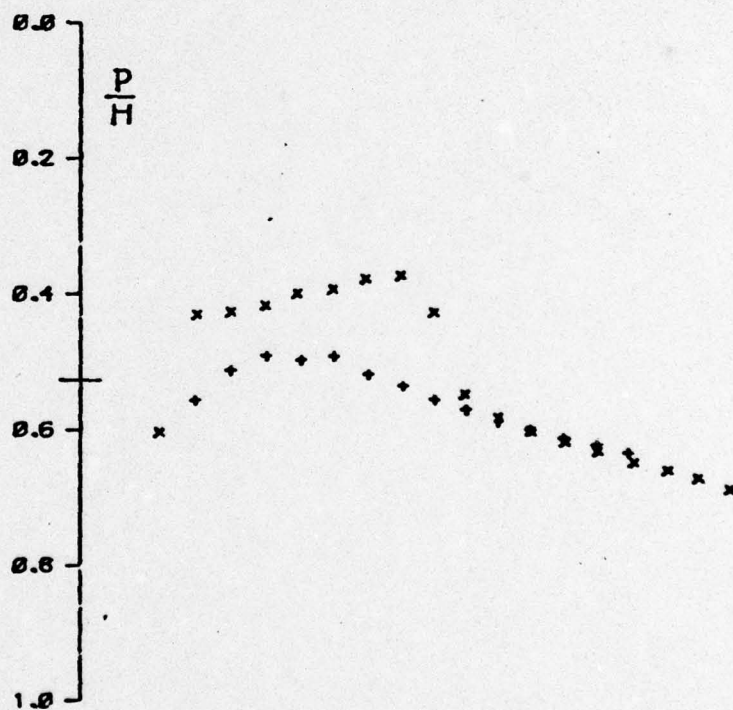
+ x NACA 0012 101.6 MM CHORD SOLID BALLS
 $M = 0.759$ $AL = 2.00$ $C_{Df} = 0.342$ $C_{Df} = 0.009$ $R = 0.812$



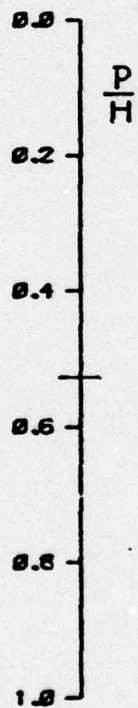
+ x NACA 0012 101.6 MM CHORD SOLID BALLS
 $M = .779$ $AL = 0.00$ $CL = 0.014$ $CM = 0.003$ $R = 0.791$



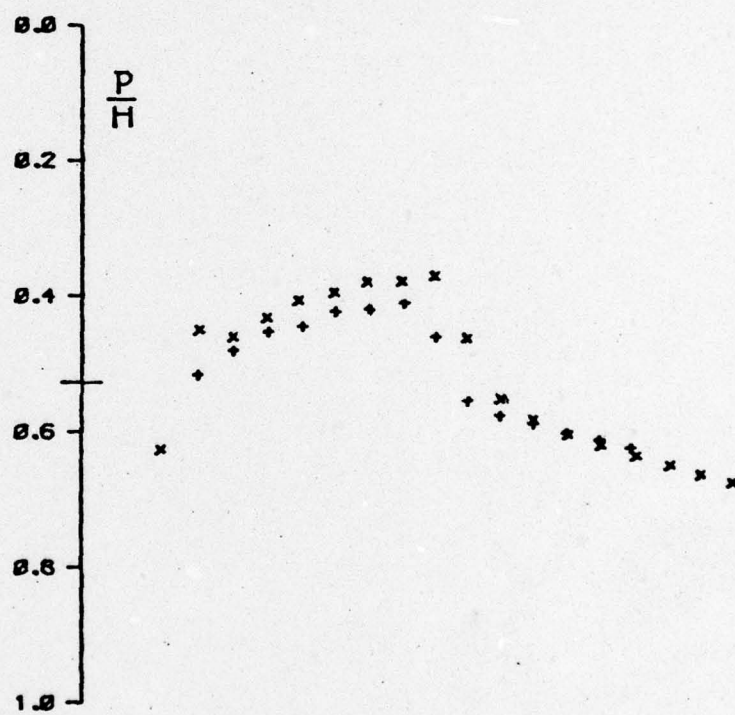
+ x NACA 0012 101.6 MM CHORD SOLID WALLS
 $\mu = .781$ $AL = 0.50$ $CM = 0.080$ $CM = 0.001$ $R = 0.794$



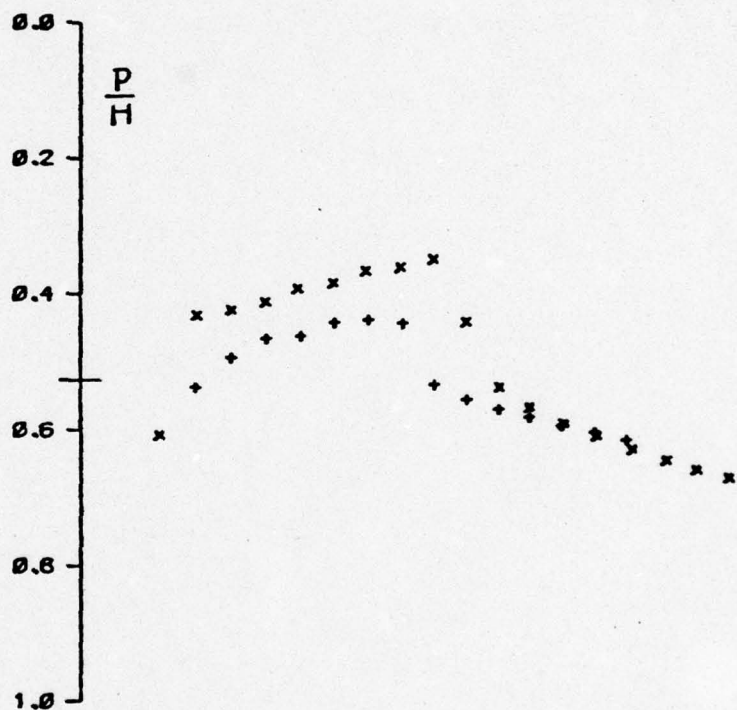
+ x NACA 0012 101.6 MM CHORD SOLID WALLS
 $M = 0.780$ $AL = 1.00$ $CN = 0.192$ $CN = 0.005$ $R = 0.784$



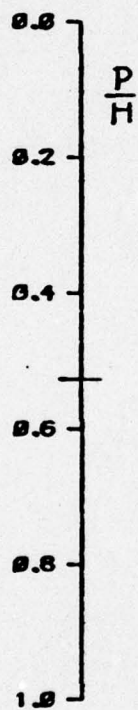
+ x NACA 0012 101.6 MM CHORD SOLID WALLS
 M= .820 AL= 0.00 CL= 0.008 CM= 0.002 R= 0.001



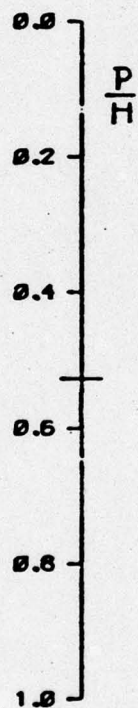
+ x NACA 0012 101.6 MM CHORD SOLID WALLS
 $M = 0.00$ $AL = 0.50$ $CL = 0.117$ $CM = 0.00$ $R = 0.001$



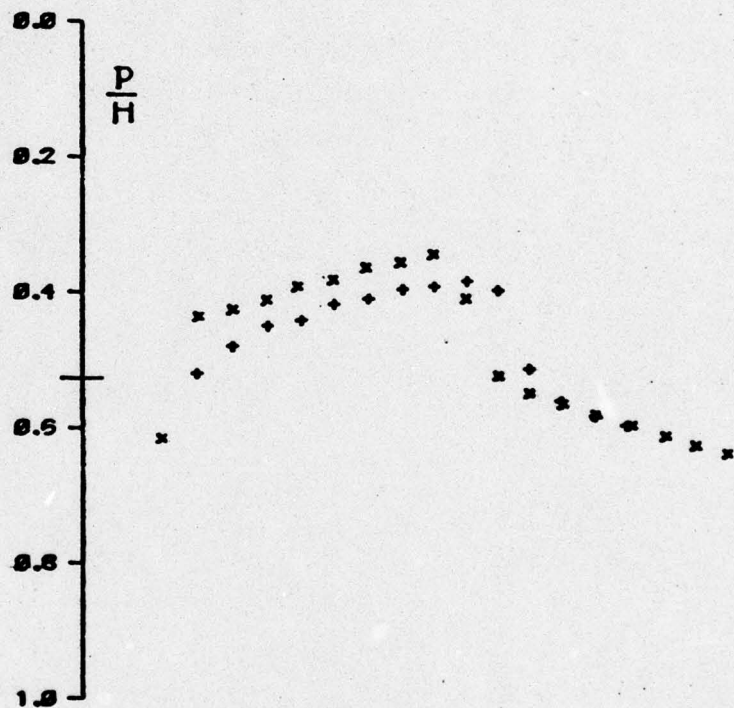
+ x NACA 0012 101.6 MM CHORD SOLID WALLS
 $M = .801$ $AL = 1.00$ $CM = 0.160$ $CM = 0.003$ $R = 0.001$



x NACA 0012 101.6 MM CHORD SOLID WALLS
 $M = 0.20$ $A = 0.02$ $C_L = 0.023$ $C_D = 0.002$ $R = 0.814$



+ x NACA 0012 101.6 MM CHORD SOLID BALLS
 $M = .820$ $AL = 0.50$ $Q = 0.057$ $Q = 0.008$ $R = 0.814$



+ x NACA 2312 101.6 MM CHORD SOLID WALLS
 $M = .519$ $AL = 1.00$ $CL = 0.070$ $CM = 0.016$ $R = 0.314$